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of Nacelle-Pylon Installation
on an Unswept Wing at Subsonic
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An Experimental Investigation
of Nacelle-Pylon Installation
on an Unswept Wing at Subsonic
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INTRODUCTION

There has been an extensive effort under way to develop efficient transport aircraft to reduce direct operating costs and conserve the nation's fuel supply. The proper integration of a transport-aircraft propulsion system with its airframe can greatly enhance the aircraft capabilities and efficiency. However, because of the complex flow in the wing-pylon-nacelle region, the propulsion integration must be carried out carefully to avoid adversely influencing the drag of the total configuration.

Numerous factors contribute to the complex flow about the nacelle, pylon, and wing. (See fig. 1.) Among these factors are disturbances due to the presence of the engine fairings and support, mutual influences of the external wing flow and inlet and exhaust flows, and changes in boundary-layer development due to regions of locally supersonic flows. These interactions greatly affect the drag of the nacelle-pylon installation. The low pressure field in the region of the wing leading edge, caused by the acceleration of the flow around the leading edge, is further enhanced by the addition of the pylon in the field. The flow, already accelerating around the wing leading edge, is further accelerated when flowing around the pylon leading edge. The nacelle, inlet, and exhaust flows are influenced by the presence of this flow. The pre-entry stream tube is influenced by the upwash flow caused by the lift generated by the wing. The stream tube is also influenced by wing side flow caused by the planform sweep and taper. The flow along the pylon planform experiences a channeling effect because the wing and nacelle act as walls and form a duct. Often, the flow becomes supersonic in the duct with a shock wave downstream. This shock typically extends from the wing lower surface, across the pylon, down to the nacelle. This shock can influence the nacelle afterbody flows by interacting with the fan jet, with the primary jet, or with the flow over the nacelle boattail. These flow interactions may be strong at transonic speeds; therefore, the effect on the total propulsion system must be taken into account when altering any part.

These interactions, and hence the determination of the installed performance of propulsion systems, have been experimentally investigated for many years (refs. 1 to 6). This type of experimentation is an expensive and time-consuming task. The construction and testing of wind-tunnel models to examine new propulsion-system installation concepts is usually not done until an airplane design is nearly complete, which allows, at best, only minor modifications to improve the design.

Experimental methods for determining the installed performance of propulsion installations will continue to be of major importance. However, in recent years, computational capabilities in the United States have increased considerably, and increasing attention is being paid to theoretical methods for preliminary analysis and design. Theoretical studies and concept modifications are much easier and less expensive than building, testing, and modifying wind-tunnel models in the early design stages.

Computer codes are being developed which do model the aerodynamic interactions of various components of the propulsion integration design problem. Present production computer codes, however, are not capable of modeling a complete transport aircraft configuration. Detailed experimental data are required for the verification

and development of these techniques and to improve the understanding of the flow phenomena involved.

The purpose of the research project reported herein was to provide static-pressure data on under-the-wing nacelle-pylon installations, tested at subsonic and transonic speeds, to be used in the verification of theoretical flow-prediction methods. In addition, a comparison of the data with predictions from a current subsonic aerodynamic panel code is made. The complex interactions and physical flow phenomena associated with engine-nacelle integration makes the development of computational techniques for their prediction an extremely difficult task. Computational techniques for a simplified version of the problem must be developed first. In this experiment, the wing-pylon-nacelle interaction was isolated by removing the wing-body and nacelle-body interactions and by removing the effects due to sweep, taper, and twist of the wing. This method also reduced the geometric requirements placed on the computational techniques. No effects due to power were investigated, as the nacelle was a flow-through type. The configurations were tested at free-stream Mach numbers from 0.20 to 0.875, over an angle-of-attack range from 0° to 5° , in the Langley 16-Foot Transonic Tunnel. Extensive static-pressure measurements were taken on the nacelle, pylon, and wing.

SYMBOLS

Dimensional quantities are presented in both the International System of Units (SI) and U.S. Customary Units. Measurements and calculations were made in U.S. Customary Units.

b	span of wing between support struts, 101.60 cm (40.00 in.)
C_p	pressure coefficient, $(p - p_\infty)/q_\infty$ (C_p in computer-generated tables)
$C_{p,\text{sonic}}$	pressure coefficient corresponding to local Mach number of 1.0
c	wing chord, 25.40 cm (10.00 in.) (C in computer-generated tables)
d_N	maximum diameter of nacelle, 11.43 cm (4.50 in.) (D_N in computer-generated tables)
L.E.	leading edge
M	Mach number
p	local static pressure, Pa (lb/in ²)
p_∞	free-stream static pressure, Pa (lb/in ²)
q_∞	free-stream dynamic pressure, Pa (lb/in ²)
R	Reynolds number
r	radius of circular arc, cm (in.)
r_e	external radius of nacelle, cm (in.)
r_i	internal radius of nacelle, cm (in.)
2	

x distance streamwise from wing, inlet, or pylon leading edge, cm (in.)
(X in computer-generated tables)

x_t location of wing boundary-layer transition strip, cm (in.)

y ordinate along span of wing measured from wing centerline, cm (in.)
(Y in computer-generated tables)

z ordinate perpendicular to x-y plane measured from wing reference plane, cm (in.)

α angle of attack, deg

ϕ nacelle angular station measured clockwise looking aft from positive Z-axis, deg

Subscripts:

le leading edge

te trailing edge

EXPERIMENTAL APPARATUS AND PROCEDURES

Wind-Tunnel Description

This investigation was conducted in the Langley 16-Foot Transonic Tunnel. The tunnel is an atmospheric transonic single-return type with continuous air exchange and is capable of operation from Mach 0.20 to 1.30. The test section is octagonal and slotted at the vertices and has an equivalent circular diameter of 4.85 m (15.9 ft). A detailed description of the tunnel is presented in reference 7.

Model and Support-System Description

The model consisted of a long-duct, flow-through nacelle mounted on a straight, unswept wing. The wing had a supercritical airfoil cross section and was mounted on the Langley 16-foot bifurcated support strut system. Figures 2(a) through 2(d) are photographs showing various views of the model and support strut. A detailed description of the strut system is presented in reference 8.

Wing.— Figure 3(a) is a sketch of the wing model. Figure 3(a) also contains the dimensions for the model. The wing model had a tip to tip span of 218.44 cm (86.00 in.), with a chord of 25.40 cm (10.00 in.), and was supported below the bifurcated support strut. Coordinates for the airfoil sections tested are given in figure 3(b). The original airfoil was an 11-percent-thick supercritical section whose coordinates are presented in reference 9. A modification of the trailing-edge region of the original section was required for structural rigidity in the cusp region. A linearly increasing thickness was added to the wing lower surface from $x/c = 0.65$ to 1.00. The trailing-edge thickness increased from 0.05 percent to 0.4 percent of the wing chord as a result of this modification. The wing upper-surface contour was left unchanged. The wing leading edge detached at $x/c = 0.175$, so that an alternate, sharp, leading-edge profile could be tested. The alternate profile design consisted

of two intersecting circular arcs forming a 60° vertex angle at the wing leading edge, as shown in figure 3(b).

Pylon.- The two pylon planforms investigated are shown in figures 4 and 5. Coordinates for each pylon, wing-pylon intersection, and nacelle-pylon intersection are given in figures 4(b) and 5(b). Both pylons are symmetrical in cross section with the center plane oriented perpendicular to the wing leading edge. The first pylon was swept in planform and was typical of transport aircraft, except that there was no cant or toe-in of the attached nacelle or inlet. The nacelle leading edge was placed 85 percent of the wing chord (21.590 cm (8.500 in.)) forward of the wing leading edge. The nacelle centerline was placed 0.956 nacelle diameters (10.922 cm (4.300 in.)) below and parallel to the wing chord line. This pylon had two detachable leading-edge sections. The first was a conventional elliptic section, and the second, alternate section, was a sharp leading-edge section designed the same as the wing alternate leading edge.

The second pylon was unswept and untapered with the leading and trailing edges aligned with the wing leading and trailing edges. The nacelle leading edge was positioned 23.75 percent of the wing chord (6.033 cm (2.375 in.)) forward of the wing leading edge. The nacelle centerline was 0.956 nacelle diameters (10.922 cm (4.300 in.)) below the wing chord, the same position as the swept-pylon configuration. This pylon has only a sharp leading edge designed by the same criteria as the wing. This pylon was designed specifically as a simple geometry to be used in the development and verification of computational models.

Inlet and nacelle.- Coordinates of the axisymmetric flow-through nacelle and inlets are given in figure 6. The conventional inlet shape was an NACA 1-83-75 inlet 8.573 cm (3.375 in.) long followed by a 2.527-cm (0.995-in.) long, constant-radius cylindrical section. The alternate inlet shape consisted of two intersecting circular arcs designed by the same criteria as the wing leading edge. The nacelle was 20.333 cm (8.005 in.) long with a 126.99-cm (49.997-in.) radius circular-arc boattail.

Configurations.- Five configurations were tested. The following table summarizes the different configurations:

Configuration	Leading-edge shape			Pylon planform
	Wing	Pylon	Nacelle	
1	Blunt	None	None	None
2	Sharp	None	None	None
3	Sharp	Sharp	Sharp	Unswept
4	Sharp	Sharp	Sharp	Swept
5	Blunt	Blunt	Blunt	Swept

Instrumentation and Test Conditions

The model was tested at Mach numbers from 0.20 to 0.875 at angles of attack from 0° to 5° . Reynolds number based on the wing chord varied from around 1.2 million at Mach 0.20 to around 3.5 million at Mach 0.875. The attitude of the model was deter-

mined by using Kistler angle-of-attack measuring devices located in each of the boom-tip fairings.

Static-pressure orifices were located on the wing upper and lower surfaces at 0.000, 0.191, 0.500, 1.500, and 3.000 nacelle diameters from the wing centerline. As shown in figure 7, static-pressure orifices were also located on the nacelle exterior surface at angular stations of 0.0°, 22.5°, 50.0°, 90.0°, 135.0°, 180.0°, and 270.0°. The swept pylon had two rows of orifices $0.200d_N$ (2.286 cm (0.90 in.)) and $0.378d_N$ (4.318 cm (1.70 in.)) below the wing chord. (See fig. 4(b).) The unswept pylon also had two rows of orifices. The upper row was $0.222d_N$ (2.54 cm (1.00 in.)) below the wing chord line. The lower row was $0.111d_N$ (1.27 cm (0.50 in.)) above the nacelle and approximately followed the nacelle contour. (See fig. 5(b).)

Several boundary-layer transition configurations were investigated. All five configurations had the boundary-layer transition on the wing upper and lower surface fixed by a 0.254-cm (0.1-in.) wide strip of No. 60 carborundum grit located 0.29c (7.4 cm (2.9 in.)) from the leading edge. Transition on each pylon was fixed by a 0.254-cm (0.1-in.) wide strip of No. 70 carborundum grit located at 10 percent of the local pylon streamwise chord from the pylon leading edge. Transition on the interior and exterior surfaces of the nacelle inlet was fixed by a 0.254-cm (0.1-in.) wide strip of No. 80 carborundum grit located 2.54 cm (1.00 in.) from the leading edge.

In addition, another location for fixing the boundary-layer transition was investigated for the clean wing (configuration 1) and for the swept pylon (configuration 5). Transition on the wing was then fixed by a 0.254-cm (0.1-in.) wide strip of No. 60 carborundum grit located 0.05c (1.27 cm (0.50 in.)) from the leading edge. The pylon boundary-layer transition was then fixed by a 0.254-cm (0.1-in.) wide strip of No. 60 carborundum grit located at 5 percent of the local streamwise pylon chord from the leading edge. The location of the nacelle boundary-layer transition strip was not changed. The grit sizing was made on the assumption that the tunnel total temperature was 49°C (120°F) at Mach 0.2. Grit drag was not considered because the model was not on a balance, and no measurement of drag was made.

For the same configurations, 1 and 5, additional data were obtained with natural transition (that is, no fixed strips) for comparison with the fixed transition data.

A matrix of test conditions and configurations investigated is presented below.

M	R × 10 ⁶ , per m (ft.)	α, deg	Configuration ¹				
			1	2	3	4	5
0.20	4.5 (1.4)	0,1,2,3,4,5	x ^{2,3}	x	x	x	x ^{2,3}
.30	6.4 (2.0)	0	x ²	x	x	x	
.40	8.2 (2.5)	0	x ²	x	x	x	
.50	9.8 (3.0)	0	x ²	x	x	x	
.60	11.2 (3.4)	0,1,2,3	x ^{2,3}	x	x	x	x ^{2,3}
.65	11.5 (3.5)	0	x ²	x	x	x	
.70	12.1 (3.7)	0	x ²	x	x	x	
.75	12.5 (3.8)	0	x ²	x	x	x	
.80	12.8 (3.9)	0,1,2	x ^{2,3}	x	x	x	x ^{2,3}
.85	13.4 (4.1)	0,1	x ^{2,3}	x	x	x	x ^{2,3}
.875	13.8 (4.2)	0	x ²	x	x	x	

¹All configurations tested with transition strips at 0.29c.

²Additional tests with transition strips at 0.05c.

³Additional tests with natural transition.

PRESENTATION OF RESULTS

Experimental results from this investigation are presented in the following figures:

	Figure
Experimental chordwise pressure distributions; configuration 1	8
Effects of boundary-layer transition-strip location; configuration 1	9
Experimental pressure distributions on wing, pylon, and nacelle; configuration 5	10
Effects of Mach number on pressure distributions for swept pylon and wing; configuration 5	11
Effects of angle of attack on pressure distributions for swept pylon and wing; configuration 5	12
Effects of Mach number on pressure distributions for unswept pylon	13
Effects of swept nacelle-pylon installation on wing pressure distributions	14

The appendix contains an index to the tabulation of basic pressure data. The data are tabulated in tables A1 to A9. The reference origin is the wing leading edge (that is, $x/c = 0$).

Numerical calculations and their comparison with the experimental results are presented in the following figures:

Configuration	M	α , deg	Figure
1	0.20	0,5	15(a), (b)
2	0.20	0	15(c)
5	0.20	0,1,5	15(d), (e), (f), (g), (h), (i)
3	0.20	0,5	15(j), (k), (l), (m)
1	.60	0,3	16(a), (b)
5	.60	0,3	16(c), (d), (e), (f)
3	.60	0	16(g), (h)
1	.80	0	17(a)
5	.80	0	17(b), (c)
3	.80	0	17(d), (e)
1,5	.60	0	18

EXPERIMENTAL RESULTS

Evaluation of Strut Interference

Plots of the wing pressures for configuration 1 (fig. 8) indicate the uniformity of the flow over the center section of the model. A spanwise variation of less than 1.5 percent in local Mach number occurred from the centerline to $1.500d_N$ at Mach numbers of 0.20 and 0.70. At Mach 0.85, a 2-percent variation in local Mach number occurred spanwise to $1.500d_N$ with a 7-percent variation by $3.000d_N$. The largest spanwise variation in pressure occurred, as would be expected, at the high transonic Mach number. The flow over the wing upper surface is almost completely supersonic at Mach 0.85. Any irregularities in the strut fairings cause disturbances in the flow field that are noticeable at transonic Mach numbers. The presence of the struts, located $4.440d_N$ from the wing centerline, and the finite nature of the wing contribute to the disturbances in the flow over the wing.

Transition-Strip Investigation

The Reynolds number at which a scaled wind-tunnel model is tested is important when the model is being used to evaluate the aerodynamic characteristics of the full-scale airplane. In particular, the boundary layer on a wind-tunnel model usually develops faster than the full-scale equivalent, because of the lower Reynolds number at which most wind tunnels operate. The boundary layer developing on the scaled model tends to separate as a result of interaction between shock and boundary layer at a lower Mach number than would occur at flight Reynolds numbers. This separation alters the lift and drag characteristics of the model compared with the full-scale airplane.

The knowledge of the characteristics of the boundary layer on the model is important. The extent of laminar flow or turbulent flow greatly influences the nature of the shock that occurs on a wing upper surface (ref. 10). The extent of laminar flow occurring on a model depends upon such factors as the physical shape of the model; the flow quality, in terms of turbulence, of the wind tunnel being used; and the physical parameters of the flow (for example, total pressure, total temperature, and specific heat). The boundary-layer transition is typically fixed by some artificial tripping method to eliminate this variability in the flow. Transition to a turbulent boundary layer on the model can be delayed by shifting the location of the transition strip rearward. This shift results in a thinner boundary layer. Therefore, a better simulation of the full-scale boundary-layer thickness, and hence a better simulation of the shock location and associated trailing-edge losses, is achieved (ref. 10). As would be expected, a limitation in the applicability of this technique requires that the boundary layer not transition naturally before the artificial trip is reached.

A preliminary series of tests was conducted to investigate the effects of boundary-layer transition-strip location on the pressure data of this model. Transition strips were placed at 5 percent of wing chord ($x/c = 0.05$) and at 29 percent of wing chord ($x/c = 0.29$) on configurations 1 and 5. Additional tests were made on the same configurations using no artificial tripping mechanism (natural transition). A representative sample of the effects of boundary-layer transition-strip location on wing pressure is presented in figure 9. The data show that, at test conditions having no shock waves on the wing, the variations of pressure coefficient with respect to transition location were insignificant.

The shock occurred in the most forward position when placement of the transition strip was at the 0.05c location for higher Mach numbers. The shock occurred farthest aft for the model with natural transition for most conditions. The data at $M = 0.80$ and $\alpha = 2^\circ$ show an exception to the above, where the shock wave, for the natural transition configuration, was approximately 5 percent farther forward than the shock for the 0.29c transition location. A change in the pressure gradients around the wing, due to the increase in angle of attack, could cause the boundary layer to thicken enough to displace the shock forward. Figure 9 shows no evidence of transition occurring before $x/c = 0.29$. Therefore, for the investigation, transition was fixed at $x/c = 0.29$, the recommended location given in reference 10. It was concluded that better correlation of full-scale effects would occur by placement of the transition strip at $x/c = 0.29$.

Static-Pressure Trends

An example of the similarity in static pressure measured in the wing, pylon, and nacelle junction region is shown in figure 10. The wing lower-surface row adjacent to the pylon, the top row of pylon pressures, and the nacelle row adjacent to the pylon are presented. Where the pressure measurements overlap, a similarity in the trends exists. This similarity is convenient for determining trends and flow characteristics on the pylon or nacelle, where bad data or no data exist, by using the wing pressure data. There are two regions of discrepancies; the wing leading edge and the nacelle trailing edge. The first pressure minimum, on the wing lower surface, is due to the initial flow expansion about the leading edge. This expansion did not show on the pylon pressure data either because the row of orifices was too far from the wing lower surface at that longitudinal station, or because of the orifice spacing. Therefore, the pressure minimum could not be resolved. The nacelle trailing-edge discrepancy is likely caused by compression of the flow to match the exit pressure of the internal-duct flow. The nacelle orifices in the exit area are less influenced by the junction region and more influenced by local conditions on the boattail. The pylon and wing had a longer longitudinal extent of similarity of pressure-coefficient trends, so the applicability is rather limited. Nevertheless, the comparison is useful in the interpretation of the data.

Effects of Mach Number on Pylon and Wing Pressures

The effects of Mach number on the swept pylon and wing lower-surface pressure coefficients (configuration 5) are presented in figure 11. In this case, there were insufficient pylon data to describe the behavior of the flow after the shock; therefore, as an indication of the character of the flow on the pylon, pressures along the wing lower surface were included in the plot.

A gentle expansion and recompression can be seen on the pylon for Mach 0.6 (fig. 11). The flow on the wing lower surface (station B; $y/d_N = 0.191$) is probably attached and experiences the typical recompression that is characteristic of the cusp of a supercritical airfoil section.

The flow pressure gradients are more severe on the pylon for Mach 0.80. A region of supersonic flow has formed on the pylon between $x/c = 0.1$ to $x/c \approx 0.30$. The shock on the pylon has possibly caused a mild flow separation. Some character of the flow has changed, as evidenced by the lack of the typical recompression of the flow in the wing lower-surface cusp region.

A similar case is seen in the data for Mach 0.85. Although the levels have shifted, the pylon flow has not changed much, in that the pressure gradients are similar. As evidenced by the flat character of the pressure distribution for x/c between 0.35 and 0.75 along wing station B, it is likely that the wing lower-surface flow has separated as the result of shock-boundary-layer interaction. By association, it is likely that the flow along the pylon has also separated past $x/c = 0.35$. This is obviously a most undesirable situation in terms of drag and lift loss of the configuration.

Effects of Angle of Attack on Pylon and Wing Pressures

The effects of angle of attack on the swept pylon and wing lower-surface pressures at Mach 0.80 are presented in figure 12. Generally, as expected near the wing bottom, the pressures become more positive. Therefore, the velocities are lower as the angle of attack increases. For $\alpha = 1^\circ$ or 2° , a good recompression trend can be seen in the wing lower-surface pressures in the cusp region. The positive pressure gradient on the pylon at $x/c \approx 0.25$ is strongest at $\alpha = 0^\circ$, which could have an adverse influence, that is, a thickening effect, on the boundary layer in that region. This thickening effect could account for the "flatter" recompression trend on the wing lower-surface pressure coefficients for $\alpha = 0^\circ$.

Effects of Mach Number on Unswept Pylon Pressures

The effects of Mach number on the pressure distributions of the unswept pylon are presented in figure 13. This flow appears to remain attached for Mach numbers of 0.70 and 0.80. However, shock-induced separation occurred for free-stream Mach numbers above 0.80. The wing and nacelle channel form a rather rapid diffuser and cause a compression shock which contributes to the tendency of the flow to separate. This separation, particularly at Mach 0.875, exists over part of the wing lower surface and around part of the nacelle. These results show that this physical configuration is undesirable, not only in terms of lift loss, but also in terms of drag due to the separation.

Effects of Nacelle and Swept-Pylon Installation on Wing Pressures

The effects of the swept-ptylon installation on the wing pressure distributions are presented in figure 14 for Mach 0.80 at $\alpha = 1^\circ$. As expected, there was a significant change in the lower-surface flow due to the presence of the pylon and nacelle. The upper-surface pressure distribution changes were of a lesser magnitude. The installation caused higher suction peaks on both wing surfaces due to the flow turning around the pylon and wing leading edges. The upper-surface shock shifted forward as a result of the altered pressure gradient on the wing. This forward movement is considered shock relief caused by the flat nature of the upper-surface contour. Therefore, separation due to shock--boundary-layer interaction becomes less likely. The double-peak structure of the wing lower-surface pressures of the installed-ptylon configuration (fig. 14) is similar to those occurring in figure 10. The first peak, as discussed previously, is caused by a mutual interference effect of the pylon and wing. The second expansion arises from the convergent nature of the nacelle and wing channel near the wing leading edge. The recompression trend of the pressure distribution after the shock at $x/c = 0.30$ indicates that the flow remained attached on the wing lower surface. No flow separation is evident on the wing upper surface.

COMPARISON OF NUMERICAL AND EXPERIMENTAL RESULTS

Description of Numerical Procedure

The numerical computations of the pressure coefficients were performed using the Hess program (ref. 11), a current panel aerodynamic flow code. The basic Hess code solves a linear equation describing potential flow. The flow is assumed to be inviscid, irrotational, and incompressible. These assumptions allow the flow to be described by Laplace's equation. The linearization of the equation due to these assumptions permits a superposition principle to describe the flow about the body. The superposition of solutions for source singularities distributed upon the body surface describes the influence on the free stream caused by that body.

In the Hess code, the exterior surface is described by an array of quadrilateral panels conforming to the body contours. The strengths of the singularities, which are located at the panel center points, are adjusted to insure potential flow and to satisfy the specified boundary conditions at infinity and on the body. This flow and body description formulation is convenient, in that complex geometries can be modeled relatively easily. For the present calculations involving flow-through nacelles, panels were distributed only on the external surface of the nacelles. Thus, the computations were for nacelles with infinitesimally thin skins, with the internal contours precisely matching the external contours. The Prandtl-Glauert compressibility correction was used for nonzero free-stream Mach number calculations. No corrections for viscous effects were in the program.

Accurate predictions of the flow character become difficult when compressibility or viscous effects are dominant factors. The linear potential-flow assumptions break down where compressibility effects are strong and where strong viscous interactions occur, such as with shock-induced boundary-layer separation. In these severe situations, linear potential-flow theory was inadequate to predict these effects.

The comparisons of theory with wind-tunnel data in the sections which follow are in order of increasing Mach number. The lowest speed, Mach 0.20, is intended to have no imbedded supersonic regions and to provide good comparison for an incompressible calculation. Data at Mach 0.60 were expected to provide very mild compressible flow data to test the upper limit of the subsonic codes. Data were also obtained at Mach 0.80, where substantial regions of sonic flow and shocks occurred, to provide verification of calculations of transonic flows.

Comparisons at Mach 0.20

Clean wing.- At Mach 0.20 for the clean wing with the conventional leading edge (configuration 1), the theory correctly predicted the trends of the data and was generally in good agreement with the magnitude. Figure 15(a) shows that the theory matched the upper-surface leading-edge pressure minimum in magnitude and position for the centerline and was slightly low for the minimum at $y/d_N = 0.500$ and 1.500 . The gradient of compression past $x/c = 0.10$ was not steep enough to match the pressure level of the experiment, but the trend was matched. Theoretically, the trailing-edge stagnation point pressurizes the aft upper surface and brings the calculation up in pressure to the experimental data past $x/c = 0.85$. The lower-surface calculation more closely matched the general level and gradients of the experiment. The theory did not match the lower-surface, leading-edge minimum level or the position of the trailing-edge-cusp recompression region. The leading-edge problem could be caused by

an angle-of-attack mismatch, as the tunnel upflow of about 0.2° was not taken into account.

The theory for the same configuration at an angle of attack of 5° (fig. 15(b)) matched the upper-surface, leading-edge pressures and pressure gradient up to $x/c = 0.25$, but again missed on pressure level between $x/c = 0.25$ and 0.85 . The predicted lower-surface pressures at $y/d_N = 0.500$ were slightly off in level until crossing over at $x/c = 0.85$. The comparison at $y/d_N = 1.500$ was even worse, but again crossed over the experiment in the aft region of the wing. An integration of the pressure coefficients over the wing surface shows that the lift tends to be consistently overpredicted. A factor leading to this result could be the lack of boundary-layer corrections in the calculations.

Configuration 2 data are presented in figure 15(c) for Mach 0.20 and $\alpha = 0^\circ$. The nature of the leading-edge pressure minimum was considerably altered as a result of the different geometry. The sharp leading edge was designed to keep the leading-edge stagnation point at a fixed location for all conditions. As before, the general trends of the pressure data were matched. However, the predicted peak magnitude and position were off, as was the level for the whole upper surface. The lower-surface, leading-edge pressure was again missed, and the cusp recompression region was overpredicted. Also, the lift was again overpredicted.

Swept-pylon configuration.— Data on the wing, generated by installation of the swept pylon and nacelle (configuration 5) at Mach 0.20 and $\alpha = 0^\circ$, are presented in figure 15(d). Results shown in figure 15(d) are similar to those with configuration 1. The leading-edge pressures on the centerline were matched, but the following recompression was still not steep enough to match the pressure level for the rest of the wing chord. The peaks were not matched in magnitude for the other two wing stations, and the pressure levels were underpredicted. The lower-surface levels were too high over the whole chord, except for the last 5 to 10 percent. The trailing-edge crossover was likely due to the previously mentioned stagnation-point calculation. The wing-pressure prediction with installed pylon was not as good as the prediction for the clean-wing pressure predictions. This difference could result from a more complex flow configuration caused by a mutual influence of bodies or from increased viscous effects as a result of more unfavorable pressure gradients.

Figure 15(e) presents the pylon and nacelle pressures of configuration 5 at Mach 0.20 and at $\alpha = 0^\circ$. The trends of the pylon pressures were well matched. The pressure gradient and peak were off in the wing leading-edge area. The lower pylon row showed the flow experiencing lower gradients, which the theory predicted, but still missed the peak magnitude. The inlet leading-edge, low-pressure minima were not predicted by the theory. A probable cause could be that improper modeling of the inlet flow occurs, since the computations were for a nacelle with a skin of zero thickness and only the external contours were modeled. Improper placement of or an insufficient number of panels in the leading-edge region could further contribute to incorrect modeling of the physical shape. The trends of the pressure distributions aft of this region are well matched and are only slightly high in pressure level. The predictions of the trailing-edge pressures are consistently high. This again was probably a result of the code forcing a flow stagnation condition on the last panel.

The results for the same configuration at $\alpha = 1^\circ$ are shown in figure 15(e). The upper-surface, leading-edge pressure decrease was predicted well. The predicted pressure level was still low for the remainder of the wing chord. The higher predicted pressure level along the first 50 percent of the lower surface, because of the movement of the stagnation point, matches better at this angle of attack. Again, the

pressure gradient in the wing leading edge was too low. The pylon top-row flow expansion peak and the pylon lower-row pressures were predicted well by the theory. The same problem with the nacelle occurs here as with the 0° case. Plots for configuration 5 at $\alpha = 5^\circ$ are presented in figure 15(f). The decrease in upper-surface pressures, caused by the increased angle of attack, was predicted by the code, and the level of the change was still too low. The lower-surface flow is generally faster in the leading-edge region of the swept-pylon configuration than in the clean-wing configuration. For example, compare figures 15(b) and 15(f). The flow on the aft portion of the wing lower surface is actually slower than with the clean wing. These trends were predicted, but the pressure level was always slightly high. A considerable portion of the pylon was influenced by the wing leading-edge stagnation region at the high angle of attack. Even this reduced flow velocity could not be matched in the region of the wing leading edge. The inlet leading-edge pressure levels were, as before, not matched. After $x/c = -0.05$, the comparison of theory and experiment was very close. The prediction of trends and levels for the latter 75 percent of the nacelle was reliable for this case.

Unswept-pylon configuration.- The pressures for the unswept-pylon and nacelle installation at Mach 0.20 and at $\alpha = 0^\circ$ are presented in figure 15(g). As pointed out in the model description, the unswept-pylon and nacelle configuration had sharp leading edges on the wing, pylon, and nacelle. An examination of the experimental data shows that the lower-surface pressures are greatly decreased at $y/d_N = 0.500$ because of the channeling effect of the junction of the wing, pylon, and nacelle. Also, the pylon and nacelle influence on the wing lower-surface pressures rapidly diminishes with increasing span and is greatly diminished even by the $y/d_N = 1.500$ wing station. As can be seen by comparing figure 15(c) with figure 15(g), these changes are predicted. The code predicts a large pressure drop at the leading edge of the upper surface of the wing that is not in the data. Because of the thin leading-edge shape, the drop is likely caused either by the proximity of the pylon panel null points or by the wing panel points. The predicted lower-surface pressures are still high in magnitude. The predicted lift would also be high. The predicted minimum pressure peaks on both rows of the pylon were too high, and results similar to those of the inlet leading-edge region occurred. The expansion-trend gradient from $x/c = 0$ to $x/c = 0.15$ on the nacelle in the channel flow was predicted well. The ability to predict pressure trends and levels for the nacelle flow was similar to the results seen for the swept-pylon configuration (fig. 15(f)). The measured flow around the unswept pylon had larger gradients in the first 20 percent of the section. As can be seen by comparing the 22.5° angular stations in figures 15(d) and 15(g), the generally higher velocities greatly influenced the nacelle flow. The interesting note, though not totally unexpected, is that by the 180° angular station on the nacelle, the effects of the wing and pylon flow were relatively small. This characteristic of the flow about this configuration was well predicted by the theory.

The last comparison for this Mach number is for configuration 3 at $\alpha = 5^\circ$ (fig. 15(h)). The pressures along the $y/d_N = 0.500$ wing station on the upper-surface display an interesting flow phenomena not predicted by the theory. It is likely that a laminar flow separation and subsequent reattachment occurs in the first 15 percent of the wing chord. The flow velocity levels came down in the junction region of the nacelle, pylon, and wing lower surface. These effects were picked up in the prediction, but the computed velocity was still low. A greatly increased pressure minimum occurred for the inlet leading-edge, zero-degree, angular station flow. This rather large gradient was not predicted, but the pressure trend downstream was soon recovered and was predicted by the code.

Overall, the subsonic prediction of flow pressure trends was good. The levels of predicted pressures on the wing were consistently low on the upper surface and consistently high on the lower surface. Therefore, the calculation of lift would always be high. The pylon and nacelle pressure levels were consistently slightly low, though occasionally the predictions matched the data quite well. The leading-edge pressure spikes, whether on the wing or the inlet, were seldom matched in magnitude. The exception was the centerline row on the wing, which was consistently matched in magnitude and position. The pressure magnitude problems could be due to a Mach number mismatch or a flow-angle mismatch. If lift matching or different Mach number calculations were performed, better agreement would be possible. In addition, the present geometry causes the program to satisfy the trailing-edge Kutta condition at the null point of the last wing panel at a given station. The enforcement of this condition occurred before the physical trailing edge and was responsible for the large pressure gradients on the aft 10 to 15 percent of the airfoil section. Repaneling the wing so that the final null point falls on the trailing edge of the physical wing could remedy the trailing-edge calculation problem. This would place the Kutta condition enforcement at the trailing edge.

Comparisons at Mach 0.60

Clean wing.— Data for configuration 1, the clean wing, at Mach 0.60 and at $\alpha = 0^\circ$ are presented in figure 16(a). The comparison of theory with experiment was very similar to the lower subsonic case. The wing upper-surface pressures were all slightly lower, as a result of the increased free-stream velocity, and retained the same character of trend as the pressures at Mach 0.20 and at $\alpha = 0^\circ$. Again, the trends were well predicted and the general level of the pressures was predicted. However, the discrepancies between the predicted and experimental pressures were greater than 0.20, particularly on the lower surface of the wing. These discrepancies could be caused by an angle-of-attack mismatch or increasing viscous or compressibility effects at the higher Mach number. A low-pressure "bulge" can be seen developing on the lower surface from $x/c = 0.05$ to 0.20 at $y/d_N = 0.500$. This bulge developed into a region of sonic flow and formed a compression shock downstream in the junction region at some higher free-stream Mach numbers (e.g., $M = 0.80$). The bulge was not picked up by the theory.

The clean-wing data for Mach 0.60 at $\alpha = 3^\circ$ are presented in figure 16(b). Very high pressure minima at the leading edge can be seen for all the wing stations. This large perturbation is caused by the blunt leading edge of the supercritical airfoil section combined with the movement of the stagnation point towards the lower surface. The theory did not predict such a low leading-edge pressure. Again, the general trend of an increased leading-edge pressure minimum was predicted. The calculations for the aft ($x/c > 0.50$) upper-surface pressures were too low for all the wing stations. This, in addition to the lower-surface pressure calculations being too high in pressure, would again cause the overestimation of lift prediction. A feature of the supercritical airfoil is the highly cambered aft region. The wing-surface curvature changes direction from concave-up to concave-down and forms the cusp region in the last quarter of the lower-surface chord. The experimental data reveal this reflex in geometry by a strong compression in the aft region of the wing lower surface. This compression can be seen in particular in figure 16(b) for the lower surface at $y/d_N = 0.500$. The gentle expansion trend from $x/c = 0.25$ to 0.35 is abruptly reversed at $x/c \approx 0.60$. The existence of this pressure trend reversal was predicted by the code. The matching of the level of the experimental pressures in the cusp region was poor. A higher panel density in the cusp region would more exactly define the geometry and would possibly result in a

better prediction of the pressures. Increasing the number of panels, though, could cause an undue burden on the code in terms of computing time or computer space limitations. Some type of compromise between geometry, space, and time must be made.

Swept-forward pylon configuration.- The pressure data and comparisons for configuration 5 at Mach 0.60 and at $\alpha = 0^\circ$ are presented in figure 16(c). The lower pressures on the wing upper surface caused by the pylon-wing junction were predicted. The spanwise variations in the minimum leading-edge pressures were also predicted. The remainder ($x/c > 0.15$) of the wing upper-surface pressure calculations were low, as seen previously. The slight forward movement of the centerline upper-surface pressure minimum in the aft region of the chord from $x/c = 0.80$, for the clean wing, to $x/c = 0.75$, for the installed swept-pylon configuration, was not discernible in the theoretical calculation. However, the overall pressure trend in the aft region of the wing upper surface was predicted. The detail of the change in pressures due to the installation was wrong. The wing lower-surface, leading-edge, low-pressure bulge for row $y/d_N = 0.500$ was enhanced considerably by the pylon installation. The theory predicted a trend of higher velocities in the region of x/c between 0.05 and 0.40, but the bulge characteristic did not even appear in the calculated pressures. The cusp pressures were poorly matched, particularly in the starting location of the higher compression rate. The theory predicted a compression in the cusp region, x/c between 0.80 and 0.95, when the experimental data were displaying a slight expansion. The compression prediction was also seen in the Mach 0.20 case for the swept-pylon installation. Local decambering of the airfoil shape due to boundary-layer growth could be a cause of this discrepancy.

The pylon and nacelle pressures are plotted in figure 16(c). The pylon upper-row pressure predictions did not capture the nature of the experimental data in the region of x/c between -0.20 and 0.10. The pressure peak at $x/c = -0.05$ and the expansion at $x/c = 0.20$ were not predicted in magnitude. The pressures on the pylon bottom row were predicted better, but still missed the magnitude of the expansion at $x/c = 0.20$. There was again the problem of predicting the low-pressure bulge. The channel of accelerated flow greatly influenced the pylon flow field, and the problem was compounded by the acceleration of the flow about the pylon leading edge. The degree of this interaction was more severe experimentally than was predicted by the code. The mismatch in predicted inlet leading-edge pressures with experiment was no worse than for the Mach 0.20 case. The overall range of pressures along the nacelle body increased. This increase indicated that stronger flow gradients were prevalent, particularly for the 22.5° angular station. A slight increase in velocities occurred for the 90.0° and 180.0° nacelle stations. These changes, however, were of a much smaller magnitude for the 22.5° station. This would indicate that the interference due to the wing and pylon flows on the nacelle flow lessens progressively around the nacelle to the bottom. This decreasing trend was expected and was predicted.

The wing pressures for configuration 5 at $\alpha = 3^\circ$ are presented in figure 16(d). The predicted leading-edge pressures were much too high for all the wing stations. All the predictions crossed the experimental data by $x/c = 0.10$ and remained low in predicted level for the rest of the wing chord. The predicted wing lower-surface comparison results were better than the comparison results in figure 16(c). The lower-surface acceleration shown previously for $y/d_N = 0.500$, with x/c between 0.05 and 0.20, was suppressed because of the movement of the leading-edge stagnation point rearward along the lower surface. The predicted pressures were still too high; therefore, the calculated flow velocity in that region was too slow. The predicted pressure change in the cusp region for both wing stations, $y/d_N = 0.500$ and 1.500 , was too far forward. The predicted pylon and nacelle pres-

tures (fig. 16(d)) show a slightly better overall comparison than the $\alpha = 0^\circ$ data. The same pressure peak and following expansion of the flow shown previously along the pylon upper-row area are observed. The average velocity of the pylon flow is reduced, compared with the 0° case, as a result of the influence of the stagnation flow of the wing leading edge. The pressure peak location was still not predicted, and the local pressure gradient at $x/c \approx 0$ was too low. This problem occurred, to a lesser degree, for the Mach 0.20 comparisons. Therefore, it was not thought that the mismatch of pylon pressures was caused by insufficient aerodynamic flow modeling. It is likely that the numerical modeling of the body geometry does not properly describe the physical configuration. The region of the junction of the pylon leading edge and wing leading edge was a highly complex intersection. The discrete intervals used in the geometry paneling might not be an accurate representation of the junction region. The details of the flow due to the given physical model would then not be predictable by the theoretical method. Comparisons of pressures on the inlet and nacelle reveal, as shown previously, leading-edge and trailing-edge mismatches. The changes in pressures on the nacelle caused by the change in angle of attack, for x/c between -0.60 and 0 , were predicted well. The levels of the predicted pressure data were still generally high for the nacelle.

Unswept-pylon configuration.- Configuration 3 pressure data for Mach 0.60 are presented in figure 16(e). There again was possible leading-edge separation in the upper-surface, leading-edge pressures at $y/d_N = 0$. The calculations for x/c between 0 and 0.10 are off because of this possibility. The general decrease in leading-edge pressure caused by the nacelle-pylon installation was predicted. The increase in upper-surface pressures caused by the nacelle-pylon installation for x/c between 0.70 and 0.90 was also predicted. The trend of the pressure changes in the central region of the wing was good. The leading-edge pressure changes were overpredicted, and the trailing-edge pressure changes were underpredicted. The prediction of the low-pressure maximum was off. Again, the trend of pylon pressures was predicted well, but was generally too high in level. The inlet leading-edge predictions of pressures were particularly bad. This may have been caused by an increase in wing, pylon, and nacelle interaction that was not predicted by the code. The trailing-edge area of the pylon and nacelle row at 22.5° may have experienced a mild flow separation. An already present adverse pressure gradient on the wing lower surface was worsened by the installation of the pylon and nacelle. A boundary layer tends to separate when developing in an adverse pressure-gradient field. Such an adverse pressure field exists in the area between $x/c = 0.50$ and 1.00 in the junction region of the wing, pylon, and nacelle. The code cannot predict the effects of boundary-layer growth or separation. Inspection of the comparison between the 22.5° , 90.0° , and 180.0° angular stations on the nacelle shows the inability to predict these viscous effects. The angular rows at 90.0° and 180.0° were predicted satisfactorily in trend for the trailing-edge region. The predicted pressures along the 22.5° row on the nacelle deviated considerably from the experimental data after $x/c = 0.85$. There were similar deviations in the predicted pressures in both the upper and lower rows of the pylon comparison plots.

Comparison at Mach 0.80

Clean wing.- Data for configuration 1, the conventional leading-edge clean wing, at Mach 0.80 and at $\alpha = 0^\circ$, are presented in figure 17(a). At this Mach number, even the character of the flow on the upper surface of the wing is very badly predicted by the theory. Except for the fact that both experiment and theory were recompressing at the wing trailing edge, the agreement was poor. The sonic pressure coefficient for Mach 0.80 is 0.435 , so about 70 percent of the wing upper-surface

flow was supersonic. The nature of the equations of motion modeling this flow is much different than the linear potential-flow modeling of the present code. The compressibility and irrotational flow assumptions were violated once supersonic flow and shocks occurred in the flow.

Only about 20 percent of the wing lower-surface flow was supersonic. This region, an enhancement of the low-pressure bulge mentioned previously, was not predicted for the wing stations $y/d_N = 0.500$ and 1.500 . The deviation between theory and experiment for the lower-surface pressures at x/c between 0.65 and 0.85 was similar to that between the pylon and nacelle in figure 16(e). It is likely that viscous effects account for much of the discrepancy in the pressure gradients. The disagreement in the levels of the pressure could be due to the Prandtl-Glauert compressibility effects.

Swept-pylon configuration.— The pressures for configuration 5 are shown in figure 17(b) for Mach 0.80 at $\alpha = 0^\circ$. The theory did predict a slight decrease in the pressure at the leading edge for $y/d_N = 0$. The experimental pressure decreased for x/c between 0.05 and 0.20 in all the wing upper-surface stations. An increase in pressure had been predicted for the $y/d_N = 0.500$ and 1.500 wing stations in the same x/c region. Therefore, the theory predicts that the flow will slow down. In reality, the flow velocity increases compared with the uninstalled case. The acceleration of the upper-surface flow as a result of the pylon installation likely feeds farther out along the span of the wing than the code predicts.

The large supersonic bubble on the lower surface caused by the effect of the pylon-wing junction, for x/c between 0.025 and 0.30 , was not predicted at all. Also, the theoretical prediction bears poor resemblance to experiment for $y/d_N = 0.500$ at $x/c > 0.5$. As mentioned previously, the boundary layer is sensitive to its surrounding pressure field. Therefore, the poor resemblance could be the result of very strong viscous effects caused by the pylon and wing interaction. The flow across a shock experiences a large positive pressure gradient. As a result, the boundary layer either separates from the surface or thickens drastically. The resulting change in the boundary-layer profile has a decambering effect on most airfoil sections and usually results in a decrease in the lift generated by the section. As mentioned previously, these types of interactions are absent from the code formulation.

The predictions of the pylon and inlet-nacelle flow presented in figure 17(b) were better than those of the wing. The comparison of the general trend was good. As expected, predictions of the extent of supersonic flow and peak local Mach number were quite poor. An interesting point, though not unexpected, was the prediction of the pressure along the 180.0° nacelle station. The theoretical agreement with experiment was comparable to that for the Mach 0.60 case for the same nacelle station. This agreement could be due to the absence of any local supersonic flow on that portion of the nacelle. The influence due to the pylon and wing is greatly diminished by the 180.0° row on the nacelle. Therefore, the flow interaction caused by the wing and pylon is not present to cause the supersonic bubble.

Unswept-pylon configuration.— Poor agreement of the predicted pressures with experiment occurred for most of configuration 3 at Mach 0.80 and at $\alpha = 0^\circ$. (See fig. 17(c).) These results are not completely unexpected, especially considering the previous comparison at this Mach number. About 70 percent of the wing upper-surface flow was supersonic. This flow and the resulting compression shock were not predicted. The lower-surface shock at $x/c \approx 0.30$ and at $y/d_N = 0.500$ and the resulting boundary-layer separation were not predicted. These same flow phenomena

can be seen on both pylon rows and along the inlet-nacelle 22.5° row. These strong interactions were not expected to be predicted. The wing lower-surface pressures at $y/d_N = 1.500$ do not display the same flat trend as the $y/d_N = 0.500$ row after $x/c = 0.400$. It is likely that the diminished effect of the nacelle at this span station allowed the flow to remain attached and to experience the usual recompression trend characteristic of supercritical airfoil sections.

The inlet-nacelle pressures for the 90.0° nacelle station were predicted reasonably well. One of the trouble spots was the inlet leading-edge mismatch discussed previously. The mismatch in the pressures in the aft region of the nacelle was likely due to the influence of the boundary-layer separation feeding around the nacelle body. There was a small region of sonic flow on the inlet-nacelle bottom row (180° at $x/c \approx -0.10$). A majority of the inlet-nacelle flow along that row was subsonic. The theory comparison was, subsequently, close in trend prediction but still high in pressure level.

Comparison of Installation Effects for Mach 0.60

The incremental pressure change ($C_{p, \text{installed}} - C_{p, \text{uninstalled}}$) on the wing due to the presence of the swept-pylon installation is shown in figure 18 for a Mach number of 0.60. The lower-surface leading-edge pressure drop as a result of the pylon installation was predicted. The magnitude and width, that is, extent over the wing chord, of the drop were off. The overall trend of the increment was predicted, except for the trailing-edge pressures past $x/c = 0.60$. The code did not predict the negative increment in pressures in the trailing-edge region. An increase in pressure was observed on the trailing-edge area of the upper surface, which can greatly influence the location of any shock waves present. The code approximated the trends, but did not adequately predict the pressure data near the leading and trailing edges.

CONCLUSIONS

A wind-tunnel investigation was conducted to obtain subsonic and transonic pressure data on wing, pylon, and nacelle configurations with supercritical wings. In addition to obtaining the data, numerical predictions of static-pressure coefficients on the model were made using a three-dimensional-panel aerodynamic code that solved the linearized potential-flow equations. An analysis of the data and a comparison of the data with the numerical solutions indicate the following results:

1. In the subsonic regime, the interactions between the wing, pylon, and nacelle flow fields are weak and can be reasonably predicted by incompressible potential-flow methods.

2. In the transonic regime, the interactions are strong and cannot be modeled by elementary, that is, incompressible and inviscid, numerical theories. More advanced theories and numerical techniques are required to adequately model compressibility and rotational effects such as the presence of boundary-layer growth, shocks, and separated flows.

This report provides experimental data for the assessment and verification of computational prediction techniques developed for wing, pylon, and nacelle studies.

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APPENDIX

TABULATED PRESSURE DATA

The following table is an index to the pressure data that are tabulated in this appendix.

Table	Pylon planform	Leading-edge type	Transition location	Configuration
A1	None	Blunt	0.29c	1
A2	None	Blunt	.05c	1
A3	None	Blunt	Natural	1
A4	None	Sharp	.29c	2
A5	Unswept	Sharp	.29c	3
A6	Swept	Sharp	.29c	4
A7	Swept	Blunt	.29c	5
A8	Swept	Blunt	.05c	5
A9	Swept	Blunt	Natural	5

APPENDIX

TABLE A1.- AERODYNAMIC PRESSURE-COEFFICIENT DATA FOR CONFIGURATION 1
AT $x_t = 0.29c$

CONFIGURATION 1 MACH = .200 ALPHA = .05										
WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.793	-.828	-.848			-.110	-.116			
.050	-.687	-.652	-.733	-.667	-.672	-.200	-.212	-.235	-.253	
.075	-.612	-.597	-.607			-.235	-.229			
.100	-.526	-.476	-.537	-.552	-.542	-.229	-.235	-.247	-.241	
.150	-.456	-.476	-.486			-.217	-.235			
.200	-.436	-.436	-.441	-.436	-.446	-.200	-.200	-.217	-.223	
.250	-.391	-.346	-.406			-.187	-.177			
.300	-.401	-.396	-.391	-.391	-.406	-.189	-.204		-.235	
.350	-.334	-.334	-.334			-.187	-.184			
.400	-.339	-.325	-.327	-.327	-.330	-.191	-.177	-.169	-.184	
.450	-.334	-.323	-.321			-.189	-.182			
.500	-.330	-.321	-.323	-.330	-.327	-.191	-.182	-.184	-.209	
.550	-.330	-.325	-.330			-.191	-.184			
.600	-.332	-.334	-.343	-.334	-.339	-.184		-.167	-.182	
.650	-.343	-.307	-.337			-.006	.001			
.700	-.361	-.349	-.349	-.349	-.355	.168	.166	.163	.153	
.750	-.367	-.355	-.361			.259	.259			
.800	-.373	-.361	-.367	-.385	-.373	.316	.318		.303	
.850	-.355	-.343	-.349			.357	.355			
.900	-.283	-.277	-.283	-.289	-.307	.389		.370	.370	
.950			-.152			.367				

CONFIGURATION 1 MACH = .200 ALPHA = 1.05										
WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.129	-1.134	-1.129			.093	.081			
.050	-.894	-.894	-.909	-.874	-.894	-.050	-.068	-.079	-.085	
.075	-.769	-.749	-.779			-.109	-.109			
.100	-.674	-.564	-.684	-.689	-.689	-.127	-.109	-.139	-.151	
.150	-.549	-.564	-.559			-.139	-.163			
.200	-.514	-.504	-.514	-.509	-.529	-.139	-.127	-.145	-.163	
.250	-.454	-.449	-.449			-.127	-.117			
.300	-.454	-.449	-.444	-.434	-.444	-.139	-.156		-.187	
.350	-.397	-.388	-.395			-.137	-.127			
.400	-.386	-.377	-.379	-.384	-.386	-.147	-.139	-.137	-.156	
.450	-.373	-.368	-.368			-.151	-.147			
.500	-.366	-.361	-.364	-.368	-.368	-.161	-.154	-.161	-.178	
.550	-.361	-.355	-.361			-.166	-.161			
.600	-.357	-.354	-.366	-.355	-.364	-.144		-.151	-.159	
.650	-.372	-.366	-.366			.006	.018			
.700	-.389	-.378	-.378	-.378	-.378	.179	.177	.174	.165	
.750	-.389	-.378	-.372			.267	.262			
.800	-.395	-.378	-.384	-.395	-.389	.321	.324		.311	
.850	-.366	-.354	-.360			.363	.360			
.900	-.294	-.282	-.282	-.288	-.312	.385		.375	.375	
.950			-.151			.370				

APPENDIX

TABLE A1.- Continued

CONFIGURATION 1 MACH = .200 ALPHA = 2.06

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.351	-1.457	-1.432			.253	.247			
.050	-1.061	-1.081	-1.121	-1.056	-1.076	.081	.075	.069	.039	
.075	-.916	-.916	-.941			-.014	-.014			
.100	-.790	-.655	-.755	-.835	-.846	-.032	-.038	-.038	-.056	
.150	-.635	-.655	-.640			-.062	-.060			
.200	-.580	-.575	-.590	-.585	-.580	-.074	-.068	-.068	-.097	
.250	-.525	-.520	-.530			-.078	-.068			
.300	-.515	-.500	-.515	-.495	-.515	-.093	-.100		-.139	
.350	-.431	-.437	-.444			-.093	-.085			
.400	-.424	-.424	-.420	-.420	-.424	-.102	-.102	-.095	-.107	
.450	-.406	-.406	-.400			-.117	-.115			
.500	-.400	-.391	-.395	-.393	-.406	-.129	-.127	-.127	-.147	
.550	-.389	-.382	-.393			-.137	-.134			
.600	-.382	-.384	-.391	-.375	-.389	-.120		-.125	-.137	
.650	-.384	-.390	-.390			.030	.035			
.700	-.396	-.396	-.396	-.390	-.396	.192	.189	.187	.180	
.750	-.396	-.396	-.396			.280	.275			
.800	-.396	-.384	-.396	-.414	-.402	.334	.336		.324	
.850	-.366	-.360	-.366			.376	.373			
.900	-.289	-.289	-.283	-.295	-.312	.398		.383	.383	
.950			-.151			.378				

CONFIGURATION 1 MACH = .200 ALPHA = 3.05

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.695	-1.750	-1.730			.431	.419			
.050	-1.329	-1.379	-1.374	-1.349	-1.329	.212	.218	.200	.182	
.075	-1.033	-1.027	-.977			.117	.123			
.100	-.897	-.736	-.887	-.912	-.897	.063	.069	.081	.057	
.150	-.726	-.736	-.767			.016	-.002			
.200	-.666	-.666	-.671	-.666	-.646	.010	.010	-.002	-.038	
.250	-.586	-.586	-.596			-.016	-.011			
.300	-.566	-.551	-.556	-.556	-.571	-.034	-.048		-.092	
.350	-.480	-.485	-.485			-.051	-.041			
.400	-.467	-.460	-.465	-.458	-.469	-.071	-.066	-.061	-.078	
.450	-.442	-.447	-.442			-.080	-.078			
.500	-.427	-.425	-.427	-.431	-.436	-.098	-.095	-.090	-.117	
.550	-.416	-.416	-.416			-.112	-.107			
.600	-.407	-.407	-.407	-.411	-.411	-.098		-.100	-.115	
.650	-.403	-.403	-.403			.043	.045			
.700	-.415	-.409	-.415	-.409	-.409	.207	.202	.202	.190	
.750	-.403	-.403	-.403			.290	.288			
.800	-.397	-.391	-.391	-.415	-.409	.344	.344		.335	
.850	-.361	-.361	-.361			.381	.381			
.900	-.283	-.277	-.277	-.283	-.301	.403		.389	.389	
.950			-.139			.384				

APPENDIX

TABLE A1.- Continued

CONFIGURATION 1

MACH = .200

ALPHA = 4.05

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C										
.025		-2.028	-1.998	-2.053			.525	.555		
.050		-1.397	-1.316	-1.321	-1.286	-1.662	.330	.318	.312	.294
.075		-1.216	-1.166	-1.181			.211	.205		
.100		-1.026	-.840	-1.016	-1.056	-1.056	.152	.152	.164	.134
.150		-.815	-.840	-.845			.087	.069		
.200		-.745	-.745	-.745	-.735	-.745	.063	.051	.045	.028
.250		-.655	-.645	-.660			.033	.045		
.300		-.620	-.610	-.615	-.600	-.630	.018	.001		-.044
.350		-.528	-.537	-.533			-.007	-.002		
.400		-.508	-.499	-.502	-.502	-.504	-.029	-.024	-.021	-.038
.450		-.484	-.473	-.475			-.051	-.043		
.500		-.457	-.453	-.455	-.457	-.464	-.063	-.063	-.061	-.088
.550		-.440	-.442	-.444			-.080	-.075		
.600		-.431	-.431	-.435	-.426	-.440	-.075		-.073	-.093
.650		-.432	-.426	-.426			.062	.065		
.700		-.438	-.426	-.432	-.426	-.438	.219	.219	.219	.207
.750		-.426	-.420	-.420			.297	.300		
.800		-.408	-.408	-.414	-.420	-.420	.356	.356		.341
.850		-.378	-.366	-.366			.395	.390		
.900		-.289	-.249	-.283	-.283	-.300	.412		.395	.398
.950				-.139			.390			

CONFIGURATION 1

MACH = .200

ALPHA = 5.05

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C										
.025		-2.410	-2.440	-2.451			.671	.647		
.050		-1.549	-1.564	-1.479	-1.444	-1.444	.445	.433	.421	.403
.075		-1.343	-1.343	-1.363			.308	.308		
.100		-1.182	-.935	-1.162	-1.162	-1.187	.230	.236	.230	.230
.150		-.950	-.935	-.935			.153	.153		
.200		-.834	-.814	-.839	-.819	-.819	.123	.117	.093	.093
.250		-.719	-.724	-.729			.085	.097		
.300		-.678	-.668	-.673	-.668	-.683	.050	.050		.004
.350		-.578	-.580	-.582			.031	.040		
.400		-.553	-.544	-.542	-.553	-.562	.011	.013	.016	-.004
.450		-.517	-.517	-.517			-.009	-.011		
.500		-.497	-.495	-.495	-.500	-.502	-.036	-.034	-.034	-.053
.550		-.477	-.473	-.475			-.058	-.048		
.600		-.459	-.457	-.464	-.455	-.466	-.051		-.061	-.071
.650		-.452	-.446	-.452			.072	.082		
.700		-.452	-.446	-.446	-.446	-.452	.235	.230	.228	.215
.750		-.446	-.434	-.434			.314	.309		
.800		-.416	-.410	-.416	-.434	-.428	.365	.365		.348
.850		-.374	-.368	-.380			.402	.400		
.900		-.284	-.278	-.278	-.284	-.308	.420		.402	.402
.950				-.140			.392			

APPENDIX

TABLE A1.- Continued

CONFIGURATION 1 MACH = .300 ALPHA = .08

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C										
.025		-.793	-.842	-.867			-.096	-.154		
.050		-.696	-.689	-.724	-.682	-.710	-.214	-.242	-.245	-.250
.075		-.615	-.634	-.627			-.247	-.253		
.100		-.564	-.488	-.557	-.553	-.555	-.225	-.245	-.272	-.267
.150		-.474	-.488	-.481			-.245	-.256		
.200		-.462	-.451	-.444	-.455	-.451	-.217	-.212	-.220	-.239
.250		-.411	-.395	-.404			-.199	-.193		
.300		-.409	-.386	-.391	-.384	-.386	-.203	-.215		-.267
.350		-.347	-.358	-.351			-.195	-.178		
.400		-.344	-.341	-.349	-.344	-.346	-.191	-.188	-.190	-.197
.450		-.337	-.335	-.338			-.193	-.191		
.500		-.340	-.333	-.335	-.343	-.343	-.191	-.199	-.197	-.218
.550		-.343	-.341	-.344			-.194	-.195		
.600		-.348	-.350	-.354	-.338	-.348	-.176		-.178	-.187
.650		-.355	-.355	-.355			-.011	.000		
.700		-.377	-.366	-.369	-.369	-.374	.171	.171	.168	.156
.750		-.383	-.377	-.380			.265	.263		
.800		-.388	-.383	-.388	-.407	-.394	.323	.325		.313
.850		-.372	-.363	-.366			.361	.365		
.900		-.300	-.292	-.294	-.303	-.327	.389		.384	.380
.950				-.162			.382			

CONFIGURATION 1 MACH = .401 ALPHA = .07

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C										
.025		-.829	-.859	-.886			-.144	-.163		
.050		-.699	-.714	-.729	-.690	-.725	-.236	-.280	-.297	-.294
.075		-.654	-.633	-.660			-.273	-.283		
.100		-.576	-.502	-.588	-.591	-.589	-.280	-.288	-.281	-.304
.150		-.484	-.502	-.504			-.268	-.281		
.200		-.477	-.456	-.467	-.462	-.452	-.242	-.233	-.247	-.271
.250		-.418	-.410	-.422			-.218	-.212		
.300		-.425	-.411	-.395	-.405	-.406	-.231	-.232		-.281
.350		-.367	-.365	-.367			-.213	-.202		
.400		-.365	-.353	-.359	-.366	-.360	-.218	-.204	-.208	-.218
.450		-.359	-.352	-.349			-.217	-.206		
.500		-.360	-.351	-.353	-.359	-.359	-.207	-.212	-.214	-.242
.550		-.361	-.352	-.360			-.214	-.216		
.600		-.362	-.363	-.366	-.356	-.369	-.190		-.193	-.205
.650		-.375	-.372	-.372			-.015	-.006		
.700		-.399	-.388	-.391	-.390	-.394	.172	.171	.166	.157
.750		-.406	-.398	-.398			.266	.268		
.800		-.414	-.401	-.407	-.430	-.419	.328	.330		.316
.850		-.391	-.378	-.386			.369	.373		
.900		-.314	-.307	-.310	-.320	-.346	.396		.390	.387
.950				-.165			.386			

APPENDIX

TABLE A1.- Continued

CONFIGURATION 1 MACH = .499 ALPHA = .08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.813	-.920	-.904			-.185	-.201		
.050	-.726	-.762	-.749	-.702	-.729	-.288	-.302	-.313	-.337
.075	-.662	-.663	-.668			-.304	-.309		
.100	-.589	-.515	-.593	-.607	-.605	-.320	-.305	-.320	-.337
.150	-.494	-.515	-.510			-.302	-.319		
.200	-.488	-.485	-.485	-.469	-.460	-.272	-.259	-.277	-.291
.250	-.432	-.432	-.421			-.245	-.232		
.300	-.444	-.421	-.434	-.417	-.438	-.245	-.237		-.309
.350	-.381	-.380	-.390			-.234	-.219		
.400	-.381	-.369	-.384	-.380	-.379	-.226	-.227	-.235	-.240
.450	-.371	-.377	-.371			-.224	-.228		
.500	-.373	-.375	-.365	-.373	-.368	-.227	-.234	-.236	-.256
.550	-.374	-.372	-.378			-.235	-.230		
.600	-.378	-.380	-.383	-.372	-.381	-.202		-.210	-.217
.650	-.393	-.387	-.393			-.013	-.003		
.700	-.417	-.411	-.407	-.406	-.407	.179	.180	.173	.166
.750	-.423	-.422	-.416			.274	.275		
.800	-.429	-.420	-.424	-.447	-.432	.334	.335		.325
.850	-.404	-.394	-.401			.377	.378		
.900	-.321	-.312	-.315	-.323	-.356	.406		.404	.398
.950			-.161			.399			

CONFIGURATION 1 MACH = .599 ALPHA = .08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.809	-.901	-.923			-.241	-.237		
.050	-.764	-.765	-.769	-.736	-.749	-.356	-.370	-.375	-.405
.075	-.721	-.711	-.707			-.360	-.340		
.100	-.642	-.537	-.637	-.623	-.641	-.360	-.360	-.374	-.379
.150	-.541	-.537	-.548			-.348	-.367		
.200	-.526	-.517	-.521	-.498	-.487	-.300	-.303	-.329	-.334
.250	-.467	-.451	-.449			-.273	-.265		
.300	-.481	-.465	-.464	-.447	-.475	-.277	-.270		-.357
.350	-.405	-.416	-.405			-.261	-.258		
.400	-.409	-.396	-.411	-.409	-.407	-.255	-.262	-.270	-.275
.450	-.403	-.400	-.395			-.255	-.264		
.500	-.407	-.399	-.400	-.405	-.392	-.261	-.264	-.264	-.299
.550	-.408	-.406	-.408			-.266	-.257		
.600	-.410	-.411	-.415	-.400	-.407	-.221		-.230	-.245
.650	-.418	-.415	-.419			-.013	-.004		
.700	-.448	-.439	-.443	-.435	-.439	.181	.181	.176	.167
.750	-.455	-.451	-.448			.275	.276		
.800	-.462	-.453	-.456	-.477	-.466	.335	.338		.325
.850	-.429	-.416	-.425			.377	.382		
.900	-.329	-.319	-.324	-.333	-.363	.410		.409	.404
.950			-.150			.409			

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TABLE A1.- Continued

CONFIGURATION 1 MACH = .599 ALPHA = 1.08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-1.180	-1.283	-1.286			-.002	.006		
.050	-1.024	-1.026	-1.053	-1.015	-1.037	-.137	-.168	-.193	-.202
.075	-.915	-.903	-.896			-.225	-.226		
.100	-.787	-.651	-.776	-.829	-.795	-.233	-.227	-.226	-.261
.150	-.645	-.651	-.651			-.228	-.251		
.200	-.624	-.614	-.614	-.595	-.583	-.228	-.211	-.228	-.254
.250	-.554	-.549	-.542			-.210	-.193		
.300	-.555	-.539	-.545	-.524	-.549	-.220	-.207		-.293
.350	-.467	-.480	-.478			-.214	-.206		
.400	-.463	-.451	-.466	-.459	-.456	-.216	-.224	-.218	-.223
.450	-.453	-.448	-.450			-.220	-.224		
.500	-.449	-.440	-.441	-.449	-.437	-.229	-.228	-.231	-.261
.550	-.446	-.440	-.444			-.236	-.229		
.600	-.445	-.442	-.450	-.434	-.443	-.199		-.202	-.224
.650	-.446	-.444	-.450			.002	.010		
.700	-.472	-.462	-.466	-.458	-.461	.196	.198	.190	.180
.750	-.475	-.467	-.466			.292	.295		
.800	-.476	-.461	-.469	-.490	-.474	.353	.355		.340
.850	-.437	-.419	-.431			.396	.399		
.900	-.327	-.316	-.322	-.332	-.356	.426		.419	.415
.950			-.144			.418			

CONFIGURATION 1 MACH = .600 ALPHA = 2.08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-1.731	-1.730	-1.705			.167	.183		
.050	-1.429	-1.456	-1.473	-1.413	-1.460	.014	.012	-.020	-.056
.075	-1.058	-.987	-1.010			-.079	-.080		
.100	-.935	-.776	-.913	-.939	-.919	-.122	-.111	-.111	-.127
.150	-.775	-.776	-.764			-.151	-.163		
.200	-.725	-.693	-.703	-.704	-.690	-.147	-.127	-.145	-.167
.250	-.627	-.616	-.623			-.141	-.108		
.300	-.624	-.594	-.608	-.605	-.623	-.157	-.144		-.222
.350	-.530	-.530	-.529			-.166	-.152		
.400	-.520	-.506	-.502	-.514	-.505	-.170	-.169	-.183	-.188
.450	-.499	-.497	-.481			-.179	-.176		
.500	-.491	-.476	-.469	-.485	-.478	-.193	-.187	-.196	-.226
.550	-.483	-.466	-.467			-.210	-.197		
.600	-.476	-.463	-.474	-.453	-.469	-.179		-.184	-.201
.650	-.473	-.471	-.460			.017	.024		
.700	-.496	-.485	-.473	-.474	-.480	.215	.205	.207	.195
.750	-.490	-.477	-.470			.310	.298		
.800	-.483	-.462	-.470	-.493	-.488	.371	.357		.358
.850	-.436	-.411	-.425			.413	.399		
.900	-.321	-.303	-.309	-.320	-.353	.440		.433	.430
.950			-.135			.422			

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TABLE A1.- Continued

CONFIGURATION 1 MACH = .600 ALPHA = 3.10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-2.190	-2.298	-2.274			.334	.356		
.050	-2.030	-2.080	-2.057	-2.051	-1.941	.161	.143	.134	.113
.075	-1.322	-1.196	-1.323			.057	.048		
.100	-1.025	-.888	-.976	-1.019	-.989	.001	.005	-.004	-.016
.150	-.886	-.888	-.879			-.042	-.059		
.200	-.821	-.787	-.798	-.779	-.757	-.062	-.061	-.070	-.096
.250	-.713	-.688	-.705			-.092	-.041		
.300	-.693	-.660	-.673	-.653	-.671	-.115	-.088		-.178
.350	-.588	-.591	-.546			-.125	-.096		
.400	-.564	-.560	-.556	-.555	-.556	-.133	-.119	-.126	-.138
.450	-.547	-.543	-.522			-.143	-.139		
.500	-.532	-.521	-.506	-.517	-.514	-.160	-.159	-.156	-.195
.550	-.516	-.504	-.493			-.177	-.172		
.600	-.502	-.502	-.490	-.487	-.492	-.156		-.155	-.178
.650	-.495	-.490	-.479			.027	.039		
.700	-.512	-.501	-.485	-.487	-.490	.224	.220	.215	.201
.750	-.500	-.488	-.471			.317	.315		
.800	-.482	-.468	-.455	-.487	-.477	.376	.375		.362
.850	-.423	-.413	-.401			.424	.416		
.900	-.297	-.293	-.287	-.300	-.325	.450		.435	.431
.950			-.124			.431			

CONFIGURATION 1 MACH = .651 ALPHA = .10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.834	-.902	-.902			-.242	-.262		
.050	-.803	-.803	-.811	-.748	-.763	-.375	-.379	-.445	-.445
.075	-.735	-.736	-.727			-.380	-.417		
.100	-.662	-.586	-.641	-.654	-.673	-.395	-.426	-.438	-.456
.150	-.562	-.586	-.566			-.371	-.419		
.200	-.559	-.545	-.523	-.519	-.509	-.331	-.351	-.378	-.389
.250	-.488	-.478	-.458			-.302	-.287		
.300	-.513	-.499	-.471	-.459	-.492	-.312	-.294		-.406
.350	-.436	-.433	-.436			-.292	-.279		
.400	-.431	-.423	-.432	-.423	-.400	-.283	-.282	-.286	-.323
.450	-.424	-.421	-.420			-.282	-.282		
.500	-.426	-.418	-.423	-.416	-.398	-.280	-.272	-.280	-.350
.550	-.426	-.423	-.423			-.282	-.268		
.600	-.433	-.431	-.437	-.409	-.405	-.232		-.238	-.287
.650	-.448	-.442	-.445			-.007	.012		
.700	-.480	-.464	-.467	-.440	-.439	.189	.194	.172	.158
.750	-.483	-.472	-.469			.276	.278		
.800	-.484	-.471	-.478	-.484	-.471	.335	.337		.320
.850	-.443	-.428	-.437			.377	.380		
.900	-.328	-.320	-.323	-.323	-.354	.412		.397	.409
.950			-.135			.416			

APPENDIX

TABLE A1.- Continued

CONFIGURATION 1 MACH = .700 ALPHA = .02

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.746	-.885	-.855			-.285	-.272		
.050	-.806	-.821	-.827	-.765	-.832	-.398	-.429	-.415	-.498
.075	-.775	-.772	-.791			-.439	-.450		
.100	-.693	-.614	-.697	-.702	-.709	-.446	-.445	-.453	-.497
.150	-.590	-.614	-.613			-.435	-.442		
.200	-.586	-.579	-.583	-.565	-.554	-.376	-.370	-.386	-.402
.250	-.511	-.508	-.516			-.338	-.319		
.300	-.538	-.516	-.528	-.496	-.525	-.352	-.328		-.436
.350	-.444	-.442	-.452			-.325	-.319		
.400	-.448	-.436	-.449	-.454	-.444	-.323	-.322	-.331	-.326
.450	-.444	-.439	-.437			-.312	-.325		
.500	-.450	-.437	-.439	-.449	-.437	-.308	-.321	-.315	-.348
.550	-.450	-.443	-.448			-.308	-.306		
.600	-.455	-.453	-.464	-.449	-.446	-.240		-.253	-.269
.650	-.467	-.464	-.465			.005	.009		
.700	-.503	-.487	-.491	-.487	-.481	.194	.188	.178	.171
.750	-.509	-.494	-.497			.270	.276		
.800	-.511	-.490	-.506	-.532	-.506	.320	.332		.316
.850	-.456	-.440	-.454			.366	.378		
.900	-.318	-.313	-.320	-.330	-.355	.409		.409	.403
.950			-.118			.418			

CONFIGURATION 1 MACH = .750 ALPHA = .02

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.656	-.785	-.723			-.335	-.327		
.050	-.799	-.781	-.797	-.759	-.788	-.475	-.488	-.497	-.521
.075	-.827	-.812	-.822			-.520	-.532		
.100	-.774	-.669	-.800	-.760	-.798	-.530	-.542	-.558	-.557
.150	-.640	-.669	-.661			-.510	-.520		
.200	-.631	-.627	-.625	-.607	-.596	-.451	-.441	-.460	-.489
.250	-.555	-.562	-.548			-.390	-.368		
.300	-.606	-.587	-.610	-.540	-.592	-.397	-.393		-.511
.350	-.477	-.477	-.475			-.373	-.373		
.400	-.479	-.464	-.474	-.468	-.459	-.372	-.369	-.383	-.379
.450	-.473	-.476	-.470			-.359	-.359		
.500	-.479	-.471	-.473	-.479	-.452	-.350	-.356	-.357	-.395
.550	-.483	-.476	-.478			-.336	-.337		
.600	-.487	-.492	-.492	-.476	-.470	-.238		-.259	-.276
.650	-.507	-.503	-.504			.024	.021		
.700	-.545	-.534	-.532	-.520	-.510	.179	.182	.175	.164
.750	-.546	-.536	-.533			.239	.251		
.800	-.532	-.519	-.526	-.555	-.534	.290	.305		.292
.850	-.451	-.438	-.446			.337	.355		
.900	-.287	-.282	-.286	-.298	-.325	.391		.393	.390
.950			-.082			.412			

APPENDIX

TABLE A1.- Continued

CONFIGURATION 1 MACH = .801 ALPHA = .08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.517	-.618	-.604			-.327	-.326		
.050	-.649	-.633	-.687	-.623	-.665	-.490	-.518	-.552	-.561
.075	-.729	-.722	-.726			-.568	-.605		
.100	-.720	-.730	-.724	-.757	-.774	-.614	-.641	-.682	-.664
.150	-.712	-.730	-.721			-.695	-.638		
.200	-.753	-.736	-.738	-.745	-.767	-.652	-.677	-.664	-.700
.250	-.731	-.721	-.725			-.509	-.491		
.300	-.760	-.744	-.725	-.700	-.818	-.462	-.482		-.750
.350	-.697	-.669	-.680			-.447	-.445		
.400	-.689	-.691	-.676	-.663	-.675	-.462	-.459	-.468	-.418
.450	-.618	-.599	-.538			-.434	-.437		
.500	-.457	-.449	-.447	-.449	-.407	-.401	-.410	-.411	-.439
.550	-.450	-.440	-.449			-.348	-.358		
.600	-.476	-.481	-.494	-.474	-.466	-.204		-.242	-.250
.650	-.507	-.513	-.527			.051	.044		
.700	-.589	-.580	-.589	-.571	-.544	.151	.167	.161	.155
.750	-.629	-.628	-.613			.202	.216		
.800	-.578	-.563	-.575	-.613	-.604	.248	.263		.252
.850	-.395	-.384	-.398			.296	.312		
.900	-.215	-.210	-.219	-.232	-.270	.351		.368	.359
.950			-.036			.388			

CONFIGURATION 1 MACH = .799 ALPHA = 1.09

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.697	-.811	-.817			-.146	-.149		
.050	-.822	-.823	-.830	-.775	-.825	-.308	-.338	-.381	-.386
.075	-.868	-.854	-.881			-.400	-.422		
.100	-.877	-.886	-.882	-.877	-.907	-.423	-.438	-.456	-.466
.150	-.861	-.886	-.881			-.461	-.480		
.200	-.907	-.892	-.903	-.882	-.924	-.420	-.419	-.418	-.455
.250	-.892	-.878	-.877			-.370	-.348		
.300	-.893	-.904	-.897	-.869	-.948	-.408	-.407		-.583
.350	-.847	-.840	-.860			-.384	-.372		
.400	-.860	-.841	-.842	-.837	-.907	-.383	-.383	-.400	-.396
.450	-.838	-.841	-.828			-.373	-.376		
.500	-.851	-.843	-.846	-.848	-.908	-.360	-.370	-.371	-.414
.550	-.854	-.853	-.857			-.338	-.342		
.600	-.862	-.860	-.875	-.859	-.540	-.220		-.231	-.243
.650	-.855	-.851	-.797			.054	.052		
.700	-.622	-.573	-.602	-.507	-.441	.190	.194	.186	.173
.750	-.404	-.395	-.401			.240	.254		
.800	-.372	-.367	-.371	-.420	-.500	.293	.299		.280
.850	-.312	-.310	-.320			.351	.360		
.900	-.180	-.182	-.189	-.202	-.272	.406		.393	.383
.950			-.038			.430			

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TABLE A1.- Continued

CONFIGURATION 1 MACH = .799 ALPHA = 2.10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.842	-.938	-.960			-.008	.004		
.050	-.941	-.971	-1.002	-.950	-.946	-.182	-.187	-.214	-.233
.075	-1.001	-.998	-1.011			-.258	-.266		
.100	-1.014	-1.010	-1.010	-.993	-1.022	-.308	-.308	-.307	-.327
.150	-.986	-1.010	-1.008			-.336	-.359		
.200	-1.015	-1.004	-1.002	-1.003	-1.017	-.317	-.293	-.335	-.353
.250	-.995	-.993	-.997			-.292	-.266		
.300	-.994	-1.009	-1.007	-.973	-1.034	-.330	-.308		-.493
.350	-.951	-.944	-.966			-.322	-.303		
.400	-.960	-.939	-.938	-.944	-.985	-.331	-.330	-.350	-.354
.450	-.936	-.935	-.933			-.338	-.338		
.500	-.941	-.928	-.946	-.933	-.967	-.346	-.350	-.355	-.395
.550	-.942	-.940	-.954			-.342	-.342		
.600	-.950	-.948	-.962	-.948	-.971	-.240		-.251	-.261
.650	-.950	-.939	-.943			.034	.030		
.700	-.957	-.951	-.966	-.934	-.637	.202	.191	.187	.175
.750	-.472	-.469	-.462			.266	.263		
.800	-.319	-.320	-.322	-.348	-.376	.322	.320		.291
.850	-.234	-.233	-.241			.376	.374		
.900	-.157	-.151	-.151	-.154	-.201	.424		.406	.398
.950			-.089			.428			

CONFIGURATION 1 MACH = .849 ALPHA = .08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.356	-.452	-.438			-.283	-.284		
.050	-.488	-.486	-.531	-.473	-.521	-.439	-.438	-.477	-.470
.075	-.581	-.573	-.571			-.539	-.540		
.100	-.578	-.596	-.584	-.608	-.617	-.558	-.593	-.610	-.597
.150	-.583	-.598	-.601			-.665	-.617		
.200	-.638	-.622	-.628	-.631	-.648	-.690	-.691	-.672	-.669
.250	-.629	-.619	-.628			-.653	-.640		
.300	-.651	-.644	-.644	-.617	-.698	-.682	-.692		-.791
.350	-.615	-.602	-.615			-.692	-.691		
.400	-.634	-.614	-.615	-.612	-.694	-.722	-.713	-.733	-.725
.450	-.627	-.621	-.624			-.767	-.763		
.500	-.643	-.635	-.636	-.641	-.665	-.821	-.816	-.835	-.860
.550	-.659	-.655	-.654			-.485	-.345		
.600	-.677	-.672	-.688	-.673	-.649	-.254		-.261	-.265
.650	-.688	-.686	-.691			-.220	-.215		
.700	-.759	-.745	-.756	-.740	-.666	-.218	-.214	-.211	-.236
.750	-.706	-.751	-.731			-.191	-.207		
.800	-.266	-.266	-.259	-.262	-.350	-.168	-.193		-.233
.850	-.130	-.129	-.127			-.115	-.148		
.900	-.053	-.053	-.047	-.049	-.132	-.010		-.089	-.148
.950			.002			.130			

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TABLE A1.- Concluded

CONFIGURATION 1 MACH = .849 ALPHA = 1.06

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.453	-.558	-.562			-.178	-.183		
.050	-.598	-.574	-.613	-.579	-.618	-.349	-.363	-.399	-.398
.075	-.661	-.648	-.675			-.435	-.468		
.100	-.674	-.694	-.670	-.686	-.709	-.489	-.510	-.542	-.504
.150	-.683	-.694	-.684			-.592	-.535		
.200	-.728	-.711	-.714	-.723	-.735	-.596	-.612	-.582	-.602
.250	-.716	-.710	-.714			-.571	-.534		
.300	-.727	-.739	-.735	-.714	-.779	-.592	-.600		-.712
.350	-.701	-.687	-.711			-.596	-.602		
.400	-.719	-.705	-.704	-.701	-.766	-.624	-.642	-.674	-.654
.450	-.711	-.705	-.702			-.682	-.697		
.500	-.730	-.721	-.725	-.728	-.806	-.758	-.752	-.758	-.804
.550	-.745	-.734	-.743			-.671	-.650		
.600	-.756	-.751	-.770	-.748	-.751	-.262		-.243	-.227
.650	-.767	-.769	-.767			-.213	-.176		
.700	-.837	-.822	-.832	-.797	-.767	-.184	-.163	-.165	-.184
.750	-.391	-.396	-.408			-.153	-.141		
.800	-.217	-.216	-.216	-.218	-.270	-.106	-.109		-.170
.850	-.150	-.146	-.145			-.042	-.053		
.900	-.108	-.103	-.099	-.089	-.080	.091		.017	-.064
.950			-.071			.207			

CONFIGURATION 1 MACH = .874 ALPHA = .08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.304	-.395	-.389			-.242	-.249		
.050	-.428	-.419	-.472	-.403	-.461	-.390	-.403	-.427	-.420
.075	-.514	-.509	-.513			-.490	-.494		
.100	-.525	-.549	-.521	-.544	-.560	-.513	-.545	-.559	-.543
.150	-.536	-.549	-.543			-.616	-.573		
.200	-.587	-.571	-.575	-.576	-.589	-.639	-.639	-.624	-.623
.250	-.579	-.574	-.571			-.610	-.608		
.300	-.605	-.603	-.600	-.569	-.654	-.637	-.647		-.730
.350	-.568	-.556	-.581			-.655	-.657		
.400	-.589	-.577	-.574	-.568	-.653	-.680	-.677	-.698	-.693
.450	-.584	-.584	-.584			-.723	-.725		
.500	-.605	-.597	-.606	-.599	-.634	-.783	-.786	-.790	-.816
.550	-.624	-.620	-.625			-.863	-.873		
.600	-.643	-.640	-.649	-.636	-.638	-.539		-.405	-.400
.650	-.655	-.659	-.658			-.392	-.348		
.700	-.723	-.712	-.724	-.705	-.694	-.384	-.376	-.375	-.364
.750	-.768	-.765	-.765			-.378	-.373		
.800	-.312	-.305	-.302	-.295	-.645	-.364	-.367		-.383
.850	-.184	-.181	-.185			-.338	-.346		
.900	-.057	-.151	-.157	-.164	-.196	-.283		-.328	-.333
.950			-.141			-.187			

APPENDIX

TABLE A2.- AERODYNAMIC PRESSURE-COEFFICIENT DATA FOR CONFIGURATION 1
AT $x_L = 0.05c$.

CONFIGURATION 1 MACH = .201 ALPHA = .03										
WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.841	-.802	-.831			-.132	-.132			
.050	-.673	-.703	-.727	-.688	-.727	-.227	-.321	-.292	-.286	
.075	-.594	-.564	-.594			-.244	-.256			
.100	-.534	-.455	-.534	-.534	-.539	-.250	-.233	-.250	-.280	
.150	-.450	-.455	-.470			-.238	-.256			
.200	-.435	-.430	-.445	-.425	-.425	-.215	-.215	-.215	-.227	
.250	-.395	-.400	-.400			-.184	-.177			
.300	-.376	-.371	-.366	-.376	-.381	-.192	-.187		-.221	
.350	-.343	-.328	-.330			-.194	-.184			
.400	-.334	-.325	-.328	-.334	-.334	-.194	-.189	-.187	-.209	
.450	-.328	-.319	-.323			-.194	-.189			
.500	-.330	-.319	-.317	-.323	-.323	-.192	-.194	-.194	-.218	
.550	-.321	-.317	-.323			-.192	-.187			
.600	-.321	-.314	-.328	-.321	-.328	-.162		-.165	-.175	
.650	-.339	-.333	-.339			-.002	.003			
.700	-.351	-.345	-.351	-.351	-.351	.155	.151	.143	.143	
.750	-.362	-.351	-.351			.235	.238			
.800	-.362	-.351	-.356	-.374	-.368	.289	.284		.277	
.850	-.345	-.327	-.339			.325	.320			
.900	-.268	-.256	-.262	-.274	-.286	.347		.342	.337	
.950			-.132			.337				

CONFIGURATION 1 MACH = .202 ALPHA = 1.05										
WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.071	-1.100	-1.105			.091	.085			
.050	-.844	-.903	-.893	-.923	-.942	-.055	-.120	-.102	-.138	
.075	-.730	-.750	-.710			-.114	-.114			
.100	-.651	-.567	-.641	-.681	-.671	-.149	-.132	-.132	-.138	
.150	-.552	-.567	-.562			-.144	-.167			
.200	-.518	-.503	-.518	-.518	-.498	-.138	-.126	-.149	-.144	
.250	-.463	-.454	-.468			-.133	-.116			
.300	-.429	-.434	-.429	-.439	-.439	-.145	-.135		-.173	
.350	-.377	-.379	-.383			-.150	-.147			
.400	-.372	-.364	-.364	-.364	-.377	-.155	-.157	-.147	-.164	
.450	-.359	-.359	-.353			-.152	-.155			
.500	-.353	-.353	-.348	-.353	-.355	-.160	-.162	-.164	-.181	
.550	-.346	-.351	-.351			-.169	-.164			
.600	-.348	-.348	-.351	-.346	-.353	-.143		-.145	-.155	
.650	-.355	-.350	-.367			.010	.017			
.700	-.373	-.373	-.367	-.367	-.367	.169	.167	.164	.157	
.750	-.367	-.373	-.367			.251	.251			
.800	-.367	-.367	-.367	-.385	-.379	.307	.302		.290	
.850	-.344	-.338	-.344			.338	.336			
.900	-.273	-.261	-.267	-.273	-.285	.362		.358	.348	
.950			-.132			.348				

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TABLE A2.- Continued

CONFIGURATION 1 MACH = .202 ALPHA = 2.05

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.399	-1.384	-1.379			.254	.260			
.050	-1.064	-1.113	-1.118	-1.138	-1.113	.108	.056	.009	.021	
.075	-.892	-.867	-.867			-.003	-.003			
.100	-.793	-.660	-.763	-.803	-.788	-.032	-.032	-.032	-.026	
.150	-.660	-.660	-.660			-.073	-.085			
.200	-.601	-.586	-.586	-.586	-.576	-.079	-.055	-.073	-.079	
.250	-.532	-.527	-.527			-.075	-.070			
.300	-.482	-.473	-.487	-.492	-.492	-.087	-.092		-.120	
.350	-.437	-.433	-.440			-.099	-.099			
.400	-.426	-.407	-.424	-.413	-.420	-.111	-.113	-.113	-.130	
.450	-.405	-.394	-.398			-.118	-.125			
.500	-.398	-.392	-.381	-.394	-.394	-.125	-.133	-.138	-.152	
.550	-.392	-.378	-.383			-.138	-.138			
.600	-.383	-.372	-.383	-.376	-.387	-.123		-.125	-.138	
.650	-.384	-.378	-.378			.027	.032			
.700	-.396	-.384	-.384	-.384	-.390	.183	.183	.179	.171	
.750	-.384	-.384	-.384			.263	.263			
.800	-.384	-.378	-.378	-.396	-.390	.316	.314		.304	
.850	-.355	-.343	-.343			.352	.347			
.900	-.273	-.273	-.267	-.273	-.296	.371		.364	.359	
.950			-.132			.355				

CONFIGURATION 1 MACH = .202 ALPHA = 3.05

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.653	-1.716	-1.643			.415	.403			
.050	-1.206	-1.260	-1.299	-1.402	-1.339	.229	.166	.177	.160	
.075	-1.049	-1.039	-1.025			.108	.113			
.100	-.912	-.745	-.897	-.902	-.907	.055	.055	.073	.038	
.150	-.730	-.745	-.730			.009	-.014			
.200	-.657	-.662	-.667	-.652	-.652	-.008	.003	-.008	-.032	
.250	-.573	-.588	-.583			-.026	-.012			
.300	-.529	-.534	-.534	-.539	-.544	-.046	-.050		-.067	
.350	-.479	-.477	-.490			-.058	-.060			
.400	-.464	-.455	-.468	-.459	-.468	-.072	-.072	-.070	-.096	
.450	-.440	-.444	-.446			-.084	-.089			
.500	-.429	-.429	-.422	-.431	-.431	-.103	-.101	-.106	-.125	
.550	-.414	-.418	-.409			-.115	-.111			
.600	-.403	-.405	-.407	-.401	-.412	-.099		-.103	-.115	
.650	-.406	-.406	-.406			.041	.050			
.700	-.418	-.406	-.418	-.412	-.412	.197	.197	.194	.187	
.750	-.406	-.406	-.394			.276	.276			
.800	-.394	-.394	-.400	-.412	-.406	.329	.329		.319	
.850	-.365	-.359	-.359			.365	.362			
.900	-.283	-.272	-.272	-.283	-.295	.382		.377	.372	
.950			-.137			.365				

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TABLE A2.- Continued

CONFIGURATION 1 MACH = .201 ALPHA = 4.06										
WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-2.010	-2.065	-2.060			.545	.539			
.050	-1.422	-1.403	-1.363	-1.308	-1.497	.363	.263	.263	.304	
.075	-1.228	-1.203	-1.199			.233	.204			
.100	-1.034	-.850	-1.019	-1.059	-1.074	.151	.151	.145	.157	
.150	-.830	-.850	-.850			.086	.068			
.200	-.741	-.741	-.746	-.741	-.741	.063	.068	.051	.033	
.250	-.656	-.656	-.656			.030	.042			
.300	-.591	-.601	-.591	-.591	-.606	.010	.010		-.014	
.350	-.532	-.523	-.532			-.002	-.014			
.400	-.512	-.499	-.501	-.507	-.505	-.031	-.029	-.026	-.051	
.450	-.485	-.477	-.479			-.051	-.051			
.500	-.461	-.457	-.457	-.461	-.461	-.068	-.065	-.068	-.092	
.550	-.444	-.441	-.446			-.085	-.078			
.600	-.430	-.430	-.433	-.426	-.437	-.073		-.080	-.095	
.650	-.429	-.429	-.429			.061	.066			
.700	-.441	-.435	-.435	-.429	-.435	.215	.215	.210	.200	
.750	-.429	-.417	-.417			.295	.293			
.800	-.406	-.406	-.400	-.423	-.411	.346	.349		.334	
.850	-.364	-.358	-.370			.380	.380			
.900	-.287	-.275	-.275	-.281	-.293	.397		.388	.388	
.950			-.133			.375				

CONFIGURATION 1 MACH = .201 ALPHA = 5.06										
WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-2.417	-2.492	-2.402			.652	.658			
.050	-1.604	-1.689	-1.589	-1.559	-1.644	.440	.387	.393	.405	
.075	-1.365	-1.355	-1.370			.310	.293			
.100	-1.175	-.961	-1.161	-1.200	-1.205	.234	.228	.234	.228	
.150	-.956	-.961	-.951			.157	.139			
.200	-.837	-.832	-.827	-.827	-.837	.104	.133	.110	.098	
.250	-.727	-.727	-.722			.084	.086			
.300	-.662	-.657	-.662	-.657	-.672	.052	.052		.021	
.350	-.577	-.572	-.585			.032	.030			
.400	-.552	-.541	-.550	-.543	-.541	.006	.008	.006	-.009	
.450	-.515	-.517	-.515			-.009	-.019			
.500	-.493	-.493	-.484	-.491	-.491	-.038	-.038	-.041	-.060	
.550	-.473	-.468	-.466			-.060	-.056			
.600	-.455	-.451	-.457	-.449	-.460	-.053		-.058	-.078	
.650	-.448	-.448	-.454			.079	.079			
.700	-.454	-.448	-.442	-.442	-.454	.227	.227	.218	.215	
.750	-.424	-.436	-.430			.303	.305			
.800	-.412	-.412	-.412	-.424	-.430	.357	.352		.330	
.850	-.370	-.359	-.365			.391	.383			
.900	-.275	-.269	-.275	-.275	-.293	.403		.396	.398	
.950			-.127			.379				

APPENDIX

TABLE A2.- Continued

CONFIGURATION 1 MACH = .300 ALPHA = .09

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.831	-.817	-.817			-.173	-.134		
.050	-.678	-.713	-.697	-.674	-.738	-.239	-.294	-.288	-.305
.075	-.600	-.607	-.586			-.266	-.266		
.100	-.538	-.464	-.524	-.549	-.528	-.269	-.272	-.250	-.280
.150	-.461	-.464	-.471			-.250	-.274		
.200	-.457	-.438	-.441	-.431	-.424	-.236	-.211	-.233	-.236
.250	-.411	-.401	-.394			-.193	-.186		
.300	-.378	-.381	-.381	-.371	-.376	-.199	-.203		-.233
.350	-.349	-.346	-.347			-.199	-.202		
.400	-.347	-.340	-.343	-.337	-.341	-.200	-.202	-.208	-.216
.450	-.335	-.335	-.335			-.197	-.194		
.500	-.334	-.333	-.333	-.331	-.331	-.197	-.203	-.210	-.223
.550	-.333	-.334	-.334			-.203	-.199		
.600	-.336	-.336	-.337	-.328	-.344	-.170		-.172	-.183
.650	-.343	-.346	-.346			-.008	.001		
.700	-.365	-.360	-.360	-.357	-.362	.156	.154	.155	.148
.750	-.368	-.362	-.360			.238	.236		
.800	-.371	-.365	-.365	-.384	-.376	.287	.284		.277
.850	-.349	-.343	-.343			.321	.315		
.900	-.277	-.277	-.266	-.277	-.299	.349		.346	.337
.950			-.131			.346			

CONFIGURATION 1 MACH = .400 ALPHA = .08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.840	-.825	-.810			-.188	-.170		
.050	-.679	-.723	-.734	-.671	-.789	-.235	-.337	-.329	-.298
.075	-.595	-.625	-.615			-.272	-.295		
.100	-.569	-.480	-.540	-.565	-.574	-.292	-.288	-.275	-.295
.150	-.491	-.480	-.491			-.262	-.287		
.200	-.459	-.445	-.456	-.440	-.436	-.235	-.238	-.249	-.274
.250	-.408	-.415	-.404			-.215	-.210		
.300	-.382	-.395	-.393	-.381	-.401	-.230	-.228		-.259
.350	-.362	-.348	-.362			-.230	-.222		
.400	-.359	-.343	-.352	-.356	-.356	-.213	-.223	-.222	-.232
.450	-.346	-.348	-.350			-.208	-.220		
.500	-.350	-.345	-.349	-.348	-.347	-.217	-.218	-.219	-.247
.550	-.351	-.348	-.348			-.223	-.215		
.600	-.348	-.349	-.357	-.345	-.358	-.183		-.189	-.195
.650	-.363	-.356	-.363			-.010	.000		
.700	-.386	-.373	-.381	-.373	-.381	.157	.155	.152	.147
.750	-.384	-.381	-.379			.239	.237		
.800	-.389	-.382	-.389	-.405	-.395	.292	.286		.278
.850	-.365	-.353	-.363			.325	.320		
.900	-.288	-.280	-.280	-.290	-.311	.351		.349	.343
.950			-.136			.349			

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TABLE A2.- Continued

CONFIGURATION 1 MACH = .499 ALPHA = .10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.821	-.821	-.823			-.204	-.193		
.050	-.710	-.756	-.753	-.679	-.789	-.292	-.416	-.390	-.347
.075	-.638	-.634	-.625			-.306	-.320		
.100	-.579	-.505	-.571	-.587	-.589	-.306	-.330	-.321	-.334
.150	-.504	-.505	-.502			-.297	-.326		
.200	-.486	-.474	-.477	-.472	-.458	-.266	-.257	-.279	-.297
.250	-.433	-.436	-.428			-.238	-.229		
.300	-.404	-.407	-.403	-.404	-.407	-.248	-.248		-.275
.350	-.373	-.368	-.380			-.239	-.242		
.400	-.374	-.363	-.373	-.366	-.375	-.238	-.241	-.248	-.257
.450	-.364	-.359	-.367			-.231	-.237		
.500	-.364	-.362	-.360	-.365	-.359	-.235	-.238	-.239	-.266
.550	-.365	-.361	-.364			-.235	-.228		
.600	-.366	-.364	-.371	-.357	-.369	-.192		-.200	-.209
.650	-.375	-.373	-.380			-.007	.003		
.700	-.402	-.388	-.393	-.386	-.394	.161	.159	.153	.147
.750	-.403	-.398	-.394			.239	.235		
.800	-.405	-.396	-.401	-.420	-.412	.289	.285		.278
.850	-.378	-.363	-.368			.326	.319		
.900	-.290	-.281	-.282	-.293	-.318	.355		.351	.346
.950			-.127			.359			

CONFIGURATION 1 MACH = .600 ALPHA = .09

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.797	-.819	-.820			-.260	-.202		
.050	-.738	-.797	-.825	-.695	-.873	-.356	-.458	-.476	-.429
.075	-.664	-.673	-.680			-.374	-.379		
.100	-.606	-.528	-.600	-.620	-.599	-.378	-.377	-.386	-.418
.150	-.523	-.528	-.525			-.347	-.369		
.200	-.523	-.505	-.513	-.495	-.489	-.323	-.305	-.312	-.339
.250	-.466	-.452	-.458			-.284	-.268		
.300	-.432	-.430	-.435	-.428	-.437	-.289	-.289		-.318
.350	-.399	-.397	-.402			-.285	-.281		
.400	-.399	-.387	-.397	-.400	-.399	-.281	-.279	-.291	-.288
.450	-.389	-.388	-.383			-.272	-.272		
.500	-.390	-.384	-.382	-.390	-.387	-.268	-.268	-.279	-.304
.550	-.388	-.387	-.392			-.267	-.259		
.600	-.393	-.387	-.404	-.379	-.398	-.210		-.217	-.227
.650	-.401	-.399	-.397			-.006	.008		
.700	-.429	-.418	-.417	-.415	-.417	.159	.158	.152	.145
.750	-.430	-.423	-.424			.232	.226		
.800	-.430	-.420	-.429	-.449	-.434	.273	.271		.266
.850	-.392	-.378	-.390			.313	.305		
.900	-.292	-.281	-.283	-.295	-.316	.349		.343	.334
.950			-.109			.359			

APPENDIX

TABLE A2.- Continued

CONFIGURATION 1 MACH = .601 ALPHA = 1.10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-1.203	-1.188	-1.187			-.033	.024		
.050	-1.042	-1.060	-1.098	-.989	-1.219	-.174	-.272	-.261	-.234
.075	-.887	-.846	-.850			-.222	-.225		
.100	-.778	-.642	-.754	-.806	-.796	-.233	-.246	-.238	-.269
.150	-.654	-.642	-.640			-.243	-.265		
.200	-.615	-.597	-.609	-.602	-.589	-.229	-.216	-.238	-.259
.250	-.546	-.531	-.553			-.208	-.182		
.300	-.509	-.503	-.507	-.515	-.509	-.223	-.216		-.258
.350	-.468	-.455	-.455			-.226	-.218		
.400	-.460	-.447	-.444	-.445	-.451	-.229	-.228	-.242	-.242
.450	-.446	-.435	-.427			-.227	-.229		
.500	-.439	-.422	-.424	-.431	-.432	-.235	-.237	-.238	-.265
.550	-.432	-.413	-.424			-.235	-.231		
.600	-.427	-.415	-.428	-.412	-.430	-.187		-.193	-.213
.650	-.436	-.431	-.422			.010	.016		
.700	-.461	-.443	-.439	-.440	-.451	.184	.170	.172	.161
.750	-.457	-.437	-.437			.261	.245		
.800	-.450	-.424	-.436	-.464	-.461	.312	.286		.288
.850	-.404	-.378	-.394			.345	.328		
.900	-.295	-.279	-.283	-.299	-.330	.377		.369	.360
.950			-.114			.374			

CONFIGURATION 1 MACH = .600 ALPHA = 2.10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-1.656	-1.661	-1.692			.192	.197		
.050	-1.329	-1.357	-1.383	-1.385	-1.535	.024	-.066	-.110	-.063
.075	-1.063	-1.084	-1.064			-.079	-.081		
.100	-.922	-.782	-.928	-.952	-.936	-.114	-.127	-.131	-.121
.150	-.762	-.782	-.786			-.134	-.149		
.200	-.722	-.704	-.699	-.678	-.666	-.145	-.148	-.159	-.165
.250	-.637	-.628	-.629			-.138	-.122		
.300	-.580	-.582	-.580	-.572	-.571	-.160	-.158		-.200
.350	-.521	-.523	-.528			-.166	-.170		
.400	-.509	-.499	-.506	-.502	-.495	-.180	-.179	-.187	-.210
.450	-.483	-.483	-.485			-.184	-.185		
.500	-.471	-.472	-.474	-.477	-.469	-.193	-.195	-.200	-.246
.550	-.462	-.466	-.467			-.206	-.199		
.600	-.461	-.455	-.464	-.451	-.460	-.165		-.173	-.207
.650	-.457	-.455	-.458			.021	.031		
.700	-.477	-.465	-.471	-.458	-.467	.193	.195	.188	.173
.750	-.464	-.462	-.461			.278	.274		
.800	-.452	-.447	-.451	-.475	-.461	.328	.324		.310
.850	-.404	-.392	-.399			.367	.363		
.900	-.294	-.284	-.283	-.300	-.316	.396		.385	.382
.950			-.114			.393			

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TABLE A2.- Continued

CONFIGURATION 1 MACH = .601 ALPHA = 3.08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-2.053	-2.282	-2.266			.335	.351		
.050	-1.951	-2.035	-2.002	-2.071	-2.164	.175	.083	.075	.124
.075	-1.277	-1.226	-1.243			.044	.048		
.100	-1.074	-.887	-1.069	-1.104	-1.100	-.001	-.008	-.007	-.006
.150	-.888	-.887	-.903			-.053	-.075		
.200	-.802	-.786	-.799	-.776	-.769	-.079	-.061	-.086	-.099
.250	-.714	-.704	-.700			-.077	-.064		
.300	-.646	-.638	-.640	-.641	-.644	-.103	-.106		-.141
.350	-.574	-.579	-.578			-.113	-.120		
.400	-.559	-.548	-.550	-.553	-.548	-.139	-.136	-.144	-.153
.450	-.530	-.524	-.527			-.147	-.148		
.500	-.513	-.508	-.504	-.513	-.503	-.160	-.161	-.170	-.195
.550	-.498	-.493	-.495			-.175	-.170		
.600	-.485	-.486	-.492	-.481	-.486	-.146		-.154	-.170
.650	-.476	-.478	-.479			.037	.042		
.700	-.491	-.481	-.482	-.479	-.482	.209	.211	.206	.191
.750	-.476	-.472	-.470			.294	.294		
.800	-.460	-.450	-.456	-.476	-.466	.351	.343		.329
.850	-.402	-.386	-.397			.387	.382		
.900	-.285	-.275	-.278	-.291	-.313	.414		.409	.400
.950			-.115			.404			

CONFIGURATION 1 MACH = .649 ALPHA = .08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.791	-.823	-.776			-.282	-.250		
.050	-.778	-.848	-.827	-.732	-.938	-.396	-.541	-.534	-.505
.075	-.689	-.696	-.691			-.415	-.426		
.100	-.625	-.563	-.622	-.645	-.610	-.413	-.412	-.440	-.423
.150	-.555	-.563	-.549			-.378	-.404		
.200	-.545	-.533	-.525	-.507	-.503	-.348	-.332	-.357	-.388
.250	-.479	-.482	-.478			-.302	-.293		
.300	-.447	-.456	-.454	-.445	-.451	-.309	-.324		-.353
.350	-.415	-.406	-.428			-.302	-.312		
.400	-.415	-.401	-.421	-.408	-.407	-.300	-.305	-.314	-.311
.450	-.402	-.407	-.409			-.289	-.300		
.500	-.402	-.406	-.407	-.407	-.402	-.286	-.296	-.299	-.323
.550	-.406	-.406	-.406			-.283	-.279		
.600	-.409	-.409	-.410	-.398	-.412	-.219		-.225	-.236
.650	-.415	-.412	-.422			-.000	.015		
.700	-.442	-.434	-.439	-.427	-.434	.155	.154	.148	.136
.750	-.440	-.442	-.437			.215	.214		
.800	-.437	-.434	-.434	-.463	-.448	.256	.252		.240
.850	-.395	-.387	-.388			.296	.289		
.900	-.283	-.277	-.273	-.288	-.308	.335		.333	.315
.950			-.093			.359			

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TABLE A2.- Continued

CONFIGURATION 1

MACH = .699

ALPHA = .09

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.730	-.752	-.731			-.326	-.267			
.050	-.836	-.902	-.952	-.769	-1.016	-.466	-.641	-.594	-.559	
.075	-.705	-.716	-.716			-.490	-.485			
.100	-.646	-.592	-.645	-.675	-.646	-.457	-.474	-.485	-.497	
.150	-.566	-.592	-.577			-.425	-.479			
.200	-.560	-.553	-.554	-.532	-.542	-.400	-.377	-.394	-.415	
.250	-.501	-.502	-.500			-.344	-.334			
.300	-.471	-.474	-.469	-.464	-.462	-.351	-.358		-.399	
.350	-.434	-.429	-.448			-.348	-.343			
.400	-.434	-.424	-.435	-.429	-.424	-.345	-.337	-.346	-.362	
.450	-.421	-.421	-.425			-.332	-.325			
.500	-.423	-.417	-.423	-.422	-.411	-.325	-.318	-.326	-.355	
.550	-.424	-.417	-.422			-.308	-.292			
.600	-.427	-.422	-.429	-.414	-.423	-.221		-.227	-.230	
.650	-.435	-.433	-.435			.008	.024			
.700	-.462	-.450	-.456	-.443	-.451	.146	.142	.134	.126	
.750	-.460	-.452	-.450			.203	.189			
.800	-.453	-.439	-.443	-.467	-.462	.235	.224		.216	
.850	-.396	-.380	-.385			.277	.260			
.900	-.266	-.257	-.251	-.268	-.289	.319		.309	.292	
.950			-.068			.351				

CONFIGURATION 1

MACH = .750

ALPHA = .08

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.597	-.656	-.633			-.355	-.283			
.050	-.964	-.986	-.975	-.833	-1.174	-.585	-.801	-.738	-.704	
.075	-.716	-.724	-.766			-.544	-.550			
.100	-.673	-.619	-.638	-.724	-.705	-.576	-.535	-.570	-.610	
.150	-.592	-.619	-.595			-.542	-.558			
.200	-.614	-.594	-.589	-.579	-.573	-.468	-.455	-.470	-.505	
.250	-.545	-.539	-.532			-.402	-.400			
.300	-.506	-.495	-.499	-.505	-.502	-.407	-.417		-.460	
.350	-.449	-.464	-.469			-.404	-.398			
.400	-.453	-.451	-.457	-.450	-.441	-.400	-.390	-.416	-.407	
.450	-.445	-.449	-.448			-.381	-.376			
.500	-.448	-.449	-.441	-.439	-.427	-.363	-.358	-.372	-.400	
.550	-.446	-.447	-.449			-.338	-.319			
.600	-.453	-.451	-.459	-.433	-.439	-.220		-.223	-.213	
.650	-.449	-.458	-.457			.015	.031			
.700	-.481	-.476	-.474	-.466	-.472	.125	.120	.114	.100	
.750	-.477	-.475	-.469			.159	.152			
.800	-.458	-.450	-.451	-.474	-.473	.194	.174		.170	
.850	-.380	-.367	-.369			.234	.210			
.900	-.231	-.221	-.215	-.233	-.255	.285		.277	.245	
.950			-.036			.332				

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TABLE A2.- Continued

CONFIGURATION 1 MACH = .801 ALPHA = .07

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.456	-.493	-.480			-.328	-.242			
.050	-.806	-.808	-.790	-.696	-1.011	-.656	-.925	-.847	-.703	
.075	-.832	-.803	-.830			-.575	-.583			
.100	-.664	-.649	-.679	-.715	-.727	-.605	-.599	-.678	-.577	
.150	-.659	-.649	-.654			-.698	-.644			
.200	-.681	-.665	-.668	-.687	-.665	-.707	-.721	-.662	-.698	
.250	-.681	-.661	-.649			-.620	-.548			
.300	-.668	-.670	-.674	-.659	-.732	-.544	-.600		-.673	
.350	-.624	-.652	-.653			-.502	-.509			
.400	-.547	-.539	-.559	-.538	-.422	-.489	-.482	-.489	-.498	
.450	-.460	-.453	-.450			-.450	-.449			
.500	-.460	-.453	-.454	-.457	-.431	-.404	-.402	-.404	-.404	
.550	-.469	-.464	-.472			-.341	-.326			
.600	-.481	-.479	-.489	-.466	-.455	-.188		-.197	-.152	
.650	-.495	-.494	-.493			.016	.029			
.700	-.530	-.517	-.518	-.511	-.494	.087	.085	.075	.063	
.750	-.506	-.496	-.495			.114	.111			
.800	-.442	-.430	-.435	-.463	-.465	.142	.135		.121	
.850	-.328	-.315	-.320			.179	.163			
.900	-.165	-.158	-.153	-.171	-.193	.232		.226	.201	
.950			-.001			.297				

CONFIGURATION 1 MACH = .800 ALPHA = 1.09

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.629	-.693	-.691			-.145	-.125			
.050	-.864	-.866	-.801	-.755	-1.050	-.392	-.582	-.549	-.509	
.075	-.920	-.890	-.942			-.426	-.419			
.100	-.920	-.902	-.923	-.927	-.949	-.472	-.453	-.466	-.488	
.150	-.893	-.902	-.878			-.481	-.507			
.200	-.899	-.869	-.890	-.899	-.885	-.451	-.404	-.447	-.483	
.250	-.875	-.848	-.844			-.393	-.377			
.300	-.856	-.839	-.856	-.872	-.894	-.413	-.413		-.479	
.350	-.826	-.837	-.837			-.408	-.428			
.400	-.844	-.829	-.832	-.829	-.894	-.418	-.418	-.434	-.435	
.450	-.822	-.819	-.824			-.398	-.396			
.500	-.830	-.837	-.824	-.834	-.781	-.373	-.374	-.388	-.406	
.550	-.828	-.843	-.843			-.334	-.319			
.600	-.647	-.769	-.747	-.616	-.398	-.195		-.197	-.178	
.650	-.410	-.425	-.416			.032	.052			
.700	-.401	-.396	-.399	-.389	-.416	.126	.122	.117	.096	
.750	-.394	-.372	-.371			.157	.152			
.800	-.373	-.342	-.353	-.404	-.426	.183	.180		.157	
.850	-.288	-.269	-.272			.224	.219			
.900	-.151	-.140	-.137	-.163	-.196	.281		.271	.240	
.950			-.012			.342				

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TABLE A2.- Continued

CONFIGURATION 1 MACH = .799 ALPHA = 2.09

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.823	-.907	-.925			-.001	.016		
.050	-.875	-.881	-.815	-.891	-1.048	-.201	-.370	-.338	-.291
.075	-.972	-.976	-.989			-.279	-.280		
.100	-1.020	-1.009	-1.026	-.987	-1.026	-.319	-.301	-.313	-.347
.150	-1.002	-1.009	-.990			-.348	-.370		
.200	-1.000	-1.017	-1.015	-.997	-.999	-.333	-.318	-.345	-.370
.250	-.984	-.977	-.990			-.308	-.290		
.300	-.967	-.962	-.965	-.981	-.999	-.346	-.338		-.389
.350	-.938	-.939	-.939			-.361	-.344		
.400	-.963	-.941	-.956	-.949	-1.003	-.380	-.355	-.377	-.378
.450	-.943	-.937	-.944			-.372	-.355		
.500	-.947	-.951	-.952	-.954	-1.009	-.362	-.355	-.361	-.399
.550	-.957	-.960	-.962			-.334	-.320		
.600	-.964	-.966	-.981	-.966	-.830	-.216		-.216	-.215
.650	-.797	-.782	-.788			.031	.049		
.700	-.474	-.466	-.461	-.423	-.402	.145	.139	.131	.116
.750	-.343	-.339	-.337			.190	.173		
.800	-.272	-.268	-.268	-.287	-.337	.224	.207		.186
.850	-.200	-.197	-.196			.265	.251		
.900	-.131	-.126	-.120	-.129	-.166	.319		.310	.276
.950			-.069			.360			

CONFIGURATION 1 MACH = .849 ALPHA = .10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.312	-.369	-.342			-.246	-.196		
.050	-.645	-.658	-.639	-.552	-.839	-.582	-.788	-.715	-.559
.075	-.682	-.654	-.691			-.653	-.700		
.100	-.572	-.573	-.577	-.599	-.648	-.570	-.600	-.642	-.635
.150	-.570	-.573	-.567			-.629	-.579		
.200	-.605	-.587	-.587	-.621	-.592	-.667	-.650	-.618	-.639
.250	-.614	-.589	-.580			-.623	-.610		
.300	-.615	-.610	-.603	-.608	-.656	-.659	-.652		-.687
.350	-.599	-.611	-.608			-.685	-.685		
.400	-.622	-.619	-.625	-.615	-.695	-.715	-.725	-.734	-.737
.450	-.616	-.623	-.629			-.759	-.766		
.500	-.639	-.639	-.636	-.635	-.657	-.816	-.814	-.830	-.723
.550	-.665	-.656	-.649			-.393	-.379		
.600	-.681	-.677	-.682	-.675	-.658	-.263		-.304	-.278
.650	-.689	-.699	-.695			-.259	-.221		
.700	-.764	-.752	-.762	-.749	-.541	-.259	-.234	-.254	-.233
.750	-.491	-.510	-.493			-.248	-.236		
.800	-.232	-.226	-.217	-.242	-.353	-.239	-.235		-.222
.850	-.125	-.127	-.125			-.220	-.202		
.900	-.061	-.067	-.065	-.055	-.144	-.169		-.181	-.165
.950			-.031			-.066			

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TABLE A2.- Concluded

CONFIGURATION 1 MACH = .850 ALPHA = 1.07

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.420	-.466	-.443			-.162	-.125			
.050	-.693	-.693	-.644	-.590	-.871	-.537	-.741	-.687	-.534	
.075	-.749	-.702	-.753			-.405	-.416			
.100	-.737	-.706	-.746	-.740	-.776	-.465	-.472	-.531	-.467	
.150	-.700	-.706	-.696			-.569	-.527			
.200	-.731	-.702	-.729	-.720	-.726	-.587	-.602	-.565	-.576	
.250	-.714	-.696	-.695			-.554	-.529			
.300	-.711	-.699	-.703	-.715	-.748	-.579	-.575		-.634	
.350	-.700	-.702	-.698			-.602	-.612			
.400	-.719	-.712	-.703	-.709	-.774	-.646	-.654	-.675	-.682	
.450	-.717	-.709	-.714			-.708	-.686			
.500	-.731	-.721	-.726	-.738	-.802	-.772	-.753	-.749	-.819	
.550	-.744	-.735	-.737			-.654	-.513			
.600	-.756	-.753	-.760	-.745	-.768	-.296		-.249	-.301	
.650	-.766	-.768	-.763			-.261	-.222			
.700	-.604	-.591	-.565	-.558	-.487	-.248	-.238	-.199	-.266	
.750	-.273	-.272	-.267			-.241	-.228			
.800	-.177	-.175	-.170	-.190	-.226	-.220	-.217		-.260	
.850	-.120	-.119	-.115			-.195	-.171			
.900	-.077	-.082	-.078	-.070	-.083	-.139		-.114	-.194	
.950			-.053			-.013				

CONFIGURATION 1 MACH = .876 ALPHA = .10

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.251	-.289	-.274			-.196	-.149			
.050	-.583	-.582	-.577	-.509	-.771	-.631	-.723	-.673	-.546	
.075	-.614	-.599	-.628			-.603	-.640			
.100	-.546	-.522	-.538	-.544	-.591	-.526	-.576	-.604	-.588	
.150	-.522	-.522	-.521			-.572	-.535			
.200	-.551	-.547	-.540	-.556	-.554	-.614	-.597	-.582	-.596	
.250	-.560	-.547	-.533			-.590	-.592			
.300	-.565	-.563	-.564	-.565	-.608	-.613	-.614		-.648	
.350	-.554	-.564	-.568			-.645	-.645			
.400	-.584	-.565	-.587	-.580	-.641	-.669	-.661	-.686	-.697	
.450	-.584	-.580	-.598			-.717	-.727			
.500	-.605	-.600	-.602	-.594	-.655	-.778	-.777	-.785	-.824	
.550	-.624	-.624	-.613			-.864	-.865			
.600	-.644	-.647	-.645	-.633	-.642	-.431		-.440	-.429	
.650	-.658	-.661	-.662			-.376	-.371			
.700	-.728	-.716	-.725	-.715	-.690	-.377	-.395	-.390	-.396	
.750	-.761	-.753	-.762			-.375	-.342			
.800	-.320	-.307	-.359	-.320	-.575	-.370	-.393		-.397	
.850	-.202	-.201	-.229			-.359	-.376			
.900	-.178	-.177	-.192	-.189	-.222	-.326		-.346	-.358	
.950			-.177			-.258				

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TABLE A3.- AERODYNAMIC PRESSURE-COEFFICIENT DATA FOR CONFIGURATION 1
WITH NATURAL TRANSITION

CONFIGURATION 1						MACH = .196		ALPHA = .05	
WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.870	-.890	-.936			-.062	-.086		
.050	-.722	-.727	-.768	-.727	-.773	-.123	-.184	-.190	-.202
.075	-.630	-.661	-.666			-.196	-.208		
.100	-.574	-.508	-.584	-.590	-.605	-.208	-.202	-.226	-.220
.150	-.483	-.508	-.503			-.202	-.214		
.200	-.462	-.462	-.447	-.467	-.625	-.196	-.178	-.196	-.208
.250	-.411	-.426	-.421			-.174	-.169		
.300	-.386	-.396	-.391	-.411	-.416	-.176	-.171		-.202
.350	-.354	-.354	-.365			-.176	-.169		
.400	-.354	-.336	-.354	-.352	-.359	-.176	-.171	-.164	-.186
.450	-.347	-.341	-.345			-.171	-.176		
.500	-.345	-.345	-.341	-.347	-.345	-.176	-.176	-.181	-.191
.550	-.345	-.347	-.347			-.171	-.169		
.600	-.345	-.350	-.372	-.345	-.354	-.139		-.134	-.141
.650	-.360	-.360	-.360			-.124	-.119		
.700	-.384	-.378	-.372	-.372	-.384	.218	.218	.228	.201
.750	-.384	-.384	-.384			.301	.298		
.800	-.390	-.390	-.390	-.415	-.402	.355	.358		.345
.850	-.378	-.366	-.378			.398	.400		
.900	-.311	-.305	-.305	-.317	-.342	.423		.418	.415
.950			-.190			.405			

CONFIGURATION 1						MACH = .199		ALPHA = 1.06	
WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-1.170	-1.190	-1.145			.064	.118		
.050	-.922	-.916	-.916	-.911	-.911	-.056	-.026	-.056	-.056
.075	-.795	-.749	-.795			-.098	-.104		
.100	-.709	-.577	-.693	-.719	-.714	-.116	-.128	-.110	-.141
.150	-.577	-.577	-.577			-.141	-.153		
.200	-.541	-.516	-.480	-.521	-.709	-.135	-.122	-.128	-.153
.250	-.470	-.470	-.485			-.113	-.113		
.300	-.435	-.440	-.445	-.419	-.435	-.128	-.138		-.153
.350	-.395	-.369	-.395			-.133	-.136		
.400	-.393	-.377	-.382	-.386	-.393	-.136	-.138	-.141	-.153
.450	-.380	-.375	-.375			-.138	-.143		
.500	-.375	-.366	-.368	-.371	-.377	-.143	-.142	-.151	-.163
.550	-.371	-.357	-.364			-.141	-.138		
.600	-.371	-.364	-.366	-.362	-.373	-.103		-.103	-.113
.650	-.376	-.370	-.376			-.086	-.088		
.700	-.401	-.388	-.388	-.394	-.394	.195	.205	.217	.153
.750	-.401	-.388	-.388			.311	.314		
.800	-.407	-.388	-.394	-.413	-.401	.361	.363		.346
.850	-.382	-.370	-.370			.398	.401		
.900	-.310	-.298	-.304	-.310	-.328	.423		.418	.413
.950			-.171			.403			

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TABLE A3.- Continued

CONFIGURATION 1 MACH = .195 ALPHA = 2.02

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C										
.025		-1.426	-1.487	-1.472			.257	.281		
.050		-1.131	-1.157	-1.121	-1.101	-1.116	.101	.083	.083	.071
.075		-.948	-.954	-.964			.022	.010		
.100		-.816	-.664	-.786	-.857	-.872	-.008	-.008	-.014	-.044
.150		-.674	-.664	-.664			-.050	-.062		
.200		-.628	-.598	-.557	-.598	-.781	-.080	-.056	-.068	-.092
.250		-.547	-.542	-.547			-.073	-.049		
.300		-.506	-.506	-.501	-.501	-.517	-.083	-.083		-.104
.350		-.441	-.448	-.450			-.088	-.093		
.400		-.434	-.427	-.427	-.432	-.432	-.096	-.098	-.098	-.113
.450		-.416	-.418	-.414			-.101	-.108		
.500		-.409	-.405	-.405	-.414	-.409	-.111	-.118	-.123	-.136
.550		-.403	-.391	-.398			-.111	-.113		
.600		-.407	-.394	-.396	-.389	-.400	-.068		-.073	-.086
.650		-.395	-.395	-.389			-.058	-.058		
.700		-.413	-.407	-.407	-.401	-.407	.126	.150	.183	.106
.750		-.413	-.407	-.401			.332	.327		
.800		-.401	-.395	-.401	-.419	-.413	.372	.376		.364
.850		-.383	-.371	-.377			.404	.406		
.900		-.304	-.298	-.304	-.304	-.322	.429		.419	.419
.950				-.165			.401			

CONFIGURATION 1 MACH = .195 ALPHA = 3.06

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C										
.025		-1.751	-1.777	-1.766			.413	.431		
.050		-1.411	-1.411	-1.411	-1.487	-1.360	.233	.239	.215	.185
.075		-1.020	-1.030	-1.045			.125	.137		
.100		-.928	-.781	-.908	-.918	-.933	.089	.089	.083	.059
.150		-.760	-.781	-.760			.028	.022		
.200		-.694	-.684	-.623	-.664	-.872	.004	.016	.004	-.020
.250		-.613	-.603	-.613			-.014	-.009		
.300		-.562	-.557	-.562	-.557	-.562	-.031	-.034		-.050
.350		-.493	-.488	-.499			-.053	-.044		
.400		-.475	-.463	-.470	-.470	-.470	-.058	-.056	-.058	-.076
.450		-.448	-.452	-.448			-.068	-.068		
.500		-.434	-.434	-.430	-.434	-.443	-.076	-.081	-.091	-.106
.550		-.425	-.423	-.421			-.083	-.086		
.600		-.416	-.414	-.416	-.405	-.423	-.044		-.046	-.061
.650		-.407	-.407	-.407			-.029	-.026		
.700		-.431	-.413	-.413	-.407	-.419	.083	.081	.126	.078
.750		-.419	-.407	-.401			.342	.339		
.800		-.407	-.407	-.401	-.425	-.413	.381	.381		.367
.850		-.371	-.365	-.371			.409	.409		
.900		-.298	-.286	-.286	-.292	-.310	.429		.424	.421
.950				-.147			.404			

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TABLE A3.- Continued

CONFIGURATION 1 MACH = .199 ALPHA = 4.02

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-2.112	-2.086	-2.137			.563	.569		
.050	-1.375	-1.370	-1.345	-1.294	-1.716	.341	.335	.323	.305
.075	-1.213	-1.197	-1.192			.227	.221		
.100	-1.055	-.862	-1.040	-1.091	-1.091	.191	.167	.161	.155
.150	-.857	-.862	-.862			.095	.083		
.200	-.776	-.755	-.715	-.745	-.964	.065	.077	.065	.047
.250	-.679	-.664	-.674			.039	.046		
.300	-.603	-.613	-.613	-.608	-.618	.014	.019		-.014
.350	-.542	-.538	-.544			-.009	.001		
.400	-.520	-.504	-.511	-.511	-.520	-.026	-.026	-.026	-.046
.450	-.488	-.488	-.484			-.041	-.039		
.500	-.470	-.463	-.468	-.468	-.475	-.049	-.051	-.056	-.071
.550	-.452	-.448	-.459			-.056	-.056		
.600	-.443	-.439	-.448	-.434	-.448	-.021		-.021	-.039
.650	-.437	-.431	-.437			-.004	-.004		
.700	-.444	-.437	-.444	-.437	-.437	.076	.076	.116	.071
.750	-.437	-.431	-.425			.349	.349		
.800	-.419	-.413	-.413	-.437	-.419	.389	.389		.372
.850	-.383	-.377	-.377			.396	.414		
.900	-.298	-.286	-.292	-.298	-.310	.434		.426	.429
.950			-.153			.406			

CONFIGURATION 1 MACH = .198 ALPHA = 5.07

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-2.451	-2.441	-2.472			.679	.703		
.050	-1.591	-1.591	-1.520	-1.474	-1.510	.474	.474	.414	.432
.075	-1.377	-1.372	-1.377			.336	.324		
.100	-1.184	-.955	-1.174	-1.220	-1.204	.245	.239	.245	.245
.150	-.940	-.955	-.965			.161	.161		
.200	-.833	-.843	-.802	-.833	-1.047	.131	.131	.119	.095
.250	-.741	-.731	-.736			.081	.093		
.300	-.670	-.670	-.665	-.670	-.665	.054	.056		.035
.350	-.588	-.583	-.588			.029	.031		
.400	-.561	-.556	-.556	-.559	-.561	.006	.009	.004	-.006
.450	-.525	-.516	-.520			-.016	-.014		
.500	-.502	-.496	-.498	-.502	-.502	-.034	-.026	-.029	-.049
.550	-.478	-.478	-.475			-.036	-.036		
.600	-.462	-.462	-.466	-.460	-.466	-.001		-.001	-.026
.650	-.456	-.456	-.456			.016	.019		
.700	-.462	-.450	-.450	-.450	-.456	.088	.086	.121	.081
.750	-.444	-.438	-.438			.357	.357		
.800	-.420	-.414	-.432	-.438	-.432	.395	.397		.377
.850	-.384	-.371	-.377			.419	.422		
.900	-.293	-.274	-.293	-.293	-.311	.437		.427	.429
.950			-.147			.407			

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TABLE A3.- Continued

CONFIGURATION 1 MACH = .600 ALPHA = .10

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.935	-.969	-.983			-.209	-.220			
.050	-.816	-.858	-.822	-.781	-.808	-.313	-.307	-.332	-.389	
.075	-.732	-.746	-.750			-.339	-.353			
.100	-.647	-.585	-.667	-.669	-.674	-.356	-.347	-.348	-.377	
.150	-.547	-.585	-.573			-.342	-.359			
.200	-.547	-.539	-.470	-.510	-.507	-.300	-.293	-.311	-.330	
.250	-.477	-.482	-.473			-.259	-.260			
.300	-.454	-.452	-.452	-.454	-.443	-.268	-.273		-.313	
.350	-.420	-.410	-.440			-.263	-.269			
.400	-.423	-.414	-.430	-.423	-.418	-.266	-.260	-.286	-.287	
.450	-.414	-.415	-.416			-.257	-.254			
.500	-.412	-.420	-.412	-.419	-.411	-.253	-.254	-.273	-.301	
.550	-.412	-.419	-.425			-.257	-.254			
.600	-.419	-.427	-.433	-.414	-.427	-.226		-.237	-.268	
.650	-.432	-.431	-.438			-.021	-.010			
.700	-.465	-.455	-.459	-.452	-.456	.202	.201	.209	.201	
.750	-.469	-.470	-.469			.303	.313			
.800	-.473	-.472	-.478	-.500	-.484	.375	.384		.388	
.850	-.441	-.437	-.446			.424	.432			
.900	-.342	-.342	-.345	-.360	-.389	.456		.468	.471	
.950			-.171			.449				

CONFIGURATION 1 MACH = .600 ALPHA = 1.09

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.351	-1.305	-1.315			.008	.022			
.050	-1.081	-1.062	-1.074	-1.064	-1.097	-.151	-.139	-.156	-.204	
.075	-.944	-.910	-.926			-.199	-.202			
.100	-.812	-.651	-.800	-.863	-.909	-.225	-.213	-.209	-.233	
.150	-.668	-.651	-.664			-.231	-.242			
.200	-.646	-.618	-.568	-.630	-.622	-.214	-.195	-.209	-.236	
.250	-.572	-.549	-.565			-.194	-.186			
.300	-.522	-.516	-.538	-.543	-.538	-.213	-.208		-.246	
.350	-.488	-.479	-.471			-.218	-.218			
.400	-.481	-.457	-.464	-.473	-.464	-.221	-.223	-.232	-.233	
.450	-.467	-.447	-.446			-.221	-.221			
.500	-.465	-.437	-.444	-.458	-.454	-.232	-.230	-.233	-.255	
.550	-.459	-.434	-.450			-.233	-.234			
.600	-.458	-.438	-.460	-.443	-.464	-.202		-.218	-.218	
.650	-.468	-.454	-.451			.010	-.008			
.700	-.494	-.467	-.474	-.473	-.492	.240	.222	.236	.238	
.750	-.496	-.468	-.477			.344	.330			
.800	-.498	-.465	-.487	-.512	-.518	.412	.406		.408	
.850	-.459	-.427	-.453			.459	.461			
.900	-.354	-.331	-.347	-.361	-.401	.488		.492	.485	
.950			-.174			.460				

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TABLE A3.- Continued

CONFIGURATION 1 MACH = .600 ALPHA = 2.10

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.870	-1.821	-1.852			.225	.220			
.050	-1.547	-1.577	-1.557	-1.589	-1.481	.040	.026	-.008	-.043	
.075	-1.067	-1.015	-1.030			-.049	-.052			
.100	-.946	-.810	-.924	-.978	-.912	-.074	-.090	-.099	-.104	
.150	-.778	-.810	-.801			-.128	-.147			
.200	-.740	-.722	-.673	-.708	-.708	-.134	-.113	-.135	-.169	
.250	-.650	-.644	-.642			-.131	-.107			
.300	-.593	-.605	-.596	-.595	-.595	-.147	-.140		-.192	
.350	-.543	-.535	-.552			-.159	-.154			
.400	-.532	-.515	-.525	-.530	-.515	-.175	-.164	-.180	-.187	
.450	-.510	-.508	-.506			-.187	-.175			
.500	-.498	-.491	-.497	-.497	-.489	-.193	-.183	-.196	-.221	
.550	-.489	-.484	-.490			-.200	-.192			
.600	-.484	-.483	-.493	-.474	-.482	-.164		-.170	-.187	
.650	-.485	-.483	-.487			-.070	.016			
.700	-.512	-.497	-.503	-.490	-.497	.244	.232	.249	.249	
.750	-.503	-.494	-.494			.344	.333			
.800	-.497	-.484	-.502	-.514	-.502	.415	.400		.409	
.850	-.454	-.435	-.449			.459	.449			
.900	-.341	-.327	-.332	-.346	-.373	.490		.487	.484	
.950			-.156			.467				

CONFIGURATION 1 MACH = .601 ALPHA = 3.10

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-2.105	-2.304	-2.317			.354	.361			
.050	-2.134	-2.099	-2.063	-2.052	-2.002	.172	.158	.134	.101	
.075	-1.483	-1.484	-1.328			.064	.040			
.100	-1.028	-.896	-.967	-1.017	-.978	-.005	.003	.009	-.005	
.150	-.882	-.896	-.875			-.064	-.061			
.200	-.827	-.809	-.746	-.781	-.783	-.070	-.052	-.060	-.092	
.250	-.727	-.719	-.711			-.081	-.046			
.300	-.661	-.663	-.648	-.650	-.663	-.094	-.090		-.126	
.350	-.596	-.594	-.605			-.111	-.104			
.400	-.578	-.567	-.573	-.554	-.549	-.130	-.119	-.152	-.151	
.450	-.551	-.549	-.546			-.142	-.132			
.500	-.532	-.529	-.527	-.515	-.513	-.156	-.150	-.173	-.189	
.550	-.518	-.513	-.509			-.173	-.158			
.600	-.507	-.507	-.500	-.486	-.496	-.137		-.147	-.157	
.650	-.507	-.501	-.502			-.116	-.084			
.700	-.523	-.513	-.507	-.490	-.499	.264	.269	.271	.270	
.750	-.505	-.500	-.490			.358	.357			
.800	-.488	-.479	-.475	-.495	-.490	.424	.417		.416	
.850	-.435	-.422	-.420			.467	.463			
.900	-.317	-.307	-.306	-.318	-.345	.497		.489	.487	
.950			-.147			.470				

APPENDIX

TABLE A3.- Continued

CONFIGURATION 1 MACH = .799 ALPHA = .08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.588	-.654	-.638			-.307	-.298		
.050	-.672	-.651	-.700	-.650	-.701	-.470	-.491	-.521	-.554
.075	-.747	-.740	-.745			-.557	-.571		
.100	-.751	-.757	-.744	-.765	-.800	-.604	-.606	-.651	-.646
.150	-.747	-.757	-.750			-.657	-.597		
.200	-.783	-.762	-.749	-.773	-.786	-.570	-.612	-.636	-.690
.250	-.760	-.756	-.752			-.488	-.471		
.300	-.745	-.743	-.731	-.743	-.792	-.433	-.415		-.610
.350	-.706	-.730	-.712			-.425	-.420		
.400	-.725	-.725	-.724	-.716	-.725	-.428	-.419	-.435	-.428
.450	-.703	-.715	-.715			-.396	-.400		
.500	-.695	-.664	-.684	-.626	-.394	-.364	-.379	-.379	-.426
.550	-.593	-.633	-.646			-.328	-.328		
.600	-.602	-.505	-.457	-.422	-.430	-.189		-.265	-.310
.650	-.481	-.479	-.460			.079	.086		
.700	-.536	-.501	-.528	-.547	-.544	.191	.199	.204	.201
.750	-.619	-.621	-.589			.243	.243		
.800	-.723	-.700	-.726	-.746	-.709	.285	.288		.367
.850	-.474	-.480	-.510			.338	.342		
.900	-.223	-.245	-.244	-.276	-.346	.390		.460	.467
.950			-.034			.425			

CONFIGURATION 1 MACH = .799 ALPHA = 1.10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.805	-.834	-.850			-.134	-.113		
.050	-.851	-.844	-.848	-.813	-.838	-.286	-.291	-.326	-.361
.075	-.886	-.865	-.891			-.366	-.376		
.100	-.898	-.902	-.900	-.898	-.928	-.400	-.401	-.396	-.445
.150	-.896	-.902	-.893			-.411	-.435		
.200	-.905	-.915	-.896	-.912	-.931	-.382	-.383	-.388	-.437
.250	-.904	-.897	-.890			-.340	-.322		
.300	-.889	-.901	-.884	-.902	-.941	-.359	-.358		-.418
.350	-.868	-.886	-.876			-.363	-.356		
.400	-.889	-.880	-.881	-.881	-.936	-.357	-.361	-.375	-.386
.450	-.868	-.873	-.867			-.340	-.352		
.500	-.877	-.877	-.874	-.884	-.926	-.328	-.344	-.352	-.410
.550	-.892	-.886	-.890			-.320	-.327		
.600	-.890	-.897	-.905	-.883	-.859	-.225		-.284	-.344
.650	-.899	-.897	-.896			.057	.042		
.700	-.978	-.951	-.962	-.924	-.406	.242	.252	.263	.271
.750	-.901	-.922	-.977			.309	.361		
.800	-.424	-.546	-.547	-.477	-.570	.377	.431		.460
.850	-.281	-.268	-.339			.449	.486		
.900	-.201	-.186	-.208	-.214	-.422	.502		.538	.545
.950			-.070			.498			

APPENDIX

TABLE A3.- Continued

CONFIGURATION 1 MACH = .799 ALPHA = 2.10										
WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.929	-.945	-.959			.028	.036			
.050	-.966	-.983	-.998	-.964	-.979	.143	.157	-.183	-.217	
.075	-1.019	-1.005	-1.019			.229	.239			
.100	-1.029	-.999	-1.011	-1.007	-1.038	.270	.269	-.287	-.297	
.150	-.999	-.999	-1.004			.305	.333			
.200	-1.012	-1.005	-.991	-1.013	-1.033	.295	.280	-.317	-.338	
.250	-.999	-.995	-.990			.274	.260			
.300	-.986	-.990	-.983	-.998	-1.020	.296	.310		-.359	
.350	-.951	-.972	-.964			.310	.311			
.400	-.975	-.965	-.954	-.970	-1.010	.326	.320	-.337	-.339	
.450	-.959	-.957	-.954			.331	.332			
.500	-.964	-.960	-.959	-.959	-1.007	.336	.338	-.334	-.371	
.550	-.967	-.969	-.971			.334	.329			
.600	-.973	-.978	-.983	-.965	-.996	.288		-.280	-.307	
.650	-.959	-.958	-.968			.036	.001			
.700	-.626	-.917	-1.026	-.990	-.937	.252	.274	.273	.273	
.750	-.404	-.473	-.825			.341	.387			
.800	-.370	-.378	-.465	-.435	-.456	.415	.460		.457	
.850	-.363	-.374	-.349			.472	.509			
.900	-.350	-.368	-.261	-.260	-.329	.515		.532	.541	
.950			-.166			.491				

CONFIGURATION 1 MACH = .850 ALPHA = .07										
WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.423	-.479	-.450			-.281	-.282			
.050	-.496	-.492	-.537	-.492	-.530	-.438	-.444	-.480	-.466	
.075	-.576	-.569	-.580			-.532	-.539			
.100	-.596	-.598	-.598	-.615	-.641	-.560	-.596	-.603	-.585	
.150	-.602	-.598	-.610			-.668	-.616			
.200	-.645	-.627	-.599	-.639	-.647	-.684	-.687	-.667	-.658	
.250	-.638	-.629	-.614			-.647	-.628			
.300	-.627	-.639	-.623	-.640	-.687	-.677	-.667		-.704	
.350	-.617	-.618	-.630			-.696	-.700			
.400	-.640	-.630	-.630	-.635	-.713	-.721	-.721	-.752	-.718	
.450	-.631	-.636	-.640			-.760	-.762			
.500	-.653	-.651	-.654	-.658	-.673	-.689	-.810	-.755	-.786	
.550	-.676	-.666	-.669			-.231	-.691			
.600	-.688	-.677	-.690	-.678	-.666	-.190		-.375	-.450	
.650	-.702	-.698	-.704			-.205	-.162			
.700	-.773	-.757	-.773	-.750	-.714	-.222	-.070	-.119	-.171	
.750	-.793	-.810	-.811			-.209	.033			
.800	-.304	-.326	-.538	-.371	-.497	-.146	.141		.093	
.850	-.208	-.201	-.243			-.005	.245			
.900	-.176	-.174	-.157	-.151	-.195	.112		.238	.257	
.950			-.115			.205				

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TABLE A3.- Concluded

CONFIGURATION 1 MACH = .849 ALPHA = 1.10									
WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.520	-.578	-.584			-.166	-.176		
.050	-.611	-.587	-.608	-.585	-.607	-.334	-.342	-.394	-.392
.075	-.676	-.663	-.683			-.425	-.439		
.100	-.694	-.711	-.681	-.695	-.731	-.472	-.493	-.528	-.497
.150	-.696	-.711	-.690			-.566	-.516		
.200	-.728	-.727	-.704	-.739	-.754	-.580	-.595	-.557	-.573
.250	-.729	-.731	-.713			-.555	-.510		
.300	-.723	-.737	-.725	-.739	-.767	-.565	-.562		-.607
.350	-.708	-.727	-.728			-.581	-.601		
.400	-.732	-.730	-.729	-.738	-.791	-.616	-.635	-.667	-.650
.450	-.730	-.724	-.731			-.671	-.673		
.500	-.745	-.737	-.744	-.748	-.820	-.728	-.708	-.663	-.667
.550	-.762	-.751	-.760			-.790	-.702		
.600	-.774	-.769	-.784	-.760	-.779	-.128		-.397	-.353
.650	-.781	-.779	-.786			-.118	-.096		
.700	-.858	-.838	-.856	-.819	-.812	-.106	-.027	.011	-.039
.750	-.648	-.797	-.886			-.047	.060		
.800	-.320	-.329	-.478	-.344	-.801	-.039	.176		.184
.850	-.270	-.276	-.312			.152	.298		
.900	-.257	-.274	-.270	-.277	-.261	.242		.349	.323
.950			-.237			.286			

APPENDIX

TABLE A4.- AERODYNAMIC PRESSURE-COEFFICIENT DATA FOR CONFIGURATION 2
AT $x_t = 0.29c$

CONFIGURATION 2 MACH = .199 ALPHA = .06										
WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.019	-.135	-.001			.459				
.050	-.312	-.302	-.307	-.282	-.302	.172	.148	.142	.118	
.075	-.484	-.484	-.489			-.056	-.038			
.100	-.636	-.636	-.621	-.636	-.626	-.225	-.213	-.219	-.237	
.150	-.611	-.636	-.636				-.327			
.200	-.484	-.479	-.479	-.504	-.504		-.243	-.249	-.267	
.250	-.413	-.408				-.207	-.200			
.300	-.388	-.373	-.398	-.413	-.413	-.217	-.210	-.215	-.255	
.350	-.347	-.350	-.350			-.180	-.185			
.400	-.345	-.334	-.341	-.341	-.345	-.190	-.183	-.190	-.193	
.450	-.334	-.338	-.329			-.190	-.188			
.500	-.334	-.332	-.329	-.332	-.338	-.193	-.190	-.190	-.210	
.550		-.327	-.329			-.193	-.185			
.600	-.334	-.334	-.338	-.332	-.341	-.165		-.173	-.185	
.650	-.345	-.339	-.339			-.006	-.004			
.700	-.363		-.339	-.351	-.357	.164	.160	.162	.155	
.750	-.363	-.351	-.351			.254	.254			
.800	-.369	-.357	-.363	-.382	-.369	.313	.308	.251	.306	
.850	-.351	-.339	-.345			.350	.348			
.900	-.279	-.297	-.273	-.279		.372		.375	.365	
.950			-.146			.360	.348			

CONFIGURATION 2 MACH = .200 ALPHA = 1.06										
WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.174	-.260	-.149			.562				
.050	-.476	-.435	-.440	-.470	-.450	.295	.253	.241	.235	
.075	-.606	-.631	-.641			.063	.057			
.100	-.767	-.726	-.746	-.772	-.752	-.110	-.122	-.122	-.145	
.150	-.701	-.726	-.726				-.253			
.200	-.566	-.566	-.551	-.566	-.571		-.175	-.181	-.199	
.250	-.496	-.481				-.149	-.142			
.300	-.455	-.440	-.476	-.476	-.465	-.167	-.164	-.162	-.205	
.350	-.405	-.414	-.405			-.144	-.147			
.400	-.389	-.387	-.389	-.389	-.396	-.154	-.147	-.154	-.157	
.450	-.371	-.374	-.371			-.154	-.157			
.500	-.367	-.363	-.365	-.367	-.376	-.159	-.164	-.164	-.184	
.550		-.356	-.358			-.162	-.169			
.600	-.358	-.358	-.363	-.354	-.371	-.157		-.154	-.169	
.650	-.367	-.367	-.367			.006	.013			
.700	-.385		-.373	-.373	-.385	.175	.175	.175	.163	
.750	-.379	-.379	-.373			.263	.263			
.800	-.379	-.373	-.379	-.391	-.385	.320	.320	.254	.310	
.850	-.355	-.349	-.355			.359	.359			
.900	-.277	-.277	-.277	-.283		.381		.379	.369	
.950			-.145			.364	.354			

APPENDIX

TABLE A4.- Continued

CONFIGURATION 2 MACH = .200 ALPHA = 2.07

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.310	-.335	-.320			.676				
.050	-.622	-.557	-.587	-.597	-.597	.390	.373	.361	.349	
.075	-.768	-.753	-.803			.165	.165			
.100	-.893	-.828	-.893	-.888	-.934	.010	-.002	-.002	-.032	
.150	-.803	-.828	-.843				-.158			
.200	-.642	-.642	-.632	-.647	-.672		-.110	-.116	-.128	
.250	-.557	-.547				-.095	-.085			
.300	-.511	-.496	-.521	-.526	-.526	-.105	-.113	-.105	-.158	
.350	-.445	-.457	-.454			-.095	-.105			
.400	-.430	-.430	-.430	-.441	-.441	-.113	-.103	-.117	-.130	
.450	-.410	-.412	-.412			-.120	-.120			
.500	-.399	-.392	-.401	-.405	-.401	-.132	-.135	-.135	-.159	
.550		-.385	-.399			-.142	-.140			
.600	-.383	-.383	-.388	-.383	-.390	-.167		-.130	-.150	
.650	-.379	-.391	-.391			.021	.026			
.700	-.397		-.391	-.391	-.397	.183	.188	.185	.175	
.750	-.391	-.385	-.385			.269	.274			
.800	-.385	-.373	-.385	-.415	-.397	.325	.330	.261	.320	
.850	-.349	-.343	-.343			.362	.367			
.900	-.271	-.265		-.277		.382		.389	.379	
.950			-.134			.367	.360			

CONFIGURATION 2 MACH = .200 ALPHA = 3.06

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.971	-1.001	-1.106			.768				
.050	-.680	-.645	-.655	-.680	-.695	.502	.454	.466	.466	
.075	-.866	-.856	-.871			.253	.276			
.100	-1.001	-.926	-.986	-.986	-.996	.081	.093	.081	.075	
.150	-.891	-.926	-.921				-.080			
.200	-.710	-.720	-.700	-.725	-.720		-.044	-.050	-.062	
.250	-.615	-.610				-.036	-.031			
.300	-.555	-.550	-.565	-.585	-.575	-.056	-.058	-.068	-.109	
.350	-.495	-.497	-.504			-.056	-.061			
.400	-.475	-.462	-.473	-.468	-.477	-.075	-.073	-.078	-.090	
.450	-.444	-.442	-.446			-.080	-.088			
.500	-.429	-.424	-.431	-.433	-.435	-.100	-.102	-.102	-.122	
.550		-.413	-.417			-.120	-.112			
.600	-.404	-.406	-.411	-.404	-.413	-.139		-.110	-.122	
.650	-.402	-.402	-.402			.040	.045			
.700	-.408		-.408	-.402	-.414	.199	.202	.194	.187	
.750	-.396	-.396	-.396			.283	.283			
.800	-.390	-.384	-.390	-.402	-.390	.339	.336	.270	.327	
.850	-.348	-.348	-.348			.373	.373			
.900	-.271	-.265		-.271		.390		.390	.385	
.950			-.133			.376	.363			

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TABLE A4.- Continued

CONFIGURATION 2 MACH = .200 ALPHA = 4.07

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-1.781	-1.776	-1.816			.820			
.050	-.875	-.880	-.945	-.814	-.865	.570	.546	.552	.552
.075	-.920	-.900	-.920			.373	.367		
.100	-1.076	-1.016	-1.066	-1.086	-1.116	.183	.201	.171	.177
.150	-.980	-1.016	-1.011				.010		
.200	-.799	-.794	-.779	-.804	-.794		.016	.010	.010
.250	-.683	-.668				.011	.023		
.300	-.623	-.598	-.623	-.633	-.633	-.014	-.009	-.019	-.062
.350	-.544	-.549	-.553			-.016	-.016		
.400	-.517	-.511	-.513	-.513	-.517	-.031	-.036	-.046	-.056
.450	-.482	-.480	-.475			-.051	-.058		
.500	-.468	-.457	-.455	-.464	-.466	-.073	-.073	-.081	-.100
.550		-.437	-.442			-.095	-.088		
.600	-.430	-.426	-.430	-.424	-.435	-.115		-.090	-.103
.650	-.422	-.428	-.416			.053	.055		
.700	-.422		-.416	-.416	-.422	.210	.205	.208	.198
.750	-.416	-.404	-.398			.289	.292		
.800	-.392	-.386	-.392	-.398	-.404	.348	.343	.272	.336
.850	-.374	-.344	-.344			.383	.378		
.900	-.266	-.260	-.254	-.266		.397		.397	.390
.950			-.134			.378	.363		

CONFIGURATION 2 MACH = .300 ALPHA = .08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	.002	-.122	.020			.424			
.050	-.350	-.286	-.290	-.300	-.290	.154	.154	.126	.088
.075	-.493	-.505	-.528			-.087	-.065		
.100	-.655	-.629	-.662	-.623	-.641	-.249	-.236	-.241	-.266
.150	-.613	-.629	-.639				.370		
.200	-.475	-.491	-.489	-.486	-.507		-.269	-.282	-.282
.250	-.419	-.422				-.222	-.216		
.300	-.408	-.383	-.431	-.410	-.396	-.190	-.215	-.221	-.274
.350	-.360	-.359	-.360			-.192	-.190		
.400	-.358	-.350	-.354	-.364	-.357	-.200	-.189	-.208	-.206
.450	-.346	-.346	-.341			-.197	-.192		
.500	-.345	-.339	-.336	-.348	-.346	-.204	-.203	-.201	-.222
.550		-.340	-.339			-.204	-.203		
.600	-.339	-.343	-.351	-.335	-.358	-.172		-.182	-.191
.650	-.359	-.357	-.359			-.008	-.007		
.700	-.381		-.368	-.373	-.379	.166	.168	.166	.157
.750	-.381	-.376	-.373			.258	.260		
.800	-.390	-.379	-.384	-.403	-.398	.316	.317	.268	.308
.850	-.365	-.354	-.362			.354	.354		
.900	-.293	-.285	-.288	-.302		.379		.381	.372
.950			-.156			.369	.359		

APPENDIX

TABLE A4.- Continued

CONFIGURATION 2 MACH = .401 ALPHA = .09

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.006	-.076	.034			.443				
.050	-.364	-.270	-.298	-.293	-.274	.156	.135	.130	.082	
.075	-.507	-.529	-.518			-.089	-.093			
.100	-.684	-.652	-.670	-.665	-.657	-.271	-.273	-.265	-.286	
.150	-.639	-.652	-.685				-.409			
.200	-.518	-.513	-.509	-.530	-.524		-.310	-.299	-.323	
.250	-.450	-.444				-.231	-.206			
.300	-.431	-.408	-.448	-.450	-.415	-.201	-.224	-.231	-.307	
.350	-.380	-.382	-.372			-.217	-.212			
.400	-.376	-.364	-.370	-.376	-.368	-.220	-.219	-.225	-.233	
.450	-.366	-.358	-.359			-.216	-.217			
.500	-.364	-.354	-.359	-.367	-.359	-.215	-.215	-.219	-.244	
.550		-.361	-.362			-.222	-.216			
.600	-.363	-.362	-.370	-.359	-.371	-.193		-.197	-.207	
.650	-.373	-.373	-.375			-.015	-.005			
.700	-.399		-.388	-.386	-.394	.168	.169	.164	.157	
.750	-.401	-.392	-.394			.262	.260			
.800	-.407	-.396	-.401	-.420	-.413	.318	.319	.285	.312	
.850	-.381	-.368	-.380			.359	.359			
.900	-.304	-.292	-.302	-.308		.385		.388	.380	
.950			-.156			.376	.364			

CONFIGURATION 2 MACH = .500 ALPHA = .08

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.047	-.053	.066			.455				
.050	-.310	-.261	-.266	-.267	-.257	.145	.118	.119	.082	
.075	-.509	-.499	-.513			-.118	-.106			
.100	-.711	-.705	-.691	-.691	-.678	-.309	-.300	-.290	-.314	
.150	-.658	-.705	-.700				-.454			
.200	-.522	-.531	-.526	-.527	-.556		-.346	-.327	-.358	
.250	-.468	-.462				-.227	-.531			
.300	-.448	-.430	-.474	-.468	-.443	-.236	-.247	-.232	-.281	
.350	-.393	-.400	-.399			-.238	-.239			
.400	-.390	-.375	-.385	-.391	-.391	-.238	-.241	-.249	-.249	
.450	-.378	-.375	-.378			-.239	-.238			
.500	-.379	-.372	-.377	-.382	-.379	-.239	-.242	-.241	-.262	
.550		-.370	-.379			-.235	-.233			
.600	-.375	-.376	-.387	-.370	-.388	-.200		-.215	-.215	
.650	-.388	-.382	-.393			-.012	-.005			
.700	-.414		-.411	-.404	-.411	.172	.174	.169	.161	
.750	-.418	-.412	-.412			.264	.270			
.800	-.422	-.410	-.421	-.441	-.427	.321	.323	.299	.315	
.850	-.397	-.381	-.395			.362	.366			
.900	-.308	-.299	-.308	-.318		.390		.399	.387	
.950			-.152			.384	.369			

APPENDIX

TABLE A4.- Continued

CONFIGURATION 2 MACH = .601 ALPHA = .09

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	.102	.001	.134			.447			
.050	-.274	-.234	-.216	-.233	-.235	.153	.117	.106	.075
.075	-.506	-.516	-.500			-.135	-.123		
.100	-.761	-.761	-.729	-.749	-.735	-.349	-.342	-.341	-.374
.150	-.742	-.761	-.778				-.506		
.200	-.566	-.571	-.562	-.585	-.607		-.435	-.386	-.395
.250	-.485	-.487				-.262	-.257		
.300	-.482	-.464	-.516	-.511	-.482	-.281	-.295	-.263	-.333
.350	-.423	-.418	-.421			-.280	-.281		
.400	-.419	-.403	-.416	-.413	-.410	-.281	-.280	-.278	-.285
.450	-.406	-.402	-.403			-.270	-.272		
.500	-.410	-.402	-.403	-.406	-.402	-.268	-.271	-.277	-.303
.550		-.404	-.406			-.273	-.263		
.600	-.408	-.408	-.414	-.404	-.417	-.220		-.232	-.240
.650	-.417	-.415	-.420			-.009	-.002		
.700	-.445		-.440	-.435	-.443	.177	.176	.173	.161
.750	-.450	-.444	-.444			.262	.264		
.800	-.456	-.443	-.449	-.478	-.467	.316	.319	.314	.310
.850	-.422	-.405	-.417			.357	.360		
.900	-.319	-.306	-.314	-.331		.389		.402	.384
.950			-.139			.390	.375		

CONFIGURATION 2 MACH = .600 ALPHA = 1.09

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.071	-.189	-.032			.591			
.050	-.483	-.401	-.403	-.436	-.417	.287	.258	.245	.237
.075	-.692	-.681	-.709			.008	.013		
.100	-.945	-.924	-.912	-.923	-.909	-.197	-.208	-.209	-.238
.150	-.900	-.924	-.937				-.387		
.200	-.686	-.672	-.677	-.686	-.684		-.300	-.304	-.315
.250	-.586	-.575				-.199	-.192		
.300	-.569	-.538	-.582	-.590	-.562	-.209	-.227	-.193	-.301
.350	-.484	-.490	-.490			-.222	-.216		
.400	-.477	-.470	-.467	-.475	-.475	-.224	-.223	-.235	-.237
.450	-.456	-.458	-.450			-.222	-.231		
.500	-.448	-.442	-.442	-.447	-.440	-.231	-.234	-.237	-.264
.550		-.435	-.438			-.240	-.235		
.600	-.443	-.435	-.445	-.438	-.444	-.198		-.212	-.222
.650	-.446	-.449	-.445			.003	.010		
.700	-.471		-.460	-.461	-.466	.192	.191	.188	.175
.750	-.470	-.461	-.456			.284	.283		
.800	-.465	-.451	-.457	-.480	-.474	.340	.340	.330	.327
.850	-.423	-.407	-.419			.381	.382		
.900	-.317	-.304	-.312	-.326		.409		.418	.405
.950			-.137			.403	.390		

APPENDIX

TABLE A4.- Continued

CONFIGURATION 2 MACH = .600 ALPHA = 2.10

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C										
.025		-.203	-.271	-.199			.708			
.050		-.652	-.591	-.604	-.618	-.597	.407	.377	.368	.340
.075		-.899	-.901	-.902			.132	.156		
.100		-1.147	-1.098	-1.162	-1.138	-1.152	-.067	-.069	-.087	-.101
.150		-1.052	-1.098	-1.101				-.264		
.200		-.788	-.793	-.780	-.802	-.809		-.180	-.194	-.216
.250		-.672	-.663				-.153	-.123		
.300		-.644	-.624	-.654	-.664	-.640	-.144	-.160	-.134	-.246
.350		-.546	-.553	-.560			-.161	-.157		
.400		-.529	-.527	-.531	-.527	-.523	-.180	-.174	-.178	-.187
.450		-.504	-.508	-.501			-.179	-.184		
.500		-.486	-.490	-.490	-.498	-.484	-.193	-.196	-.191	-.224
.550			-.480	-.479			-.208	-.199		
.600		-.470	-.472	-.477	-.468	-.473	-.177		-.183	-.199
.650		-.472	-.475	-.473			.016	.025		
.700		-.489		-.483	-.475	-.482	.204	.206	.201	.191
.750		-.482	-.479	-.473			.297	.299		
.800		-.469	-.463	-.463	-.485	-.481	.356	.356	.330	.345
.850		-.418	-.411	-.416			.398	.401		
.900		-.307	-.300	-.304	-.316		.425	-.015	.425	.416
.950				-.133			.416	.398		

CONFIGURATION 2 MACH = .599 ALPHA = 3.02

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C										
.025		-.682	-.853	-.723			.784			
.050		-.787	-.709	-.723	-.744	-.766	.517	.486	.477	.465
.075		-1.083	-1.056	-1.069			.254	.263		
.100		-1.380	-1.232	-1.364	-1.357	-1.383	.050	.056	.061	.026
.150		-1.196	-1.232	-1.242				-.164		
.200		-.885	-.892	-.883	-.901	-.912		-.101	-.102	-.127
.250		-.751	-.751				-.095	-.079		
.300		-.699	-.692	-.725	-.725	-.712	-.075	-.107	-.078	-.187
.350		-.603	-.609	-.609			-.109	-.108		
.400		-.581	-.564	-.578	-.577	-.575	-.134	-.129	-.140	-.150
.450		-.549	-.544	-.547			-.141	-.146		
.500		-.527	-.522	-.527	-.530	-.521	-.156	-.164	-.164	-.194
.550			-.508	-.510			-.177	-.173		
.600		-.494	-.499	-.501	-.489	-.498	-.153		-.160	-.173
.650		-.480	-.485	-.496			.034	.038		
.700		-.504		-.499	-.489	-.493	.215	.217	.212	.204
.750		-.489	-.483	-.482			.308	.312		
.800		-.470	-.463	-.463	-.485	-.478	.369	.369	.339	.357
.850		-.410	-.402	-.410			.408	.411		
.900		-.294	-.288	-.292	-.300		.436		.437	.427
.950				-.129			.420	.406		

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TABLE A4.- Continued

CONFIGURATION 2 MACH = .651 ALPHA = .02

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	.163	.037	.176			.447			
.050	-.254	-.189	-.167	-.199	-.179	.129	.116	.096	.062
.075	-.485	-.509	-.486			-.161	-.146		
.100	-.780	-.819	-.758	-.758	-.737	-.374	-.383	-.364	-.411
.150	-.803	-.819	-.819				-.570		
.200	-.592	-.593	-.598	-.623	-.640		-.500	-.426	-.439
.250	-.505	-.502				-.294	-.275		
.300	-.505	-.488	-.539	-.542	-.505	-.310	-.323	-.303	-.370
.350	-.436	-.435	-.441			-.309	-.311		
.400	-.438	-.416	-.435	-.428	-.430	-.303	-.306	-.314	-.313
.450	-.423	-.419	-.421			-.296	-.296		
.500	-.425	-.415	-.420	-.427	-.417	-.289	-.293	-.303	-.325
.550		-.423	-.423			-.285	-.281		
.600	-.427	-.427	-.430	-.418	-.429	-.224		-.239	-.250
.650	-.437	-.429	-.440			-.000	.003		
.700	-.470		-.459	-.454	-.460	.179	.176	.173	.164
.750	-.472	-.461	-.462			.258	.258		
.800	-.474	-.461	-.465	-.497	-.483	.306	.307	.306	.300
.850	-.432	-.416	-.425			.350	.350		
.900	-.317	-.305	-.312	-.328		.384		.396	.376
.950			-.124			.391	.373		

CONFIGURATION 2 MACH = .701 ALPHA = .09

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	.216	.106	.239			.456			
.050	-.176	-.131	-.131	-.122	-.122	.143	.123	.084	.077
.075	-.438	-.441	-.447			-.158	-.160		
.100	-.781	-1.008	-.757	-.734	-.713	-.407	-.406	-.423	-.428
.150	-.965	-1.008	-.957				-.643		
.200	-.592	-.605	-.686	-.681	-.681		-.583	-.495	-.467
.250	-.510	-.510				-.321	-.306		
.300	-.538	-.516	-.579	-.580	-.535	-.358	-.367	-.345	-.422
.350	-.457	-.452	-.465			-.350	-.346		
.400	-.456	-.437	-.453	-.455	-.446	-.334	-.339	-.348	-.346
.450	-.442	-.442	-.440			-.325	-.330		
.500	-.446	-.442	-.439	-.453	-.433	-.317	-.325	-.323	-.357
.550		-.442	-.444			-.310	-.308		
.600	-.444	-.447	-.458	-.447	-.447	-.234		-.244	-.254
.650	-.460	-.454	-.460			.006	.014		
.700	-.492		-.482	-.476	-.477	.174	.174	.170	.154
.750	-.494	-.489	-.484			.242	.244		
.800	-.489	-.479	-.486	-.514	-.497	.291	.288	.290	.278
.850	-.433	-.419	-.431			.332	.335		
.900	-.299	-.291	-.297	-.312		.372		.382	.363
.950			-.100			.386	.372		

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TABLE A4.- Continued

CONFIGURATION 2 MACH = .750 ALPHA = .09

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	.299	.175	.317			.449			
.050	-.089	-.046	-.033	-.043	-.037	.151	.123	.087	.071
.075	-.337	-.347	-.355			-.165	-.149		
.100	-.666	-1.044	-.639	-.635	-.616	-.420	-.432	-.428	-.446
.150	-1.016	-1.044	-1.015				-.817		
.200	-.986	-1.049	-1.014	-.962	-.961		-.737	-.667	-.529
.250	-.554	-.526				-.331	-.310		
.300	-.498	-.478	-.523	-.571	-.468	-.416	-.428	-.420	-.514
.350	-.441	-.448	-.447			-.405	-.406		
.400	-.458	-.445	-.457	-.458	-.445	-.391	-.389	-.408	-.405
.450	-.454	-.458	-.452			-.373	-.380		
.500	-.461	-.460	-.459	-.464	-.442	-.360	-.369	-.372	-.392
.550		-.460	-.468			-.332	-.331		
.600	-.464	-.469	-.484	-.463	-.456	-.231		-.238	-.240
.650	-.480	-.480	-.483			.019	.025		
.700	-.518		-.508	-.502	-.490	.157	.159	.153	.127
.750	-.516	-.514	-.510			.214	.219		
.800	-.501	-.489	-.500	-.522	-.504	.251	.251	.241	.224
.850	-.419	-.407	-.422			.298	.295		
.900	-.265	-.258	-.263	-.272		.340		.339	.312
.950			-.063			.362	.353		

CONFIGURATION 2 MACH = .800 ALPHA = .07

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	.393	.288	.409			.466			
.050	.014	.067	.077	.052	.067	.173	.151	.115	.116
.075	-.225	-.224	-.226			-.116	-.109		
.100	-.526	-.885	-.503	-.507	-.478	-.372	-.377	-.357	-.368
.150	-.871	-.885	-.877				-.763		
.200	-.921	-.940	-.932	-.928	-.942		-.918	-.898	-.895
.250	-.900	-.890				-.830	-.775		
.300	-.915	-.885	-.898	-.883	-.907	-.821	-.838	-.815	-.859
.350	-.803	-.805	-.796			-.650	-.684		
.400	-.813	-.795	-.803	-.791	-.823	-.362	-.346	-.358	-.565
.450	-.784	-.775	-.786			-.343	-.361		
.500	-.756	-.746	-.714	-.748	-.513	-.367	-.376	-.371	-.366
.550		-.540	-.420			-.343	-.340		
.600	-.406	-.392	-.400	-.387	-.375	-.225		-.244	-.175
.650	-.414	-.418	-.419			.037	.049		
.700	-.476		-.488	-.480	-.479	.177	.175	.169	.129
.750	-.511	-.523	-.533			.230	.227		
.800	-.506	-.496	-.524	-.535	-.544	.267	.263	.282	.221
.850	-.389	-.381	-.396			.313	.310		
.900	-.219	-.213	-.222	-.236		.364		.375	.315
.950			-.036			.395	.375		

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TABLE A4.- Continued

CONFIGURATION 2 MACH = .801 ALPHA = 1.10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	.295	.191	.313			.552			
.050	-.073	-.013	-.011	-.033	-.029	.263	.242	.208	.207
.075	-.302	-.303	-.310			-.037	-.021		
.100	-.599	-.956	-.566	-.576	-.554	-.291	-.290	-.283	-.292
.150	-.943	-.956	-.939				-.677		
.200	-1.002	-1.010	-1.014	-1.003	-1.023		-.787	-.729	-.799
.250	-.982	-.970				-.394	-.385		
.300	-.974	-.932	-.957	-.960	-.999	-.382	-.390	-.375	-.556
.350	-.890	-.900	-.907			-.396	-.391		
.400	-.897	-.896	-.896	-.887	-.903	-.412	-.403	-.421	-.407
.450	-.883	-.886	-.886			-.396	-.394		
.500	-.888	-.884	-.891	-.877	-.905	-.379	-.384	-.393	-.420
.550		-.889	-.891			-.353	-.345		
.600	-.899	-.905	-.905	-.879	-.855	-.231		-.239	-.204
.650	-.895	-.901	-.892			.038	.047		
.700	-.652		-.695	-.583	-.405	.170	.167	.164	.128
.750	-.392	-.387	-.387			.218	.221		
.800	-.318	-.309	-.314	-.343	-.425	.255	.267	.264	.231
.850	-.235	-.225	-.243			.313	.310		
.900	-.132	-.129	-.139	-.153		.359		.363	.336
.950			-.032			.379	.364		

CONFIGURATION 2 MACH = .799 ALPHA = 2.11

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	.216	.101	.225			.627			
.050	-.139	-.096	-.090	-.117	-.114	.354	.320	.289	.297
.075	-.368	-.373	-.381			.048	.065		
.100	-.663	-1.021	-.627	-.641	-.614	-.201	-.206	-.199	-.223
.150	-1.002	-1.021	-.997				-.570		
.200	-1.062	-1.071	-1.063	-1.064	-1.088		-.584	-.470	-.465
.250	-1.031	-1.002				-.266	-.231		
.300	-1.002	-.983	-1.027	-1.008	-1.025	-.379	-.382	-.362	-.534
.350	-.938	-.959	-.977			-.373	-.370		
.400	-.959	-.962	-.971	-.967	-.989	-.387	-.380	-.390	-.388
.450	-.954	-.956	-.964			-.384	-.386		
.500	-.967	-.960	-.973	-.963	-.990	-.390	-.387	-.395	-.420
.550		-.975	-.974			-.374	-.362		
.600	-.982	-.983	-.988	-.964	-1.008	-.249		-.271	-.250
.650	-.968	-.959	-.964			.027	.034		
.700	-.614		-.550	-.584	-.519	.159	.163	.161	.140
.750	-.380	-.374	-.367			.211	.220		
.800	-.292	-.293	-.290	-.306	-.327	.256	.263	.276	.251
.850	-.228	-.228	-.229			.307	.307		
.900	-.175	-.178	-.183	-.165		.353		.374	.350
.950			-.143			.366	.353		

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TABLE A4.- Continued

CONFIGURATION 2 MACH = .852 ALPHA = .02

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	.491	.375	.488			.496				
.050	.124	.177	.183	.158	.169	.230	.207	.175	.152	
.075	-.098	-.100	-.107			-.047	-.031			
.100	-.367	-.723	-.354	-.355	-.343	-.295	-.289	-.281	-.272	
.150	-.711	-.723	-.707				-.659			
.200	-.769	-.773	-.772	-.761	-.782		-.808	-.791	-.796	
.250	-.755	-.741				-.744	-.735			
.300	-.776	-.748	-.754	-.743	-.769	-.758	-.766	-.774	-.808	
.350	-.684	-.698	-.691			-.753	-.770			
.400	-.701	-.686	-.684	-.687	-.713	-.780	-.766	-.777	-.757	
.450	-.686	-.687	-.685			-.810	-.802			
.500	-.689	-.687	-.694	-.699	-.743	-.853	-.861	-.853	-.811	
.550		-.701	-.706			-.402	-.448			
.600	-.714	-.712	-.714	-.708	-.740	-.281		-.306	-.300	
.650	-.723	-.715	-.713			-.274	-.277			
.700	-.790		-.787	-.758	-.729	-.281	-.295	-.281	-.284	
.750	-.596	-.619	-.601			-.285	-.293			
.800	-.254	-.244	-.260	-.240	-.346	-.287	-.286	-.275	-.289	
.850	-.142	-.144	-.148			-.264	-.257			
.900	-.086	-.090	-.102	-.085		-.212		-.196	-.210	
.950			-.071			-.130	-.101			

CONFIGURATION 2 MACH = .850 ALPHA = 1.02

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	.414	.303	.422			.558				
.050	.061	.106	.111	.102	.103	.282	.259	.231	.234	
.075	-.160	-.161	-.161			.001	.011			
.100	-.447	-.781	-.413	-.408	-.395	-.242	-.241	-.224	-.231	
.150	-.771	-.781	-.766				-.607			
.200	-.831	-.842	-.837	-.821	-.847		-.763	-.733	-.749	
.250	-.822	-.815				-.699	-.686			
.300	-.840	-.811	-.818	-.817	-.837	-.700	-.724	-.712	-.734	
.350	-.749	-.763	-.761			-.697	-.708			
.400	-.766	-.757	-.757	-.744	-.767	-.729	-.724	-.728	-.713	
.450	-.750	-.752	-.749			-.750	-.756			
.500	-.758	-.760	-.759	-.760	-.787	-.799	-.813	-.809	-.829	
.550		-.765	-.771			-.503	-.824			
.600	-.776	-.779	-.789	-.774	-.794	-.270		-.309	-.329	
.650	-.783	-.786	-.785			-.238	-.276			
.700	-.735		-.774	-.623	-.720	-.229	-.282	-.283	-.287	
.750	-.297	-.294	-.300			-.225	-.278			
.800	-.181	-.183	-.188	-.199	-.246	-.213	-.271	-.272	-.269	
.850	-.123	-.124	-.135			-.198	-.228			
.900	-.082	-.084	-.094	-.097		-.134		-.189	-.172	
.950			-.073			-.020	-.021			

APPENDIX

TABLE A4.- Concluded

CONFIGURATION 2 MACH = .875 ALPHA = .10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	.511	.415	.513			.530			
.050	.171	.210	.212	.194	.195	.256	.245	.201	.205
.075	-.056	-.059	-.064			-.012	-.001		
.100	-.336	-.658	-.308	-.312	-.291	-.248	-.237	-.230	-.236
.150	-.650	-.658	-.647				-.595		
.200	-.711	-.715	-.719	-.701	-.721		-.746	-.721	-.736
.250	-.698	-.687				-.694	-.692		
.300	-.720	-.694	-.732	-.682	-.702	-.706	-.717	-.734	-.760
.350	-.635	-.644	-.641			-.708	-.719		
.400	-.651	-.641	-.641	-.640	-.671	-.729	-.725	-.730	-.716
.450	-.645	-.637	-.643			-.762	-.758		
.500	-.652	-.645	-.653	-.657	-.696	-.804	-.815	-.807	-.817
.550		-.660	-.660			-.868	-.897		
.600	-.674	-.673	-.679	-.677	-.705	-.511		-.496	-.465
.650	-.687	-.681	-.678			-.431	-.410		
.700	-.754		-.745	-.735	-.749	-.434	-.434	-.424	-.416
.750	-.788	-.781	-.783			-.430	-.434		
.800	-.395	-.361	-.345	-.348	-.419	-.419	-.427	-.422	-.413
.850	-.240	-.235	-.223			-.392	-.408		
.900	-.207	-.203	-.200	-.204		-.353		-.369	-.358
.950			-.188			-.284	-.277		

APPENDIX

TABLE A5.- AERODYNAMIC PRESSURE-COEFFICIENT DATA FOR CONFIGURATION 3
AT $x_t = 0.29c$

CONFIGURATION 3 MACH = .203 ALPHA = .07

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.416	-.218	-.097			.616			
.050	-.416	-.441	-.373	-.290	-.295	.283	.129	.106	.123
.075	-.615	-.625	-.566			-.072	-.107		
.100	-.750	-.683	-.687	-.625	-.625	-.390	-.343	-.280	-.274
.150	-.644	-.683	-.683				-.540		
.200	-.518	-.518	-.508	-.499	-.499	-.713	-.465	-.320	-.297
.250	-.460	-.431	-.431			-.554	-.409		
.300	-.436	-.421	-.421	-.426	-.465	-.504	-.436	-.392	-.280
.350	-.362	-.379	-.368			-.454	-.385		
.400	-.353	-.359	-.351	-.334	-.340	-.428	-.383	-.240	-.216
.450	-.355	-.355	-.334			-.404	-.376		
.500	-.344	-.336	-.327	-.338	-.334	-.385	-.361	-.259	-.233
.550	-.338	-.334	-.325			-.345	-.345		
.600	-.336	-.332	-.332	-.327	-.340	-.240		-.231	-.202
.650	-.320	-.338	-.332			-.124	-.117		
.700	-.384	-.349	-.343	-.355	-.361	-.091	.054	.125	.142
.750	-.343	-.338	-.343			-.026	.151		
.800	-.332	-.332	-.343	-.384	-.378	.061	.218	.222	.293
.850	-.297	-.297	-.315			.142	.274		
.900	-.211	-.216	-.245	-.286	-.834	.218		.350	.360
.950			-.113			.298			

NACELLE							PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	
X/C	CP						X/C	CP
-.2265	-.260	-.214		-.309	-.340	-.689	.0125	.523
-.2115	-.179	-.237		-.286	-.298	-.325	.0625	-.134
-.1765	-.083	-.122	-.191	-.286	-.298	-.321	.1125	
-.1165	.027	.004	-.106	-.267	-.233	-.340	.1625	-.750
-.0515	.201	.076	-.080	-.168	-.194	-.202	.2375	-.600
.0235		.004		-.175		-.168	.3225	-.533
.0985		-.260	-.191	-.171	-.160	-.145	.3875	
.1735		-.470		-.171		-.145	.4625	-.528
.2485		-.371	-.293			-.108	.5375	-.346
.3235		-.400		-.170		-.120	.6125	-.294
.3985		-.371	-.293	-.170			.6875	-.210
.4735			-.257	-.155	-.112	-.105	.7625	
.5485		-.262		-.135	-.097	-.100	.8375	.084
.6235		-.193				-.077	.9125	.062
.6985		-.108	-.088	-.068	-.057	-.062	.9625	.165
.7735		-.010	-.017	-.028	-.032			.129
.8485		.070	.038	.014	.007	-.000		
.9235		.107	.087	.061		.045		
.9735		.135	.128	.109	.098	.098	.103	
.9935		.144			.122	.117		

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TABLE A5.- Continued

CONFIGURATION 3

MACH = .199

ALPHA = 1.06

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.921	-.563	-.241			.690			
.050	-.528	-.533	-.518	-.422	-.452	.380	.249	.201	.225
.075	-.689	-.714	-.699			.034	-.014		
.100	-.825	-.780	-.815	-.750	-.750	-.284	-.236	-.194	-.134
.150	-.704	-.780	-.780				-.440		
.200	-.583	-.604	-.599	-.583	-.553	-.633	-.380	-.266	-.230
.250	-.508	-.508	-.513			-.477	-.323		
.300	-.472	-.483	-.493	-.498	-.503	-.442	-.377	-.335	-.242
.350	-.409	-.406	-.413			-.400	-.333		
.400	-.397	-.384	-.395	-.382	-.391	-.377	-.336	-.197	-.177
.450	-.377	-.380	-.373			-.360	-.333		
.500	-.364	-.360	-.360	-.364	-.373	-.355	-.326	-.227	-.197
.550	-.355	-.348	-.355			-.333	-.316		
.600	-.351	-.346	-.351	-.353	-.364	-.244		-.202	-.182
.650	-.350	-.350	-.362			-.095	-.098		
.700	-.362	-.362	-.362	-.374	-.374	-.046	.070	.139	.152
.750	-.350	-.344	-.356			.018	.171		
.800	-.338	-.338	-.356	-.386	-.392	.102	.238	.159	.302
.850	-.296	-.296	-.314			.179	.292		
.900	-.206	-.212	-.248	-.290	-.464	.248		.356	.366
.950			-.116			.314			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.2265	-.894	-.616				.818		.0125	.561
-.2115	-.257	-.285		-.273	-.261	-.257		.0625	-.053
-.1765	-.093	-.129	-.193	-.245	-.241	-.257	-.241	.1125	
-.1165	.045	.014	-.081	-.237	-.193	-.301		.1625	-.640
-.0515	.227	.101	-.042	-.133	-.157	-.173		.2375	-.514
.0235		.049		-.133		-.141	-.141	.3225	
.0985		-.193	-.133	-.137	-.133	-.125		.3875	
.1735		-.385		-.137		-.121		.4625	-.461
.2485		-.297	-.234			-.074		.5375	-.315
.3235		-.345		-.139		-.096		.6125	-.262
.3985		-.329	-.245	-.143			-.143	.6875	-.190
.4735			-.220	-.137	-.095	-.081		.7625	
.5485		-.227		-.116	-.083	-.076		.8375	.103
.6235		-.167				-.056		.9125	.061
.6985		-.088	-.079	-.053	-.046	-.046	-.061	.9625	.190
.7735		-.007	-.009	-.016	-.023				.153
.8485		.074	.045	.022	.013	.009			
.9235		.126	.092	.070		.054			
.9735		.146	.133	.115	.099	.106	.111		
.9935		.155			.126	.123			

APPENDIX

TABLE A5.- Continued

CONFIGURATION 3

MACH = .199

ALPHA = 2.06

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.400	-1.143	-.518			.756				
.050	-.730	-.573	-.619	-.573	-.548	.458	.362	.332	.332	
.075	-.770	-.785	-.820			.135	.094			
.100	-.881	-.835	-.921	-.861	-.866	-.158	-.134	-.062	-.038	
.150	-.775	-.835	-.856				-.356			
.200	-.624	-.649	-.649	-.634	-.634	-.524	-.302	-.182	-.152	
.250	-.558	-.548	-.548			-.400	-.266			
.300	-.513	-.518	-.518	-.528	-.553	-.375	-.318	-.271	-.188	
.350	-.455	-.458	-.451			-.343	-.291			
.400	-.435	-.424	-.424	-.429	-.429	-.340	-.296	-.152	-.142	
.450	-.406	-.402	-.404			-.328	-.298			
.500	-.391	-.384	-.393	-.395	-.397	-.330	-.301	-.192	-.167	
.550	-.377	-.375	-.382			-.323	-.293			
.600	-.366	-.364	-.375	-.375	-.389	-.254		-.182	-.162	
.650	-.362	-.362	-.374			-.095	-.085			
.700	-.368	-.362	-.374	-.386	-.398	-.019	.085	.152	.169	
.750	-.356	-.356	-.362			.060	.184			
.800	-.332	-.332	-.362	-.392	-.392	.132	.253	.124	.312	
.850	-.290	-.290	-.314			.196	.300			
.900	-.194	-.206	-.236	-.272	-.362	.267		.364	.373	
.950			-.104			.322				

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CP	
-.2265	1.230	-.971				.936		.0125	.584	.509
-.2115	-.273	-.313		-.261	-.205	-.145		.0625		-.006
-.1765	-.109	-.145	-.197	-.221	-.201	-.185	-.213	.1125		
-.1165	.053	.022	-.073	-.209	-.165	-.257		.1625	-.543	-.494
-.0515	.242	.120	-.014	-.105	-.129	-.137		.2375	-.436	-.389
.0235		.101		-.105		-.113	-.105	.3225		
.0985		-.109	-.097	-.101	-.105	-.101		.3875		-.374
.1735		-.317		-.101		-.105		.4625		
.2485		-.232	-.192			-.061		.5375	-.287	-.237
.3235		-.285		-.109		-.076		.6125		-.175
.3985		-.283	-.213	-.118			-.116	.6875		
.4735			-.197	-.113	-.081	-.066		.7625		
.5485		-.208		-.095	-.069	-.066		.8375	.108	.054
.6235		-.146				-.046		.9125		
.6985		-.076	-.065	-.044	-.034	-.036	-.046	.9625	.210	.188
.7735		-.000	-.005	-.005	-.014					
.8485		.061	.049	.031	.018	.014				
.9235		.128	.101	.077		.051				
.9735		.164	.142	.122	.106	.108	.116			
.9935		.178			.131	.128				

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TABLE A5.- Continued

CONFIGURATION 3

MACH = .199

ALPHA = 3.05

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-1.700	-1.721	-1.190			.817			
.050	-1.145	-.832	-.680	-.675	-.685	.524	.434	.440	.423
.075	-.842	-.847	-.908			.213	.190		
.100	-.953	-.928	-1.044	-.928	-1.003	-.044	-.038	.052	.046
.150	-.847	-.924	-.948				-.254		
.200	-.695	-.736	-.726	-.716	-.736	-.441	-.230	-.116	-.086
.250	-.625	-.635	-.620			-.341	-.219		
.300	-.579	-.574	-.579	-.594	-.620	-.316	-.274	-.207	-.140
.350	-.496	-.499	-.501			-.294	-.239		
.400	-.474	-.470	-.472	-.467	-.470	-.291	-.249	-.120	-.104
.450	-.443	-.438	-.443			-.286	-.252		
.500	-.420	-.416	-.420	-.429	-.434	-.301	-.259	-.162	-.134
.550	-.403	-.403	-.407			-.299	-.262		
.600	-.389	-.389	-.396	-.394	-.405	-.249		-.155	-.135
.650	-.381	-.381	-.387			-.081	-.061		
.700	-.381	-.381	-.393	-.393	-.405	.023	.105	.167	.182
.750	-.363	-.363	-.381			.105	.204		
.800	-.339	-.333	-.363	-.393	-.393	.164	.270	.132	.322
.850	-.290	-.290	-.315			.226	.312		
.900	-.194	-.200	-.230	-.246	-.327	.290		.374	.382
.950			-.094			.332			

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CP	
-.2265	-1.132	-1.156				.965		.0125	.595	.540
-.2115	-.537	-.405		-.221	-.161	-.058		.0625		.061
-.1765	-.125	-.149	-.201	-.201	-.149	-.125	-.189	.1125		
-.1165	.065	.034	-.062	-.189	-.157	-.225		.1625	-.455	-.410
-.0515	.259	.144	.010	-.082	-.105	-.105		.2375	-.360	-.318
.0235		.140		-.074		-.085	-.070	.3225		
.0985		-.050	-.066	-.070	-.085	-.078		.3875		-.318
.1735		-.237		-.074		-.082		.4625		
.2485		-.181	-.134			-.043		.5375	-.255	-.205
.3235		-.232		-.083		-.058		.6125		-.153
.3985		-.239	-.167	-.093			-.088	.6875		
.4735			-.153	-.088	-.062	-.051		.7625		
.5485		-.174		-.076	-.051	-.051		.8375	.121	.054
.6235		-.127				-.036		.9125		
.6985		-.054	-.032	-.030	-.023	-.026	-.031	.9625	.226	.211
.7735		.014	.014	.004	-.005					
.8485		.072	.061	.038	.025	.021				
.9235		.133	.113	.081		.064				
.9735		.178	.154	.126	.113	.114	.124			
.9935		.192			.140	.136				

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TABLE A5.- Continued

CONFIGURATION 3 MACH = .195 ALPHA = 4.06

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
N										
X/C										
.025	-1.995	-1.975	-1.879			.863				
.050	-1.612	-1.562	-.997	-.825	-.795	.619	.529	.511	.523	
.075	-1.178	-1.042	-.931			.320	.291			
.100	-1.027	-.987	-1.098	-1.093	-1.083	.046	.082	.124	.141	
.150	-.901	-.987	-1.027				-.152			
.200	-.770	-.790	-.805	-.785	-.790	-.356	-.152	-.050	-.014	
.250	-.679	-.674	-.674			-.269	-.155			
.300	-.619	-.624	-.634	-.649	-.669	-.259	-.212	-.170	-.086	
.350	-.534	-.536	-.538			-.234	-.189			
.400	-.509	-.498	-.502	-.502	-.511	-.249	-.219	-.088	-.068	
.450	-.476	-.473	-.469			-.261	-.222			
.500	-.449	-.444	-.447	-.458	-.455	-.271	-.239	-.135	-.115	
.550	-.424	-.426	-.424			-.281	-.244			
.600	-.406	-.411	-.415	-.415	-.429	-.246		-.140	-.120	
.650	-.392	-.398	-.398			-.080	-.051			
.700	-.386	-.392	-.398	-.404	-.422	.055	.120	.171	.186	
.750	-.368	-.368	-.374			.132	.213			
.800	-.338	-.332	-.350	-.392	-.398	.189	.275	.117	.327	
.850	-.284	-.284	-.314			.243	.322			
.900	-.194	-.188	-.224	-.260	-.320	.297		.373	.381	
.950			-.098			.332				

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CP	
-.2265	-.911	-1.078				.964		.0125	.631	.541
-.2115	-.859	-.807		-.213	-.073	.034		.0625		.126
-.1765	-.177	-.153	-.201	-.185	-.097	-.061	-.173	.1125		
-.1165	.097	.053	-.042	-.173	-.113	-.149		.1625	-.357	-.297
-.0515	.270	.175	.045	-.057	-.065	-.065		.2375	-.290	-.235
.0235		.191		-.046		-.054	-.046	.3225		
.0985		.026	-.022	-.034	-.057	-.046		.3875		-.265
.1735		-.153		-.030		-.057		.4625		
.2485		-.123	-.081			-.024		.5375	-.230	-.178
.3235		-.178		-.055		-.041		.6125		-.128
.3985		-.190	-.132	-.062			-.058	.6875		
.4735			-.130	-.065	-.046	-.039		.7625		
.5485		-.155		-.058	-.037	-.041		.8375	.126	.059
.6235	-.104					-.026		.9125		
.6985	-.044	-.030	-.011	-.014	-.011	-.016		.9825	.240	.220
.7735	.022	.027	.016	.004						
.8485	.077	.072	.047	.034	.029					
.9235	.144	.122	.086		.071					
.9735	.187	.158	.133	.115	.123	.128				
.9935	.203			.144	.141					

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TABLE A5.- Continued

CONFIGURATION 3 MACH = .199 ALPHA = 5.06

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
N									
X/C									
.025	-2.203	-2.153	-2.239			.897			
.050	-2.042	-2.072	-1.685	-1.463	-1.307	.665	.600	.606	.588
.075	-1.544	-1.388	-1.066			.379	.367		
.100	-1.181	-1.025	-1.106	-1.136	-1.146	.135	.165	.219	.219
.150	-.955	-1.025	-1.096				-.068		
.200	-.819	-.834	-.859	-.854	-.874	-.272	-.074	.016	.040
.250	-.728	-.733	-.733			-.189	-.078		
.300	-.663	-.668	-.673	-.693	-.718	-.194	-.152	-.105	-.038
.350	-.580	-.577	-.584			-.184	-.142		
.400	-.546	-.537	-.539	-.546	-.553	-.199	-.162	-.046	-.026
.450	-.504	-.504	-.504			-.209	-.182		
.500	-.475	-.475	-.477	-.486	-.493	-.236	-.201	-.098	-.080
.550	-.450	-.446	-.453			-.251	-.216		
.600	-.428	-.433	-.433	-.441	-.453	-.226		-.115	-.093
.650	-.404	-.410	-.422			-.068	-.033		
.700	-.398	-.398	-.410	-.422	-.434	.072	.134	.188	.203
.750	-.368	-.374	-.392			.151	.230		
.800	-.344	-.338	-.356	-.398	-.404	.208	.292	.134	.338
.850	-.284	-.290	-.302			.262	.336		
.900	-.194	-.188	-.212	-.254	-.308	.314		.383	.390
.950			-.098			.341			

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CF	
-.2265	-.511	-1.073				.970		.0125	.647	.545
-.2115	-.535	-.742		-.193	.022	.144		.0625		.203
-.1765	-.424	-.360	-.209	-.169	-.046	-.006	-.153	.1125		
-.1165	.081	.057	-.030	-.161	-.049	-.101		.1625	-.260	-.225
-.0515	.281	.195	.069	-.038	-.046	-.038		.2375	-.212	-.168
.0235		.222		-.022		-.030	-.018	.3225		
.0985		.081	.026	-.010	-.034	-.022		.3875		-.197
.1735		-.073		-.006		-.034		.4625		
.2485		-.051	-.037			-.004		.5375	-.202	-.145
.3235		-.111		-.028		-.021		.6125		-.103
.3985		-.141	-.095	-.037			-.038	.6875		
.4735			-.095	-.039	-.030	-.024		.7625		
.5485		-.122		-.034	-.018	-.029		.8375	.126	.063
.6235		-.083				-.011		.9125		
.6985		-.028	-.005	-.000	-.002	-.006	-.009	.9625	.245	.225
.7735		.031	.043	.029	.016					
.8485		.083	.083	.058	.043	.031				
.9235		.146	.128	.097		.071				
.9735		.189	.166	.142	.121	.126	.130			
.9935		.207			.148	.143				

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TABLE A5.- Continued

CONFIGURATION 3

MACH = .303

ALPHA = .09

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.421	-.212	-.097			.619				
.050	-.385	-.437	-.376	-.302	-.228	.269	.125	.075	.080	
.075	-.604	-.622	-.577			-.080	-.142			
.100	-.748	-.689	-.714	-.658	-.628	-.394	-.368	-.325	-.290	
.150	-.620	-.689	-.687				-.577			
.200	-.505	-.525	-.518	-.496	-.514	-.727	-.486	-.343	-.317	
.250	-.451	-.455	-.446			-.562	-.412			
.300	-.430	-.442	-.437	-.448	-.453	-.535	-.458	-.312	-.309	
.350	-.372	-.372	-.367			-.474	-.414			
.400	-.363	-.354	-.357	-.351	-.354	-.446	-.405	-.264	-.222	
.450	-.351	-.347	-.349			-.417	-.387			
.500	-.342	-.341	-.340	-.344	-.343	-.396	-.375	-.274	-.239	
.550	-.335	-.337	-.339			-.342	-.358			
.600	-.332	-.336	-.344	-.334	-.349	-.228		-.236	-.208	
.650	-.335	-.338	-.346			-.131	-.119			
.700	-.354	-.351	-.354	-.368	-.376	-.089	.055	.127	.144	
.750	-.351	-.346	-.354			-.024	.151			
.800	-.343	-.338	-.360	-.394	-.394	.065	.213	.246	.299	
.850	-.303	-.306	-.333			.132	.273			
.900	-.215	-.223	-.255	-.295	-.419	.209		.360	.367	
.950			-.121			.298				

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CF	
-.2265	-.251	-.101				.731		.0125	.500	.463
-.2115	-.160	-.217		-.315	-.383	-.326		.0625		-.154
-.1765	-.059	-.110	-.199	-.288	-.313	-.331	-.278	.1125		
-.1165	.039	.009	-.096	-.280	-.253	-.353		.1625	-.797	-.725
-.0515	.217	.076	-.075	-.176	-.205	-.215		.2375	-.626	-.572
.0235		.005		-.180		-.176	-.182	.3225		
.0985		-.271	-.205	-.182	-.167	-.153		.3875		-.464
.1735		-.508		-.185		-.155		.4625		
.2485		-.395	-.319			-.107		.5375	-.344	-.300
.3235		-.427		-.187		-.120		.6125		-.220
.3985		-.397	-.314	-.187			-.179	.6875		
.4735			-.275	-.170	-.121	-.102		.7625		
.5485		-.274		-.148	-.104	-.098		.8375	.083	.065
.6235		-.200				-.077		.9125		
.6985		-.107	-.098	-.077	-.062	-.060	-.079	.9625	.160	.133
.7735		-.008	-.025	-.037	-.035					
.8485		.069	.034	.008	.002	-.000				
.9235		.102	.086	.060		.046				
.9735		.132	.126	.108	.096	.101	.105			
.9935		.143			.122	.120				

APPENDIX

TABLE A5.- Continued

CONFIGURATION 3

MACH = .399

ALPHA = .09

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.464	-.197	-.065			.629			
.050	-.388	-.433	-.377	-.267	-.254	.298	.140	.066	.060
.075	-.618	-.639	-.592			-.066	-.136		
.100	-.770	-.726	-.737	-.664	-.634	-.418	-.379	-.340	-.325
.150	-.641	-.726	-.730				-.621		
.200	-.535	-.551	-.539	-.509	-.521	-.723	-.510	-.366	-.337
.250	-.471	-.460	-.453			-.618	-.442		
.300	-.456	-.461	-.453	-.476	-.484	-.579	-.506	-.322	-.283
.350	-.391	-.394	-.389			-.506	-.446		
.400	-.383	-.373	-.378	-.366	-.364	-.479	-.436	-.290	-.250
.450	-.367	-.362	-.359			-.443	-.418		
.500	-.358	-.353	-.355	-.360	-.354	-.412	-.398	-.290	-.263
.550	-.353	-.352	-.353			-.354	-.376		
.600	-.352	-.350	-.359	-.355	-.364	-.226		-.254	-.219
.650	-.356	-.358	-.361			-.135	-.123		
.700	-.373	-.364	-.373	-.379	-.389	-.091	.048	.127	.145
.750	-.368	-.363	-.371			-.039	.141		
.800	-.360	-.355	-.374	-.415	-.410	.040	.209	.256	.294
.850	-.319	-.319	-.347			.118	.271		
.900	-.219	-.228	-.266	-.304	-.601	.193		.364	.364
.950			-.121			.289			

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CP	
-.2265	-.233	-.087				.739		.0125	.552	.476
-.2115	-.159	-.227		-.319	-.398	-.314		.0625		-.151
-.1765	-.059	-.116	-.208	-.300	-.322	-.342	-.291	.1125		
-.1165	.040	.011	-.099	-.291	-.265	-.371		.1625	-.862	-.763
-.0515	.223	.089	-.078	-.187	-.212	-.228		.2375	-.675	-.605
.0235		.012		-.194		-.188	-.192	.3225		
.0985		-.301	-.222	-.199	-.177	-.165		.3875		-.459
.1735		-.557		-.205		-.171		.4625		
.2485		-.439	-.349			-.122		.5375	-.361	-.318
.3235		-.468		-.206		-.135		.6125		-.228
.3985		-.429	-.340	-.207			-.201	.6875		
.4735			-.299	-.190	-.131	-.118		.7625		
.5485		-.293		-.163	-.116	-.113		.8375	.077	.062
.6235		.210				-.088		.9125		
.6985		-.112	-.104	-.084	-.072	-.070	-.089	.9625	.151	.131
.7735		-.007	-.028	-.038	-.042	-.013				
.8485		.069	.031	.009	-.000	-.007				
.9235		.101	.085	.061		.045				
.9735		.132	.129	.113	.098	.102	.106			
.9935		.143			.124	.121				

APPENDIX

TABLE A5.- Continued

CONFIGURATION 3 MACH = .500 ALPHA = .09

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.448	-.173	-.041			.647				
.050	-.395	-.424	-.362	-.274	-.228	.311	.150	.047	.060	
.075	-.640	-.628	-.610			-.086	-.153			
.100	-.825	-.767	-.767	-.690	-.642	-.444	-.435	-.387	-.375	
.150	-.690	-.767	-.744				-.698			
.200	-.558	-.560	-.559	-.541	-.546	-.752	-.552	-.410	-.367	
.250	-.480	-.465	-.477			-.676	-.476			
.300	-.480	-.478	-.484	-.515	-.517	-.635	-.555	-.362	-.315	
.350	-.408	-.400	-.401			-.557	-.484			
.400	-.400	-.387	-.379	-.385	-.378	-.518	-.462	-.329	-.274	
.450	-.387	-.372	-.362			-.473	-.442			
.500	-.379	-.360	-.363	-.370	-.372	-.435	-.421	-.317	-.293	
.550	-.375	-.355	-.363			-.354	-.393			
.600	-.370	-.355	-.371	-.361	-.385	-.218		-.270	-.240	
.650	-.371	-.369	-.363			-.125	-.128			
.700	-.392	-.373	-.379	-.393	-.416	-.094	.035	.126	.145	
.750	-.385	-.370	-.379			-.047	.124			
.800	-.376	-.356	-.383	-.431	-.437	.037	.191	.264	.298	
.850	-.330	-.312	-.352			.110	.258			
.900	-.216	-.216	-.262	-.311	-.454	.176		.369	.369	
.950			-.110			.273				

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C								X/C	CP	
-.2265	-.114	.026				.736		.0125	.577	.506
-.2115	-.148	-.203		-.309	-.428	-.329		.0625		-.150
-.1765	-.054	-.106	-.195	-.300	-.350	-.366	-.299	.1125		
-.1165	.051	.029	-.089	-.295	-.289	-.394		.1625	-.949	-.863
-.0515	.247	.113	-.071	-.191	-.231	-.238		.2375	-.721	-.688
.0235		.023		-.207		-.203	-.205	.3225		
.0985		-.314	-.244	-.221	-.199	-.178		.3875		-.482
.1735		-.595		-.236		-.184		.4625		
.2485		-.489	-.374			-.137		.5375	-.361	-.331
.3235		-.515		-.230		-.150		.6125		-.237
.3985		-.475	-.361	-.236			-.214	.6875		
.4735			-.316	-.218	-.145	-.131		.7625		
.5485		-.313		-.186	-.127	-.125		.8375	.070	.068
.6235		-.222				-.097		.9125		
.6985		-.115	-.108	-.096	-.079	-.078	-.086	.9625	.131	.131
.7735		-.003	-.029	-.046	-.049					
.8485		.077	.035	.005	-.004	-.004				
.9235		.105	.088	.064		.049				
.9735		.135	.130	.118	.100	.110	.114			
.9935		.147			.130	.131				

APPENDIX

TABLE A5.- Continued

CONFIGURATION 3

MACH = .599

ALPHA = .09

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.448	-.193	-.028			.699				
.050	-.377	-.427	-.361	-.210	-.188	.364	.164	.048	.033	
.075	-.659	-.673	-.636			-.024	-.155			
.100	-.883	-.827	-.867	-.749	-.677	-.418	-.447	-.422	-.417	
.150	-.749	-.827	-.871				-.803			
.200	-.584	-.617	-.609	-.587	-.582	-.835	-.649	-.464	-.413	
.250	-.513	-.524	-.515			-.735	-.575			
.300	-.516	-.544	-.529	-.549	-.562	-.692	-.664	-.402	-.363	
.350	-.436	-.432	-.429			-.593	-.569			
.400	-.428	-.412	-.421	-.422	-.416	-.549	-.543	-.370	-.314	
.450	-.405	-.406	-.412			-.496	-.505			
.500	-.396	-.395	-.410	-.418	-.405	-.438	-.471	-.351	-.333	
.550	-.388	-.393	-.411			-.341	-.429			
.600	-.387	-.392	-.415	-.408	-.413	-.206		-.289	-.258	
.650	-.394	-.391	-.408			-.128	-.116			
.700	-.410	-.404	-.425	-.433	-.437	-.110	.033	.126	.145	
.750	-.399	-.403	-.425			-.071	.108			
.800	-.386	-.389	-.423	-.469	-.463	-.001	.165	.272	.286	
.850	-.328	-.334	-.380			.059	.234			
.900	-.201	-.219	-.271	-.321	-.348	.125		.366	.360	
.950			-.098			.226				

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CF	
-.2265	-.042	.144				.794		.0125	.619	.555
-.2115	-.130	-.194		-.306	-.436	-.342		.0625		-.107
-.1765	-.034	-.094	-.196	-.323	-.367	-.386	-.316	.1125		
-.1165	.077	.041	-.090	-.316	-.306	-.419		.1625	1.160	-1.021
-.0515	.280	.137	-.064	-.207	-.239	-.262		.2375	-.834	-.769
.0235		.051		-.229		-.219	-.222	.3225		
.0985		-.353	-.288	-.252	-.212	-.194		.3675		-.515
.1735		-.757		-.272		-.206		.4625		
.2485		-.548	-.471			-.149		.5375	-.377	-.346
.3235		-.566		-.275		-.162		.6125		-.240
.3985		-.514	-.428	-.279			-.266	.6875		
.4735			-.370	-.250	-.167	-.143		.7625		
.5485		-.334		-.211	-.148	-.136		.8375	.058	.064
.6235		-.226				-.108		.9125		
.6985		-.114	-.127	-.101	-.095	-.089	-.110	.9625	.108	.126
.7735		-.006	-.041	-.047	-.060					
.8485		.070	.025	.010	-.010	-.012				
.9235		.096	.085	.070		.042				
.9735		.125	.129	.124	.102	.105	.109			
.9935		.137			.132	.127				

APPENDIX

TABLE A5.- Continued

CONFIGURATION 3

MACH = .600

ALPHA = 1.10

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
N										
X/C										
.025	-.907	-.548	-.169			.766				
.050	-.539	-.550	-.527	-.418	-.374	.454	.272	.199	.189	
.075	-.797	-.843	-.816			.081	-.026			
.100	-1.038	-1.009	-1.031	-.915	-.883	-.293	-.311	-.294	-.277	
.150	-.891	-1.009	-1.000				-.647			
.200	-.694	-.730	-.712	-.672	-.661	-.726	-.532	-.388	-.340	
.250	-.601	-.596	-.591			-.634	-.470			
.300	-.592	-.606	-.588	-.604	-.621	-.626	-.567	-.331	-.311	
.350	-.495	-.498	-.508			-.546	-.491			
.400	-.482	-.469	-.483	-.475	-.469	-.518	-.476	-.326	-.270	
.450	-.461	-.453	-.450			-.479	-.464			
.500	-.442	-.437	-.441	-.449	-.445	-.440	-.439	-.322	-.295	
.550	-.432	-.431	-.435			-.361	-.404			
.600	-.424	-.425	-.435	-.430	-.438	-.219		-.274	-.240	
.650	-.418	-.422	-.428			-.117	-.104			
.700	-.435	-.427	-.436	-.448	-.455	-.087	.061	.136	.151	
.750	-.421	-.419	-.428			-.030	.146			
.800	-.397	-.394	-.421	-.473	-.461	.040	.211	.273	.299	
.850	-.330	-.333	-.373			.104	.276			
.900	-.197	-.210	-.262	-.313	-.313	.175		.381	.377	
.950			-.092			.281				

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CP	
-.2265	-.614	-.292				.883		.0125	.645	.590
-.2115	-.249	-.289		-.262	-.361	-.270		.0625		-.025
-.1765	-.075	-.121	-.206	-.285	-.293	-.322	-.278	.1125		
-.1165	.076	.048	-.068	-.285	-.253	-.376		.1625	-.961	-.815
-.0515	.292	.161	-.019	-.160	-.193	-.222		.2375	-.720	-.640
.0235		.102		-.172		-.186	-.176	.3225		
.0985		-.251	-.195	-.186	-.173	-.165		.3875		-.465
.1735		-.616		-.206		-.178		.4625		
.2485		-.448	-.387			-.123		.5375	-.366	-.333
.3235		-.502		-.232		-.142		.6125		-.233
.3985		-.473	-.381	-.233			-.226	.6875		
.4735			-.334	-.212	-.144	-.129		.7625		
.5485		-.331		-.176	-.129	-.125		.8375	.084	.068
.6235		-.218				-.097		.9125		
.6985		-.110	-.105	-.079	-.080	-.079	-.094	.9625	.148	.154
.7735		-.010	-.023	-.031	-.047					
.8485		.079	.043	.019	-.005	-.004				
.9235		.120	.100	.075		.049				
.9735		.148	.145	.127	.104	.116	.125			
.9935		.161			.139	.138				

APPENDIX

TABLE A5.- Continued

CONFIGURATION 3

MACH = .600

ALPHA = 2.08

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
N										
X/C										
.025	-1.311	-1.140	-.383			.828				
.050	-.783	-.628	-.667	-.605	-.566	.523	.375	.322	.330	
.075	-.904	-.957	-.988			.169	.086			
.100	-1.192	-1.131	-1.254	-1.131	-1.106	-.188	-.184	-.146	-.124	
.150	-1.031	-1.131	-1.134				-.515			
.200	-.799	-.817	-.794	-.794	-.777	-.631	-.434	-.275	-.252	
.250	-.680	-.673	-.669			-.548	-.382			
.300	-.650	-.664	-.668	-.693	-.701	-.545	-.485	-.276	-.242	
.350	-.555	-.552	-.557			-.484	-.426			
.400	-.530	-.513	-.527	-.523	-.515	-.461	-.414	-.280	-.223	
.450	-.500	-.499	-.493			-.435	-.404			
.500	-.481	-.471	-.474	-.491	-.484	-.412	-.400	-.284	-.257	
.550	-.465	-.457	-.464			-.362	-.383			
.600	-.451	-.447	-.458	-.454	-.475	-.222		-.248	-.223	
.650	-.441	-.439	-.446			-.106	-.096			
.700	-.448	-.442	-.448	-.462	-.479	-.062	.077	.151	.170	
.750	-.428	-.425	-.435			.003	.163			
.800	-.396	-.390	-.420	-.479	-.479	.079	.230	.288	.326	
.850	-.319	-.321	-.362			.146	.296			
.900	-.184	-.192	-.247	-.310	-.306	.222		.398	.401	
.950			-.084			.319				

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C				CP				X/C	CP	
-.2265	-1.047	-.734				.987		.0125	.673	.602
-.2115	-.332	-.336		-.229	-.295	-.210		.0625		.028
-.1765	-.097	-.136	-.211	-.256	-.240	-.246	-.247	.1125		
-.1165	.085	.057	-.048	-.252	-.221	-.328		.1625	-.791	-.696
-.0515	.309	.186	.015	-.125	-.163	-.186		.2375	-.605	-.558
.0235		.156		-.127		-.154	-.127	.3225		
.0985		-.159	-.122	-.138	-.145	-.134		.3875		-.418
.1735		-.494		-.161		-.151		.4625		
.2485		-.364	-.305			-.098		.5375	-.349	-.308
.3235		-.432		-.187		-.119		.6125		-.220
.3985		-.421	-.324	-.197			-.187	.6875		
.4735			-.293	-.188	-.124	-.109		.7625		
.5485		-.306		-.160	-.108	-.106		.8375	.100	.067
.6235		-.197				-.080		.9125		
.6985		-.100	-.091	-.072	-.065	-.065	-.076	.9625	.178	.196
.7735		-.008	-.012	-.023	-.035					
.8485		.072	.054	.027	.007	.004				
.9235		.139	.115	.083		.057				
.9735		.179	-.159	.139		.118	.132			
.9935		.192				.151	.146			

APPENDIX

TABLE A5.- Continued

CONFIGURATION 3 MACH = .599 ALPHA = 3.10

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.680	-1.569	-1.204			.875				
.050	-1.215	-.944	-.772	-.743	-.740	.596	.474	.434	.441	
.075	-1.101	-1.072	-1.132			.271	.202			
.100	-1.324	-1.274	-1.427	-1.372	-1.363	-.068	-.061	-.019	.001	
.150	-1.166	-1.274	-1.278				-.387			
.200	-.892	-.898	-.901	-.907	-.890	-.541	-.330	-.182	-.162	
.250	-.764	-.744	-.749			-.454	-.305			
.300	-.719	-.728	-.724	-.754	-.774	-.462	-.388	-.211	-.168	
.350	-.608	-.613	-.615			-.416	-.353			
.400	-.578	-.567	-.566	-.574	-.572	-.401	-.362	-.230	-.172	
.450	-.542	-.535	-.530			-.391	-.363			
.500	-.517	-.512	-.512	-.517	-.521	-.383	-.361	-.245	-.221	
.550	-.495	-.494	-.495			-.344	-.351			
.600	-.474	-.475	-.481	-.481	-.496	-.214		-.228	-.196	
.650	-.457	-.458	-.463			-.096	-.082			
.700	-.459	-.454	-.465	-.478	-.490	-.044	.093	.170	.186	
.750	-.429	-.434	-.437			.026	.190			
.800	-.389	-.391	-.415	-.476	-.476	.110	.259	.305	.342	
.850	-.308	-.311	-.353			.184	.313			
.900	-.178	-.182	-.233	-.293	-.305	.261		.412	.413	
.950			-.081			.345				

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CF	
-.2265	-1.177	-.964				1.041		.0125	.695	.624
-.2115	-.582	-.538		-.206	-.195	-.112		.0625		.110
-.1765	-.121	-.153	-.208	-.224	-.181	-.179	-.219	.1125		
-.1165	.101	.067	-.033	-.212	-.181	-.276		.1625	-.638	-.562
-.0515	.322	.209	.053	-.086	-.126	-.140		.2375	-.503	-.451
.0235		.199		-.082		-.115	-.080	.3225		
.0985		-.072	-.044	-.095	-.111	-.102		.3875		-.361
.1735		-.370		-.109		-.121		.4625		
.2485		-.284	-.227			-.074		.5375	-.322	-.282
.3235		-.356		-.143		-.094		.6125		-.195
.3985		-.360	-.271	-.160			-.150	.6875		
.4735			-.249	-.153	-.097	-.088		.7625		
.5485		-.282		-.129	-.086	-.088		.8375	.121	.072
.6235		-.179				-.064		.9125		
.6985		-.083	-.067	-.057	-.049	-.051	-.058	.9625	.201	.227
.7735		.009	.006	-.009	-.020					
.8485		.079	.066	.037	.020	.012				
.9235		.153	.127	.091		.064				
.9735		.197	.171	.147	.126	.130	.143			
.9935		.213			.158	.154				

APPENDIX

TABLE A5.- Continued

CONFIGURATION 3

MACH = .651

ALPHA = .09

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.407	-.127	.025			.725			
.050	-.352	-.396	-.327	-.174	-.147	.395	.190	.053	.038
.075	-.654	-.679	-.628			.022	-.136		
.100	-.959	-.922	-.898	-.734	-.711	-.377	.457	-.472	-.458
.150	-.834	-.922	-.911				-.911		
.200	-.619	-.650	-.627	-.625	-.608	-1.438	-.767	-.520	-.452
.250	-.536	-.529	-.525			-.807	-.623		
.300	-.550	-.569	-.553	-.592	-.609	-.742	-.723	-.469	-.417
.350	-.456	-.456	-.460			-.633	-.592		
.400	-.442	-.433	-.442	-.432	-.421	-.577	-.551	-.429	-.350
.450	-.424	-.423	-.419			-.511	-.511		
.500	-.419	-.413	-.421	-.424	-.410	-.418	-.470	-.399	-.358
.550	-.416	-.412	-.416			-.304	-.414		
.600	-.410	-.413	-.417	-.411	-.423	-.201		-.310	-.263
.650	-.412	-.412	-.417			-.155	-.110		
.700	-.428	-.422	-.434	-.438	-.447	-.136	.007	.119	.141
.750	-.418	-.419	-.428			-.093	.065		
.800	-.401	-.398	-.419	-.476	-.471	-.021	.120	.259	.263
.850	-.332	-.335	-.369			.033	.194		
.900	-.189	-.206	-.254	-.311	-.348	.087		.358	.347
.950			-.078			.163			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CF
-.2265	.049	.223				.798		.0125	.645
-.2115	-.104	-.174		-.290	-.442	-.412		.0625	-.581
-.1765	-.020	-.079	-.186	-.318	-.380	-.421	-.313	.1125	-.077
-.1165	.091	.060	-.073	-.312	-.324	-.446		.1625	1.232
-.0515	.301	.162	-.041	-.207	-.257	-.277		.2375	-.164
.0235		.077		-.236		-.238	-.234	.3225	-.830
.0985		-.350	-.309	-.273	-.237	-.213		.3875	
.1735		-.429		-.311		-.220		.4625	-.548
.2485		-.615	-.550			-.169		.5375	
.3235		-.615		-.312		-.187		.6125	-.347
.3985		-.555	-.469	-.309			-.302	.6875	-.352
.4735			-.394	-.278	-.188	-.165		.7625	-.245
.5485		-.337		-.232	-.167	-.159		.8375	
.6235		-.238				-.126		.9125	.036
.6985		-.129	-.128	-.116	-.104	-.103	-.120	.9625	.057
.7735		-.020	-.040	-.059	-.065				.128
.8485		.053	.029	-.001	-.014	-.021			
.9235		.092	.085	.060		.037			
.9735		.121	.130	.117	.102	.104	.111		
.9935		.134			.133	.127			

APPENDIX

TABLE A5.- Continued

CONFIGURATION 3

MACH = .700

ALPHA = .10

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.369	-.074	.073			.750				
.050	-.315	-.341	-.270	-.150	-.123	.454	.241	.044	.022	
.075	-.622	-.634	-.574			.098	-.076			
.100	-.935	-1.290	-.879	-.746	-.723	-.276	-.387	-.486	-.499	
.150	-1.218	-1.290	-1.228				-.882			
.200	-.624	-.640	-.621	-.654	-.656	-1.357	-1.106	-.602	-.518	
.250	-.539	-.530	-.529			-1.413	-.923			
.300	-.572	-.588	-.574	-.629	-.660	-.871	-.823	-.536	-.488	
.350	-.473	-.474	-.478			-.645	-.604			
.400	-.463	-.446	-.457	-.445	-.440	-.519	-.551	-.484	-.387	
.450	-.447	-.435	-.435			-.411	-.496			
.500	-.436	-.427	-.435	-.441	-.428	-.333	-.453	-.438	-.390	
.550	-.428	-.430	-.432			-.263	-.391			
.600	-.427	-.428	-.432	-.430	-.433	-.214		-.315	-.263	
.650	-.426	-.425	-.432			-.202	-.119			
.700	-.445	-.434	-.445	-.455	-.463	-.184	-.032	.107	.131	
.750	-.431	-.428	-.439			-.150	.015			
.800	-.400	-.397	-.423	-.489	-.483	-.086	.056	.228	.229	
.850	-.310	-.318	-.360			-.010	.105			
.900	-.148	-.171	-.230	-.292	-.325	.048		.338	.316	
.950			-.056			.107				

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CP	
-.2265	.182	.369				.814		.0125	.682	.621
-.2115	-.070	-.127		-.290	-.448	-.481		.0625		-.008
-.1765	.008	-.049	-.158	-.314	-.398	-.446	-.303	.1125		
-.1165	.128	.099	-.044	-.311	-.336	-.458		.1625	1.095	-1.047
-.0515	.346	.207	-.011	-.205	-.264	-.290		.2375	1.610	-1.363
.0235		.131		-.245		-.252	-.238	.3225		
.0985		-.291	-.319	-.298	-.258	-.228		.3875		-.509
.1735		-.899		-.358		-.238		.4625		
.2485		-1.065	-.814			-.192		.5375	-.294	-.328
.3235		-.658		-.366		-.213		.6125		-.246
.3985		-.528	-.497	-.345			-.341	.6875		
.4735			-.399	-.301	-.211	-.169		.7625		
.5485		-.343		-.247	-.187	-.179		.8375	-.011	.030
.6235		-.248				-.144		.9125		
.6985		-.154	-.143	-.127	-.116	-.117	-.136	.9625	.053	.107
.7735		-.053	-.058	-.070	-.074					
.8485		.025	.011	-.011	-.021	-.027				
.9235		.087	.075	.053		.034				
.9735		.121	.120	.113	.098	.102	.107			
.9935		.134			.130	.126				

APPENDIX

TABLE A5.- Continued

CONFIGURATION 3

MACH = .750

ALPHA = .08

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/L										
.025	-.307	-.038	.129				.797			
.050	-.233	-.274	-.209	-.069	-.024		.512	.298	.101	.085
.075	-.526	-.542	-.504				.180	.012		
.100	-.828	-1.183	-.763	-.671	-.629		-.166	-.284	-.436	-.446
.150	-1.129	-1.183	-1.145					-.746		
.200	-1.155	-1.204	-1.199	-1.095	-1.013		-.798	-.986	-.805	-.642
.250	-1.131	-1.136	-1.093				-.771	-.659		
.300	-.701	-.621	-.580	-.609	-.623		-.675	-.592	-.589	-.550
.350	-.442	-.465	-.446				-.584	-.480		
.400	-.443	-.439	-.433	-.450	-.457		-.527	-.460	-.502	-.435
.450	-.449	-.447	-.441				-.451	-.392		
.500	-.459	-.453	-.458	-.473	-.454		-.371	-.345	-.445	-.428
.550	-.469	-.466	-.469				-.328	-.301		
.600	-.479	-.485	-.488	-.477	-.463		-.289		-.310	-.238
.650	-.488	-.501	-.491				-.267	-.240		
.700	-.527	-.533	-.525	-.521	-.499		-.230	-.223	.086	.098
.750	-.529	-.536	-.532				-.203	-.172		
.800	-.524	-.520	-.530	-.548	-.515		-.177	-.132	.188	.192
.850	-.451	-.465	-.455				-.122	-.080		
.900	-.303	-.318	-.324	-.295	-.273		-.078		.295	.284
.950			-.135				-.043			

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CP	
-.2265	.452	.593				.882		.0125	.726	.659
-.2115	.021	-.051		-.264	-.380	-.444		.0625		.076
-.1765	.057	-.004	-.111	-.283	-.379	-.441	-.277	.1125		
-.1165	.176	.145	.004	-.282	-.332	-.469		.1625	-.935	-.890
-.0515	.393	.261	.043	-.174	-.254	-.289		.2375	-.751	-.813
.0235		.200		-.216		-.254	-.215	.3225		
.0985		-.195	-.244	-.282	-.255	-.233		.3875		-.448
.1735		-.763		-.361		-.246		.4625		
.2485		-.767	-.670			-.199		.5375	-.326	-.321
.3235		-.596		-.357		-.224		.6125		-.269
.3985		-.473	-.434	-.335			-.336	.6875		
.4735			-.361	-.299	-.220	-.200		.7625		
.5485		-.312		-.245	-.196	-.190		.8375	-.129	-.097
.6235		-.250				-.153		.9125		
.6985		-.191	-.162	-.136	-.127	-.129	-.151	.9625	-.048	-.025
.7735		-.129	-.100	-.086	-.087					
.8485		-.076	-.049	-.034	-.036	-.043				
.9235		-.034	-.019	.018		.017				
.9735		-.011	.006	.069		.077	.066			
.9935		-.015			.106	.105				

APPENDIX

TABLE A5.- Continued

CONFIGURATION 3

MACH = .800

ALPHA = .07

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C										
.025		-.220	.079	.198			.836			
.050		-.130	-.164	-.098	.011	.050	.569	.358	.161	.129
.075		-.405	-.417	-.377			.258	.094		
.100		-.688	-1.024	-.620	-.531	-.493	-.064	-.185	-.345	-.357
.150		-.976	-1.024	-.982				-.616		
.200		-1.018	-1.061	-1.058	-.963	-.946	-.524	-.859	-.852	-.897
.250		-1.036	-1.020	-1.000			-.380	-.630		
.300		-1.031	-1.025	-1.001	-.995	-.998	-.392	-.528	-.726	-.827
.350		-.942	-.950	-.938			-.394	-.360		
.400		-.941	-.925	-.917	-.871	-.861	-.366	-.315	-.482	-.572
.450		-.906	-.897	-.895			-.354	-.300		
.500		-.870	-.872	-.895	-.850	-.861	-.327	-.317	-.437	-.392
.550		-.875	-.881	-.885			-.303	-.295		
.600		-.870	-.877	-.870	-.853	-.474	-.297		-.310	-.169
.650		-.821	-.813	-.809			-.278	-.243		
.700		-.625	-.566	-.720	-.479	-.462	-.260	-.244	.083	.096
.750		-.474	-.469	-.469			-.237	-.226		
.800		-.467	-.457	-.474	-.460	-.529	-.217	-.213	.193	.181
.850		-.424	-.435	-.454			-.196	-.189		
.900		-.314	-.328	-.359	-.263	-.238	-.180		.300	.282
.950				-.219			-.153			

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CP	
-.2265	.689	.806				.938		.0125	.763	.706
-.2115	.125	.058		-.206	-.294	-.390		.0625		.160
-.1765	.123	.066	-.037	-.234	-.355	-.424	-.231	.1125		
-.1165	.242	.209	.061	-.245	-.323	-.488		.1625	-.787	-.745
-.0515	.454	.323	.105	-.132	-.238	-.281		.2375	-.464	-.529
.0235		.265		-.181		-.253	-.179	.3225		
.0985		-.107	-.166	-.260	-.256	-.238		.3875		-.314
.1735		-.635		-.371		-.258		.4625		
.2485		-.546	-.640			-.220		.5375	-.315	-.281
.3235		-.344		-.380		-.253		.6125		-.276
.3985		-.321	-.381	-.325			-.344	.6875		
.4735			-.332	-.278	-.230	-.217		.7625		
.5485		-.281		-.235	-.201	-.204		.8375	-.203	-.152
.6235		-.267				-.164		.9125		
.6985		-.241	-.241	-.150	-.134	-.135	-.170	.9625	-.138	-.110
.7735		-.181	-.206	-.116	-.094					
.8485		-.125	-.158	-.074	-.050	-.050				
.9235		-.114	-.133	-.040		.002				
.9735		-.111	-.122	-.008	.047	.062	-.010			
.9935		-.113			.073	.081				

APPENDIX

TABLE A5.- Continued

CONFIGURATION 3 MACH = .799 ALPHA = 1.08

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.401	-.036	-.136			.861				
.050	-.181	-.214	-.166	-.062	-.022	.599	.405	.232	.195	
.075	-.442	-.468	-.439			.286	.140			
.100	-.728	-.076	-.673	-.600	-.551	-.033	-.144	-.277	-.297	
.150	-1.022	-1.076	-1.042				-.579			
.200	-1.075	-1.123	-1.115	-1.028	-1.022	-.703	-.814	-.766	-.828	
.250	-1.097	-1.086	-1.062			-.548	-.752			
.300	-1.094	-1.092	-1.077	-1.059	-1.066	-.516	-.624	-.612	-.610	
.350	-1.012	-1.026	-1.016			-.521	-.439			
.400	-1.015	-.999	-.998	-.957	-.943	-.452	-.390	-.505	-.452	
.450	-.985	-.980	-.982			-.429	-.359			
.500	-.967	-.960	-.981	-.970	-.957	-.381	-.353	-.488	-.454	
.550	-.965	-.966	-.974			-.346	-.340			
.600	-.962	-.966	-.979	-.973	-.958	-.321		-.307	-.181	
.650	-.758	-.792	-.875			-.301	-.274			
.700	-.511	-.480	-.532	-.462	-.418	-.270	-.265	.077	.097	
.750	-.348	-.341	-.365			-.246	-.231			
.800	-.269	-.274	-.297	-.339	-.391	-.222	-.192	.181	.188	
.850	-.198	-.200	-.231			-.190	-.155			
.900	-.143	-.145	-.164	-.166	-.188	-.154		.296	.292	
.950			-.112			-.117				

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/L	CP							X/C	CP	
-.2265	.305	.493				1.001		.0125	.772	.711
-.2115	.008	-.054		-.202	-.274	-.285		.0625		.179
-.1765	.086	.031	-.069	-.225	-.310	-.370	-.218	.1125		
-.1165	.228	.196	.059	-.226	-.290	-.441		.1625	-.757	-.714
-.0515	.451	.325	.116	-.108	-.209	-.251		.2375	-.642	-.745
.0235		.281		-.148		-.231	-.141	.3225		
.0985		-.079	-.123	-.219	-.228	-.220		.3875		-.387
.1735		-.603		-.320		-.241		.4625		
.2485		-.735	-.640			-.198		.5375	-.351	-.326
.3235		-.480		-.393		-.235		.6125		-.296
.3985		-.444	-.426	-.357			-.342	.6875		
.4735			-.354	-.314	-.247	-.214		.7625		
.5485		-.317		-.256	-.220	-.205		.8375	-.194	-.164
.6235		-.286				-.165		.9125		
.6985		-.240	-.194	-.149	-.143	-.137	-.166	.9625	-.109	-.087
.7735		-.181	-.146	-.109	-.101					
.8485		-.121	-.105	-.062	-.050	-.051				
.9235		-.080	-.078	-.011		.009				
.9735		-.066	-.051	.036		.064	.076	.034		
.9935		-.065			.095	.099				

APPENDIX

TABLE A5.- Continued

CONFIGURATION 3

MACH = .800

ALPHA = 2.09

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
N										
X/C										
.025	-.577	-.106	.082			.883				
.050	-.259	-.254	-.228	-.122	-.087	.625	.450	.288	.269	
.075	-.473	-.516	-.498			.314	.183			
.100	-.760	-1.115	-.724	-.650	-.602	-.003	-.103	-.227	-.236	
.150	-1.068	-1.115	-1.089				-.539			
.200	-1.126	-1.172	-1.165	-1.078	-1.068	-.947	-.766	-.678	-.712	
.250	-1.152	-1.139	-1.114			-1.177	-.791			
.300	-1.146	-1.147	-1.130	-1.122	-1.111	-1.196	-1.003	-.609	-.594	
.350	-1.062	-1.071	-1.057			-.850	-1.009			
.400	-1.071	-1.055	-1.063	-1.021	-1.025	-.626	-.872	-.697	-.467	
.450	-1.046	-1.036	-1.051			-.512	-.592			
.500	-1.030	-1.021	-1.047	-1.047	-1.040	-.441	-.402	-.742	-.499	
.550	-.787	-.804	-.945			-.396	-.373			
.600	-.518	-.492	-.563	-1.009	-1.046	-.367		-.268	-.208	
.650	-.400	-.398	-.409			-.344	-.270			
.700	-.353	-.358	-.360	-.422	-.459	-.293	-.250	.034	.072	
.750	-.294	-.297	-.295			-.235	-.192			
.800	-.244	-.248	-.245	-.287	-.317	-.184	-.145	.135	.162	
.850	-.207	-.211	-.199			-.140	-.086			
.900	-.183	-.183	-.166	-.153	-.171	-.086		.277	.270	
.950			-.141			-.038				

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C								X/C	CP	
-.2265	-.169	.139				1.055		.0125	.774	.712
-.2115	-.109	-.144		-.198	-.263	-.216		.0625		.195
-.1765	.045	-.005	-.095	-.217	-.274	-.312	-.210	.1125		
-.1165	.218	.189	.055	-.213	-.268	-.416		.1625	-.723	-.683
-.0515	.448	.327	.125	-.089	-.192	-.229		.2375	1.185	-.944
.0235		.293		-.123		-.211	-.118	.3225		
.0985		-.056	-.112	-.191	-.217	-.203		.3875		-.882
.1735		-.572		-.290		-.235		.4625		
.2485		-.771	-.619			-.195		.5375	-.371	-.372
.3235		-1.146		-.485		-.244		.6125		-.331
.3985		-.905	-.866	-.545			-.435	.6875		
.4735			-.485	-.463	-.306	-.237		.7625		
.5485		-.349		-.313	-.260	-.229		.8375	-.133	-.110
.6235		-.306				-.187		.9125		
.6985		-.231	-.193	-.188	-.170	-.157	-.160	.9625	-.061	-.053
.7735		-.146	-.122	-.137	-.127					
.8485		-.077	-.054	-.079	-.072	-.064				
.9235		-.013	.003	-.009		.004				
.9735		.032	.059	.062	.063	.085	.059			
.9935		.047			.103	.111				

APPENDIX

TABLE A5.- Continued.

CONFIGURATION 3 MACH = .849 ALPHA = .02

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C										
.025		-.046	.172	.299			.862			
.050		-.024	-.052	.017	.115	.153	.617	.406	.216	.191
.075		-.278	-.286	-.239			.315	.161		
.100		-.545	-.861	-.475	-.397	-.352	.005	-.106	-.260	-.274
.150		-.816	-.861	-.820				-.515		
.200		-.861	-.900	-.894	-.805	-.806	-.563	-.741	-.748	-.791
.250		-.883	-.870	-.844			-.600	-.758		
.300		-.879	-.873	-.860	-.849	-.873	-.519	-.746	-.670	-.794
.350		-.804	-.820	-.808			-.439	-.536		
.400		-.808	-.792	-.794	-.743	-.731	-.395	-.416	-.799	-.745
.450		-.784	-.777	-.782			-.387	-.366		
.500		-.762	-.768	-.782	-.762	-.758	-.365	-.343	-.870	-.887
.550		-.777	-.774	-.778			-.345	-.318		
.600		-.782	-.787	-.784	-.782	-.774	-.354		-.439	-.337
.650		-.784	-.783	-.778			-.361	-.297		
.700		-.845	-.829	-.831	-.823	-.806	-.352	-.331	-.239	-.297
.750		-.661	-.672	-.821			-.358	-.330		
.800		-.389	-.401	-.472	-.424	-.352	-.352	-.336	-.097	-.294
.850		-.307	-.303	-.342			-.345	-.348		
.900		-.289	-.290	-.311	-.274	-.171	-.336		.122	-.231
.950				-.316			-.333			

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CP	
-.2265	.935	.997				.981		.0125	.806	.752
-.2115	.251	.182		-.132	-.204	-.406		.0625		.231
-.1765	.211	.159	.049	-.173	-.300	-.640	-.175	.1125		
-.1165	.304	.279	.134	-.197	-.301	-.512		.1625	-.655	-.619
-.0515	.512	.387	.173	-.082	-.212	-.263		.2375	-.615	-.641
.0235		.333		-.133		-.248	-.131	.3225		
.0985		-.023	-.085	-.221	-.251	-.241		.3875		-.326
.1735		-.517		-.356		-.275		.4625		
.2485		-.734	-.598			-.265		.5375	-.349	-.326
.3235		-.533		-.579		-.364		.6125		-.328
.3985		-.455	-.586	-.604			-.660	.6875		
.4735			-.502	-.519	-.513	-.486		.7625		
.5485		-.337		-.395	-.451	-.513		.8375	-.364	-.355
.6235		-.339				-.310		.9125		
.6985		-.343	-.420	-.247	-.240	-.236	-.307	.9625	-.315	-.292
.7735		-.368	-.391	-.237	-.200					
.8485		-.350	-.361	-.221	-.163	-.153				
.9235		-.319	-.324	-.207		-.107				
.9735		-.304	-.293	-.174	-.067	-.043	-.190			
.9935		-.306			-.043	-.074				

APPENDIX

TABLE A5.- Continued

CONFIGURATION 3

MACH = .849

ALPHA = 1.10

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D										
N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.211	.098	.239			.888				
.050	-.062	-.091	-.041	.063	.101	.643	.452	.271	.236	
.075	-.314	-.330	-.300			.345	.205			
.100	-.581	-.897	-.519	-.445	-.402	.041	-.069	-.208	-.230	
.150	-.856	-.897	-.868				-.474			
.200	-.910	-.948	-.945	-.862	-.852	-.727	-.701	-.697	-.752	
.250	-.932	-.917	-.901			-.738	-.739			
.300	-.931	-.927	-.919	-.905	-.928	-.768	-.830	-.617	-.708	
.350	-.858	-.874	-.859			-.584	-.755			
.400	-.864	-.846	-.848	-.814	-.805	-.457	-.548	-.742	-.696	
.450	-.843	-.832	-.839			-.417	-.414			
.500	-.830	-.822	-.844	-.836	-.828	-.382	-.375	-.838	-.824	
.550	-.844	-.841	-.846			-.358	-.353			
.600	-.851	-.849	-.858	-.846	-.853	-.355		-.440	-.320	
.650	-.813	-.830	-.839			-.361	-.337			
.700	-.614	-.671	-.787	-.743	-.856	-.353	-.365	-.274	-.277	
.750	-.397	-.403	-.443			-.356	-.377			
.800	-.321	-.322	-.333	-.318	-.308	-.355	-.383	-.152	-.251	
.850	-.302	-.302	-.306			-.347	-.382			
.900	-.300	-.300	-.294	-.254	-.153	-.336		.026	-.161	
.950			-.296			-.336				

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CF	
-.2265	.700	.814				1.038		.0125	.811	.748
-.2115	.155	.094		-.126	-.200	-.229		.0625		.245
-.1765	.166	.118	.016	-.163	-.275	-.326	-.165	.1125		
-.1165	.297	.269	.127	-.180	-.277	-.463		.1625	-.633	-.596
-.0515	.511	.393	.180	-.061	-.187	-.237		.2375	-.890	-.817
.0235		.348		-.105		-.225	-.104	.3225		
.0985		.002	-.068	-.186	-.226	-.222		.3875		-.514
.1735		-.488		-.308		-.256		.4625		
.2485		-.717	-.562			-.248		.5375	-.347	-.341
.3235		-.774		-.544		-.345		.6125		-.333
.3985		-.689	-.745	-.654			-.622	.6875		
.4735			-.548	-.655	-.501	-.466		.7625		
.5485		-.357		-.509	-.466	-.519		.8375	-.365	-.334
.6235		-.357				-.401		.9125		
.6985		-.358	-.355	-.253	-.238	-.259	-.280	.9625	-.325	-.280
.7735		-.359	-.335	-.232	-.197					
.8485		-.346	-.310	-.202	-.155	-.155				
.9235		-.318	-.266	-.165		-.101				
.9735		-.293	-.216	-.123	-.042	-.034	-.145			
.9935		-.291			-.013	-.012				

APPENDIX

TABLE A5.- Concluded

CONFIGURATION 3

MACH = .875

ALPHA = .18

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
N									
X/C									
.025	-.010	.211	.333			.892			
.050	.022	-.007	.056	.156	.196	.644	.443	.255	.227
.075	-.227	-.241	-.195			.354	.205		
.100	-.487	-.791	-.417	-.337	-.298	.061	-.054	-.205	-.221
.150	-.748	-.791	-.751				-.452		
.200	-.795	-.836	-.825	-.738	-.733	-.795	-.666	-.692	-.730
.250	-.819	-.805	-.776			-.958	-.723		
.300	-.813	-.811	-.792	-.787	-.808	-1.088	-.911	-.605	-.736
.350	-.739	-.755	-.745			-.851	-.932		
.400	-.744	-.729	-.738	-.696	-.687	-.620	-.971	-.735	-.710
.450	-.728	-.717	-.725			-.520	-.605		
.500	-.715	-.713	-.729	-.713	-.707	-.461	-.464	-.839	-.829
.550	-.732	-.732	-.727			-.433	-.438		
.600	-.739	-.741	-.738	-.729	-.734	-.425		-.819	-.648
.650	-.743	-.743	-.732			-.430	-.431		
.700	-.810	-.790	-.789	-.784	-.785	-.429	-.467	-.525	-.491
.750	-.814	-.822	-.826			-.434	-.476		
.800	-.546	-.628	-.856	-.777	-.617	-.442	-.478	-.476	-.460
.850	-.367	-.401	-.547			-.445	-.478		
.900	-.335	-.361	-.414	-.380	-.077	-.440		-.364	-.414
.950			-.416			-.453			

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CF	
-.2265	.962	1.024				1.006		.0125	.818	.767
-.2115	.291	.217		-.091	-.201	-.274		.0625		.272
-.1765	.245	.186	.086	-.137	-.310	-.656	-.140	.1125		
-.1165	.338	.312	.166	-.163	-.267	-.478		.1625	-.588	-.552
-.0515	.539	.418	.204	-.043	-.172	-.225		.2375	1.010	-.867
.0235		.366		-.094		-.211	-.090	.3225		
.0985		.020	-.056	-.178	-.211	-.204		.3875		-.922
.1735		-.455		-.313		-.236		.4625		
.2485		-.683	-.545			-.227		.5375	-.405	-.448
.3235		-.935		-.529		-.328		.6125		-.429
.3985		-1.012	-.870	-.678			-.647	.6875		
.4735			-.925	-.763	-.532	-.456		.7625		
.5485		-.457		-.773	-.625	-.553		.8375	-.458	-.431
.6235		-.442				-.592		.9125		
.6985		-.458	-.506	-.513	-.661	-.633	-.473	.9625	-.466	-.432
.7735		-.459	-.518	-.499	-.576					
.8485		-.451	-.491	-.331	-.309	-.276				
.9235		-.428	-.382	-.219		-.143				
.9735		-.392	-.313	-.182	-.101	-.077	-.253			
.9935		-.389			-.082	-.064				

APPENDIX

TABLE A6.- AERODYNAMIC PRESSURE-COEFFICIENT DATA FOR CONFIGURATION 4
AT $x_t = 0.29c$

CONFIGURATION 4 MACH = .200 ALPHA = .06

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.137	-.196	.034			.260				
.050	-.398	-.349	-.265	-.260	-.280	-.078	-.002	.074	.109	
.075	-.522	-.541	-.497			-.402	-.237			
.100	-.655	-.655	-.620	-.601	-.655	-.596	-.425	-.278	-.255	
.150	-.620	-.655	-.620			-.601	-.513			
.200	-.487	-.502	-.482	-.492	-.532	-.519	-.402	-.296	-.284	
.250	-.423	-.418	-.418			-.399	-.307			
.300	-.408	-.384	-.369	-.403	-.398	-.338	-.263	-.205	-.284	
.350	-.433	-.348	-.359			-.278	-.227			
.400	-.365	-.332	-.343	-.346	-.348	-.210	-.193	-.210	-.203	
.450	-.332	-.332	-.332			-.140	-.174			
.500	-.328	-.328	-.330	-.332	-.337	-.084	-.147	-.200	-.217	
.550	-.328	-.326	-.330			-.040	-.137			
.600	-.330	-.332	-.337	-.324	-.343	.020		-.171	-.193	
.650	-.331	-.331	-.337			.124				
.700	-.349	-.343	-.349	-.349	-.361	.198	.189	.160	.150	
.750	-.355	-.349	-.349			.235	.263			
.800	-.349	-.343	-.343	-.384	-.378	.266	.309	.249	.300	
.850	-.325	-.319	-.337			.292	.343			
.900	-.249	-.243	-.267	-.284	-2.047	.309		.370	.362	
.950			-.125			.304	.350			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.864	-.747				.868		-.5600	
-.8350	-.174	-.415	-.388	-.302	-.302	-.267		-.3700	-.099
-.8000	-.236	-.271			-.271	-.275	-.302	-.2900	-.087
-.7400	-.111	-.143	-.178	-.174	-.182	-.189		-.2100	.049
-.6750		-.111		-.158	-.170	-.178		-.1300	.028
-.6000		-.076						-.0500	-.004
-.5250		-.061	-.092		-.131	-.283		.0300	-.094
-.4500		-.068		-.096		-.084		.1100	-.393
-.3750		-.061	-.059			-.089		.1900	
-.3000		-.061				-.099		.2700	-.294
-.2250		-.050	-.043	-.075	-.073		-.062	.3500	-.281
-.1500		-.041	-.052	-.082	-.086	-.087		.4300	-.162
-.0750			-.073	-.086	-.077	-.087		.5100	-.009
.0000		-.116				-.067			
.0750				-.095	-.061	-.087	-.165		
.1500		-.277	-.175	-.082	-.052				
.2250		-.266	-.157	-.061	-.025	-.014			
.3000		-.177	-.102	-.018		.025			
.3500		-.061	-.025	.032	.052		.025		
.3700		-.014			.081				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4

MACH = .201

ALPHA = 1.06

WING CP									
UPPER SURFACE					LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.443	-.380	-.282			.540			
.050	-.633	-.604	-.570	-.609	-.599	.199	.245	.338	.349
.075	-.770	-.775	-.779			-.119	.015		
.100	-.882	-.804	-.867	-.896	-.901	-.305	-.165	-.055	-.055
.150	-.779	-.804	-.818			-.392	-.328		
.200	-.623	-.633	-.653	-.653	-.662	-.363	-.241	-.148	-.154
.250	-.536	-.531	-.545			-.258	-.172		
.300	-.516	-.497	-.521	-.511	-.526	-.220	-.196	-.110	-.188
.350	-.428	-.441	-.446			-.172	-.127		
.400	-.446	-.424	-.422	-.431	-.441	-.129	-.112	-.127	-.131
.450	-.400	-.409	-.400			-.083	-.105		
.500	-.390	-.392	-.392	-.394	-.405	-.043	-.091	-.136	-.158
.550	-.379	-.379	-.381			-.019	-.086		
.600	-.372	-.377	-.379	-.374	-.390	.022		-.134	-.146
.650	-.386	-.386	-.386			.143			
.700	-.398	-.398	-.392	-.398	-.403	.246	.220	.184	.174
.750	-.386	-.386	-.386			.289	.291		
.800	-.374	-.380	-.380	-.398	-.403	.317	.341	.253	.317
.850	-.345	-.339	-.351			.344	.372		
.900	-.264	-.252	-.264	-.276	-1.240	.360		.382	.377
.950			-.130			.346			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-1.491	-1.502				.961		-.5600	
-.8350	-.176	-.523	-.431	-.303	-.199	-.107		-.3700	-.069
-.8000	-.311	-.353			-.195	-.172	-.288	-.2900	-.048
-.7400	-.164	-.199	-.199	-.172	-.134	-.126		-.2100	.056
-.6750		-.130		-.157	-.141	-.134		-.1300	.080
-.6000		-.083						-.0500	.140
-.5250		-.056	-.087		-.107	-.230		.0300	.121
-.4500		-.056		-.080		-.068		.1100	-.156
-.3750		-.036	-.036			-.060		.1900	
-.3000		-.023				-.069		.2700	-.120
-.2250		.006	-.001	-.032	-.041		-.036	.3500	-.178
-.1500		.025	.012	-.030	-.054	-.060		.4300	-.093
-.0750			.012	-.023	-.045	-.060		.5100	.036
.0000		-.003				-.043			
.0750				-.025	-.028	-.065	-.161		
.1500		-.126	-.061	-.016	-.012				
.2250		-.137	-.057	-.003	.012	.010			
.3000		-.077	-.019	.032		.049			
.3500		.008	.038	.077	.088		.068		
.3700		.049			.115				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4 MACH = .201 ALPHA = 2.07

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.843	-1.408	-1.291			.730				
.050	-.906	-.740	-.731	-.809	-.916	.459	.459	.540	.545	
.075	-.994	-.945	-.955			.176	.234			
.100	-1.087	-.979	-1.077	-1.087	-1.116	-.049	.055	.165	.165	
.150	-.931	-.979	-.989			-.200	-.124			
.200	-.745	-.775	-.784	-.789	-.804	-.165	-.084	.003	.003	
.250	-.658	-.662	-.658			-.127	-.055			
.300	-.604	-.594	-.614	-.614	-.628	-.100	-.093	-.016	-.061	
.350	-.521	-.532	-.536			-.072	-.040			
.400	-.521	-.497	-.502	-.506	-.519	-.045	-.045	-.060	-.060	
.450	-.476	-.474	-.474			-.026	-.038			
.500	-.452	-.448	-.446	-.456	-.463	-.002	-.038	-.076	-.103	
.550	-.428	-.431	-.435			.024	-.038			
.600	-.418	-.418	-.422	-.418	-.433	.050		-.083	-.103	
.650	-.409	-.415	-.415			.162	.105			
.700	-.409	-.409	-.409	-.409	-.421	.277	.246	.210	.201	
.750	-.403	-.392	-.386			.332	.320			
.800	-.374	-.351	-.380	-.392	-.403	.367	.365	.248	.336	
.850	-.334	-.334	-.339			.394	.394			
.900	-.252	-.246	-.246	-.264	-.269	.406		.396	.389	
.950			-.119			.384	.375			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-1.209	-1.375				.965		-.5600	
-.8350	-.180	-1.248	-.793	-.299	-.087	.047		-.3700	-.038
-.8000	-.481	-.431			-.134	-.072	-.296	-.2900	-.007
-.7400	-.184	-.215	-.230	-.176	-.103	-.103		-.2100	.063
-.6750		-.145		-.161	-.107	-.091		-.1300	.123
-.6000		-.087						-.0500	.114
-.5250		-.053	-.087		-.083	-.184		.0300	.246
-.4500		-.041		-.076		-.041		.1100	.275
-.3750		-.016	-.021			-.036		.1900	.015
-.3000		.006				-.041		.2700	.024
-.2250		.049	.032	-.012	-.019		-.012	.3500	-.091
-.1500		.093	.064	.006	-.030	-.028		.4300	-.041
-.0750			.082	.014	-.021	-.031		.5100	.073
.0000		.095				-.016			
.0750				.032	.001	-.041	-.163		
.1500		-.003	.027	.038	.014				
.2250		-.023	.019	.051	.043	.034			
.3000		.003	.047	.071		.070			
.3500		.071	.093	.117	.117		.109		
.3700		.104			.141				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4 MACH = .201 ALPHA = 3.07

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.237	-2.362	-2.230			.807				
.050	-1.168	-1.188	-1.780	-2.156	-2.195	.599	.634	.674	.686	
.075	-1.198	-1.100	-1.124			.374	.420			
.100	-1.256	-1.149	-1.144	-1.158	-1.163	.183	.235	.327	.327	
.150	-1.090	-1.149	-1.110			.003	.050			
.200	-.885	-.909	-.909	-.899	-.924	.119	.055	.107	.119	
.250	-.772	-.762	-.767			.027	.051			
.300	-.699	-.679	-.708	-.704	-.723	.012	.010	.070	.026	
.350	-.583	-.605	-.618			.015	.048			
.400	-.585	-.564	-.570	-.579	-.592	.027	.039	.017	.008	
.450	-.531	-.527	-.527			.043	.031			
.500	-.499	-.495	-.497	-.510	-.518	.060	.022	-.016	-.048	
.550	-.473	-.475	-.475			.065	.008			
.600	-.449	-.456	-.456	-.451	-.469	.086		-.043	-.069	
.650	-.434	-.440	-.440			.187	.130			
.700	-.434	-.428	-.434	-.428	-.440	.302	.268	.235	.220	
.750	-.405	-.410	-.393			.357	.338			
.800	-.370	-.375	-.375	-.387	-.393	.390	.383	.299	.352	
.850	-.317	-.317	-.329			.414	.409			
.900	-.230	-.236	-.236	-.247	-.620	.421		.409	.400	
.950			-.136			.395	.383			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.753	-.977				.953		-.5600	
-.8350	-.173	-.946	-1.306	-.320	.020	.223		-.3700	.003
-.8000	-.768	-.877			-.068	.040	-.316	-.2900	.044
-.7400	-.470	-.362	-.262	-.200	-.060	-.029		-.2100	.071
-.6750		-.122		-.176	-.080	-.037		-.1300	.160
-.6000		-.076						-.0500	.314
-.5250		-.049	-.095		-.072	-.146		.0300	.392
-.4500		-.029		-.080		-.010		.1100	.194
-.3750		.003	-.010			.005		.1900	.162
-.3000		.036				-.007		.2700	.162
-.2250		.084	.065	-.001	.001		-.002	.3500	.010
-.1500		.139	.106	.023	-.005	.000		.4300	.037
-.0750			.137	.045	.006	.000		.5100	.114
.0000		.185				.015			
.0750				.069	.030	-.012	-.164		
.1500		.104	.115	.082	.041				
.2250		.071	.100	.102	.065	.061			
.3000		.080	.115	.117		.099			
.3500		.133	.148	.159	.139		.148		
.3700		.157			.165				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4

MACH = .200

ALPHA = 4.06

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
N										
X/C										
.025	-1.847	-2.844	-2.481			.812				
.050	-1.351	-2.048	-2.515	-2.279	-2.299	.724	.724	.777	.777	
.075	-1.321	-1.410	-2.171			.538	.550			
.100	-1.356	-1.238	-1.660	-2.107	-2.147	.370	.364	.434	.451	
.150	-1.179	-1.238	-1.174			.172	.172			
.200	-.987	-1.016	-.967	-1.139	-1.080	.155	.166	.195	.195	
.250	-.859	-.859	-.835			.121	.150			
.300	-.771	-.756	-.766	-.756	-.766	.094	.094	.125	.091	
.350	-.658	-.669	-.666			.085	.109			
.400	-.636	-.612	-.614	-.599	-.612	.077	.089	.070	.061	
.450	-.573	-.569	-.564			.089	.077			
.500	-.529	-.529	-.525	-.525	-.534	.097	.061	.020	-.007	
.550	-.497	-.495	-.497			.099	.041			
.600	-.464	-.464	-.466	-.451	-.464	.113		-.016	-.038	
.650	-.436	-.447	-.441			.212	.150			
.700	-.418	-.424	-.424	-.418	-.400	.323	.284	.248	.229	
.750	-.389	-.395	-.389			.375	.354			
.800	-.348	-.348	-.359	-.365	-.348	.407	.392	.311	.351	
.850	-.283	-.295	-.301			.426	.419			
.900	-.207	-.219	-.219	-.231	-.506	.428		.411	.399	
.950			-.125			.392	.383			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.682	-1.028				.812		-.5600	
-.8350	-.216	-.920	-1.122	-.368	.090	.301		-.3700	.037
-.8000	-.749	-.908			-.045	.078	-.364	-.2900	.081
-.7400	-.686	-.624	-.344	-.255	-.072	-.010		-.2100	.076
-.6750		-.259		-.235	-.099	-.026		-.1300	.158
-.6000		-.115						-.0500	.343
-.5250		-.061	-.146		-.096	-.158		.0300	.483
-.4500		-.041		-.138		-.018		.1100	.333
-.3750		.023	-.001			.032		.1900	
-.3000		.059				.020		.2700	.253
-.2250		.111	.092	-.003	.015		.003	.3500	.083
-.1500		.168	.147	.030	.010	.025		.4300	.088
-.0750			.188	.061	.019	.027		.5100	.154
.0000		.232				.039			
.0750				.100	.043	.013	-.169		
.1500		.173	.166	.118	.059				
.2250		.129	.147	.129	.083	.081			
.3000		.122	.151	.151		.117			
.3500		.164	.175	.182	.153		.183		
.3700		.182			.179				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4

MACH = .302

ALPHA = .09

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
N									
X/C									
.025	-.126	-.179	.076			.260			
.050	-.386	-.327	-.260	-.271	-.271	-.115	-.042	.069	.080
.075	-.509	-.518	-.478			-.457	-.251		
.100	-.641	-.630	-.632	-.605	-.643	-.646	-.465	-.310	-.296
.150	-.599	-.630	-.628			-.657	-.593		
.200	-.473	-.491	-.480	-.487	-.511	-.566	-.454	-.315	-.315
.250	-.413	-.408	-.397			-.423	-.282		
.300	-.397	-.388	-.401	-.388	-.422	-.358	-.285	-.234	-.283
.350	-.440	-.348	-.353			-.285	-.238		
.400	-.382	-.331	-.340	-.341	-.349	-.217	-.205	-.227	-.235
.450	-.332	-.333	-.332			-.139	-.176		
.500	-.323	-.331	-.335	-.340	-.342	-.084	-.157	-.222	-.242
.550	-.322	-.329	-.333			-.040	-.142		
.600	-.325	-.330	-.340	-.332	-.351	.013		-.197	-.204
.650	-.342	-.334	-.347			.125	.049		
.700	-.363	-.347	-.360	-.363	-.368	.200	.193	.153	.156
.750	-.358	-.355	-.363			.235	.262		
.800	-.352	-.355	-.363	-.395	-.390	.264	.309	.241	.313
.850	-.326	-.326	-.339			.288	.339		
.900	-.246	-.251	-.267	-.291	-1.266	.307		.375	.381
.950			-.131			.300			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CF
-.8500	-.739	-.606				.864		-.5600	
-.8350	-.163	-.413	-.406	-.312	-.335	-.292		-.3700	-.106
-.8000	-.234	-.273			-.273	-.296	-.301	-.2900	-.090
-.7400	-.112	-.147	-.174	-.188	-.195	-.221		-.2100	.043
-.6750		-.110		-.172	-.179	-.202		-.1300	.016
-.6000		-.081						-.0500	-.024
-.5250		-.067	-.099		-.145	-.220		.0300	-.123
-.4500		-.073		-.103		-.126		.1100	-.412
-.3750		-.062	-.064			-.091		.1900	
-.3000		-.061				-.099		.2700	-.320
-.2250		-.047	-.054	-.075	-.098		-.074	.3500	-.289
-.1500		-.045	-.060	-.083	-.100	-.085		.4300	-.160
-.0750			-.077	-.089	-.095	-.083		.5100	-.067
.0000		-.123				-.068			
.0750				-.097	-.078	-.080	-.155		
.1500		-.294	-.190	-.096	-.061				
.2250		-.283	-.176	-.081	-.030	-.016			
.3000		-.188	-.115	-.036		.023			
.3500		-.070	-.035	.019	.056		.017		
.3700		-.020			.081				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4 MACH = .399 ALPHA = .09

WING CP									
UPPER SURFACE					LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.113	-.159	-.093			.282			
.050	-.390	-.318	-.237	-.261	-.261	-.125	-.043	.076	.088
.075	-.541	-.533	-.490			-.489	-.300		
.100	-.685	-.658	-.660	-.634	-.642	-.709	-.500	-.345	-.332
.150	-.627	-.658	-.658			-.718	-.638		
.200	-.502	-.509	-.497	-.503	-.505	-.609	-.502	-.337	-.332
.250	-.432	-.432	-.412			-.454	-.323		
.300	-.417	-.407	-.420	-.405	-.404	-.395	-.310	-.259	-.284
.350	-.406	-.369	-.370			-.310	-.264		
.400	-.400	-.356	-.362	-.366	-.376	-.238	-.226	-.248	-.248
.450	-.360	-.354	-.354			-.165	-.195		
.500	-.351	-.347	-.353	-.360	-.363	-.098	-.168	-.228	-.252
.550	-.349	-.345	-.354			-.054	-.153		
.600	-.351	-.352	-.357	-.352	-.369	.004		-.190	-.212
.650	-.359	-.359	-.370			.119	.051		
.700	-.383	-.372	-.382	-.385	-.386	.201	.201	.164	.154
.750	-.385	-.375	-.383			.241	.272		
.800	-.383	-.372	-.385	-.420	-.407	.272	.314	.266	.309
.850	-.351	-.345	-.361			.297	.349		
.900	-.264	-.260	-.282	-.308	-.824	.312		.378	.377
.950			-.133			.307	.357		

NACELLE							PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	
X/C	CP						X/C	CP
-.8500	-.786	-.568				.859		-.5600
-.8350	-.172	-.439	-.430	-.329	-.363	-.313		-.3700
-.8000	-.256	-.290			-.292	-.308	-.306	-.2900
-.7400	-.132	-.162	-.190	-.204	-.210	-.227		-.2100
-.6750		-.115		-.185	-.188	-.201		-.1300
-.6000		-.086						-.0500
-.5250		-.072	-.104		-.149	-.181		.0300
-.4500		-.082		-.106		-.139		.1100
-.3750		-.066	-.067			-.101		.1900
-.3000		-.070				-.112		.2700
-.2250		-.057	-.059	-.083	-.098		-.078	.3500
-.1500		-.052	-.065	-.087	-.099	-.101		.4300
-.0750			-.087	-.097	-.097	-.101		.5100
.0000		-.144				-.083		
.0750				-.112	-.082	-.081	-.158	
.1500		-.346	-.223	-.110	-.065			
.2250		-.326	-.206	-.092	-.037	-.028		
.3000		-.219	-.137	-.048		.015		
.3500		-.085	-.049	.015	.049		.007	
.3700		-.029			.074			

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4 MACH = .501 ALPHA = .10

WING CP									
UPPER SURFACE					LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.084	-.137	.116			.287			
.050	-.386	-.305	-.231	-.226	-.218	-.144	-.048	.047	.099
.075	-.562	-.542	-.510			-.527	-.313		
.100	-.736	-.702	-.704	-.669	-.681	-.763	-.530	-.354	-.350
.150	-.673	-.702	-.697			-.797	-.680		
.200	-.519	-.538	-.514	-.545	-.527	-.660	-.540	-.375	-.383
.250	-.456	-.449	-.432			-.491	-.341		
.300	-.446	-.427	-.439	-.430	-.432	-.421	-.335	-.287	-.330
.350	-.489	-.391	-.392			-.321	-.286		
.400	-.416	-.377	-.382	-.385	-.391	-.241	-.240	-.265	-.264
.450	-.374	-.373	-.366			-.164	-.201		
.500	-.370	-.366	-.366	-.378	-.371	-.104	-.174	-.241	-.270
.550	-.367	-.365	-.366			-.052	-.153		
.600	-.367	-.366	-.375	-.364	-.381	.007		-.199	-.225
.650	-.376	-.378	-.380			.121	.054		
.700	-.396	-.390	-.393	-.397	-.408	.198	.200	.171	.156
.750	-.397	-.394	-.393			.237	.274		
.800	-.395	-.391	-.401	-.435	-.431	.271	.318	.291	.309
.850	-.361	-.354	-.373			.299	.351		
.900	-.262	-.260	-.284	-.311	-.601	.319		.386	.383
.950			-.124			.317	.364		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.687	-.449				.899		-.5600	
-.8350	-.164	-.446	-.440	-.326	-.377	-.306		-.3700	-.110
-.8000	-.260	-.307			-.307	-.319	-.317	-.2900	-.092
-.7400	-.138	-.171	-.190	-.205	-.221	-.238		-.2100	.051
-.6750		-.117		-.184	-.194	-.210		-.1300	.033
-.6000		-.086						-.0500	.000
-.5250		-.068	-.101		-.150	-.159		-.0300	-.130
-.4500		-.077		-.112		-.146		.1100	-.510
-.3750		-.063	-.066			-.104		.1900	
-.3000		-.061				-.114		.2700	-.382
-.2250		-.046	-.051	-.088	-.104		-.082	.3500	-.324
-.1500		-.037	-.054	-.093	-.101	-.101		.4300	-.175
-.0750			-.078	-.102	-.099	-.105		.5100	-.010
.0000		-.145				-.088			
.0750				-.129	-.087	-.084	-.154		
.1500		-.379	-.251	-.129	-.075				
.2250		-.363	-.235	-.108	-.047	-.035			
.3000		-.247	-.157	-.060		.008			
.3500		-.105	-.060	.010	.045		.002		
.3700		-.045			.074				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4

MACH = .600

ALPHA = .09

WING CP									
UPPER SURFACE					LOWER SURFACE				
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
N									
X/C									
.025	-.042	-.095	.165			.306			
.050	-.352	-.263	-.198	-.197	-.188	-.124	-.058	.030	.031
.075	-.553	-.545	-.504			-.571	-.358		
.100	-.773	-.772	-.718	-.684	-.704	-.854	-.618	-.434	-.424
.150	-.724	-.772	-.776			-.906	-.816		
.200	-.550	-.570	-.535	-.593	-.560	-.769	-.690	-.449	-.453
.250	-.484	-.470	-.466			-.552	-.401		
.300	-.478	-.457	-.481	-.458	-.470	-.483	-.392	-.334	-.391
.350	-.451	-.418	-.425			-.362	-.320		
.400	-.443	-.404	-.407	-.407	-.416	-.263	-.261	-.306	-.308
.450	-.400	-.400	-.390			-.177	-.224		
.500	-.393	-.388	-.392	-.398	-.404	-.106	-.188	-.273	-.308
.550	-.392	-.389	-.396			-.056	-.168		
.600	-.393	-.392	-.403	-.397	-.412	.009		-.218	-.248
.650	-.402	-.406	-.407			.116	.061		
.700	-.426	-.420	-.422	-.430	-.436	.185	.205	.172	.155
.750	-.427	-.421	-.425			.228	.268		
.800	-.420	-.411	-.427	-.471	-.456	.261	.310	.310	.304
.850	-.376	-.368	-.386			.292	.344		
.900	-.262	-.260	-.285	-.319	-.459	.310		.385	.377
.950			-.109			.310	.368		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.583	-.273				.937		-.5600	
-.8350	-.167	-.441	-.455	-.329	-.393	-.314		-.3700	-.108
-.8000	-.269	-.322			-.326	-.343	-.339	-.2900	-.088
-.7400	-.147	-.181	-.198	-.220	-.238	-.267		-.2100	.060
-.6750		-.119		-.199	-.205	-.224		-.1300	.044
-.6000		-.087						-.0500	.013
-.5250		-.069	-.105		-.161	-.157		.0300	-.155
-.4500		-.079		-.115		-.158		.1100	-.615
-.3750		-.060	-.065			-.109		.1900	-.464
-.3000		-.060				-.124		.2700	-.464
-.2250		-.044	-.051	-.090	-.114		-.084	.3500	-.352
-.1500		-.032	-.057	-.100	-.111	-.113		.4300	-.181
-.0750			-.087	-.114	-.114	-.119		.5100	-.014
.0000		-.160				-.102			
.0750				-.154	-.108	-.099	-.154		
.1500		-.465	-.315	-.164	-.097				
.2250		-.440	-.290	-.143	-.069	-.052			
.3000		-.287	-.200	-.088		-.006			
.3500		-.130	-.086	-.010	.029		-.021		
.3700		-.066			.059				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4 MACH = .599 ALPHA = 1.10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.347	-.357	-.186			.580			
.050	-.670	-.615	-.570	-.580	-.585	.183	.231	.318	.336
.075	-.874	-.884	-.865			-.187	-.034		
.100	-1.126	-1.056	-1.136	-1.110	-1.140	-.461	-.275	-.117	-.110
.150	-1.004	-1.056	-1.060			-.580	-.487		
.200	-.758	-.780	-.784	-.784	-.794	-.510	-.369	-.236	-.232
.250	-.649	-.650	-.651			-.368	-.245		
.300	-.619	-.604	-.628	-.625	-.623	-.323	-.245	-.202	-.247
.350	-.515	-.539	-.550			-.254	-.203		
.400	-.555	-.509	-.520	-.521	-.523	-.185	-.179	-.199	-.204
.450	-.494	-.490	-.491			-.123	-.143		
.500	-.477	-.476	-.476	-.492	-.486	-.067	-.129	-.200	-.232
.550	-.466	-.462	-.469			-.033	-.119		
.600	-.461	-.460	-.468	-.457	-.471	.025		-.169	-.204
.650	-.453	-.458	-.464			.161	.087		
.700	-.474	-.464	-.470	-.472	-.479	.261	.242	.205	.185
.750	-.465	-.457	-.460			.303	.320		
.800	-.448	-.438	-.450	-.484	-.479	.339	.368	.322	.340
.850	-.393	-.384	-.403			.369	.403		
.900	-.273	-.267	-.287	-.310	-.385	.386		.420	.413
.950			-.111			.376	.397		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-1.424	-1.131				1.037		-.5600	
-.8350	-.184	-.602	-.513	-.312	-.265	-.172		-.3700	-.079
-.8000	-.352	-.399			-.241	-.229	-.336	-.2900	-.046
-.7400	-.198	-.224	-.226	-.219	-.189	-.195		-.2100	.072
-.6750		-.140		-.187	-.167	-.171		-.1300	.102
-.6000		-.091						-.0500	.166
-.5250		-.064	-.097		-.129	-.122		.0300	.100
-.4500		-.062		-.089		-.125		.1100	-.305
-.3750		-.040	-.040			-.075		.1900	
-.3000		-.026				-.087		.2700	-.217
-.2250		.006	-.001	-.041	-.072		-.038	.3500	-.251
-.1500		.037	.019	-.035	-.064	-.075		.4300	-.130
-.0750			.016	-.034	-.062	-.079		.5100	.022
.0000		-.011				-.063			
.0750				-.056	-.050	-.057	-.153		
.1500		-.243	-.141	-.062	-.039				
.2250		-.261	-.152	-.053	-.015	-.010			
.3000		-.163	-.089	-.009		.038			
.3500		-.042	-.005	.052	.077		.046		
.3700		.012			.109				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4 MACH = .600 ALPHA = 2.10

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-.770	-1.051	-.953			.771				
.050	-.987	-.846	-.830	-.836	-.844	.446	.464	.539	.547	
.075	-1.213	-1.203	-1.204			.122	.220			
.100	-1.538	-1.306	-1.536	-1.542	-1.548	-.131	-.012	.109	.118	
.150	-1.283	-1.306	-1.329			-.317	-.233			
.200	-.940	-.949	-.973	-.985	-.994	-.258	-.161	-.068	-.073	
.250	-.800	-.787	-.804			-.189	-.086			
.300	-.748	-.728	-.760	-.753	-.757	-.169	-.107	-.081	-.131	
.350	-.618	-.650	-.653			-.134	-.101			
.400	-.651	-.607	-.602	-.620	-.626	-.100	-.091	-.110	-.111	
.450	-.568	-.574	-.561			-.056	-.078			
.500	-.544	-.545	-.541	-.566	-.558	-.017	-.071	-.130	-.164	
.550	-.523	-.521	-.524			.004	-.072			
.600	-.506	-.506	-.516	-.514	-.518	.046		-.131	-.159	
.650	-.481	-.497	-.491			.180	.112			
.700	-.491	-.493	-.491	-.503	-.501	.300	.270	.230	.210	
.750	-.471	-.474	-.470			.353	.354			
.800	-.449	-.439	-.452	-.481	-.469	.393	.405	.354	.367	
.850	-.387	-.373	-.392			.424	.436			
.900	-.268	-.251	-.269	-.281	-.336	.439		.443	.433	
.950			-.110			.417	.413			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-1.155	-1.436				1.048		-.5600	
-.8350	-.160	-1.213	-.885	-.325	-.139	.024		-.3700	-.042
-.8000	-.617	-.574			-.171	-.113	-.335	-.2900	.003
-.7400	-.216	-.243	-.248	-.224	-.120	-.120		-.2100	.082
-.6750		-.147		-.190	-.133	-.121		-.1300	.152
-.6000		-.090						-.0500	.283
-.5250		-.053	-.095		-.106	-.083		.0300	.276
-.4500		-.041		-.078		-.089		.1100	-.064
-.3750		-.008	-.015			-.035		.1900	-.023
-.3000		.017				-.046		.2700	-.023
-.2250		.059	.046	-.009	-.041		-.004	.3500	-.144
-.1500		.103	.080	.010	-.030	-.040		.4300	-.061
-.0750			.098	.025	-.019	-.043		.5100	.059
.0000		.108				-.025			
.0750				.023	-.001	-.019	-.142		
.1500		-.061	-.009	.020	.008				
.2250		-.098	-.032	.025	.030	.027			
.3000		-.049	.004	.053		.070			
.3500		.041	.064	.108	.119		.102		
.3700		.084			.149				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4 MACH = .599 ALPHA = 3.10

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.240	-1.953	-2.076			.830				
.050	-1.326	-1.287	-1.442	-1.701	-1.676	.587	.605	.676	.673	
.075	-1.500	-1.417	-1.401			.343	.400			
.100	-1.802	-2.261	-1.700	-1.730	-1.729	.117	.185	.271	.295	
.150	-2.129	-2.261	-2.193			-.092	-.042			
.200	-1.108	-1.168	-1.141	-1.116	-1.174	-.023	-.004	.066	.063	
.250	-.924	-.922	-.907			-.043	.032			
.300	-.848	-.836	-.834	-.835	-.847	-.043	.001	.018	-.020	
.350	-.690	-.739	-.748			-.027	-.001			
.400	-.706	-.682	-.695	-.696	-.692	-.011	.004	-.030	-.044	
.450	-.634	-.636	-.641			.013	.005			
.500	-.596	-.600	-.602	-.608	-.597	.035	-.004	-.064	-.106	
.550	-.564	-.572	-.571			.053	-.011			
.600	-.536	-.540	-.544	-.534	-.533	.076		-.087	-.127	
.650	-.506	-.513	-.523			.198	.145			
.700	-.498	-.502	-.505	-.490	-.483	.322	.302	.249	.225	
.750	-.463	-.477	-.471			.383	.378			
.800	-.424	-.432	-.428	-.430	-.423	.423	.423	.365	.377	
.850	-.355	-.354	-.351			.449	.452			
.900	-.245	-.244	-.238	-.241	-.320	.464		.448	.440	
.950			-.107			.438	.417			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.758	-1.321				1.041		-.5600	
-.8350	-.157	-.863	-1.152	-.348	-.003	.189		-.3700	
-.8000	-.740	-.832			-.092	-.007	-.353	-.2900	.013
-.7400	-.549	-.464	-.288	-.223	-.078	-.036		-.2100	.056
-.6750		.165		-.190	-.101	-.070		-.1300	.118
-.6000		-.076						-.0500	.210
-.5250		-.035	-.090		-.085	-.051		.0300	.367
-.4500		-.018		-.071		-.060		.0100	.405
-.3750		.018	.003			-.007		.1100	.250
-.3000		.054				-.016		.1900	.132
-.2250		.106	.084	.020	-.015		.011	.2700	
-.1500		.162	.133	.049	-.002	-.004		.3500	-.132
-.0750			.170	.069	.007	-.006		.4300	-.028
.0000		.195				.010		.5100	.014
.0750				.083	.029	.017	-.151		.110
.1500		.085	.096	.085	.039				
.2250		.028	.076	.085	.061	.058			
.3000		.037	.092	.110		.100			
.3500		.106	.136	.157	.146		.149		
.3700		.140			.179				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4 MACH = .650 ALPHA = .10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/L									
.025	.002	-.060	.208			.316			
.050	-.332	-.242	-.172	-.174	-.146	-.129	-.065	.016	.036
.075	-.553	-.534	-.490			-.594	-.377		
.100	-.821	-.823	-.778	-.730	-.719	-.943	-.668	-.468	-.431
.150	-.792	-.823	-.814			-.985	-.910		
.200	-.575	-.587	-.530	-.625	-.584	-.866	-.827	-.497	-.489
.250	-.501	-.486	-.484			-.608	-.430		
.300	-.500	-.480	-.500	-.480	-.500	-.508	-.427	-.377	-.435
.350	-.450	-.436	-.442			-.374	-.338		
.400	-.461	-.420	-.421	-.426	-.431	-.270	-.281	-.326	-.326
.450	-.419	-.416	-.406			-.178	-.228		
.500	-.415	-.407	-.410	-.425	-.416	-.104	-.200	-.298	-.337
.550	-.413	-.405	-.414			-.053	-.174		
.600	-.412	-.414	-.420	-.410	-.424	.018		-.223	-.256
.650	-.420	-.423	-.423			.117	.065		
.700	-.446	-.435	-.439	-.447	-.450	.174	.207	.172	.154
.750	-.444	-.441	-.443			.216	.266		
.800	-.438	-.425	-.440	-.487	-.474	.252	.300	.290	.293
.850	-.384	-.374	-.395			.279	.335		
.900	-.255	-.252	-.280	-.313	-.363	.299		.380	.370
.950			-.095			.303	.363		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.467	-.235				.966		-.5600	
-.8350	-.162	-.438	-.457	-.326	-.385	-.310		-.3700	-.108
-.8000	-.277	-.326			-.335	-.357	-.349	-.2900	-.084
-.7400	-.154	-.182	-.203	-.227	-.247	-.280		-.2100	.070
-.6750		-.118		-.203	-.212	-.230		-.1300	.057
-.6000		-.084						-.0500	.026
-.5250		-.068	-.104		-.166	-.153		.0300	-.154
-.4500		-.076		-.111		-.163		.1100	-.695
-.3750		-.053	-.063			-.114		.1900	-.531
-.3000		-.055				-.128		.2700	
-.2250		-.035	-.045	-.087	-.118		-.084	.3500	-.360
-.1500		-.021	-.049	-.096	-.116	-.119		.4300	-.180
-.0750			-.084	-.116	-.119	-.128		.5100	-.017
.0000		-.164				-.114			
.0750				-.176	-.121	-.110	-.153		
.1500		-.540	-.365	-.195	-.113				
.2250		-.490	-.339	-.174	-.085	-.060			
.3000		-.316	-.225	-.108		-.015			
.3500		-.147	-.107	-.026	.020		-.035		
.3700		-.079			.054				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4

MACH = .700

ALPHA = .10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	.047	-.004	.257			.350			
.050	-.284	-.198	-.124	-.115	-.108	-.089	-.030	.009	.057
.075	-.521	-.515	-.452			-.546	-.360		
.100	-.829	-.944	-.759	-.730	-.715	-.938	-.668	-.489	-.484
.150	-.993	-.944	-.906			-1.111	-1.113		
.200	-.591	-.601	-.520	-.716	-.613	-1.144	-1.102	-.584	-.536
.250	-.513	-.506	-.487			-.656	-.449		
.300	-.528	-.508	-.533	-.506	-.528	-.518	-.458	-.426	-.501
.350	-.487	-.452	-.454			-.373	-.355		
.400	-.483	-.440	-.440	-.443	-.445	-.267	-.284	-.366	-.369
.450	-.439	-.434	-.428			-.176	-.233		
.500	-.434	-.427	-.430	-.435	-.421	-.102	-.198	-.320	-.370
.550	-.430	-.431	-.433			-.039	-.164		
.600	-.430	-.432	-.445	-.427	-.439	.029		-.224	-.262
.650	-.437	-.442	-.442			.098	.080		
.700	-.463	-.456	-.460	-.462	-.472	.152	.185	.172	.153
.750	-.460	-.455	-.458			.191	.232		
.800	-.444	-.437	-.453	-.495	-.492	.217	.269	.272	.275
.850	-.375	-.369	-.388			.240	.301		
.900	-.227	-.225	-.255	-.295	-.230	.267		.366	.359
.950			-.065			.273	.345		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.402	-.107				.981		-.5600	
-.8350	-.156	-.424	-.458	-.328	-.381	-.309		-.3700	-.097
-.8000	-.285	-.336			-.346	-.370	-.361	-.2900	-.075
-.7400	-.158	-.186	-.206	-.232	-.256	-.291		-.2100	.084
-.6750		-.117		-.200	-.214	-.238		-.1300	.075
-.6000		-.082						-.0500	.053
-.5250		-.063	-.101		-.167	-.151		.0300	-.131
-.4500		-.070		-.111		-.166		.1100	-.768
-.3750		-.050	-.058			-.114		.1900	-.609
-.3000		-.043				-.132		.2700	
-.2250		-.019	-.034	-.089	-.123		-.083	.3500	-.362
-.1500		-.000	-.036	-.098	-.123	-.125		.4300	-.183
-.0750			-.075	-.121	-.129	-.136		.5100	-.028
.0000		-.146				-.125			
.0750				-.205	-.141	-.123	-.148		
.1500		-.685	-.461	-.232	-.136				
.2250		-.570	-.400	-.213	-.106	-.081			
.3000		-.338	-.256	-.139		-.032			
.3500		-.161	-.125	-.046	.009		-.055		
.3700		-.100			.042				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4

MACH = .750

ALPHA = .09

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	.111	.057	.325			.412				
.050	-.203	-.131	-.045	-.020	-.022	-.016	.025	.038	.060	
.075	-.436	-.423	-.361			-.445	-.273			
.100	-.743	-1.090	-.660	-.623	-.619	-.797	-.570	-.463	-.456	
.150	-1.079	-1.090	-1.044			-.858	-1.019			
.200	-1.039	-1.081	-1.077	-.974	-1.023	-.586	-1.109	-.903	-.728	
.250	-.530	-.659	-.496			-.467	-.347			
.300	-.490	-.461	-.489	-.505	-.516	-.446	-.379	-.461	-.598	
.350	-.439	-.451	-.457			-.428	-.344			
.400	-.490	-.445	-.456	-.453	-.444	-.412	-.319	-.397	-.416	
.450	-.452	-.455	-.448			-.357	-.280			
.500	-.454	-.459	-.451	-.464	-.434	-.272	-.231	-.351	-.401	
.550	-.453	-.467	-.458			-.182	-.153			
.600	-.453	-.473	-.471	-.455	-.444	-.081		-.232	-.243	
.650	-.460	-.469	-.477			-.050	.025			
.700	-.485	-.485	-.489	-.487	-.473	-.005	.035	.145	.128	
.750	-.475	-.486	-.480			.044	.085			
.800	-.449	-.456	-.458	-.505	-.491	.099	.128	.241	.225	
.850	-.353	-.353	-.369			.134	.152			
.900	-.184	-.180	-.207	-.260	-.015	.182		.332	.310	
.950			-.009			.217	.214			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.245	.005				1.003		-.5600	
-.8350	-.147	-.411	-.456	-.335	-.355	-.311		-.3700	-.080
-.8000	-.282	-.341			-.350	-.381	-.366	-.2900	-.054
-.7400	-.158	-.189	-.204	-.232	-.262	-.306		-.2100	.119
-.6750		-.116		-.196	-.215	-.241		-.1300	.123
-.6000		-.079						-.0500	.116
-.5250		-.056	-.089		-.163	-.144		.0300	-.046
-.4500		-.058		-.093		-.164		.1100	-.626
-.3750		-.030	-.045			-.111		.1900	
-.3000		-.018				-.129		.2700	-.518
-.2250		.009	-.007	-.065	-.121		-.074	.3500	-.416
-.1500		.034	-.000	-.077	-.124	-.131		.4300	-.372
-.0750			-.030	-.106	-.136	-.144		.5100	-.268
.0000		-.088				-.137			
.0750				-.221	-.160	-.140	-.150		
.1500		-.681	-.478	-.266	-.161				
.2250		-.516	-.403	-.241	-.134	-.110			
.3000		-.360	-.270	-.161		-.065			
.3500		-.225	-.152	-.068	-.017		-.111		
.3700		-.162			.019				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4 MACH = .800 ALPHA = .10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	.184	.150	.385			.472			
.050	-.112	-.036	.051	.055	.065	.071	.090	.094	.108
.075	-.324	-.312	-.261			-.328	-.185		
.100	-.609	-.949	-.538	-.496	-.483	-.594	-.459	-.382	-.371
.150	-.933	-.949	-.911			-.555	-.877		
.200	-.932	-.973	-.979	-.927	-.942	-.396	-1.019	-.944	-.920
.250	-.927	-.923	-.895			-.371	-.500		
.300	-.901	-.887	-.900	-.871	-.877	-.362	-.368	-.888	-.958
.350	-.838	-.819	-.834			-.357	-.298		
.400	-.850	-.825	-.842	-.804	-.818	-.370	-.305	-.351	-.683
.450	-.803	-.812	-.801			-.373	-.318		
.500	-.743	-.806	-.791	-.803	-.572	-.357	-.304	-.367	-.368
.550	-.559	-.672	-.540			-.325	-.264		
.600	-.412	-.413	-.389	-.377	-.368	-.276		-.279	-.209
.650	-.402	-.396	-.402			-.208	-.162		
.700	-.446	-.426	-.443	-.461	-.457	-.160	-.136	.137	.129
.750	-.465	-.430	-.461			-.120	-.097		
.800	-.423	-.397	-.455	-.539	-.528	-.067	-.044	.261	.218
.850	-.296	-.283	-.332			-.013	-.006		
.900	-.119	-.118	-.167	-.255	-.243	.041		.359	.325
.950			-.003			.081	.099		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.110	.142				1.039		-.5600	
-.8350	-.133	-.385	-.429	-.306	-.309	-.300		-.3700	-.045
-.8000	-.273	-.332			-.352	-.385	-.360	-.2900	-.011
-.7400	-.144	-.178	-.193	-.231	-.270	-.314		-.2100	.158
-.6750		-.094		-.189	-.210	-.242		-.1300	.168
-.6000		-.052						-.0500	.171
-.5250		-.027	-.066		-.149	-.145		.0300	.025
-.4500		-.025		-.076		-.145		.1100	-.444
-.3750		.002	-.011			-.097		.1900	
-.3000		.018				-.116		.2700	-.400
-.2250		.051	.034	-.042	-.107		-.039	.3500	-.357
-.1500		.085	.042	-.051	-.111	-.126		.4300	-.358
-.0750			.017	-.082	-.127	-.147		.5100	-.332
.0000		-.017				-.150			
.0750				-.230	-.176	-.165	-.134		
.1500		-.560	-.480	-.323	-.198				
.2250		-.464	-.457	-.309	-.180	-.143			
.3000		-.295	-.270	-.194		-.096			
.3500		-.209	-.162	-.093	-.057		-.149		
.3700		-.173			-.017				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4

MACH = .800

ALPHA = 1.2

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	.129	.092	.311			.523				
.050	-.167	-.090	-.025	-.021	-.004	.129	.165	.170	.194	
.075	-.376	-.363	-.317			-.264	-.112			
.100	-.659	-1.004	-.587	-.566	-.555	-.605	-.403	-.304	-.301	
.150	-.985	-1.004	-.965			-.778	-.827			
.200	-.996	-1.020	-1.045	-.999	-1.024	-.581	-1.041	-.848	-.832	
.250	-.990	-.992	-.964			-.409	-.745			
.300	-.966	-.967	-.981	-.965	-.977	-.395	-.356	-.617	-.711	
.350	-.905	-.903	-.931			-.380	-.281			
.400	-.935	-.902	-.920	-.884	-.917	-.391	-.294	-.365	-.399	
.450	-.910	-.892	-.886			-.380	-.286			
.500	-.906	-.897	-.896	-.901	-.932	-.319	-.245	-.367	-.420	
.550	-.906	-.898	-.912			-.242	-.174			
.600	-.910	-.910	-.928	-.904	-.898	-.141		-.262	-.237	
.650	-.801	-.819	-.851			-.085	-.033			
.700	-.450	-.438	-.466	-.491	-.396	-.017	-.008	.171	.152	
.750	-.307	-.304	-.316			.044	.037			
.800	-.223	-.228	-.253	-.340	-.414	.079	.091	.298	.255	
.850	-.129	-.135	-.180			.148	.129			
.900	-.048	-.044	-.076	-.169	-.204	.179		.391	.352	
.950			-.003			.203	.202			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CF
-.8500	-.457	-.156				1.080		-.5600	
-.8350	-.157	-.450	-.433	-.297	-.292	-.243		-.3700	-.038
-.8000	-.324	-.376			-.311	-.328	-.357	-.2900	.002
-.7400	-.169	-.197	-.209	-.226	-.245	-.279		-.2100	.161
-.6750		-.100		-.181	-.194	-.215		-.1300	.182
-.6000		-.050					.693	-.0500	.208
-.5250		-.021	-.063		-.140	-.131		.0300	.075
-.4500		-.017		-.066		-.134		.1100	-.507
-.3750		.007	-.001			-.085		.1900	-.389
-.3000		.027				-.103		.2700	-.389
-.2250		.064	.047	-.022	-.086		-.019	.3500	-.380
-.1500		.100	.060	-.028	-.090	-.108		.4300	-.378
-.0750			.042	-.052	-.102	-.125		.5100	-.327
.0000		.010				-.124			
.0750				-.190	-.145	-.136	-.145		
.1500		-.579	-.451	-.283	-.167				
.2250		-.725	-.586	-.302	-.150	-.117			
.3000		-.331	-.264	-.181		-.067			
.3500		-.205	-.137	-.070	-.018		-.107		
.3700		-.153			.019				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4

MACH = .800

ALPHA = 2.2

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	.069	.014	.248			.577				
.050	-.218	-.158	-.091	-.086	-.084	.190	.222	.241	.272	
.075	-.431	-.420	-.390			-.199	-.062			
.100	-.708	-1.057	-.634	-.617	-.608	-.552	-.347	-.243	-.238	
.150	-1.038	-1.057	-1.012			-.773	-.778			
.200	-1.053	-1.086	-1.071	-1.050	-1.064	-1.072	-.978	-.762	-.668	
.250	-1.041	-1.050	-1.006			-1.032	-.862			
.300	-1.014	-1.012	-1.048	-1.034	-1.030	-.638	-.877	-.501	-.649	
.350	-.959	-.952	-1.008			-.424	-.294			
.400	-.986	-.966	-.996	-.978	-.993	-.270	-.253	-.402	-.422	
.450	-.975	-.976	-.964			-.169	-.233			
.500	-.984	-.979	-.970	-.973	-.990	-.091	-.213	-.369	-.436	
.550	-.988	-.980	-.975			-.040	-.183			
.600	-.992	-.990	-.998	-.976	-1.002	.021		-.251	-.241	
.650	-.749	-.794	-.905			.075	.060			
.700	-.464	-.457	-.482	-.589	-.551	.120	.143	.163	.128	
.750	-.335	-.327	-.334			.160	.160			
.800	-.260	-.253	-.259	-.301	-.327	.199	.239	.284	.240	
.850	-.192	-.188	-.202			.231	.290			
.900	-.139	-.145	-.150	-.153	-.156	.261		.376	.333	
.950			-.120			.255	.327			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.761	-.400				1.116		-.5600	
-.8350	-.193	-.510	-.396	-.298	-.245	-.160		-.3700	-.026
-.8000	-.355	-.408			-.267	-.268	-.355	-.2900	.019
-.7400	-.184	-.216	-.221	-.229	-.222	-.247		-.2100	.160
-.6750		-.107		-.181	-.181	-.197		-.1300	.193
-.6000		-.051						-.0500	.237
-.5250		-.017	-.057		-.130	-.121		.0300	.118
-.4500		-.013		-.058		-.125		.1100	-.447
-.3750		.016	.004			-.071		.1900	
-.3000		.036				-.090		.2700	-.349
-.2250		.073	.059	-.010	-.074	-.006	-.006	.3500	-.415
-.1500		.111	.074	-.012	-.074	-.090		.4300	-.217
-.0750			.063	-.029	-.087	-.104		.5100	-.087
.0000		.031				-.101			
.0750				-.154	-.127	-.113	-.174		
.1500		-.533	-.401	-.251	-.150				
.2250		-.826	-.606	-.316	-.145				
.3000		-.407	-.372	-.227		-.091			
.3500		-.235	-.172	-.091	-.010	-.043	-.073		
.3700		-.170			.029				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4

MACH = .850

ALPHA = .12

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
- X/C									
.025	.273	.252	.476			.502			
.050	-.003	.073	.149	.158	.165	.130	.161	.153	.168
.075	-.202	-.186	-.140			-.257	-.102		
.100	-.472	-.793	-.401	-.362	-.359	-.602	-.365	-.291	-.289
.150	-.781	-.793	-.754			-.803	-.768		
.200	-.784	-.814	-.820	-.773	-.796	-1.018	-.985	-.826	-.812
.250	-.787	-.784	-.755			-.853	-.901		
.300	-.776	-.763	-.773	-.743	-.747	-.552	-.895	-.844	-.882
.350	-.730	-.717	-.730			-.459	-.510		
.400	-.750	-.722	-.727	-.703	-.735	-.409	-.412	-.272	-.777
.450	-.724	-.721	-.714			-.400	-.400		
.500	-.719	-.717	-.730	-.724	-.761	-.397	-.389	-.460	-.705
.550	-.727	-.723	-.736			-.393	-.372		
.600	-.739	-.743	-.744	-.746	-.779	-.390		-.418	-.297
.650	-.752	-.749	-.739			-.369	-.322		
.700	-.817	-.790	-.806	-.794	-.758	-.351	-.323	-.306	-.270
.750	-.573	-.666	-.682			-.318	-.303		
.800	-.287	-.294	-.305	-.309	-.338	-.283	-.268	-.151	-.264
.850	-.198	-.187	-.192			-.260	-.177		
.900	-.153	-.148	-.144	-.154	-.176	-.199		.122	-.232
.950			-.117			-.152	-.107		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.132	.370				1.084		-.5600	
-.8350	-.128	-.297	-.354	-.248	-.244	-.244		-.3700	-.000
-.8000	-.239	-.292			-.310	-.428	-.298	-.2900	.037
-.7400	-.116	-.152	-.170	-.215	-.260	-.330		-.2100	.216
-.6750		-.057		-.165	-.192	-.226		-.1300	.224
-.6000		-.014						-.0500	.225
-.5250		.011	-.028		-.126	-.118		.0300	.084
-.4500		.016		-.040		-.119		.1100	-.479
-.3750		.046	.022			-.072		.1900	
-.3000		.068				-.092		.2700	-.341
-.2250		.108	.088	.002	-.076		.002	.3500	-.440
-.1500		.147	.099	-.008	-.083	-.102		.4300	-.393
-.0750			.076	-.041	-.104	-.128		.5100	-.376
.0000		.052				-.143			
.0750				-.212	-.193	-.179	-.123		
.1500		-.532	-.447	-.349	-.262				
.2250		-.835	-.689	-.495	-.332	-.271			
.3000		-1.060	-.889	-.629		-.292			
.3500		-.689	-.770	-.599	-.305		-.574		
.3700		-.525			-.227				

APPENDIX

TABLE A6.- Continued

CONFIGURATION 4 MACH = .850 ALPHA = 1.2

WING CP									
UPPER SURFACE					LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	.238	.211	.416			.548			
.050	-.035	.038	.098	.103	.113	.182	.206	.202	.224
.075	-.237	-.221	-.185			-.195	-.059		
.100	-.501	-.829	-.436	-.413	-.403	-.531	-.326	-.244	-.242
.150	-.812	-.829	-.797			-.747	-.729		
.200	-.820	-.858	-.866	-.827	-.854	-.993	-.949	-.782	-.764
.250	-.824	-.835	-.815			-.961	-.883		
.300	-.822	-.815	-.832	-.813	-.818	-.672	-.938	-.792	-.824
.350	-.777	-.773	-.785			-.541	-.618		
.400	-.802	-.776	-.790	-.760	-.793	-.474	-.452	-.821	-.714
.450	-.783	-.772	-.776			-.442	-.413		
.500	-.783	-.778	-.787	-.797	-.825	-.430	-.384	-.925	-.863
.550	-.787	-.785	-.802			-.424	-.359		
.600	-.794	-.798	-.820	-.803	-.840	-.416		-.365	-.281
.650	-.773	-.800	-.808			-.376	-.307		
.700	-.440	-.503	-.664	-.761	-.574	-.331	-.321	-.255	-.245
.750	-.291	-.277	-.303			-.291	-.304		
.800	-.236	-.211	-.216	-.254	-.248	-.247	-.268	-.096	-.239
.850	-.198	-.170	-.174			-.197	-.215		
.900	-.170	-.143	-.139	-.146	-.116	-.147		.201	-.176
.950			-.119			-.108	-.098		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.107	.169				1.121		-.5600	
-.8350	-.137	-.354	-.383	-.244	-.245	-.197		-.3700	.007
-.8000	-.254	-.311			-.274	-.306	-.305	-.2900	.046
-.7400	-.135	-.166	-.179	-.210	-.245	-.296		-.2100	.216
-.6750		-.061		-.163	-.182	-.212		-.1300	.236
-.6000		-.011						-.0500	.255
-.5250		.018	-.028		-.120	-.114		.0300	.125
-.4500		.024		-.037		-.117		.1100	-.432
-.3750		.053	.029			-.066		.1900	
-.3000		.076				-.087		.2700	-.313
-.2250		.116	.099	.010	-.069		.016	.3500	-.525
-.1500		.157	.112	.004	-.075	-.097		.4300	-.394
-.0750			.092	-.024	-.094	-.121		.5100	-.374
.0000		.074				-.135			
.0750				-.184	-.176	-.169	-.117		
.1500		-.496	-.408	-.314	-.242				
.2250		-.794	-.650	-.456	-.309	-.259			
.3000		-1.051	-.848	-.587		-.281			
.3500		-.830	-.912	-.578	-.278		-.589		
.3700		-.595			-.211				

APPENDIX

TABLE A6.- Concluded

CONFIGURATION 4

MACH = .874

ALPHA = .12

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	.315	.290	.503			.528				
.050	.050	.114	.186	.202	.205	.167	.196	.182	.202	
.075	-.152	-.139	-.091			-.209	-.056			
.100	-.410	-.729	-.344	-.310	-.300	-.540	-.318	-.249	-.244	
.150	-.711	-.729	-.689			-.752	-.711			
.200	-.717	-.749	-.756	-.707	-.734	-.951	-.920	-.773	-.752	
.250	-.722	-.723	-.696			-.933	-.870			
.300	-.717	-.703	-.714	-.685	-.692	-.956	-.913	-.800	-.832	
.350	-.671	-.663	-.675			-.655	-.951			
.400	-.692	-.669	-.680	-.650	-.676	-.567	-.607	-.827	-.750	
.450	-.674	-.671	-.667			-.519	-.521			
.500	-.670	-.676	-.682	-.683	-.711	-.491	-.500	-.913	-.862	
.550	-.679	-.683	-.697			-.490	-.488			
.600	-.694	-.700	-.703	-.700	-.733	-.497		-.461	-.430	
.650	-.707	-.705	-.701			-.494	-.431			
.700	-.773	-.752	-.761	-.755	-.763	-.475	-.433	-.426	-.389	
.750	-.800	-.793	-.794			-.455	-.421			
.800	-.431	-.395	-.405	-.427	-.429	-.427	-.409	-.398	-.398	
.850	-.276	-.256	-.256			-.386	-.373			
.900	-.237	-.230	-.233	-.249	-.265	-.345		-.301	-.382	
.950			-.231			-.300	-.268			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.216	.416				1.100		-.5600	
-.8350	-.152	-.262	-.324	-.215	-.189	-.203		-.3700	.029
-.8000	-.216	-.271			-.536	-.607	-.343	-.2900	.070
-.7400	-.102	-.136	-.156	-.203	-.261	-.336		-.2100	.244
-.6750		-.037		-.146	-.176	-.205		-.1300	.256
-.6000		.008						-.0500	.261
-.5250		.036	-.005		-.105	-.097		.0300	.122
-.4500		.043		-.014		-.096		.1100	-.424
-.3750		.075	.034			-.047		.1900	
-.3000		.100				-.064		.2700	-.293
-.2250		.140	.120	.032	-.045		.030	.3500	-.715
-.1500		.181	.130	.023	-.053	-.072		.4300	-.498
-.0750			.109	-.010	-.075	-.095		.5100	-.467
.0000		.090		.001		-.110			
.0750				-.179	-.162	-.145	-.153		
.1500		-.478	-.397	-.314	-.231				
.2250		-.776	-.636	-.457	-.304	-.247			
.3000		-1.015	-.834	-.593		-.298			
.3500		-1.049	-.912	-.634	-.373		-.626		
.3700		-1.004			-.317				

APPENDIX

TABLE A7.- AERODYNAMIC PRESSURE-COEFFICIENT DATA FOR CONFIGURATION 5
AT $x_t = 0.29c$

CONFIGURATION 5 MACH = .201 ALPHA = .06

WING CP									
UPPER SURFACE					LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.956	-.970	-.862			-.493	-.289		
.050	-.769	-.760	-.730	-.681	-.691	-.593	-.394	-.271	-.265
.075	-.661	-.652	-.627			-.558	-.423		
.100	-.558	-.465	-.544	-.563	-.534	-.523	-.417	-.295	-.248
.150	-.451	-.465	-.455			-.482	-.400		
.200	-.426	-.416	-.416	-.421	-.436	-.488	-.336	-.265	-.230
.250	-.382	-.372	-.397			-.356	-.279		
.300	-.387	-.372	-.362	-.402	-.397	-.313	-.284	-.226	-.254
.350	-.333	-.338	-.329			-.248	-.204		
.400	-.331	-.327	-.324	-.331	-.331	-.195	-.185	-.202	-.195
.450	-.329	-.327	-.318			-.132	-.171		
.500	-.322	-.320	-.322	-.327	-.329	-.082	-.145	-.195	-.209
.550	-.320	-.320	-.324			-.043	-.134		
.600	-.322	-.322	-.333	-.320	-.331	.017	-.110	-.168	-.185
.650	-.330	-.336	-.336			.142	.043		
.700	-.353	-.347	-.347	-.353	-.359	.192	.197	.166	.154
.750	-.365	-.353	-.353			.228	.269		
.800	-.359	-.353	-.359	-.388	-.382	.257	.319	.214	.300
.850	-.330	-.318	-.341			.286	.353		
.900	-.248	-.242	-.265	-.289	-.306	.314		.370	.365
.950						.319	.355		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.430	.365				.687		-.5600	
-.8350			-.449		-.452			-.3700	
-.8000		-.216	-.247		-.262	-.251	-.251	-.2900	.085
-.7400	-.057		-.162	-.173		-.181		-.2100	.005
-.6750		-.092	-.142		-.162	-.162		-.1300	-.009
-.6000		-.076		-.142		-.158	-.138	-.0500	.020
-.5250		-.092	-.096	-.103	-.123	-.119		.0300	-.212
-.4500		-.084		-.080		-.092		.1100	-.377
-.3750		-.059	-.059		-.089	-.092		.1900	-.418
-.3000		-.059				-.101		.2700	-.304
-.2250		-.046	-.050		-.086		-.077	.3500	-.256
-.1500		-.043	-.059	-.052	-.086	-.089		.4300	-.133
-.0750		-.077		-.086		-.089		.5100	.008
.0000		-.145				-.072			
.0750		-.237	-.154	-.093			-.094		
.1500			-.170		-.052	-.041			
.2250		-.249	-.156	-.068	-.025	-.017			
.3000		-.165	-.100	-.025		.022			
.3500		-.055	-.025	.028	.054	.068	.022		
.3700		-.010		.054	.078	.085			

APPENDIX

TABLE A7.- Continued

CONFIGURATION 5

MACH = .202

ALPHA = 1.06

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
N									
X/C									
.025	-1.125	-1.218	-1.120			-.235	-.101		
.050	-.906	-.921	-.886	-.882	-.886	-.374	-.241	-.165	-.107
.075	-.765	-.769	-.745			-.398	-.281		
.100	-.653	-.526	-.667	-.667	-.696	-.363	-.281	-.183	-.136
.150	-.531	-.526	-.540			-.363	-.299		
.200	-.487	-.482	-.497	-.492	-.521	-.386	-.281	-.183	-.165
.250	-.438	-.428	-.433			-.285	-.220		
.300	-.443	-.414	-.419	-.419	-.424	-.261	-.227	-.163	-.212
.350	-.379	-.383	-.383			-.206	-.158		
.400	-.368	-.372	-.370	-.370	-.379	-.158	-.146	-.160	-.165
.450	-.361	-.366	-.353			-.100	-.136		
.500	-.355	-.355	-.349	-.361	-.361	-.064	-.122	-.165	-.191
.550	-.349	-.346	-.351			-.026	-.115		
.600	-.349	-.344	-.355	-.349	-.361	.024	-.088	-.151	-.170
.650	-.351	-.363	-.351			.141	.060		
.700	-.374	-.363	-.363	-.369	-.380	.224	.207	.176	.167
.750	-.374	-.369	-.363			.253	.284		
.800	-.369	-.363	-.363	-.392	-.392	.281	.334	.198	.312
.850	-.334	-.328	-.339			.312	.367		
.900	-.247	-.247	-.264	-.281	-.299	.334		.379	.374
.950						.334	.362		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.199	.161				.785		-.5600	
-.8350			-.693		-.380			-.3700	
-.8000		-.257	-.269		-.230	-.215	-.257	-.2900	.080
-.7400	-.080		-.165	-.168		-.161		-.2100	.031
-.6750		-.103	-.145		-.145	-.141		-.1300	.043
-.6000		-.087		-.134		-.138	-.141	-.0500	.101
-.5250		-.087	-.091	-.095	-.107	-.103		.0300	-.086
-.4500		-.080		-.072		-.080		.1100	-.267
-.3750		-.050	-.050		-.077	-.074		.1900	-.329
-.3000		-.039				-.082		.2700	-.211
-.2250		-.012	-.028		-.075		-.050	.3500	-.214
-.1500		-.012	-.028	-.041	-.070	-.072		.4300	-.106
-.0750		-.030		-.059		-.070		.5100	.034
.0000		-.086				-.058			
.0750		-.171	-.097	-.064			-.055		
.1500			-.115		-.039	-.026			
.2250		-.194	-.113	-.039	-.010	-.021			
.3000		-.122	-.059	.003		.036			
.3500		-.021	.003	.053	.071	.087	.051		
.3700		.021		.075	.095	.104			

APPENDIX

TABLE A7.- Continued

CONFIGURATION 5 MACH = .202 ALPHA = 2.05

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C										
.025	-1.291	-1.442	-1.393				-.008	.107		
.050	-1.052	-1.076	-1.081	-1.076	-1.101		-.194	-.090	.026	.049
.075	-.882	-.911	-.886				-.241	-.159		
.100	-.765	-.614	-.769	-.828	-.862		-.218	-.159	-.067	-.032
.150	-.614	-.614	-.618				-.258	-.200		
.200	-.560	-.555	-.589	-.560	-.604		-.287	-.183	-.090	-.078
.250	-.501	-.501	-.501				-.232	-.153		
.300	-.492	-.477	-.472	-.487	-.506		-.208	-.175	-.112	-.130
.350	-.422	-.433	-.430				-.158	-.117		
.400	-.411	-.413	-.411	-.413	-.426		-.117	-.112	-.119	-.119
.450	-.398	-.400	-.396				-.074	-.105		
.500	-.385	-.385	-.385	-.394	-.392		-.036	-.091	-.134	-.148
.550	-.381	-.377	-.383				-.012	-.084		
.600	-.374	-.377	-.383	-.381	-.385		.024	-.067	-.127	-.136
.650	-.369	-.380	-.380				.153	.074		
.700	-.386	-.380	-.380	-.386	-.392		.248	.224	.188	.188
.750	-.386	-.374	-.380				.286	.305		
.800	-.374	-.374	-.380	-.398	-.392		.315	.348	.210	.336
.850	-.345	-.334	-.345				.346	.381		
.900	-.252	-.252	-.264	-.281	-.293		.369		.391	.391
.950							.367	.369		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.060	-.103				.877		-.5600	
-.8350			-.689		-.311			-.3700	
-.8000		-.296	-.307		-.195	-.165	-.242	-.2900	.068
-.7400	-.107		-.184	-.172		-.145		-.2100	.034
-.6750		-.111	-.188		-.126	-.118		-.1300	.072
-.6000		-.087		-.138		-.118	-.126	-.0500	.166
-.5250		-.087	-.091	-.091	-.099	-.087		.0300	.024
-.4500		-.064		-.064		-.060		.1100	-.183
-.3750		-.041	-.039		-.066	-.065		.1900	-.264
-.3000		-.021				-.070		.2700	-.132
-.2250		.005	-.003		-.061		-.038	.3500	-.168
-.1500		.029	.010	-.019	-.055	-.067		.4300	-.086
-.0750		.029		-.032		-.060		.5100	.048
.0000		-.014				-.045			
.0750		-.090	-.041	-.032			-.031		
.1500			-.068		-.010	-.012			
.2250		-.135	-.066	-.008	.016	.010			
.3000		-.059	-.023	.029		.051			
.3500		.014	.036	.079	.099	.096	.070		
.3700		.049		.101	.119	.113			

APPENDIX

TABLE A7.- Continued

CONFIGURATION 5

MACH = .201

ALPHA = 3.05

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
N										
X/C										
.025	-1.485	-1.658	-1.628			.202	.278			
.050	-1.179	-1.317	-1.277	-1.356	-1.322	-.020	.068	.155	.155	
.075	-.986	-.971	-.991			-.114	-.043			
.100	-.853	-.705	-.853	-.897	-.912	-.108	-.073	.015	.039	
.150	-.680	-.705	-.710			-.185	-.132			
.200	-.626	-.636	-.651	-.636	-.665	-.185	-.108	-.049	-.043	
.250	-.562	-.562	-.557			-.174	-.096			
.300	-.537	-.522	-.517	-.537	-.547	-.167	-.123	-.065	-.096	
.350	-.473	-.477	-.479			-.121	-.077			
.400	-.449	-.449	-.455	-.462	-.464	-.082	-.072	-.084	-.084	
.450	-.433	-.429	-.431			-.048	-.065			
.500	-.418	-.416	-.412	-.427	-.427	-.014	-.058	-.108	-.118	
.550	-.405	-.407	-.403			.005	-.060			
.600	-.396	-.401	-.401	-.396	-.409	.034	-.050	-.106	-.123	
.650	-.396	-.396	-.396			.150	.090			
.700	-.414	-.402	-.402	-.402	-.414	.266	.235	.203	.186	
.750	-.402	-.396	-.390			.317	.312			
.800	-.390	-.379	-.385	-.408	-.402	.353	.360	.169	.331	
.850	-.349	-.349	-.355			.379	.391			
.900	-.255	-.255	-.267	-.285	-.296	.399		.387	.389	
.950						.384	.372			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C								X/C	CP
-.8500	-.400	-.393				.919		-.5600	
-.8350			-.623		-.248			-.3700	
-.8000		-.330	-.338		-.151	-.115	-.240	-.2900	.054
-.7400	-.123		-.197	-.170		-.104		-.2100	.049
-.6750		-.119	-.158		-.108	-.088		-.1300	.113
-.6000		-.088		-.127	-.100	-.100	-.127	-.0500	.227
-.5250		-.080	-.088	-.088	-.084	-.069		.0300	.118
-.4500		-.057		-.057		-.041		.1100	-.087
-.3750		-.030	-.028		-.048	-.046		.1900	-.177
-.3000		-.005				-.055		.2700	-.058
-.2250		.026	.021		-.046		-.019	.3500	-.119
-.1500		.057	.039	-.005	-.041	-.041		.4300	-.053
-.0750		.065		-.007		-.043		.5100	.069
.0000		.048				-.029			
.0750		-.014	.004	-.003			.006		
.1500			-.019		-.001	.003			
.2250		-.082	-.019	.024	.024	.025			
.3000		-.032	.019	.052		.064			
.3500		.048	.070	.094	.101	.113	.098		
.3700		.083		.118	.125	.132			

APPENDIX

TABLE A7.- Continued

CONFIGURATION 5 MACH = .201 ALPHA = 4.05

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
N										
X/C										
.025	-1.655	-1.897	-1.947			.378	.419			
.050	-1.314	-1.388	-1.413	-1.329	-1.690	.138	.220	.273	.273	
.075	-1.106	-1.116	-1.121			.039	.091			
.100	-.963	-.775	-.978	-1.057	-1.077	.009	.033	.132	.132	
.150	-.765	-.775	-.795			-.085	-.026			
.200	-.686	-.686	-.736	-.716	-.755	-.108	-.049	.009	.027	
.250	-.617	-.602	-.622			-.101	-.031			
.300	-.587	-.578	-.568	-.597	-.607	-.106	-.062	-.009	-.055	
.350	-.511	-.517	-.515			-.072	-.024			
.400	-.484	-.487	-.487	-.493	-.511	-.038	-.029	-.046	-.050	
.450	-.460	-.460	-.452			-.014	-.029			
.500	-.447	-.447	-.445	-.454	-.454	.008	-.029	-.077	-.096	
.550	-.430	-.432	-.432			.027	-.029			
.600	-.417	-.419	-.423	-.423	-.427	.049	-.021	-.079	-.099	
.650	-.415	-.421	-.415			.165	.109			
.700	-.421	-.415	-.415	-.421	-.426	.286	.252	.206	.204	
.750	-.415	-.409	-.409			.339	.327			
.800	-.403	-.391	-.391	-.415	-.415	.373	.373	.153	.339	
.850	-.362	-.350	-.350			.397	.402			
.900	-.261	-.261	-.261	-.273	-.291	.414		.399	.395	
.950						.392	.385			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.557	-.585				.952		-.5600	
-.8350			-.690		-.182			-.3700	
-.8000		-.358	-.350		-.120	-.073	-.249	-.2900	.045
-.7400	-.147		-.209	-.174		-.073		-.2100	.057
-.6750		-.131	-.170		-.096	-.061		-.1300	.132
-.6000		-.092		-.131		-.073	-.135	-.0500	.276
-.5250		-.080	-.088	-.088	-.073	-.053		.0300	.220
-.4500		-.053		-.057		-.026		.1100	-.002
-.3750		-.012	-.021		-.034	-.031		.1900	-.090
-.3000		.017				-.041		.2700	.018
-.2250		.057	.035		-.028		-.014	.3500	-.077
-.1500		.094	.063	.013	-.023	-.031		.4300	-.026
-.0750		.114		.015		-.031		.5100	.081
.0000		.099				-.016			
.0750		.044	.059	.028			.020		
.1500			.037		.017	.015			
.2250		-.021	.032	.055	.044	.035			
.3000		.021	.057	.079		.074			
.3500		.086	.101	.119	.116	.125	.111		
.3700		.114		.141	.143	.140			

APPENDIX

TABLE A7.- Continued

CONFIGURATION 5 MACH = .200 ALPHA = 5.06

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.825	-2.198	-2.347			.510	.563			
.050	-1.452	-1.377	-1.382	-1.457	-1.397	.310	.292	.392	.416	
.075	-1.223	-1.263	-1.278			.169	.204			
.100	-1.069	-.895	-1.094	-1.168	-1.183	.092	.133	.198	.233	
.150	-.865	-.895	-.885			-.002	.051			
.200	-.765	-.780	-.800	-.800	-.820	-.002	.027	.069	.092	
.250	-.676	-.676	-.691			-.033	.032			
.300	-.646	-.636	-.631	-.656	-.671	-.048	.003	.028	.004	
.350	-.560	-.556	-.578			-.024	.020			
.400	-.529	-.527	-.534	-.534	-.549	.003	.008	-.007	-.014	
.450	-.501	-.498	-.492			.023	.008			
.500	-.476	-.474	-.476	-.481	-.494	.037	.003	-.033	-.063	
.550	-.461	-.456	-.465			.045	-.007			
.600	-.443	-.441	-.450	-.443	-.452	.071	-.004	-.060	-.080	
.650	-.435	-.435	-.435			.176	.122			
.700	-.447	-.447	-.435	-.435	-.441	.298	.264	.227	.210	
.750	-.417	-.417	-.417			.354	.337			
.800	-.399	-.399	-.399	-.417	-.417	.390	.380	.139	.346	
.850	-.358	-.352	-.358			.410	.412			
.900	-.263	-.257	-.257	-.269	-.281	.424		.402	.402	
.950						.400	.388			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.313	-.462				.958		-.5600	
-.8350			-1.171		-.140			-.3700	
-.8000		-.675	-.415		-.089	-.014	-.252	-.2900	.035
-.7400	-.203		-.234	-.175		-.045		-.2100	.070
-.6750		-.116	-.183		-.077	-.034		-.1300	.158
-.6000		-.085		-.136		-.057	-.140	-.0500	.310
-.5250		-.069	-.089	-.089	-.061	-.034		.0300	.296
-.4500		-.049		-.053		-.006		.1100	.097
-.3750		-.007	-.014		-.028	.008		.1900	-.002
-.3000		.026				-.021		.2700	.094
-.2250		.073	.051		-.018		-.004	.3500	-.031
-.1500		.122	.089	.020	-.009	-.009		.4300	.013
-.0750		.155		.031		-.011		.5100	.107
.0000		.149				.003			
.0750		.102	.106	.055			.057		
.1500			.089		.033	.035			
.2250		.028	.068	.073	.055	.052			
.3000		.055	.091	.102		.089			
.3500		.115	.129	.144	.135	.141	.138		
.3700		.142		.160	.160	.163			

APPENDIX

TABLE A7.- Continued

CONFIGURATION 5

MACH = .601

ALPHA = .10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-1.080	-1.019	-.918			-.647	-.484		
.050	-.867	-.833	-.806	-.751	-.771	-.824	-.606	-.434	-.429
.075	-.743	-.750	-.735			-.753	-.639		
.100	-.638	-.521	-.654	-.639	-.633	-.717	-.615	-.430	-.410
.150	-.512	-.521	-.528			-.666	-.594		
.200	-.486	-.488	-.517	-.488	-.492	-.630	-.586	-.395	-.361
.250	-.445	-.448	-.435			-.494	-.368		
.300	-.462	-.455	-.443	-.449	-.449	-.430	-.375	-.275	-.308
.350	-.404	-.402	-.408			-.325	-.308		
.400	-.397	-.386	-.398	-.408	-.400	-.237	-.260	-.285	-.289
.450	-.392	-.383	-.391			-.152	-.229		
.500	-.388	-.378	-.391	-.405	-.390	-.087	-.187	-.273	-.308
.550	-.386	-.384	-.400			-.031	-.164		
.600	-.386	-.389	-.408	-.394	-.405	.050	-.122	-.221	-.246
.650	-.401	-.396	-.410			.128	.060		
.700	-.432	-.411	-.425	-.428	-.431	.164	.207	.182	.165
.750	-.435	-.416	-.430			.206	.272		
.800	-.427	-.413	-.434	-.471	-.457	.249	.318	.304	.315
.850	-.382	-.367	-.398			.288	.351		
.900	-.259	-.259	-.290	-.323	-.343	.313		.407	.394
.950						.315	.374		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.601	.550				.792		-.5600	
-.8350			-.370		-.417			-.3700	
-.8000		-.240	-.266		-.315	-.312	-.291	-.2900	.100
-.7400	-.076		-.189	-.213		-.239		-.2100	.019
-.6750		-.091	-.159		-.195	-.204		-.1300	.017
-.6000		-.089		-.167		-.190	-.165	-.0500	.056
-.5250		-.103	-.110	-.118	-.148	-.146		.0300	-.348
-.4500		-.094		-.092		-.112		.1100	-.595
-.3750		-.060	-.065		-.107	-.113		.1900	-.618
-.3000		-.052				-.125		.2700	-.473
-.2250		-.032	-.050		-.112		-.088	.3500	-.322
-.1500		-.027	-.062	-.100	-.112	-.114		.4300	-.157
-.0750		-.074		-.116		-.120		.5100	-.005
.0000		-.198				-.103			
.0750		-.365	-.270	-.162			-.158		
.1500			-.317		-.098	-.074			
.2250		-.391	-.286	-.143	-.067	-.051			
.3000		-.258	-.188	-.082		-.003			
.3500		-.117	-.083	-.009	.033	.054	-.019		
.3700		-.061		.025	.064	.074			

APPENDIX

TABLE A7.- Continued

CONFIGURATION 5

MACH = .600

ALPHA = 1.09

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
N										
X/C										
.025	-1.366	-1.371	-1.245			-.371	-.220			
.050	-1.055	-1.106	-1.031	-.991	-1.029	-.561	-.367	-.259	-.222	
.075	-.882	-.894	-.901			-.572	-.434			
.100	-.764	-.634	-.768	-.816	-.789	-.544	-.465	-.311	-.279	
.150	-.626	-.634	-.623			-.534	-.468			
.200	-.579	-.576	-.608	-.577	-.591	-.521	-.457	-.300	-.296	
.250	-.518	-.513	-.517			-.409	-.278			
.300	-.531	-.510	-.512	-.525	-.527	-.366	-.299	-.211	-.248	
.350	-.456	-.459	-.470			-.282	-.245			
.400	-.443	-.438	-.448	-.447	-.455	-.207	-.209	-.240	-.239	
.450	-.431	-.433	-.427			-.129	-.186			
.500	-.426	-.426	-.420	-.436	-.434	-.078	-.156	-.238	-.272	
.550	-.425	-.420	-.419			-.032	-.143			
.600	-.423	-.422	-.428	-.424	-.435	.039	-.110	-.198	-.228	
.650	-.427	-.423	-.430			.155	.068			
.700	-.452	-.438	-.438	-.451	-.459	.214	.221	.197	.174	
.750	-.451	-.446	-.434			.244	.294			
.800	-.441	-.432	-.437	-.484	-.475	.289	.340	.311	.330	
.850	-.393	-.378	-.395			.325	.379			
.900	-.269	-.266	-.288	-.321	-.344	.349		.419	.408	
.950						.347	.388			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.438	.392				.870		-.5600	
-.8350			-.409		-.397			-.3700	
-.8000		-.283	-.294		-.284	-.261	-.291	-.2900	.093
-.7400	-.098		-.195	-.211		-.201		-.2100	.038
-.6750		-.107	-.157		-.177	-.174		-.1300	.062
-.6000		-.092		-.159		-.169	-.161	-.0500	.128
-.5250		-.100	-.097	-.113	-.131	-.129		.0300	.184
-.4500		-.088		-.084		-.096		.1100	.448
-.3750		-.045	-.051		-.089	-.092		.1900	.513
-.3000		-.033				-.104		.2700	.352
-.2250		-.010	-.020		-.087		-.059	.3500	.285
-.1500		.008	-.018	-.071	-.088	-.095		.4300	.136
-.0750		-.014		-.081		-.100		.5100	.026
.0000		-.112				-.084			
.0750		-.262	-.175	-.108			-.105		
.1500			-.224		-.073	-.054			
.2250		-.317	-.209	-.096	-.044	-.030			
.3000		-.202	-.133	-.045		.017			
.3500		-.075	-.039	.021	.057	.074	.017		
.3700		-.021		.053	.089	.095			

APPENDIX

TABLE A7.- Continued

CONFIGURATION 5

MACH = .601

ALPHA = 2.09

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-1.840	-1.792	-1.734			-.119	.037		
.050	-1.316	-1.514	-1.383	-1.492	-1.377	-.337	-.181	-.092	-.070
.075	-1.060	-1.010	-1.021			-.392	-.284		
.100	-.911	-.736	-.881	-.912	-.910	-.379	-.312	-.176	-.160
.150	-.734	-.736	-.743			-.424	-.362		
.200	-.667	-.662	-.686	-.664	-.711	-.401	-.329	-.210	-.191
.250	-.598	-.593	-.591			-.332	-.238		
.300	-.606	-.580	-.580	-.588	-.590	-.301	-.229	-.142	-.187
.350	-.515	-.521	-.524			-.231	-.193		
.400	-.499	-.498	-.497	-.497	-.494	-.165	-.172	-.194	-.201
.450	-.484	-.481	-.475			-.096	-.151		
.500	-.469	-.467	-.463	-.477	-.467	-.057	-.124	-.201	-.230
.550	-.459	-.460	-.461			-.025	-.115		
.600	-.453	-.454	-.461	-.454	-.460	.032	-.086	-.171	-.203
.650	-.455	-.458	-.457			.169	.088		
.700	-.477	-.465	-.465	-.465	-.469	.251	.248	.210	.188
.750	-.472	-.460	-.457			.294	.321		
.800	-.454	-.445	-.450	-.483	-.473	.326	.368	.320	.345
.850	-.395	-.386	-.402			.360	.404		
.900	-.271	-.268	-.288	-.309	-.334	.388		.430	.417
.950						.383	.400		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.259	.223				.940		-.5600	
-.8350			-.490		-.333			-.3700	
-.8000		-.325	-.330		-.239	-.204	-.287	-.2900	.085
-.7400	-.119		-.210	-.207		-.164		-.2100	.056
-.6750		-.116	-.167		-.157	-.151		-.1300	.107
-.6000		-.094		-.151		-.148	-.153	-.0500	.204
-.5250		-.094	-.095	-.099	-.116	-.112		.0300	-.039
-.4500		-.077		-.068		-.077		.1100	-.300
-.3750		-.036	-.038		-.071	-.076		.1900	-.390
-.3000		-.016				-.087		.2700	-.237
-.2250		.017	.007		-.068		-.039	.3500	-.234
-.1500		.046	.020	-.035	-.064	-.074		.4300	-.110
-.0750		.045		-.039		-.077		.5100	.047
.0000		-.029				-.062			
.0750		-.159	-.099	-.061			-.059		
.1500			-.149		-.043	-.035			
.2250		-.243	-.153	-.050	-.016	-.012			
.3000		-.150	-.087	-.010		.035			
.3500		-.028	-.001	.050	.078	.095	.048		
.3700		.018		.081	.108	.116			

APPENDIX

TABLE A7.- Continued

CONFIGURATION 5 MACH = .601 ALPHA = 3.08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-2.078	-2.266	-2.212			.100	.192		
.050	-2.007	-2.084	-1.989	-2.052	-1.950	-.148	-.033	.067	.088
.075	-1.184	-1.146	-1.252			-.218	-.131		
.100	-1.011	-.836	-.952	-.990	-.963	-.257	-.175	-.040	-.039
.150	-.824	-.836	-.825			-.303	-.235		
.200	-.744	-.750	-.766	-.748	-.767	-.291	-.216	-.116	-.112
.250	-.670	-.657	-.669			-.263	-.163		
.300	-.662	-.634	-.641	-.648	-.656	-.240	-.173	-.096	-.126
.350	-.563	-.575	-.584			-.183	-.141		
.400	-.543	-.538	-.551	-.550	-.543	-.128	-.118	-.146	-.154
.450	-.521	-.521	-.513			-.076	-.107		
.500	-.503	-.499	-.493	-.512	-.506	-.036	-.094	-.162	-.198
.550	-.491	-.485	-.486			-.009	-.090		
.600	-.477	-.479	-.483	-.475	-.484	.034	-.069	-.150	-.177
.650	-.473	-.473	-.474			.178	.099		
.700	-.491	-.477	-.475	-.476	-.479	.280	.262	.220	.205
.750	-.480	-.465	-.461			.326	.339		
.800	-.455	-.440	-.448	-.478	-.469	.364	.389	.333	.358
.850	-.395	-.381	-.389			.397	.420		
.900	-.268	-.261	-.269	-.291	-.312	.423		.437	.425
.950						.409	.406		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.137	.095				.994		-.5600	
-.8350			-.836			-.288		-.3700	
-.8000		-.356	-.369			-.195		-.2900	
-.7400	-.132		-.219	-.207		-.126	-.288	-.2100	.079
-.6750		-.122	-.172		-.138	-.127		-.1300	.073
-.6000		-.092		-.148		-.129	-.147	-.0500	.146
-.5250		-.088	-.091	-.094	-.103	-.092		.0300	.267
-.4500		-.071		-.059		-.060		.1100	.089
-.3750		-.022	-.025		-.057	-.061		.1900	-.175
-.3000		.004				-.070		.2700	-.283
-.2250		.043	.031		-.053		-.020	.3500	-.137
-.1500		.081	.054	-.007	-.046	-.057		.4300	-.182
-.0750		.088		-.004		-.061		.5100	-.097
.0000		.037				-.044			
.0750		-.067	-.030	-.022			-.019		
.1500			-.080		-.018	-.017			
.2250		-.167	-.081	-.016	.011	.009			
.3000		-.092	-.033	.022		.055			
.3500		.011	.038	.082	.099	.114	.076		
.3700		.055		.107	.129	.134			

APPENDIX

TABLE A7.- Continued

CONFIGURATION 5 MACH = .800 ALPHA = .10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.900	-.854	-.689			-.435	-.381		
.050	-.933	-.887	-.756	-.652	-.686	-.817	-.588	-.580	-.557
.075	-.876	-.888	-.845			-.938	-.716		
.100	-.838	-.841	-.849	-.782	-.792	-.853	-.805	-.706	-.670
.150	-.830	-.841	-.846			-.873	-.887		
.200	-.814	-.776	-.785	-.770	-.752	-1.092	-.985	-.799	-.744
.250	-.757	-.734	-.740			-.739	-.950		
.300	-.772	-.705	-.714	-.789	-.780	-.491	-.712	-.793	-.767
.350	-.679	-.712	-.692			-.354	-.343		
.400	-.600	-.574	-.608	-.561	-.431	-.281	-.266	-.347	-.470
.450	-.454	-.467	-.456			-.236	-.254		
.500	-.441	-.433	-.437	-.432	-.414	-.199	-.223	-.349	-.403
.550	-.460	-.447	-.476			-.178	-.205		
.600	-.477	-.483	-.496	-.468	-.452	-.123	-.144	-.242	-.232
.650	-.508	-.513	-.508			-.066	-.034		
.700	-.568	-.546	-.551	-.553	-.529	-.034	.015	.170	.146
.750	-.557	-.540	-.542			.008	.058		
.800	-.451	-.444	-.472	-.570	-.568	.056	.112	.285	.244
.850	-.301	-.295	-.344			.116	.176		
.900	-.112	-.111	-.174	-.234	-.253	.167		.377	.348
.950						.202	.271		

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CP	
-.8500	.820	.778				.935		-.5600		
-.8350			-.676		-.901			-.3700		
-.8000		-.236	-.270		-.342	-.323	-.288	-.2900	.182 .028	
-.7400	-.070		-.179	-.221		-.268		-.2100	.123 .093	
-.6750		-.072	-.138		-.207	-.221		-.1300	.146 .126	
-.6000		-.063		-.159		-.206	-.161	-.0500	.214 .082	
-.5250		-.072	-.073	-.101	-.144	-.151		.0300	-.289 -.210	
-.4500		-.049		-.062		-.105		.1100	-.720 -.641	
-.3750		.001	-.021		-.101	-.110		.1900	-.991 -.991	
-.3000		.022				-.130		.2700	-.641	
-.2250		.057	.030		-.113		-.054	.3500	-.382 -.368	
-.1500		.087	.029	-.076	-.133	-.138		.4300	-.275	
-.0750		.051		-.120		-.164		.5100	-.206	
.0000		-.117				-.169				
.0750		-.450	-.392	-.303			-.301			
.1500			-.648		-.248	-.188				
.2250		-.854	-.804	-.441	-.224	-.166				
.3000		-.402	-.330	-.255		-.102				
.3500		-.281	-.177	-.120	-.060	-.025	-.113			
.3700		-.241		-.065	-.016	.002				

APPENDIX

TABLE A7.- Continued

CONFIGURATION 5

MACH = .799

ALPHA = 1.02

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
N										
X/C										
.025	-.967	-.950	-.844			-.349	-.269			
.050	-1.040	-1.002	-.870	-.805	-.832	-.719	-.477	-.436	-.404	
.075	-1.008	-1.006	-.936			-.778	-.613			
.100	-.983	-.963	-.956	-.904	-.915	-.713	-.710	-.554	-.494	
.150	-.949	-.963	-.968			-.770	-.730			
.200	-.930	-.900	-.931	-.883	-.918	-1.001	-.874	-.681	-.508	
.250	-.892	-.864	-.895			-.908	-.852			
.300	-.911	-.874	-.875	-.923	-.936	-.577	-.713	-.447	-.574	
.350	-.839	-.858	-.847			-.428	-.274			
.400	-.856	-.850	-.838	-.879	-.930	-.225	-.223	-.370	-.414	
.450	-.839	-.846	-.834			-.119	-.207			
.500	-.845	-.838	-.840	-.869	-.945	-.060	-.169	-.332	-.407	
.550	-.857	-.847	-.857			-.011	-.139			
.600	-.851	-.850	-.840	-.829	-.395	.034	-.081	-.211	-.232	
.650	-.545	-.591	-.588			.090	.098			
.700	-.405	-.401	-.395	-.376	-.419	.127	.185	.206	.163	
.750	-.386	-.367	-.366			.173	.228			
.800	-.353	-.343	-.381	-.423	-.470	.208	.262	.312	.265	
.850	-.260	-.263	-.293			.246	.322			
.900	-.110	-.110	-.161	-.210	-.253	.278		.406	.376	
.950						.289	.369			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.721	.677				.980		-.5600	
-.8350			-.845		-.749			-.3700	
-.8000		-.256	-.285		-.316	-.287	-.278	-.2900	.176
-.7400	-.088		-.185	-.214		-.230		-.2100	.129
-.6750		-.077	-.141		-.186	-.192		-.1300	.170
-.6000		-.062		-.147		-.185	-.148	-.0500	.247
-.5250		-.067	-.066	-.088	-.126	-.133		.0300	-.215
-.4500		-.041		-.049		-.087		.1100	-.604
-.3750		.008	-.006		-.077	-.090		.1900	-.885
-.3000		.032				-.107		.2700	-.543
-.2250		.069	.047		-.085		-.029	.3500	-.367
-.1500		.100	.051	-.042	-.100	-.110		.4300	-.149
-.0750		.072		-.077		-.129		.5100	-.034
.0000		-.079				-.128			
.0750		-.377	-.307	-.220			-.225		
.1500			-.528		-.174	-.132			
.2250		-.832	-.644	-.315	-.147	-.108			
.3000		-.360	-.299	-.185		-.051			
.3500		-.211	-.131	-.063	-.003	.024	-.074		
.3700		-.161		-.013	.039	.050			

APPENDIX

TABLE A7.- Continued

CONFIGURATION 5 MACH = .800 ALPHA = 2.02

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-1.002	-1.007	-.930			-.230	-.148		
.050	-1.099	-1.068	-.989	-.942	-.976	-.581	-.371	-.283	-.243
.075	-1.089	-1.084	-1.030			-.591	-.499		
.100	-1.070	-1.042	-1.042	-1.001	-1.027	-.578	-.556	-.390	-.333
.150	-1.038	-1.042	-1.061			-.666	-.599		
.200	-1.020	-.996	-1.031	-.992	-1.021	-.897	-.794	-.476	-.369
.250	-.983	-.967	-1.007			-.924	-.723		
.300	-1.005	-.972	-.990	-1.026	-1.035	-.536	-.427	-.366	-.408
.350	-.941	-.949	-.938			-.337	-.260		
.400	-.956	-.954	-.939	-.971	-1.016	-.192	-.223	-.344	-.366
.450	-.948	-.943	-.942			-.104	-.201		
.500	-.955	-.941	-.957	-.983	-1.041	-.052	-.163	-.320	-.385
.550	-.963	-.951	-.967			-.013	-.144		
.600	-.972	-.964	-.981	-.981	-1.020	.044	-.098	-.214	-.237
.650	-.967	-.974	-.974			.118	.099		
.700	-.636	-.651	-.653	-.623	-.410	.169	.238	.205	.179
.750	-.372	-.372	-.376			.215	.298		
.800	-.267	-.267	-.275	-.306	-.353	.252	.346	.320	.290
.850	-.184	-.186	-.194			.289	.384		
.900	-.117	-.119	-.119	-.128	-.202	.320		.409	.394
.950						.316	.396		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.640	.598				1.025		-.5600	
-.8350			-.946		-.543			-.3700	
-.8000		-.238	-.259		-.268	-.237	-.266	-.2900	.180
-.7400	-.090		-.182	-.203		-.195		-.2100	.151
-.6750		-.076	-.137		-.165	-.167		-.1300	.204
-.6000		-.053		-.132		-.160	-.135	-.0500	.286
-.5250		-.052	-.051	-.072	-.111	-.111		.0300	-.117
-.4500		-.024		-.031		-.067		.1100	-.504
-.3750		.022	.013		-.061	-.073		.1900	-.782
-.3000		.052				-.089		.2700	-.444
-.2250		.093	.072		-.063		-.005	.3500	-.284
-.1500		.127	.082	-.010	-.068	-.085		.4300	-.099
-.0750		.108		-.034		-.100		.5100	.015
.0000		-.023				-.097			
.0750		-.291	-.223	-.152			-.158		
.1500			-.412		-.119	-.102			
.2250		-.742	-.500	-.229	-.097	-.083			
.3000		-.306	-.239	-.136		-.029			
.3500		-.157	-.090	-.033	.026	.045	-.044		
.3700		-.110		.016	.067	.071			

APPENDIX

TABLE A7.- Continued

CONFIGURATION 5

MACH = .848

ALPHA = .09

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.740	-.702	-.549			-.345	-.310		
.050	-.781	-.737	-.610	-.500	-.528	-.701	-.491	-.489	-.477
.075	-.753	-.743	-.692			-.852	-.617		
.100	-.712	-.720	-.703	-.625	-.647	-.762	-.697	-.634	-.604
.150	-.709	-.720	-.710			-.768	-.799		
.200	-.705	-.666	-.675	-.646	-.661	-.961	-.898	-.746	-.689
.250	-.662	-.648	-.649			-.595	-.862		
.300	-.692	-.661	-.662	-.693	-.690	-.477	-.928	-.769	-.749
.350	-.634	-.638	-.635			-.423	-.588		
.400	-.648	-.638	-.633	-.668	-.720	-.411	-.458	-.857	-.738
.450	-.640	-.642	-.649			-.411	-.428		
.500	-.653	-.650	-.659	-.674	-.696	-.396	-.398	-.885	-.586
.550	-.671	-.673	-.684			-.368	-.368		
.600	-.686	-.697	-.698	-.690	-.730	-.336	-.331	-.306	-.205
.650	-.702	-.700	-.708			-.313	-.296		
.700	-.777	-.756	-.769	-.758	-.674	-.290	-.277	-.198	-.181
.750	-.674	-.659	-.661			-.260	-.248		
.800	-.274	-.258	-.276	-.301	-.361	-.218	-.188	-.100	-.180
.850	-.139	-.144	-.157			-.180	-.134		
.900	-.076	-.093	-.108	-.114	-.145	-.116		.132	-.105
.950						-.060	-.030		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.907	.855				.989		-.5600	
-.8350			-.542		-.784			-.3700	
-.8000		-.206	-.229		-.284	-.270	-.252	-.2900	.235 .082
-.7400	-.044		-.159	-.210		-.262		-.2100	.179 .154
-.6750		-.040	-.113		-.189	-.205		-.1300	.210 .189
-.6000		-.032		-.130		-.187	-.133	-.0500	.283 .152
-.5250		-.041	-.040	-.065	-.117	-.125		.0300	-.199 -.126
-.4500		-.012		-.022		-.078		.1100	-.629 -.545
-.3750		.048	.009		-.077	-.084		.1900	-.881 -.891
-.3000		.073				-.104		.2700	-.545
-.2250		.114	.082		-.081		-.018	.3500	-.413 -.479
-.1500		.148	.085	-.030	-.102	-.119		.4300	
-.0750		.119		-.073		-.150		.5100	-.392
.0000		-.036				-.169			-.372
.0750		-.357	-.320	-.284			-.283		
.1500			-.583		-.304	-.258			
.2250		-.888	-.770	-.556	-.365	-.301			
.3000		-.797	-.884	-.647		-.270			
.3500		-.485	-.664	-.436	-.229	-.173	-.558		
.3700		-.414		-.294	-.158	-.138			

APPENDIX

TABLE A7.- Concluded

CONFIGURATION 5 MACH = .849 ALPHA = 1.08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.788	-.757	-.641			-.280	-.233		
.050	-.849	-.810	-.682	-.597	-.635	-.635	-.430	-.413	-.392
.075	-.832	-.819	-.757			-.748	-.550		
.100	-.798	-.798	-.776	-.708	-.731	-.686	-.636	-.562	-.514
.150	-.776	-.798	-.795			-.710	-.725		
.200	-.767	-.744	-.768	-.723	-.747	-.926	-.830	-.674	-.619
.250	-.739	-.721	-.738			-.714	-.823		
.300	-.766	-.738	-.734	-.775	-.780	-.502	-.951	-.709	-.680
.350	-.715	-.721	-.721			-.464	-.678		
.400	-.733	-.729	-.717	-.749	-.790	-.448	-.483	-.794	-.687
.450	-.733	-.727	-.723			-.460	-.435		
.500	-.739	-.732	-.739	-.760	-.825	-.435	-.374	-.900	-.835
.550	-.755	-.747	-.755			-.384	-.342		
.600	-.764	-.768	-.784	-.772	-.784	-.320	-.297	-.300	-.225
.650	-.777	-.780	-.784			-.274	-.247		
.700	-.744	-.725	-.793	-.806	-.650	-.233	-.231	-.218	-.168
.750	-.313	-.291	-.311			-.178	-.199		
.800	-.192	-.185	-.194	-.238	-.247	-.147	-.147	-.099	-.149
.850	-.140	-.141	-.140			-.115	-.034		
.900	-.104	-.111	-.105	-.120	-.093	-.036		.200	-.073
.950						.026	.051		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.836	.798				1.029		-.5600	
-.8350			-.684		-.651			-.3700	
-.8000		-.216	-.227		-.303	-.286	-.246	-.2900	.226 .087
-.7400	-.060		-.165	-.206		-.244		-.2100	.185 .159
-.6750		-.046	-.115		-.179	-.194		-.1300	.221 .198
-.6000		-.030		-.129		-.181	-.127	-.0500	.298 .168
-.5250		-.032	-.036	-.065	-.113	-.120		.0300	-.158 -.095
-.4500		-.003		-.020		-.071		.1100	-.564 -.490
-.3750		.051	.020		-.071	-.077		.1900	-.834 -.830
-.3000		.079				-.097		.2700	-.490
-.2250		.119	.092		-.073		-.003	.3500	-.461 -.515
-.1500		.154	.097	-.016	-.091	-.112		.4300	-.436
-.0750		.131		-.055		-.142		.5100	-.397
.0000		-.015				-.159			
.0750		-.321	-.279	-.242			-.242		
.1500			-.525		-.275	-.235			
.2250		-.836	-.707	-.501	-.329	-.272			
.3000		-.941	-.802	-.572		-.240			
.3500		-.506	-.595	-.474	-.222	-.144	-.364		
.3700		-.406		-.315	-.160	-.108			

APPENDIX

TABLE A8.- AERODYNAMIC PRESSURE-COEFFICIENT DATA FOR CONFIGURATION 5
AT $x_t = 0.05c$

CONFIGURATION 5 MACH = .202 ALPHA = .07

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.942	-.854	-.761			-.468	-.328		
.050	-.727	-.663	-.693	-.580	-.566	-.619	-.439	-.398	-.264
.075	-.610	-.644	-.571			-.567	-.427		
.100	-.561	-.454	-.517	-.532	-.517	-.538	-.421	-.293	-.259
.150	-.463	-.454	-.439			-.491	-.398		
.200	-.419	-.414	-.439	-.414	-.429	-.445	-.340	-.253	-.241
.250	-.375	-.375	-.380			-.359	-.287		
.300	-.361	-.356	-.361	-.366	-.366	-.323	-.266	-.213	-.230
.350	-.332	-.323	-.336			-.259	-.235		
.400	-.321	-.315	-.323	-.319	-.334	-.194	-.204	-.216	-.206
.450	-.315	-.312	-.317			-.127	-.182		
.500	-.317	-.317	-.315	-.319	-.319	-.081	-.156	-.201	-.213
.550	-.317	-.319	-.315			-.041	-.132		
.600	-.312	-.317	-.317	-.312	-.319	.024	-.098	-.163	-.180
.650	-.323	-.317	-.328			.119	.053		
.700	-.340	-.323	-.334	-.340	-.340	.177	.179	.153	.143
.750	-.340	-.340	-.328			.210	.246		
.800	-.340	-.340	-.340	-.398	-.352	.246	.286	.165	.277
.850	-.317	-.305	-.311			.277	.317		
.900	-.230	-.230	-.235	-.253	-.276	.296		.334	.334
.950						.291	.327		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.394	.356				.688		-.5600	
-.8350			-.447		-.470			-.3700	
-.8000		-.219	-.246		-.261	-.257	-.246	-.2900	.092
-.7400	-.049		-.180	-.176		-.188		-.2100	-.005
-.6750		-.192	-.138		-.165	-.161		-.1300	-.024
-.6000		-.061		-.142		-.153	-.138	-.0500	.022
-.5250		-.095	-.095	-.103	-.122	-.118		.0300	-.202
-.4500		-.088		-.084		-.088		.1100	-.392
-.3750		-.052	-.061		-.088	-.086		.1900	-.426
-.3000		-.052				-.096		.2700	-.306
-.2250		-.043	-.048		-.088		-.074	.3500	-.260
-.1500		-.037	-.061	-.055	-.086	-.084		.4300	-.135
-.0750		-.064		-.091		-.086		.5100	.005
.0000		-.151				-.070			
.0750		-.248	-.169	-.097			-.099		
.1500			-.189		-.052	-.041			
.2250		-.261	-.174	-.075	-.025	-.021			
.3000		-.169	-.113	-.028		.019			
.3500		-.055	-.034	.027	.053	.068	.022		
.3700		-.010		.053	.077	.085			

APPENDIX

TABLE A8.- Continued

CONFIGURATION S MACH = .202 ALPHA = 1.06

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-1.065	-1.090	-1.016			-.241	-.113		
.050	-.855	-.767	-.919	-.752	-.723	-.451	-.265	-.247	-.131
.075	-.723	-.738	-.747			-.405	-.288		
.100	-.660	-.528	-.640	-.630	-.650	-.387	-.300	-.201	-.166
.150	-.537	-.528	-.523			-.387	-.323		
.200	-.484	-.479	-.493	-.479	-.503	-.370	-.282	-.189	-.177
.250	-.440	-.435	-.435			-.302	-.223		
.300	-.405	-.410	-.415	-.410	-.430	-.271	-.211	-.163	-.177
.350	-.363	-.376	-.376			-.216	-.175		
.400	-.354	-.358	-.367	-.376	-.369	-.158	-.151	-.177	-.175
.450	-.347	-.350	-.352			-.103	-.139		
.500	-.347	-.341	-.343	-.358	-.352	-.057	-.122	-.173	-.192
.550	-.347	-.341	-.343			-.026	-.105		
.600	-.341	-.337	-.350	-.339	-.347	.022	-.079	-.146	-.163
.650	-.341	-.341	-.346			.136	.065		
.700	-.358	-.346	-.352	-.352	-.358	.206	.201	.163	.153
.750	-.352	-.352	-.346			.244	.265		
.800	-.352	-.352	-.352	-.451	-.364	.275	.311	.199	.292
.850	-.323	-.311	-.323			.301	.337		
.900	-.241	-.236	-.247	-.265	-.276	.323		.347	.352
.950						.316	.342		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.230	.131				.784		-.5600	
-.8350			-.637		-.378			-.3700	
-.8000		-.246	-.273		-.227	-.219	-.250	-.2900	.077
-.7400	-.076		-.165	-.169		-.157		-.2100	.022
-.6750		-.165	-.146		-.142	-.142		-.1300	.041
-.6000		-.068		-.134	-.146	-.146	-.138	-.0500	.097
-.5250		-.088	-.092	-.095	-.111	-.111		.0300	-.101
-.4500		-.076		-.068		-.084		.1100	-.289
-.3750		-.041	-.046		-.082	-.072		.1900	-.340
-.3000		-.032				-.082		.2700	-.224
-.2250		-.019	-.025		-.077		-.050	.3500	-.214
-.1500		-.005	-.025	-.039	-.075	-.072		.4300	-.108
-.0750		-.019		-.059		-.072		.5100	.034
.0000		-.077				-.055			
.0750		-.165	-.111	-.064			-.060		
.1500			-.124		-.041	-.024			
.2250		-.201	-.111	-.041	-.010	-.005			
.3000		-.124	-.059	-.003		.034			
.3500		-.023	.005	.047	.069	.085	.049		
.3700		.019		.073	.095	.099			

APPENDIX

TABLE A8.- Continued

CONFIGURATION 5

MACH = .203

ALPHA = 2.06

WING CP									
UPPER SURFACE					LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-1.264	-1.540	-1.486			.112	.186		
.050	-1.065	-1.094	-1.186	-.842	-.707	-.130	-.020	-.078	-.061
.075	-.920	-1.022	-.930			-.124	-.089		
.100	-.847	-.755	-.804	-.649	-.600	-.136	-.113	-.095	-.136
.150	-.707	-.755	-.649			-.159	-.176		
.200	-.668	-.673	-.552	-.421	-.431	-.159	-.176	-.159	-.193
.250	-.629	-.605	-.445			-.257	-.212		
.300	-.586	-.571	-.387	-.334	-.349	-.252	-.200	-.000	-.211
.350	-.430	-.539	-.533			-.226	-.176		
.400	-.439	-.512	-.507	-.454	-.379	-.193	-.143	-.026	-.102
.450	-.443	-.494	-.482			-.157	-.119		
.500	-.447	-.486	-.464	-.413	-.327	-.119	-.067	-.038	-.148
.550	-.449	-.475	-.452			-.083	-.043		
.600	-.458	-.469	-.434	-.364	-.293	-.048	-.003	-.031	-.143
.650	-.378	-.476	-.459			.080	.151		
.700	-.413	-.476	-.459	-.343	-.274	.182	.301	.225	.137
.750	-.424	-.476	-.441			.234	.376		
.800	-.430	-.465	-.424	-.338	-.251	.270	.417	.244	.253
.850	-.407	-.424	-.378			.308	.438		
.900	-.320	-.326	-.268	-.199	-.170	.334		.364	.305
.950						.327	.417		

NACELLE							PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	
X/C	CP						X/C	CF
-.8500	-.095	-.183				.060	-.5600	
-.8350			-.646		-.175		-.3700	
-.8000		-.397	-.386		-.072	-.068	-.2900	.055
-.7400	-.160		-.256	-.145		-.060	-.2100	.031
-.6750		-.244	-.217		-.018	-.049	-.1300	.096
-.6000		-.156		-.083		-.064	-.0500	.213
-.5250		-.164	-.129	-.007	.008	-.053	.0300	.062
-.4500		-.141		.035		-.045	.1100	-.124
-.3750		-.054	-.101		.014	-.086	.1900	-.177
-.3000		-.052				-.108	.2700	-.081
-.2250		-.050	-.059		.003		.3500	-.057
-.1500		-.028	-.043	.081	.001	-.122	.4300	.026
-.0750		-.041		.074		-.131	.5100	.151
.0000		-.090				-.122		
.0750		-.159	-.088	.077				
.1500			-.108		-.021	-.086		
.2250		-.217	-.095	.087	-.010	-.067		
.3000		-.155	-.032	.113		-.024		
.3500		-.054	.059	.148	.042	.033	.019	
.3700		-.006		.150	.061	.055		

APPENDIX

TABLE A8.- Continued

CONFIGURATION 5

MACH = .199

ALPHA = 3.06

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-1.294	-1.415	-1.592			.130	.291		
.050	-.992	-.946	-1.299	-1.405	-1.325	-.134	.088	.201	.231
.075	-.815	-.830	-1.088			-.164	.010		
.100	-.689	-.629	-.982	-1.082	-1.067	-.212	-.020	.135	.124
.150	-.533	-.629	-.830			-.254	-.062		
.200	-.467	-.578	-.775	-.780	-.775	-.224	-.020	.064	.052
.250	-.422	-.543	-.699			-.068	-.113		
.300	-.397	-.533	-.664	-.659	-.644	-.043	-.135	-.123	.022
.350	-.355	-.335	-.413			-.004	-.140		
.400	-.319	-.328	-.418	-.527	-.574	.023	-.145	-.145	-.026
.450	-.295	-.319	-.415			.060	-.135		
.500	-.279	-.313	-.424	-.513	-.545	.078	-.147	-.137	-.033
.550	-.270	-.308	-.433			.087	-.150		
.600	-.264	-.319	-.449	-.498	-.529	.117	-.135	-.108	-.014
.650	-.284	-.278	-.386			.211	.016		
.700	-.284	-.284	-.410	-.512	-.536	.295	.176	.230	.280
.750	-.260	-.290	-.422			.324	.255		
.800	-.248	-.284	-.428	-.524	-.518	.346	.307	.300	.403
.850	-.212	-.266	-.410			.356	.346		
.900	-.134	-.200	-.332	-.374	-.392	.364		.440	.445
.950						.344	.344		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.261	-.233				.983		-.5600	
-.8350			-.572		-.361			-.3700	
-.8000		-.229	-.317		-.253	-.137	-.149	-.2900	.001
-.7400	-.006		-.213			-.113		-.2100	-.019
-.6750		-.093	-.185	-.245	-.185	-.065		-.1300	.039
-.6000		-.018		-.221		-.057	-.030	-.0500	.153
-.5250		-.034	-.133	-.181	-.149	-.010		.0300	.044
-.4500		-.034		-.145		.026		.1100	-.178
-.3750		.070	-.000		-.044	.049		.1900	-.255
-.3000		.099				.051		.2700	-.016
-.2250		.133	.025		-.023		.006	.3500	-.220
-.1500		.160	.031	-.083	.009	.061		.4300	-.131
-.0750		.167		-.088		.059		.5100	.014
.0000		.128				.066			
.0750		.065	-.028	-.069			.011		
.1500			-.065		.090	.083			
.2250		.004	-.079	-.032	.122	.093			
.3000		.034	-.041	.016		.118			
.3500		.092	-.002	.079	.192	.153	.086		
.3700		.113		.108	.210	.158			

APPENDIX

TABLE A8.- Continued

CONFIGURATION 5

MACH = .201

ALPHA = 4.05

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
N										
X/C										
.025	-1.822	-1.669	-1.591			.305	.393			
.050	-1.439	-1.055	-1.208	-1.350	-1.429	.032	.137	.143	.323	
.075	-1.168	-.928	-1.011			-.043	.044			
.100	-.997	-.594	-.888	-1.105	-1.193	-.108	-.003	.108	.178	
.150	-.746	-.594	-.756			-.166	-.084			
.200	-.648	-.510	-.687	-.800	-.884	-.178	-.084	.032	.090	
.250	-.554	-.456	-.628			-.140	.048			
.300	-.466	-.432	-.604	-.677	-.721	-.127	.024	-.084	.061	
.350	-.597	-.429	-.358			-.077	.017			
.400	-.549	-.373	-.332	-.416	-.486	-.019	.007	-.127	-.099	
.450	-.495	-.334	-.319			.029	.000			
.500	-.449	-.305	-.316	-.399	-.469	.077	-.021	-.147	-.140	
.550	-.405	-.284	-.312			.113	-.045			
.600	-.369	-.269	-.327	-.390	-.464	.152	-.055	-.147	-.135	
.650	-.488	-.301	-.278			.260	.058			
.700	-.477	-.283	-.283	-.401	-.471	.368	.188	.168	.204	
.750	-.442	-.266	-.283			.411	.253			
.800	-.389	-.242	-.289	-.424	-.471	.433	.293	.265	.365	
.850	-.324	-.201	-.272			.445	.325			
.900	-.213	-.120	-.201	-.295	-.348	.445		.375	.425	
.950						.416	.313			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.636	-.562				.938		-.5600	
-.8350			-.535		-.251			-.3700	
-.8000		-.263	-.228		-.193	-.135	-.298	-.2900	.076
-.7400	-.127		-.115	-.177		-.142		-.2100	.073
-.6750		-.053	-.084		-.173	-.127		-.1300	.131
-.6000		.028		-.150		-.138	-.154	-.0500	.245
-.5250		.032	-.026	-.138	-.154	-.103		.0300	.170
-.4500		.059		-.119		-.072		.1100	-.048
-.3750		-.053	.078		-.102	-.065		.1900	-.152
-.3000		-.005				-.065		.2700	-.029
-.2250		.058	.118		-.095		.088	.3500	-.160
-.1500		.120	.133	-.055	-.080	-.009		.4300	-.109
-.0750		.155		-.057		.008		.5100	.003
.0000		.157				.042			
.0750		.122	.109	-.053			.112		
.1500			.078		-.014	.102			
.2250		.072	.063	-.028	.028	.127			
.3000		.115	.078	.008		.168			
.3500		.179	.083	.056	.142	.216	.182		
.3700		.197		.102	.175	.226			

APPENDIX

TABLE A8.- Continued

CONFIGURATION 5

MACH = .202

ALPHA = 5.07

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C										
.025	-1.755	-2.214	-2.282				.495	.570		
.050	-1.422	-1.412	-1.598	-1.500	-1.500		.229	.310	.321	.385
.075	-1.241	-1.251	-1.305				.159	.194		
.100	-1.075	-.879	-1.095	-1.153	-1.192		.090	.125	.223	.217
.150	-.850	-.879	-.894				.009	.044		
.200	-.757	-.767	-.796	-.787	-.831		-.008	.015	.096	.090
.250	-.679	-.689	-.694				-.033	.029		
.300	-.620	-.635	-.635	-.645	-.669		-.043	.012	.038	.026
.350	-.549	-.557	-.566				-.007	.012		
.400	-.516	-.521	-.527	-.540	-.544		-.005	.003	-.007	-.014
.450	-.501	-.492	-.499				.012	-.002		
.500	-.475	-.469	-.477	-.484	-.486		.034	-.005	-.036	-.067
.550	-.451	-.444	-.458				.046	-.007		
.600	-.436	-.438	-.451	-.443	-.449		.077	-.002	-.057	-.077
.650	-.428	-.434	-.434				.184	.124		
.700	-.434	-.428	-.434	-.434	-.440		.292	.261	.225	.213
.750	-.416	-.410	-.410				.347	.330		
.800	-.393	-.399	-.399	-.416	-.410		.380	.373	.139	.337
.850	-.346	-.341	-.352				.402	.402		
.900	-.247	-.247	-.259	-.265	-.276		.411		.395	.392
.950							.385	.378		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.308	-.505				.956		-.5600	
-.8350			-1.016		-.134			-.3700	
-.8000		-.641	-.424		-.099	-.014	-.258	-.2900	.034
-.7400	-.235		-.250	-.184		-.045		-.2100	.063
-.6750		-.134	-.188		-.084	-.045		-.1300	.157
-.6000		-.092		-.142		-.061	-.138	-.0500	.307
-.5250		-.080	-.095	-.095	-.068	-.037		.0300	.290
-.4500		-.049		-.061		-.014		.1100	.097
-.3750		-.008	-.014		-.028	-.014		.1900	-.009
-.3000		.023				-.024		.2700	.087
-.2250		.075	.054		-.021		-.002	.3500	-.041
-.1500		.126	.089	.016	-.012	-.012		.4300	.003
-.0750		.152		.027		-.014		.5100	.102
.0000		.141				.000			
.0750		.102	.099	.047			.053		
.1500			.082		.029	.029			
.2250		.032	.067	.069	.056	.051			
.3000		.056	.084	.099		.090			
.3500		.113	.123	.139	.132	.138	.138		
.3700		.139		.161	.154	.157			

APPENDIX

TABLE A8.- Continued

CONFIGURATION 5

MACH = .602

ALPHA = .10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.979	-.917	-.754			-.629	-.456		
.050	-.866	-.797	-.833	-.671	-.682	-.960	-.618	-.655	-.458
.075	-.703	-.681	-.677			-.777	-.632		
.100	-.619	-.509	-.593	-.610	-.600	-.717	-.629	-.446	-.427
.150	-.497	-.509	-.497			-.675	-.618		
.200	-.473	-.470	-.487	-.469	-.490	-.633	-.523	-.379	-.366
.250	-.436	-.434	-.432			-.496	-.405		
.300	-.416	-.411	-.418	-.421	-.420	-.420	-.363	-.302	-.335
.350	-.380	-.381	-.391			-.315	-.302		
.400	-.371	-.375	-.380	-.380	-.382	-.231	-.253	-.299	-.303
.450	-.371	-.371	-.369			-.154	-.223		
.500	-.368	-.368	-.367	-.374	-.371	-.091	-.184	-.275	-.310
.550	-.365	-.365	-.367			-.028	-.155		
.600	-.366	-.373	-.372	-.366	-.380	.050	-.102	-.191	-.228
.650	-.375	-.376	-.379			.102	.064		
.700	-.399	-.390	-.393	-.395	-.403	.137	.180	.153	.140
.750	-.401	-.396	-.388			.173	.229		
.800	-.393	-.388	-.386	-.424	-.420	.224	.267	.242	.250
.850	-.348	-.339	-.347			.258	.306		
.900	-.227	-.227	-.243	-.265	-.292	.286		.322	.320
.950						.293	.347		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CF
-.8500	.634	.588				.796		-.5600	
-.8350			-.373		-.409			-.3700	
-.8000		-.232	-.256		-.312	-.304	-.294	-.2900	.117
-.7400	-.060		-.186	-.206		-.235		-.2100	.019
-.6750		-.100	-.146		-.190	-.203		-.1300	.018
-.6000		-.084		-.161		-.191	-.166	-.0500	.056
-.5250		-.099	-.096	-.116	-.144	-.149		.0300	-.360
-.4500		-.090		-.089		-.114		.1100	-.602
-.3750		-.044	-.062		-.109	-.104		.1900	-.624
-.3000		-.037				-.117		.2700	-.489
-.2250		-.023	-.047		-.114		-.086	.3500	-.317
-.1500		-.018	-.059	-.101	-.117	-.109		.4300	-.158
-.0750		-.067		-.120		-.115		.5100	-.018
.0000		-.204				-.100			
.0750		-.382	-.273	-.161			-.166		
.1500			-.317		-.107	-.076			
.2250		-.394	-.284	-.145	-.075	-.053			
.3000		-.257	-.186	-.086		-.006			
.3500		-.127	-.083	-.015	.023	.049	-.025		
.3700		-.070		.020	.054	.069			

APPENDIX

TABLE A8.- Continued

CONFIGURATION 5 MACH = .600 ALPHA = 1.11

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.282	-1.219	-1.086			-.345	-.236			
.050	-1.157	-1.043	-1.124	-.935	-.921	-.668	-.381	-.422	-.238	
.075	-.895	-.879	-.858			-.582	-.437			
.100	-.784	-.629	-.759	-.752	-.765	-.562	-.467	-.305	-.286	
.150	-.626	-.629	-.617			-.560	-.491			
.200	-.577	-.566	-.577	-.569	-.597	-.534	-.412	-.293	-.271	
.250	-.527	-.524	-.525			-.423	-.329			
.300	-.496	-.491	-.495	-.501	-.501	-.371	-.304	-.248	-.265	
.350	-.454	-.451	-.458			-.289	-.259			
.400	-.442	-.440	-.438	-.438	-.433	-.211	-.220	-.257	-.255	
.450	-.434	-.428	-.425			-.138	-.193			
.500	-.427	-.417	-.418	-.429	-.420	-.081	-.162	-.242	-.273	
.550	-.419	-.413	-.417			-.038	-.137			
.600	-.412	-.412	-.414	-.405	-.419	.037	-.097	-.179	-.214	
.650	-.418	-.419	-.419			.131	.073			
.700	-.442	-.427	-.428	-.420	-.433	.180	.204	.174	.156	
.750	-.439	-.425	-.422			.227	.260			
.800	-.428	-.413	-.413	-.444	-.441	.268	.301	.254	.280	
.850	-.374	-.356	-.368			.302	.340			
.900	-.249	-.241	-.254	-.275	-.304	.327		.352	.351	
.950						.326	.362			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.433	.392				.869		-.5600	
-.8350			-.405		-.411			-.3700	
-.8000		-.286	-.297		-.280	-.264	-.290	-.2900	.097
-.7400	-.100		-.208	-.202		-.204		-.2100	.033
-.6750		-.113	-.162		-.180	-.179		-.1300	.063
-.6000		-.094		-.156		-.174	-.161	-.0500	.140
-.5250		-.101	-.099	-.110	-.136	-.134		.0300	-.171
-.4500		-.086		-.082		-.098		.1100	-.437
-.3750		-.048	-.052		-.091	-.099		.1900	-.507
-.3000		-.035				-.111		.2700	-.362
-.2250		-.010	-.021		-.091		-.065	.3500	-.287
-.1500		.006	-.018	-.072	-.094	-.102		.4300	-.138
-.0750		-.019		-.084		-.107		.5100	.008
.0000		-.121				-.088			
.0750		-.273	-.188	-.119			-.117		
.1500			-.238		-.077	-.064			
.2250		-.323	-.224	-.108	-.048	-.039			
.3000		-.210	-.142	-.056		.008			
.3500		-.082	-.047	.013	.047	.066	.006		
.3700		-.028		.044	.078	.087			

APPENDIX

TABLE A8.- Continued

CONFIGURATION 5

MACH = .600

ALPHA = 2.08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
N									
X/C									
.025	-1.655	-1.584	-1.426			-.076	-.007		
.050	-1.462	-1.318	-1.450	-1.205	-1.314	-.387	-.199	-.234	-.098
.075	-1.054	-1.036	-1.038			-.389	-.286		
.100	-.915	-.731	-.904	-.935	-.932	-.398	-.308	-.181	-.149
.150	-.723	-.731	-.735			-.437	-.354		
.200	-.655	-.648	-.675	-.664	-.691	-.427	-.314	-.210	-.195
.250	-.594	-.593	-.595			-.350	-.251		
.300	-.556	-.550	-.564	-.569	-.563	-.306	-.240	-.181	-.205
.350	-.507	-.502	-.513			-.233	-.204		
.400	-.488	-.484	-.487	-.487	-.488	-.169	-.178	-.214	-.209
.450	-.471	-.465	-.465			-.114	-.157		
.500	-.458	-.452	-.453	-.468	-.457	-.065	-.132	-.208	-.236
.550	-.449	-.443	-.444			-.021	-.121		
.600	-.438	-.442	-.444	-.440	-.452	.036	-.089	-.162	-.194
.650	-.443	-.439	-.441			.160	.082		
.700	-.462	-.443	-.446	-.450	-.455	.228	.226	.191	.173
.750	-.450	-.437	-.433			.273	.292		
.800	-.433	-.421	-.424	-.461	-.455	.307	.334	.275	.304
.850	-.379	-.365	-.373			.340	.377		
.900	-.252	-.247	-.258	-.282	-.306	.361		.381	.374
.950						.357	.382		

NACELLE							PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	
								UPPER ROW LOWER ROW
X/C	CP						X/C	CP
-.8500	.267	.266				.940	-.5600	
-.8350			-.477		-.360		-.3700	
-.8000		-.316	-.329		-.238	-.214	-.2900	.089
-.7400	-.115		-.218	-.207		-.171	-.2100	.052
-.6750		-.118	-.164		-.157	-.154	-.1300	.100
-.6000		-.093		-.154		-.149	-.0500	.203
-.5250		-.094	-.094	-.105	-.118	-.112	.0300	-.056
-.4500		-.078		-.072		-.078	.1100	-.316
-.3750		-.035	-.039		-.076	-.080	.1900	-.409
-.3000		-.012				-.091	.2700	-.247
-.2250		.021	.001		-.072		.3500	-.244
-.1500		.047	.017	-.040	-.067	-.076	.4300	-.117
-.0750		.035		-.046		-.084	.5100	.036
.0000		-.040				-.067		
.0750		-.167	-.108	-.065				
.1500			-.159		-.046	-.038		
.2250		-.243	-.157	-.059	-.020	-.014		
.3000		-.151	-.091	-.019		.032		
.3500		-.033	-.008	.043	.069	.090	.037	
.3700		.012		.074	.101	.109		

APPENDIX

TABLE A8.- Continued

CONFIGURATION 5

MACH = .601

ALPHA = 3.08

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-2.072	-2.225	-2.017			.102	.196			
.050	-1.904	-1.922	-1.956	-1.923	-1.885	-.167	.000	-.062	.089	
.075	-1.219	-1.214	-1.230			-.199	-.136			
.100	-1.046	-.839	-1.030	-1.074	-1.097	-.253	-.181	-.050	-.045	
.150	-.842	-.839	-.833			-.295	-.234			
.200	-.749	-.742	-.762	-.754	-.765	-.307	-.220	-.124	-.129	
.250	-.673	-.669	-.673			-.266	-.175			
.300	-.624	-.608	-.627	-.632	-.630	-.240	-.181	-.127	-.148	
.350	-.563	-.560	-.568			-.189	-.148			
.400	-.539	-.536	-.541	-.540	-.539	-.136	-.124	-.155	-.163	
.450	-.514	-.508	-.508			-.077	-.111			
.500	-.496	-.491	-.486	-.501	-.496	-.036	-.099	-.166	-.204	
.550	-.483	-.479	-.474			-.011	-.094			
.600	-.469	-.471	-.472	-.465	-.477	.038	-.068	-.141	-.174	
.650	-.466	-.463	-.463			.167	.098			
.700	-.480	-.463	-.462	-.467	-.472	.260	.246	.207	.191	
.750	-.464	-.452	-.446			.314	.320			
.800	-.442	-.429	-.432	-.464	-.456	.350	.366	.300	.327	
.850	-.378	-.369	-.373			.385	.400			
.900	-.254	-.247	-.255	-.274	-.296	.406		.398	.397	
.950						.393	.399			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.126	.093				.994		-.5600	
-.8350			-.778		-.306			-.3700	
-.8000		-.361	-.369		-.203	-.156	-.282	-.2900	.083
-.7400	-.128		-.227	-.209		-.122		-.2100	.069
-.6750		-.123	-.169		-.136	-.129		-.1300	.141
-.6000		-.095		-.150		-.134	-.148	-.0500	.263
-.5250		-.090	-.090	-.098	-.103	-.096		.0300	.067
-.4500		-.067		-.063		-.063		.1100	-.195
-.3750		-.022	-.025		-.059	-.061		.1900	-.300
-.3000		.006				-.071		.2700	-.148
-.2250		.044	.030		-.057		-.020	.3500	-.191
-.1500		.079	.056	-.013	-.052	-.059		.4300	-.090
-.0750		.092		-.007		-.061		.5100	.048
.0000		.041				-.042			
.0750		-.070	-.039	-.021			-.021		
.1500			-.083		-.021	-.016			
.2250		-.170	-.086	-.019	.006	.006			
.3000		-.102	-.039	.019		.051			
.3500		.009	.032	.077	.096	.111	.074		
.3700		.050		.103	.125	.130			

APPENDIX

TABLE A8.- Continued

CONFIGURATION 5

MACH = .801

ALPHA = .09

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
N									
X/C									
.025	-.828	-.676	-.462			-.406	-.328		
.050	-.907	-.851	-.929	-.765	-.791	-.907	-.753	-1.066	-.831
.075	-.936	-.933	-.935			-1.036	-.834		
.100	-.966	-.915	-.911	-.812	-.822	-.983	-.768	-.699	-.617
.150	-.862	-.915	-.830			-.835	-.874		
.200	-.790	-.731	-.782	-.720	-.666	-.836	-.970	-.778	-.741
.250	-.699	-.692	-.603			-.551	-.750		
.300	-.551	-.530	-.569	-.638	-.707	-.517	-.522	-.794	-.754
.350	-.460	-.458	-.491			-.488	-.418		
.400	-.453	-.455	-.458	-.449	-.423	-.452	-.373	-.493	-.587
.450	-.461	-.457	-.456			-.387	-.335		
.500	-.465	-.465	-.463	-.457	-.431	-.301	-.266	-.339	-.373
.550	-.467	-.465	-.464			-.229	-.214		
.600	-.470	-.472	-.478	-.460	-.444	-.175	-.170	-.169	-.134
.650	-.480	-.481	-.482			-.120	-.115		
.700	-.508	-.492	-.494	-.491	-.477	-.090	-.070	.072	.050
.750	-.474	-.467	-.458			-.041	-.044		
.800	-.405	-.398	-.406	-.451	-.451	.005	.001	.124	.097
.850	-.278	-.272	-.297			.047	.042		
.900	-.109	-.107	-.133	-.164	-.176	.084		.209	.162
.950						.124	.138		

NACELLE							PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	
								UPPER ROW LOWER ROW
X/C								X/C CP
-.8500	.842	.793				.933		-.5600
-.8350			-.641		-.902			-.3700
-.8000		-.232	-.265		-.346	-.324	-.289	-.2900
-.7400	-.061		-.197	-.222		-.268		-.2100
-.6750		-.051	-.137		-.210	-.221		-.1300
-.6000		-.068		-.157		-.206	-.164	-.0500
-.5250		-.070	-.073	-.100	-.147	-.151		.0300
-.4500		-.048		-.063		-.108		.1100
-.3750		.007	-.023		-.103	-.112		.1900
-.3000		.028				-.131		.2700
-.2250		.063	.030		-.117		-.056	.3500
-.1500		.091	.028	-.083	-.135	-.143		.4300
-.0750		.054		-.129		-.170		.5100
.0000		-.123				-.179		
.0750		-.478	-.412	-.323			-.320	
.1500			-.671		-.268	-.203		
.2250		-.814	-.765	-.456	-.248	-.182		
.3000		-.425	-.340	-.286		-.122		
.3500		-.305	-.192	-.146	-.086	-.046	-.135	
.3700		-.254		-.085	-.044	-.018		

APPENDIX

TABLE A8.- Continued

CONFIGURATION 5

MACH = .800

ALPHA = 1.10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.949	-.873	-.602			-.299	-.244		
.050	-1.013	-.899	-.962	-.813	-.856	-.865	-.697	-.883	-.643
.075	-.959	-.979	-1.012			-.948	-.598		
.100	-1.018	-.987	-1.001	-.956	-.962	-.693	-.710	-.579	-.511
.150	-.984	-.987	-.976			-.768	-.748		
.200	-.950	-.923	-.930	-.887	-.916	-1.011	-.865	-.673	-.566
.250	-.887	-.873	-.896			-.777	-.849		
.300	-.868	-.855	-.852	-.886	-.906	-.568	-.788	-.519	-.509
.350	-.827	-.834	-.839			-.492	-.362		
.400	-.852	-.830	-.853	-.837	-.904	-.359	-.287	-.407	-.451
.450	-.830	-.842	-.829			-.221	-.252		
.500	-.830	-.830	-.837	-.866	-.756	-.123	-.188	-.339	-.407
.550	-.738	-.807	-.782			-.075	-.128		
.600	-.489	-.518	-.432	-.397	-.389	-.017	-.045	-.155	-.173
.650	-.376	-.385	-.379			.033	.047		
.700	-.389	-.377	-.380	-.381	-.404	.064	.078	.113	.095
.750	-.384	-.372	-.368			.107	.117		
.800	-.351	-.329	-.354	-.379	-.410	.155	.168	.169	.145
.850	-.251	-.234	-.264			.182	.218		
.900	-.107	-.101	-.126	-.149	-.183	.218		.252	.220
.950						.233	.289		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.726	.680				.980		-.5600	
-.8350			-.823		-.725			-.3700	
-.8000		-.260	-.289		-.316	-.294	-.280	-.2900	.176
-.7400	-.084		-.209	-.223		-.243		-.2100	.119
-.6750		-.070	-.146		-.190	-.202		-.1300	.164
-.6000		-.073		-.154		-.192	-.152	-.0500	.247
-.5250		-.072	-.071	-.094	-.132	-.139		.0300	-.222
-.4500		-.044		-.055		-.094		.1100	-.580
-.3750		.007	-.014		-.089	-.097		.1900	-.899
-.3000		.032				-.115		.2700	-.549
-.2250		.069	.042		-.097		-.038	.3500	-.466
-.1500		.098	.046	-.053	-.110	-.122		.4300	-.231
-.0750		.069		-.085		-.145		.5100	-.095
.0000		-.090				-.145			
.0750		-.404	-.329	-.242			-.250		
.1500			-.553		-.197	-.160			
.2250		-.854	-.687	-.359	-.174	-.141			
.3000		-.421	-.348	-.230		-.084			
.3500		-.267	-.166	-.097	-.031	-.008	-.104		
.3700		-.207		-.044	.012	.021			

APPENDIX

TABLE A8.- Continued

CONFIGURATION 5

MACH = .799

ALPHA = 2.11

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D										
N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.003	-.998	-.821			-.206	-.126			
.050	-1.151	-1.039	-.991	-.835	-.876	-.826	-.437	-.550	-.301	
.075	-.999	-1.006	-1.058			-.531	-.503			
.100	-1.028	-1.049	-1.059	-1.005	-1.040	-.581	-.566	-.404	-.355	
.150	-1.054	-1.049	-1.069			-.678	-.608			
.200	-1.050	-1.024	-1.033	-.994	-1.029	-.916	-.791	-.461	-.404	
.250	-.995	-.976	-.989			-.868	-.765			
.300	-.971	-.966	-.985	-.989	-1.009	-.502	-.462	-.386	-.409	
.350	-.931	-.938	-.945			-.360	-.321			
.400	-.941	-.934	-.944	-.970	-1.006	-.230	-.267	-.370	-.390	
.450	-.936	-.916	-.945			-.134	-.235			
.500	-.932	-.940	-.946	-.970	-.996	-.076	-.190	-.325	-.387	
.550	-.956	-.954	-.959			-.029	-.159			
.600	-.961	-.963	-.977	-.964	-.693	.032	-.097	-.164	-.198	
.650	-.561	-.596	-.633			.099	.078			
.700	-.394	-.407	-.413	-.391	-.378	.147	.176	.144	.119	
.750	-.307	-.311	-.315			.183	.220			
.800	-.247	-.245	-.248	-.270	-.321	.216	.259	.201	.188	
.850	-.178	-.172	-.178			.253	.293			
.900	-.092	-.099	-.106	-.114	-.156	.277		.284	.272	
.950						.276	.345			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CF
-.8500	.631	.598				1.026		-.5600	
-.8350			-.943		-.491			-.3700	
-.8000		-.246	-.272		-.274	-.242	-.270	-.2900	.174
-.7400	-.087		-.205	-.217		-.196		-.2100	.141
-.6750		-.072	-.146		-.173	-.172		-.1300	.193
-.6000		-.062		-.142		-.169	-.141	-.0500	.280
-.5250		-.056	-.063	-.081	-.117	-.117		.0300	-.160
-.4500		-.031		-.038		-.073		.1100	-.529
-.3750		.024	.005		-.065	-.071		.1900	-.806
-.3000		.051				-.088		.2700	-.460
-.2250		.090	.063		-.071		-.013	.3500	-.374
-.1500		.120	.071	-.020	-.081	-.090		.4300	-.147
-.0750		.099		-.044		-.107		.5100	-.030
.0000		-.035				-.103			
.0750		-.302	-.240	-.178			-.173		
.1500			-.438		-.137	-.108			
.2250		-.745	-.556	-.262	-.119	-.091			
.3000		-.318	-.276	-.160		-.039			
.3500		-.172	-.122	-.053	.006	.032	-.067		
.3700		-.129		-.007	.045	.056			

APPENDIX

TABLE A8.- Continued

CONFIGURATION 5

MACH = .851

ALPHA = .10

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.709	-.538	-.344			-.305	-.244		
.050	-.758	-.702	-.769	-.617	-.648	-.764	-.622	-.924	-.682
.075	-.785	-.785	-.789			-.881	-.726		
.100	-.822	-.781	-.776	-.682	-.689	-.851	-.702	-.706	-.577
.150	-.748	-.781	-.721			-.781	-.779		
.200	-.718	-.666	-.692	-.640	-.643	-.954	-.892	-.705	-.651
.250	-.655	-.633	-.658			-.768	-.837		
.300	-.639	-.643	-.639	-.656	-.674	-.615	-.900	-.742	-.713
.350	-.633	-.619	-.647			-.576	-.753		
.400	-.636	-.628	-.634	-.654	-.704	-.503	-.505	-.857	-.761
.450	-.644	-.628	-.636			-.474	-.461		
.500	-.647	-.644	-.648	-.676	-.698	-.429	-.408	-.483	-.466
.550	-.660	-.656	-.661			-.385	-.369		
.600	-.680	-.682	-.684	-.687	-.685	-.350	-.333	-.295	-.257
.650	-.697	-.692	-.696			-.318	-.318		
.700	-.771	-.744	-.760	-.755	-.631	-.297	-.296	-.252	-.238
.750	-.581	-.593	-.566			-.269	-.272		
.800	-.266	-.258	-.254	-.267	-.367	-.238	-.232	-.210	-.236
.850	-.156	-.159	-.161			-.209	-.194		
.900	-.097	-.107	-.103	-.106	-.177	-.164		-.102	-.196
.950						-.103	-.088		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.895	.851				.978		-.5600	
-.8350			-.526		-.786			-.3700	
-.8000		-.212	-.241		-.327	-.287	-.271	-.2900	.232
-.7400	-.044		-.188	-.216		-.269		-.2100	.161
-.6750		.005	-.115		-.197	-.211		-.1300	.200
-.6000		-.044		-.137		-.191	-.135	-.0500	.282
-.5250		-.039	-.042	-.074	-.124	-.130		.0300	-.170
-.4500		-.012		-.030		-.081		.1100	-.691
-.3750		.043	.005		-.082	-.091		.1900	-.902
-.3000		.070				-.110		.2700	-.576
-.2250		.111	.079		-.086		-.018	.3500	-.564
-.1500		.146	.081	-.034	-.107	-.124		.4300	-.490
-.0750		.119		-.084		-.156		.5100	-.409
.0000		-.041				-.177			
.0750		-.377	-.340	-.296			-.291		
.1500			-.598		-.315	-.270			
.2250		-.893	-.789	-.570	-.378	-.318			
.3000		-.817	-.860	-.657		-.287			
.3500		-.537	-.605	-.494	-.250	-.180	-.435		
.3700		-.450		-.322	-.179	-.143			

APPENDIX

TABLE A8.- Concluded

CONFIGURATION 5

MACH = .849

ALPHA = 1.12

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.776	-.687	-.426			-.244	-.170		
.050	-.806	-.731	-.797	-.658	-.689	-.730	-.589	-.887	-.663
.075	-.800	-.817	-.845			-.858	-.651		
.100	-.855	-.830	-.829	-.784	-.779	-.799	-.594	-.517	-.488
.150	-.828	-.830	-.817			-.689	-.705		
.200	-.800	-.780	-.785	-.730	-.746	-.896	-.808	-.648	-.601
.250	-.755	-.739	-.760			-.917	-.804		
.300	-.747	-.733	-.732	-.752	-.769	-.806	-.927	-.691	-.648
.350	-.704	-.729	-.727			-.622	-.924		
.400	-.737	-.716	-.739	-.739	-.800	-.529	-.763	-.800	-.699
.450	-.729	-.736	-.731			-.488	-.568		
.500	-.747	-.741	-.744	-.764	-.805	-.383	-.394	-.737	-.796
.550	-.764	-.751	-.759			-.310	-.337		
.600	-.768	-.777	-.781	-.778	-.808	-.286	-.300	-.279	-.268
.650	-.777	-.774	-.779			-.266	-.279		
.700	-.495	-.456	-.474	-.428	-.367	-.248	-.249	-.225	-.225
.750	-.254	-.251	-.265			-.213	-.218		
.800	-.169	-.176	-.188	-.200	-.233	-.172	-.184	-.183	-.216
.850	-.124	-.128	-.138			-.126	-.136		
.900	-.079	-.092	-.106	-.111	-.111	-.069		-.021	-.173
.950						-.013	.006		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.828	.778				1.025		-.5600	
-.8350			-.673		-.643			-.3700	
-.8000		-.223	-.252		-.315	-.285	-.266	-.2900	.224 .088
-.7400	-.056		-.195	-.216		-.239		-.2100	.170 .162
-.6750		-.016	-.121		-.185	-.193		-.1300	.216 .199
-.6000		-.041		-.134		-.182	-.134	-.0500	.302 .167
-.5250		-.034	-.041	-.069	-.117	-.124		.0300	-.142 -.107
-.4500		-.004		-.024		-.076		.1100	-.596 -.521
-.3750		.049	.012		-.073	-.083		.1900	-.808 -.823
-.3000		.079				-.102		.2700	-.521
-.2250		.121	.089		-.075		-.007	.3500	-.653 -.524
-.1500		.155	.094	-.021	-.095	-.113		.4300	-.482
-.0750		.131		-.060		-.143		.5100	-.381
.0000		-.021				-.159			
.0750		-.338	-.296	-.253			-.254		
.1500			-.539		-.284	-.235			
.2250		-.814	-.713	-.510	-.339	-.268			
.3000		-.855	-.829	-.573		-.236			
.3500		-.541	-.795	-.486	-.243	-.149	-.391		
.3700		-.455		-.351	-.174	-.116			

APPENDIX

TABLE A9.- AERODYNAMIC PRESSURE-COEFFICIENT DATA FOR CONFIGURATION 5
WITH NATURAL TRANSITION

CONFIGURATION 5 MACH = .202 ALPHA = .03

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.987	-.972	-.898			-.371	-.295		
.050	-.790	-.776	-.726	-.731	-.712	-.524	-.377	-.254	-.248
.075	-.663	-.672	-.648			-.524	-.377		
.100	-.584	-.496	-.555	-.569	-.594	-.494	-.389	-.266	-.260
.150	-.471	-.496	-.471			-.465	-.371		
.200	-.442	-.422	-.437	-.437	-.461	-.465	-.319	-.231	-.237
.250	-.407	-.387	-.412			-.345	-.272		
.300	-.383	-.378	-.387	-.402	-.402	-.299	-.250	-.200	-.219
.350	-.345	-.351	-.353			-.236	-.219		
.400	-.336	-.336	-.338	-.340	-.353	-.178	-.188	-.205	-.195
.450	-.338	-.334	-.327			-.120	-.152		
.500	-.329	-.329	-.329	-.336	-.340	-.070	-.130	-.197	-.207
.550	-.329	-.334	-.334			-.031	-.123		
.600	-.334	-.336	-.340	-.332	-.345	.022	-.106	-.169	-.164
.650	-.336	-.342	-.348			.147	.051		
.700	-.360	-.354	-.354	-.360	-.371	.207	.212	.202	.226
.750	-.360	-.365	-.360			.231	.291		
.800	-.360	-.360	-.371	-.401	-.389	.267	.341	.224	.351
.850	-.336	-.330	-.354			.298	.377		
.900	-.260	-.260	-.272	-.307	-.330	.322		.421	.421
.950						.325	.377		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.362	.293				.688		-.5600	
-.8350			-.465		-.430			-.3700	
-.8000		-.216	-.243		-.251	-.259	-.247	-.2900	.088
-.7400	-.049		-.158	-.177		-.197		-.2100	-.002
-.6750		-.084	-.138		-.154	-.158		-.1300	-.012
-.6000		-.084		-.135		-.154	-.142	-.0500	.032
-.5250		-.088	-.088	-.100	-.119	-.115		.0300	-.181
-.4500		-.080		-.072		-.084		.1100	-.356
-.3750		-.050	-.059		-.086	-.087		.1900	-.404
-.3000		-.041				-.094		.2700	-.288
-.2250		-.030	-.046		-.086		-.067	.3500	-.247
-.1500		-.030	-.057	-.044	-.080	-.080		.4300	-.128
-.0750		-.059		-.080		-.082		.5100	.017
.0000		-.132				-.065			
.0750		-.172	-.147	-.086			-.087		
.1500			-.168		-.046	-.034			
.2250		-.247	-.150	-.062	-.016	-.012			
.3000		-.156	-.086	-.016		.025			
.3500		-.044	-.014	.036	.065	.076	.032		
.3700		.001		.067	.087	.090			

APPENDIX

TABLE A9.- Continued

CONFIGURATION 5

MACH = .202

ALPHA = 1.03

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.158	-1.188	-1.090			-.183	-.113			
.050	-.938	-.924	-.914	-.894	-.928	-.358	-.212	-.102	-.119	
.075	-.777	-.787	-.782			-.358	-.242			
.100	-.674	-.552	-.679	-.689	-.713	-.364	-.271	-.166	-.160	
.150	-.547	-.552	-.557			-.376	-.300			
.200	-.508	-.503	-.523	-.498	-.523	-.393	-.259	-.177	-.160	
.250	-.454	-.444	-.449			-.285	-.218			
.300	-.425	-.425	-.425	-.435	-.449	-.254	-.211	-.153	-.166	
.350	-.384	-.382	-.402			-.204	-.173			
.400	-.380	-.371	-.380	-.380	-.386	-.156	-.149	-.170	-.165	
.450	-.371	-.363	-.360			-.101	-.132			
.500	-.360	-.358	-.360	-.367	-.367	-.053	-.117	-.165	-.180	
.550	-.358	-.354	-.354			-.019	-.115			
.600	-.356	-.363	-.365	-.358	-.367	.029	-.096	-.134	-.127	
.650	-.358	-.358	-.352			.153	.057			
.700	-.381	-.370	-.370	-.376	-.387	.232	.227	.230	.220	
.750	-.381	-.370	-.370			.265	.306			
.800	-.376	-.364	-.376	-.410	-.399	.299	.359	.196	.354	
.850	-.346	-.341	-.352			.325	.397			
.900	-.259	-.259	-.276	-.300	-.317	.347		.416	.416	
.950						.347	.395			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.131	.070				.784		-.5600	
-.8350			-.745		-.370			-.3700	
-.8000		-.254	-.273		-.227	-.215	-.254	-.2900	.075
-.7400	-.076		-.173	-.177		-.173		-.2100	.020
-.6750		-.103	-.146		-.146	-.138		-.1300	.044
-.6000		-.088		-.134		-.138	-.138	-.0500	.116
-.5250		-.088	-.092	-.095	-.103	-.107		.0300	-.072
-.4500		-.080		-.072		-.076		.1100	-.277
-.3750		-.046	-.048		-.068	-.075		.1900	-.337
-.3000		-.030				-.082		.2700	-.205
-.2250		-.012	-.021		-.068		-.048	.3500	-.207
-.1500		-.001	-.019	-.028	-.061	-.072		.4300	-.106
-.0750		-.017		-.052		-.072		.5100	.034
.0000		-.073				-.053			
.0750		-.075	-.102	-.059			-.053		
.1500			-.115		-.030	-.017			
.2250		-.192	-.100	-.032	.001	.000			
.3000		-.115	-.050	.005		.039			
.3500		-.012	.016	.058	.078	.090	.056		
.3700		.032		.088	.102	.106			

APPENDIX

TABLE A9.- Continued

CONFIGURATION 5

MACH = .199

ALPHA = 2.03

WING CP										
UPPER SURFACE							LOWER SURFACE			
Y/D	N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C										
.025	-1.314	-1.435	-1.395				.004	.154		
.050	-1.030	-1.116	-1.096	-1.152	-1.096		.182	.062	.034	.040
.075	-.879	-.934	-.919				.219	.152		
.100	-.788	-.621	-.767	-.828	-.863		.255	.176	-.074	-.050
.150	-.626	-.621	-.621				.297	.200		
.200	-.570	-.565	-.585	-.590	-.605		.291	.182	-.110	-.098
.250	-.525	-.514	-.519				.220	.153		
.300	-.474	-.474	-.499	-.499	-.509		.202	.143	-.108	-.122
.350	-.417	-.428	-.435				.158	.120		
.400	-.401	-.403	-.412	-.414	-.417		.105	.110	-.140	-.118
.450	-.392	-.390	-.392				.056	.096		
.500	-.387	-.385	-.385	-.394	-.396		.021	.086	-.130	-.140
.550	-.381	-.379	-.383				.001	.083		
.600	-.374	-.374	-.383	-.374	-.392		.033	.061	-.088	-.096
.650	-.375	-.375	-.381				.160	.011		
.700	-.393	-.387	-.387	-.393	-.412		.259	.256	.241	.172
.750	-.387	-.375	-.387				.301	.330		
.800	-.387	-.375	-.381	-.412	-.412		.330	.375	.182	.360
.850	-.351	-.345	-.351				.358	.409		
.900	-.261	-.255	-.279	-.297	-.321		.375		.424	.427
.950							.365	.405		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.138	-.242				.868		-.5600	
-.8350			-.610		-.310			-.3700	
-.8000		-.286	-.302		-.198	-.158	-.254	-.2900	.061
-.7400	-.094		-.190	-.174		-.130		-.2100	.034
-.6750		-.106	-.158		-.130	-.118		-.1300	.076
-.6000		-.090		-.134		-.118	-.134	-.0500	.169
-.5250		-.078	-.086	-.094	-.098	-.090		.0300	.036
-.4500		-.066		-.066		-.058		.1100	-.153
-.3750		-.025	-.030		-.051	-.051		.1900	-.231
-.3000		-.011				-.061		.2700	-.136
-.2250		.011	.002		-.053		-.034	.3500	-.163
-.1500		.036	.011	-.014	-.046	-.051		.4300	-.078
-.0750		.041		-.025		-.054		.5100	.051
.0000		-.002				-.039			
.0750		-.041	-.037	-.032			-.021		
.1500			-.055		-.009	-.006			
.2250		-.130	-.053	-.007	.011	.011			
.3000		-.065	-.011	.025		.054			
.3500		.025	.045	.081	.100	.106	.076		
.3700		.063		.115	.120	.121			

APPENDIX

TABLE A9.- Continued

CONFIGURATION 5 MACH = .198 ALPHA = 3.04

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
N										
X/C										
.025	-1.500	-1.709	-1.643			.197	.270			
.050	-1.195	-1.317	-1.282	-1.399	-1.383	-.013	.089	.149	.197	
.075	-1.012	-.981	-1.007			-.068	-.026			
.100	-.874	-.701	-.854	-.894	-.940	-.135	-.050	.029	.059	
.150	-.691	-.701	-.716			-.196	-.129			
.200	-.635	-.625	-.670	-.645	-.660	-.202	-.129	-.044	-.038	
.250	-.563	-.563	-.574			-.161	-.081			
.300	-.533	-.538	-.533	-.543	-.553	-.143	-.101	-.058	-.068	
.350	-.469	-.471	-.478			-.108	-.088			
.400	-.451	-.455	-.453	-.457	-.471	-.081	-.071	-.091	-.086	
.450	-.439	-.435	-.430			-.043	-.061			
.500	-.424	-.417	-.421	-.428	-.430	-.006	-.053	-.098	-.116	
.550	-.410	-.408	-.410			.014	-.046			
.600	-.401	-.401	-.408	-.403	-.417	.041	-.026	-.058	-.071	
.650	-.402	-.408	-.402			.161	-.006			
.700	-.426	-.414	-.414	-.420	-.426	.278	.285	.196	.101	
.750	-.414	-.408	-.402			.323	.343			
.800	-.402	-.396	-.402	-.420	-.420	.355	.390	.153	.368	
.850	-.365	-.359	-.372			.383	.425			
.900	-.274	-.274	-.287	-.299	-.323	.400		.422	.425	
.950						.388	.410			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.489	-.461				.918		-.5600	
-.8350			-.618		-.255			-.3700	
-.8000		-.332	-.336		-.162	-.118	-.251	-.2900	.054
-.7400	-.126		-.215	-.179		-.098		-.2100	.047
-.6750		-.118	-.162		-.110	-.090		-.1300	.110
-.6000		-.090		-.130		-.098	-.138	-.0500	.238
-.5250		-.078	-.090	-.090	-.078	-.074		.0300	.147
-.4500		-.062		-.062		-.046		.1100	-.074
-.3750		-.067	-.027		-.042	-.039		.1900	-.164
-.3000		-.004				-.049		.2700	-.051
-.2250		.032	.014		-.042		-.021	.3500	-.124
-.1500		.064	.041	-.002	-.035	-.044		.4300	-.051
-.0750		.071		-.009		-.049		.5100	.077
.0000		.046				-.026			
.0750		.009	.007	.003			.002		
.1500			-.016		.000	.007			
.2250		-.065	-.013	.023	.018	.024			
.3000		-.020	.021	.053		.062			
.3500		.053	.078	.100	.114	.117	.097		
.3700		.082		.130	.137	.135			

APPENDIX

TABLE A9.- Continued

CONFIGURATION 5 MACH = .198 ALPHA = 4.04

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.669	-1.954	-1.964			.396	.414			
.050	-1.338	-1.404	-1.429	-1.353	-1.709	.149	.209	.276	.288	
.075	-1.124	-1.124	-1.139			.017	.095			
.100	-.976	-.793	-.981	-1.042	-1.063	-.020	.029	.125	.137	
.150	-.777	-.793	-.798			-.092	-.056			
.200	-.701	-.701	-.732	-.721	-.747	-.104	-.044	.011	.017	
.250	-.640	-.625	-.630			-.113	-.028			
.300	-.584	-.574	-.584	-.604	-.614	-.096	-.038	-.006	-.026	
.350	-.518	-.527	-.527			-.066	-.028			
.400	-.494	-.494	-.498	-.500	-.507	-.046	-.026	-.048	-.053	
.450	-.478	-.471	-.466			-.013	-.026			
.500	-.460	-.451	-.451	-.460	-.464	.011	-.026	-.066	-.083	
.550	-.444	-.437	-.437			.029	-.026			
.600	-.428	-.426	-.428	-.428	-.439	.051	-.003	-.026	-.048	
.650	-.420	-.432	-.426			.163	.021			
.700	-.432	-.432	-.426	-.426	-.438	.298	.305	.111	.076	
.750	-.426	-.420	-.414			.353	.360			
.800	-.414	-.408	-.402	-.426	-.426	.385	.400	.121	.370	
.850	-.372	-.359	-.359			.417	.435			
.900	-.281	-.274	-.287	-.293	-.311	.430		.425	.425	
.950						.405	.412			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	-.626	-.682				.946		-.5600	
-.8350			-.743		-.195			-.3700	
-.8000		-.364	-.364		-.126	-.078	-.259	-.2900	
-.7400	-.138		-.223	-.183		-.070		-.2100	.044
-.6750		-.130	-.171		-.098	-.062		-.1300	.054
-.6000		-.094		-.134		-.082	-.138	-.0500	.135
-.5250		-.078	-.082	-.094	-.074	-.062		.0300	.273
-.4500		-.054		-.058		-.030		.1100	.215
-.3750		-.056	-.018		-.030	-.031		.1900	.007
-.3000		.012				-.036		.2700	.017
-.2250		.055	.034		-.030		-.008	.3500	-.081
-.1500		.096	.064	.012	-.020	-.031		.4300	-.023
-.0750		.121		.014		-.034		.5100	.090
.0000		.105				-.013			
.0750		.082	.062	.030			.027		
.1500			.043		.018	.022			
.2250		-.009	.039	.048	.037	.039			
.3000		.025	.064	.078		.077			
.3500		.089	.105	.123	.125	.130	.120		
.3700		.119		.151	.148	.147			

APPENDIX

TABLE A9.- Continued

CONFIGURATION 5

MACH = .198

ALPHA = 5.03

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.843	-2.258	-2.274			.496	.562			
.050	-1.458	-1.432	-1.453	-1.458	-1.422	.268	.334	.394	.388	
.075	-1.230	-1.275	-1.296			.148	.214			
.100	-1.093	-.880	-1.128	-1.220	-1.199	.088	.124	.214	.214	
.150	-.870	-.880	-.900			-.008	.046			
.200	-.774	-.784	-.814	-.809	-.829	-.032	.052	.082	.100	
.250	-.693	-.693	-.708			-.043	.026			
.300	-.637	-.637	-.652	-.657	-.672	-.043	.006	.041	.028	
.350	-.556	-.561	-.565			-.023	.001			
.400	-.529	-.534	-.538	-.541	-.550	.001	.006	-.014	-.009	
.450	-.505	-.502	-.502			.016	.011			
.500	-.487	-.478	-.480	-.493	-.496	.038	.004	-.036	-.053	
.550	-.469	-.455	-.464			.046	.001			
.600	-.451	-.446	-.453	-.449	-.460	.063	.026	-.006	-.028	
.650	-.442	-.448	-.448			.165	.053			
.700	-.448	-.442	-.448	-.448	-.448	.319	.304	.078	.083	
.750	-.436	-.430	-.430			.371	.376			
.800	-.418	-.406	-.406	-.430	-.430	.410	.405	.098	.378	
.850	-.376	-.364	-.370			.430	.437			
.900	-.273	-.273	-.273	-.285	-.297	.445		.425	.428	
.950						.423	.410			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CF
-.8500	-.382	-.551				.953		-.5600	
-.8350			-1.208		-.122			-.3700	
-.8000		-.763	-.438		-.094	-.034	-.258	-.2900	.034 .039
-.7400	-.238		-.250	-.194		-.042		-.2100	.064 .094
-.6750		-.118	-.190		-.082	-.042		-.1300	.157 .152
-.6000		-.090		-.138		-.062	-.142	-.0500	.309 .207
-.5250		-.074	-.090	-.098	-.062	-.034		.0300	.299 .194
-.4500		-.054		-.058		-.010		.1100	.114 .092
-.3750		-.053	-.014		-.021	-.011		.1900	.001 .006
-.3000		.023				-.021		.2700	.092
-.2250		.073	.052		-.018		-.001	.3500	-.019 .041
-.1500		.122	.093	.020	-.011	-.016		.4300	.014
-.0750		.154		.032		-.019		.5100	.109
.0000		.152				.001			
.0750		.104	.104	.050			.052		
.1500			.079		.034	.034			
.2250		.039	.070	.073	.052	.052			
.3000		.063	.095	.100		.089			
.3500		.120	.134	.141	.138	.144	.136		
.3700		.150		.168	.163	.162			

APPENDIX

TABLE A9.- Continued

CONFIGURATION 5 MACH = .601 ALPHA = .06

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-1.097	-1.097	-.914			-.640	-.480		
.050	-.881	-.886	-.824	-.779	-.788	-.815	-.609	-.465	-.397
.075	-.759	-.752	-.742			-.789	-.636		
.100	-.660	-.527	-.639	-.641	-.619	-.724	-.615	-.442	-.415
.150	-.530	-.527	-.539			-.676	-.595		
.200	-.517	-.493	-.520	-.487	-.497	-.611	-.567	-.375	-.352
.250	-.472	-.450	-.453			-.493	-.354		
.300	-.444	-.428	-.449	-.443	-.450	-.418	-.340	-.286	-.329
.350	-.409	-.413	-.412			-.319	-.281		
.400	-.403	-.400	-.402	-.404	-.400	-.235	-.238	-.281	-.292
.450	-.402	-.401	-.386			-.154	-.206		
.500	-.399	-.397	-.386	-.401	-.396	-.087	-.179	-.262	-.294
.550	-.398	-.390	-.395			-.034	-.153		
.600	-.405	-.395	-.407	-.401	-.416	.053	-.118	-.221	-.257
.650	-.412	-.411	-.407			.145	.062		
.700	-.442	-.429	-.422	-.432	-.447	.181	.218	.190	.187
.750	-.444	-.437	-.427			.219	.285		
.800	-.441	-.428	-.433	-.481	-.473	.268	.329	.319	.380
.850	-.396	-.375	-.396			.306	.367		
.900	-.274	-.265	-.294	-.334	-.377	.331		.426	.458
.950						.333	.386		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.585	.508				.780		-.5600	
-.8350					-.408			-.3700	
-.8000		-.247	-.381		-.321	-.314	-.292	-.2900	.106
-.7400	-.070		-.183	-.210		-.242		-.2100	.011
-.6750		-.098	-.152		-.196	-.205		-.1300	.012
-.6000		-.092		-.166		-.192	-.165	-.0500	.062
-.5250		-.095	-.101	-.119	-.147	-.147		.0300	-.327
-.4500		-.092		-.090		-.109		.1100	-.574
-.3750		-.058	-.063		-.106	-.112		.1900	-.607
-.3000		-.049				-.122		.2700	-.477
-.2250		-.030	-.047		-.113		-.086	.3500	-.315
-.1500		-.025	-.054	-.095	-.112	-.113		.4300	-.152
-.0750		-.074		-.114		-.121		.5100	.002
.0000		-.194				-.103			
.0750		-.359	-.260	-.155			-.161		
.1500			-.297		-.095	-.076			
.2250		-.386	-.264	-.139	-.063	-.048			
.3000		-.250	-.178	-.081		-.001			
.3500		-.115	-.075	-.004	.039	.058	-.012		
.3700		-.059		.030	.068	.079			

APPENDIX

TABLE A9.- Continued

CONFIGURATION 5

MACH = .601

ALPHA = 1.06

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-1.401	-1.424	-1.305			-.338	-.216			
.050	-1.095	-1.141	-1.057	-1.007	-1.041	-.567	-.392	-.249	-.217	
.075	-.913	-.930	-.937			-.580	-.448			
.100	-.800	-.646	-.813	-.842	-.819	-.554	-.446	-.309	-.293	
.150	-.642	-.646	-.646			-.552	-.450			
.200	-.602	-.602	-.622	-.599	-.605	-.502	-.450	-.302	-.284	
.250	-.550	-.547	-.541			-.416	-.290			
.300	-.512	-.506	-.517	-.530	-.532	-.361	-.285	-.248	-.275	
.350	-.469	-.472	-.482			-.280	-.249			
.400	-.460	-.457	-.469	-.470	-.466	-.201	-.208	-.241	-.250	
.450	-.451	-.452	-.449			-.131	-.174			
.500	-.444	-.446	-.439	-.462	-.446	-.074	-.156	-.231	-.287	
.550	-.440	-.439	-.443			-.032	-.140			
.600	-.438	-.438	-.446	-.442	-.453	.036	-.106	-.210	-.256	
.650	-.442	-.443	-.448			.160	.068			
.700	-.470	-.459	-.459	-.465	-.472	.217	.233	.200	.212	
.750	-.467	-.462	-.458			.255	.304			
.800	-.457	-.450	-.455	-.500	-.497	.295	.350	.323	.394	
.850	-.408	-.396	-.415			.332	.391			
.900	-.281	-.278	-.305	-.341	-.386	.358		.446	.475	
.950						.355	.396			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.397	.338				.857		-.5600	
-.8350			-.414		-.381			-.3700	
-.8000		-.289	-.309		-.288	-.266	-.300	-.2900	.087
-.7400	-.097		-.206	-.213		-.216		-.2100	.021
-.6750		-.116	-.166		-.183	-.183		-.1300	.053
-.6000		-.104		-.164		-.175	-.165	-.0500	.135
-.5250		-.101	-.105	-.114	-.137	-.132		.0300	-.176
-.4500		-.094		-.084		-.099		.1100	-.445
-.3750		-.052	-.059		-.095	-.099		.1900	-.508
-.3000		-.039				-.110		.2700	-.351
-.2250		-.013	-.025		-.095		-.068	.3500	-.282
-.1500		.009	-.018	-.069	-.091	-.097		.4300	-.137
-.0750		-.017		-.080		-.104		.5100	.019
.0000		-.111				-.085			
.0750		-.245	-.180	-.114			-.111		
.1500			-.231		-.072	-.058			
.2250		-.320	-.215	-.102	-.044	-.034			
.3000		-.204	-.136	-.049		.013			
.3500		-.075	-.040	.021	.052	.072	.016		
.3700		-.025		.052	.080	.091			

APPENDIX

TABLE A9.- Continued

CONFIGURATION 5 MACH = .601 ALPHA = 2.06

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D	N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C										
.025	-1.921	-1.841	-1.781				-.089	.004		
.050	-1.363	-1.542	-1.431	-1.484	-1.457		-.339	-.181	-.082	-.080
.075	-1.085	-1.026	-1.039				-.385	-.274		
.100	-.943	-.764	-.904	-.960	-.958		-.387	-.298	-.190	-.163
.150	-.760	-.764	-.759				-.428	-.348		
.200	-.695	-.695	-.709	-.690	-.717		-.399	-.310	-.215	-.204
.250	-.629	-.624	-.623				-.337	-.243		
.300	-.586	-.588	-.584	-.599	-.599		-.301	-.228	-.186	-.218
.350	-.536	-.532	-.545				-.238	-.198		
.400	-.519	-.513	-.523	-.523	-.510		-.174	-.165	-.213	-.212
.450	-.505	-.497	-.497				-.105	-.142		
.500	-.491	-.483	-.491	-.494	-.488		-.057	-.121	-.207	-.245
.550	-.483	-.474	-.483				-.025	-.115		
.600	-.473	-.472	-.482	-.468	-.478		.032	-.088	-.191	-.219
.650	-.476	-.474	-.481				.171	.085		
.700	-.499	-.483	-.492	-.485	-.491		.248	.249	.221	.227
.750	-.492	-.478	-.479				.289	.326		
.800	-.477	-.465	-.471	-.508	-.501		.325	.373	.344	.398
.850	-.420	-.406	-.420				.363	.410		
.900	-.290	-.284	-.303	-.336	-.375		.388		.466	.475
.950							.375	.405		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.183	.164				.927		-.5600	
-.8350			-.520		-.315			-.3700	
-.8000		-.334	-.343		-.248	-.216	-.303	-.2900	.074
-.7400	-.126		-.230	-.210		-.178		-.2100	.038
-.6750		-.126	-.181		-.164	-.159		-.1300	.096
-.6000		-.108		-.158		-.157	-.165	-.0500	.206
-.5250		-.100	-.107	-.107	-.125	-.121		.0300	-.030
-.4500		-.089		-.075		-.086		.1100	-.299
-.3750		-.045	-.049		-.078	-.085		.1900	-.398
-.3000		-.026				-.097		.2700	-.240
-.2250		.006	-.004		-.078		-.049	.3500	-.234
-.1500		.036	.008	-.038	-.072	-.087		.4300	-.114
-.0750		.032		-.043		-.089		.5100	.041
.0000		-.038				-.070			
.0750		-.166	-.109	-.067			-.068		
.1500			-.157		-.049	-.042			
.2250		-.246	-.150	-.057	-.023	-.017			
.3000		-.156	-.089	-.015		.029			
.3500		-.035	-.001	.048	.070	.088	.038		
.3700		.011		.078	.099	.108			

APPENDIX

TABLE A9.- Continued

CONFIGURATION 5 MACH = .600 ALPHA = 3.06

WING CP										
UPPER SURFACE						LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000	
X/C										
.025	-2.106	-2.240	-2.246			.115	.187			
.050	-2.045	-2.112	-2.025	-2.063	-1.985	-.135	-.028			
.075	-1.224	-1.356	-1.276			-.228	-.134	.073	.075	
.100	-1.037	-.851	-.970	-1.021	-.989	-.260	-.172	-.062	-.051	
.150	-.840	-.851	-.850			-.308	-.242			
.200	-.773	-.769	-.790	-.770	-.795	-.301	-.217	-.138	-.120	
.250	-.698	-.692	-.689			-.263	-.167			
.300	-.636	-.638	-.645	-.661	-.658	-.237	-.175	-.130	-.159	
.350	-.579	-.583	-.594			-.177	-.147			
.400	-.556	-.546	-.567	-.563	-.561	-.132	-.123	-.166	-.170	
.450	-.539	-.530	-.534			-.078	-.103			
.500	-.521	-.515	-.517	-.529	-.517	-.033	-.095	-.179	-.212	
.550	-.506	-.507	-.509			-.007	-.088			
.600	-.496	-.495	-.507	-.495	-.505	.039	-.074	-.174	-.181	
.650	-.490	-.484	-.495			.176	.092			
.700	-.507	-.493	-.499	-.496	-.505	.281	.264	.245	.248	
.750	-.493	-.482	-.486			.330	.343			
.800	-.473	-.461	-.472	-.504	-.500	.364	.392	.357	.402	
.850	-.413	-.398	-.410			.397	.430			
.900	-.285	-.275	-.289	-.326	-.356	.418		.479	.475	
.950						.403	.413			

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.050	.019				.981		-.5600	
-.8350			-.783		-.311			-.3700	
-.8000		-.361	-.386		-.214	-.173	-.300	-.2900	.068
-.7400	-.134		-.238	-.213		-.152		-.2100	.058
-.6750		-.130	-.185		-.148	-.137		-.1300	.133
-.6000		-.106		-.156		-.137	-.162	-.0500	.266
-.5250		-.092	-.103	-.105	-.113	-.101		.0300	.087
-.4500		-.077		-.069		-.067		.1100	-.187
-.3750		-.032	-.035		-.067	-.066		.1900	-.292
-.3000		-.004				-.077		.2700	-.143
-.2250		.036	.020		-.062		-.029	.3500	-.189
-.1500		.074	.046	-.016	-.053	-.064		.4300	-.088
-.0750		.079		-.014		-.071		.5100	.058
.0000		.034				-.049			
.0750		-.052	-.039	-.027			-.027		
.1500			-.087		-.025	-.020			
.2250		-.174	-.089	-.021	.001	.003			
.3000		-.097	-.041	.016		.049			
.3500		.012	.032	.075	.090	.109	.070		
.3700		.052		.101	.119	.129			

APPENDIX

TABLE A9.- Continued

CONFIGURATION 5

MACH = .79P

ALPHA = .04

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.905	-.867	-.713			-.435	-.391		
.050	-.941	-.880	-.769	-.642	-.694	-.814	-.580	-.586	-.557
.075	-.892	-.884	-.853			-.937	-.710		
.100	-.856	-.848	-.857	-.781	-.799	-.853	-.792	-.710	-.656
.150	-.844	-.848	-.854			-.873	-.879		
.200	-.826	-.805	-.800	-.773	-.769	-1.083	-.975	-.806	-.738
.250	-.762	-.742	-.750			-.714	-.942		
.300	-.726	-.730	-.727	-.768	-.794	-.493	-.751	-.724	-.724
.350	-.683	-.695	-.720			-.384	-.329		
.400	-.626	-.607	-.704	-.687	-.685	-.315	-.245	-.285	-.433
.450	-.478	-.515	-.480			-.231	-.210		
.500	-.446	-.440	-.467	-.421	-.365	-.147	-.169	-.338	-.405
.550	-.440	-.445	-.418			-.110	-.119		
.600	-.471	-.448	-.475	-.457	-.454	-.052	-.051	-.278	-.300
.650	-.519	-.525	-.524			-.007	-.007		
.700	-.599	-.573	-.578	-.575	-.551	.014	.012	.214	.207
.750	-.610	-.596	-.606			.065	.067		
.800	-.488	-.484	-.542	-.664	-.678	.114	.115	.393	.381
.850	-.303	-.303	-.368			.161	.178		
.900	-.099	-.094	-.192	-.268	-.332	.197		.459	.462
.950						.214	.273		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.807	.762				.929		-.5600	
-.8350			-.700		-.887			-.3700	
-.8000		-.241	-.276		-.348	-.327	-.291	-.2900	.182
-.7400	-.060		-.179	-.214		-.264		-.2100	.103
-.6750		-.070	-.142		-.206	-.219		-.1300	.138
-.6000		-.068		-.159		-.204	-.160	-.0500	.222
-.5250		-.069	-.074	-.101	-.144	-.148		.0300	-.290
-.4500		-.052		-.062		-.104		.1100	-.723
-.3750		-.001	-.026		-.097	-.110		.1900	-.993
-.3000		.021				-.129		.2700	-.644
-.2250		.055	.028		-.112		-.054	.3500	-.401
-.1500		.084	.028	-.075	-.126	-.139		.4300	-.279
-.0750		.048		-.119		-.161		.5100	-.162
.0000		.116				-.161			
.0750		-.402	-.391	-.301			-.297		
.1500			-.641		-.236	-.178			
.2250		-.919	-.798	-.430	-.205	-.154			
.3000		-.422	-.333	-.250		-.092			
.3500		-.292	-.168	-.109	-.046	-.016	-.115		
.3700		-.243		-.052	-.002	.013			

APPENDIX

TABLE A9.- Continued

CONFIGURATION 5

MACH = .801

ALPHA = 1.08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.965	-.948	-.842			-.335	-.258		
.050	-1.038	-1.005	-.890	-.812	-.821	-.704	-.463	-.423	-.384
.075	-1.018	-.985	-.933			-.750	-.591		
.100	-.986	-.951	-.957	-.905	-.910	-.682	-.666	-.546	-.473
.150	-.946	-.951	-.967			-.756	-.683		
.200	-.928	-.911	-.929	-.896	-.930	-.992	-.836	-.655	-.481
.250	-.889	-.875	-.897			-.929	-.826		
.300	-.880	-.863	-.877	-.902	-.940	-.568	-.727	-.425	-.457
.350	-.851	-.859	-.858			-.415	-.253		
.400	-.871	-.855	-.859	-.883	-.938	-.227	-.216	-.345	-.413
.450	-.858	-.853	-.867			-.120	-.185		
.500	-.864	-.861	-.864	-.882	-.920	-.048	-.156	-.315	-.419
.550	-.874	-.865	-.867			.009	-.121		
.600	-.877	-.883	-.884	-.874	-.856	.063	-.052	-.211	-.301
.650	-.886	-.890	-.877			.112	.119		
.700	-.881	-.877	-.876	-.862	-.356	.159	.213	.234	.242
.750	-.440	-.429	-.537			.191	.252		
.800	-.282	-.277	-.272	-.386	-.517	.230	.289	.361	.444
.850	-.172	-.169	-.248			.265	.339		
.900	-.077	-.077	-.147	-.186	-.399	.295		.449	.531
.950						.298	.394		

NACELLE								PYLON		
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW	
X/C	CP							X/C	CP	
-.8500	.683	.667				.979		-.5600		
-.8350			-.858		-.735			-.3700		
-.8000		-.254	-.287		-.318	-.287	-.276	-.2900	.178 .048	
-.7400	-.076		-.181	-.206		-.228		-.2100	.123 .115	
-.6750		-.076	-.140		-.181	-.189		-.1300	.170 .147	
-.6000		-.065		-.144		-.178	-.145	-.0500	.257 .115	
-.5250		-.059	-.061	-.082	-.120	-.125		.0300	-.209 -.148	
-.4500		-.039		-.042		-.080		.1100	-.596 -.521	
-.3750		.011	-.004		-.073	-.085		.1900	-.864 -.855	
-.3000		.035				-.103		.2700	-.521	
-.2250		.073	.050		-.081		-.024	.3500	-.353 -.226	
-.1500		.105	.058	-.031	-.087	-.106		.4300	-.130	
-.0750		.075		-.064		-.127		.5100	-.020	
.0000		-.069				-.121				
.0750		-.322	-.299	-.212			-.216			
.1500			-.502		-.157	-.117				
.2250		-.813	-.630	-.289	-.133	-.094				
.3000		-.343	-.291	-.165		-.035				
.3500		-.196	-.123	-.049	.012	.037	-.058			
.3700		-.142		.000	.052	.063				

APPENDIX

TABLE A9.- Continued

CONFIGURATION 5 MACH = .798 ALPHA = 2.08

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-1.021	-1.038	-.951			-.228	-.136		
.050	-1.124	-1.098	-1.010	-.970	-.980	-.593	-.351	-.270	-.236
.075	-1.113	-1.088	-1.045			-.586	-.481		
.100	-1.092	-1.058	-1.061	-1.022	-1.038	-.574	-.536	-.382	-.322
.150	-1.044	-1.058	-1.073			-.665	-.575		
.200	-1.031	-1.017	-1.029	-1.005	-1.037	-.895	-.761	-.444	-.377
.250	-.988	-.974	-1.001			-.891	-.668		
.300	-.981	-.962	-.985	-1.006	-1.030	-.435	-.342	-.342	-.393
.350	-.944	-.957	-.964			-.278	-.258		
.400	-.973	-.951	-.963	-.987	-1.033	-.183	-.216	-.337	-.353
.450	-.960	-.960	-.962			-.104	-.183		
.500	-.970	-.959	-.969	-.990	-1.031	-.047	-.159	-.307	-.383
.550	-.977	-.967	-.975			-.002	-.143		
.600	-.986	-.984	-.985	-.973	-1.004	.073	-.098	-.209	-.332
.650	-.902	-.839	-.937			.165	.108		
.700	-.521	-.456	-.844	-1.008	-.717	.206	.249	.234	.274
.750	-.373	-.360	-.385			.241	.307		
.800	-.322	-.323	-.297	-.378	-.398	.283	.348	.353	.467
.850	-.270	-.277	-.277			.321	.389		
.900	-.229	-.241	-.252	-.167	-.341	.342		.460	.547
.950						.326	.401		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.587	.563				1.028		-.5600	
-.8350			-.972		-.573			-.3700	
-.8000		-.246	-.269		-.276	-.237	-.268	-.2900	.176
-.7400	-.080		-.181	-.199		-.190		-.2100	.138
-.6750		-.075	-.138		-.163	-.161		-.1300	.196
-.6000		-.057		-.132		-.155	-.134	-.0500	.288
-.5250		-.044	-.053	-.070	-.109	-.103		.0300	-.112
-.4500		-.024		-.029		-.060		.1100	-.494
-.3750		.027	.015		-.053	-.064		.1900	-.773
-.3000		.054				-.080		.2700	-.438
-.2250		.094	.072		-.056		.001	.3500	-.274
-.1500		.126	.083	-.004	-.060	-.079		.4300	-.103
-.0750		.103		-.028		-.095		.5100	.016
.0000		-.020				-.085			
.0750		-.257	-.218	-.153			-.137		
.1500			-.393		-.108	-.081			
.2250		-.646	-.445	-.218	-.085	-.059			
.3000		-.253	-.215	-.131		-.004			
.3500		-.116	-.082	-.021	.038	.065	-.016		
.3700		-.067		.028	.075	.092			

APPENDIX

TABLE A9.- Continued

CONFIGURATION S

MACH = .848

ALPHA = -.02

WING CP									
UPPER SURFACE					LOWER SURFACE				
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.740	-.690	-.544			-.358	-.322		
.050	-.757	-.724	-.613	-.515	-.512	-.715	-.504	-.500	-.485
.075	-.744	-.737	-.692			-.850	-.628		
.100	-.696	-.717	-.708	-.645	-.650	-.773	-.703	-.646	-.613
.150	-.699	-.717	-.715			-.786	-.807		
.200	-.703	-.658	-.680	-.649	-.668	-.966	-.909	-.743	-.681
.250	-.661	-.648	-.658			-.609	-.845		
.300	-.650	-.647	-.655	-.667	-.702	-.452	-.736	-.754	-.730
.350	-.633	-.650	-.656			-.401	-.402		
.400	-.649	-.653	-.645	-.660	-.732	-.368	-.343	-.869	-.752
.450	-.647	-.643	-.640			-.344	-.341		
.500	-.656	-.662	-.654	-.659	-.694	-.339	-.319	-.852	-.860
.550	-.666	-.687	-.671			-.326	-.285		
.600	-.690	-.687	-.688	-.685	-.686	-.312	-.257	-.282	-.418
.650	-.702	-.701	-.696			-.288	-.232		
.700	-.779	-.752	-.759	-.757	-.713	-.254	-.215	.007	-.126
.750	-.601	-.649	-.769			-.212	-.191		
.800	-.264	-.247	-.467	-.437	-.463	-.170	-.154	.200	.102
.850	-.179	-.156	-.242			-.125	-.101		
.900	-.150	-.128	-.121	-.170	-.230	-.067		.333	.267
.950						-.023	-.006		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.909	.856				.972		-.5600	
-.8350			-.529		-.794			-.3700	
-.8000		-.217	-.268		-.334	-.298	-.277	-.2900	.232
-.7400	-.028		-.149	-.194		-.261		-.2100	.157
-.6750		-.051	-.110		-.191	-.213		-.1300	.192
-.6000		-.034		-.135		-.194	-.136	-.0500	.276
-.5250		-.033	-.040	-.072	-.123	-.131		.0300	-.211
-.4500		-.011		-.030		-.081		.1100	-.642
-.3750		.047	.008		-.079	-.085		.1900	-.883
-.3000		.074				-.105		.2700	-.559
-.2250		.074	.079		-.089		-.017	.3500	-.367
-.1500		.151	.081	-.035	-.107	-.117		.4300	-.339
-.0750		.120		-.084		-.149		.5100	-.333
.0000		-.033				-.166			
.0750		-.278	-.327	-.287			-.283		
.1500			-.589		-.310	-.250			
.2250		-.879	-.776	-.555	-.364	-.289			
.3000		-.771	-.872	-.625		-.245			
.3500		-.417	-.352	-.325	-.203	-.144	-.503		
.3700		-.348		-.198	-.134	-.109			

APPENDIX

TABLE A9.- Concluded

CONFIGURATION 5 MACH = .849 ALPHA = 1.07

WING CP									
UPPER SURFACE						LOWER SURFACE			
Y/D N	0.0	0.191	0.500	1.500	3.000	0.191	0.500	1.500	3.000
X/C									
.025	-.789	-.762	-.648			-.281	-.222		
.050	-.849	-.817	-.687	-.612	-.627	-.626	-.422	-.406	-.391
.075	-.826	-.820	-.758			-.731	-.542		
.100	-.807	-.794	-.783	-.715	-.747	-.675	-.624	-.551	-.502
.150	-.782	-.794	-.796			-.701	-.704		
.200	-.775	-.737	-.779	-.733	-.763	-.916	-.803	-.661	-.611
.250	-.746	-.732	-.751			-.733	-.807		
.300	-.737	-.731	-.746	-.754	-.797	-.481	-.896	-.679	-.655
.350	-.722	-.735	-.729			-.445	-.585		
.400	-.747	-.718	-.729	-.753	-.813	-.429	-.415	-.793	-.684
.450	-.737	-.735	-.741			-.417	-.380		
.500	-.748	-.748	-.745	-.772	-.832	-.385	-.333	-.750	-.702
.550	-.764	-.753	-.764			-.340	-.286		
.600	-.777	-.771	-.781	-.777	-.806	-.296	-.242	-.279	-.387
.650	-.788	-.783	-.775			-.246	-.215		
.700	-.672	-.711	-.770	-.837	-.756	-.198	-.192	.056	-.116
.750	-.303	-.282	-.715			-.169	-.160		
.800	-.216	-.205	-.257	-.331	-.602	-.108	-.103	.240	.093
.850	-.190	-.185	-.188			-.054	-.039		
.900	-.182	-.183	-.188	-.212	-.254	-.008		.341	.254
.950						.036	.077		

NACELLE								PYLON	
ANGULAR STATION	0	22.5	50	90	135	180	270	UPPER ROW	LOWER ROW
X/C	CP							X/C	CP
-.8500	.803	.772				1.022		-.5600	
-.8350			-.677		-.651			-.3700	
-.8000		-.226	-.253		-.316	-.291	-.266	-.2900	.226 .086
-.7400	-.048		-.158	-.192		-.232		-.2100	.172 .162
-.6750		-.045	-.115		-.175	-.191		-.1300	.217 .198
-.6000		-.035		-.127		-.179	-.131	-.0500	.306 .166
-.5250		-.028	-.034	-.063	-.110	-.120		.0300	-.153 -.099
-.4500		-.003		-.019		-.070		.1100	-.553 -.492
-.3750		.049	.016		-.067	-.076		.1900	-.828 -.827
-.3000		.078				-.097		.2700	-.492
-.2250		.119	.091		-.073		-.000	.3500	-.405 -.479
-.1500		.156	.098	-.011	-.088	-.110		.4300	-.388
-.0750		.131		-.050		-.138		.5100	-.367
.0000		-.011				-.150			
.0750		-.262	-.276	-.234			-.241		
.1500			-.518		-.263	-.225			
.2250		-.821	-.700	-.481	-.310	-.251			
.3000		-.942	-.821	-.528		-.207			
.3500		-.460	-.494	-.342	-.199	-.120	-.402		
.3700		-.362		-.217	-.135	-.085			

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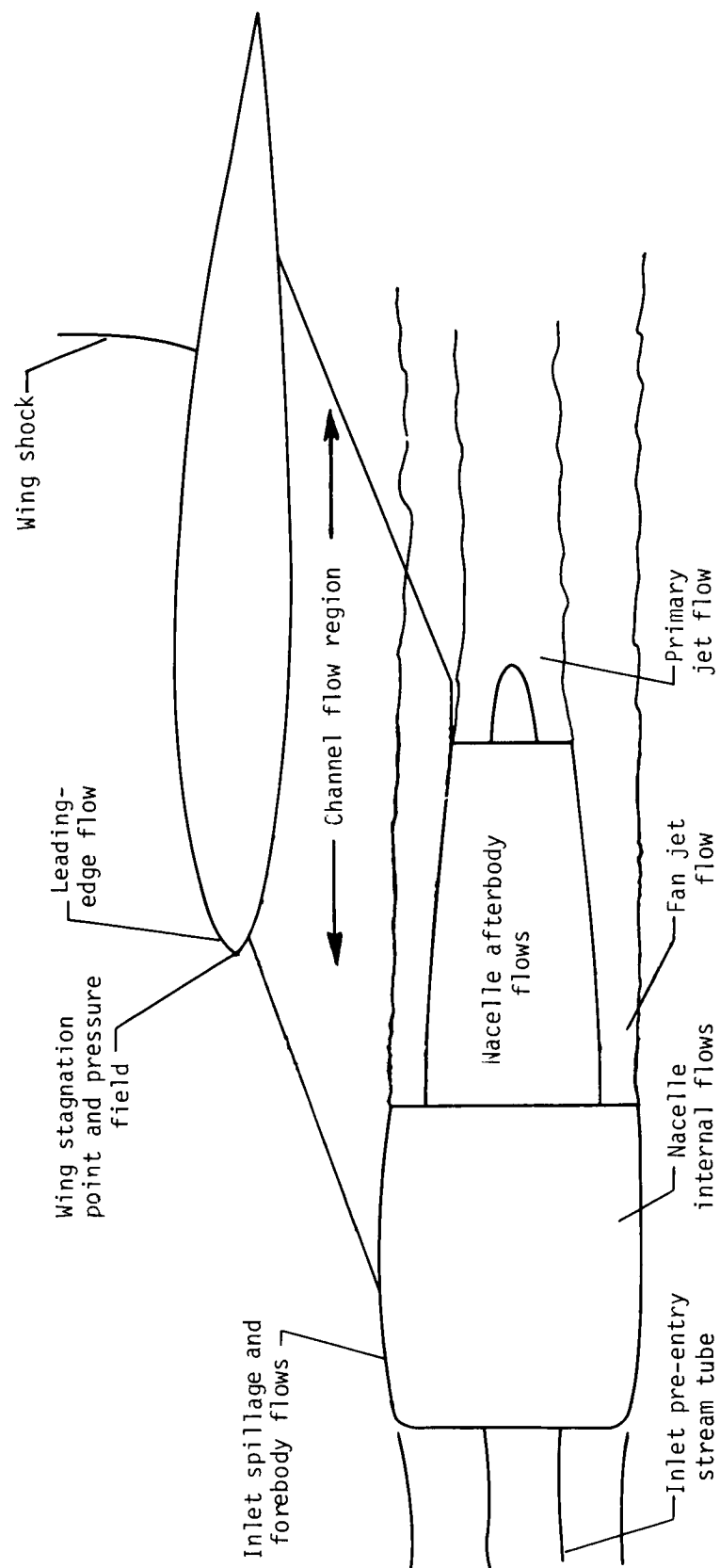
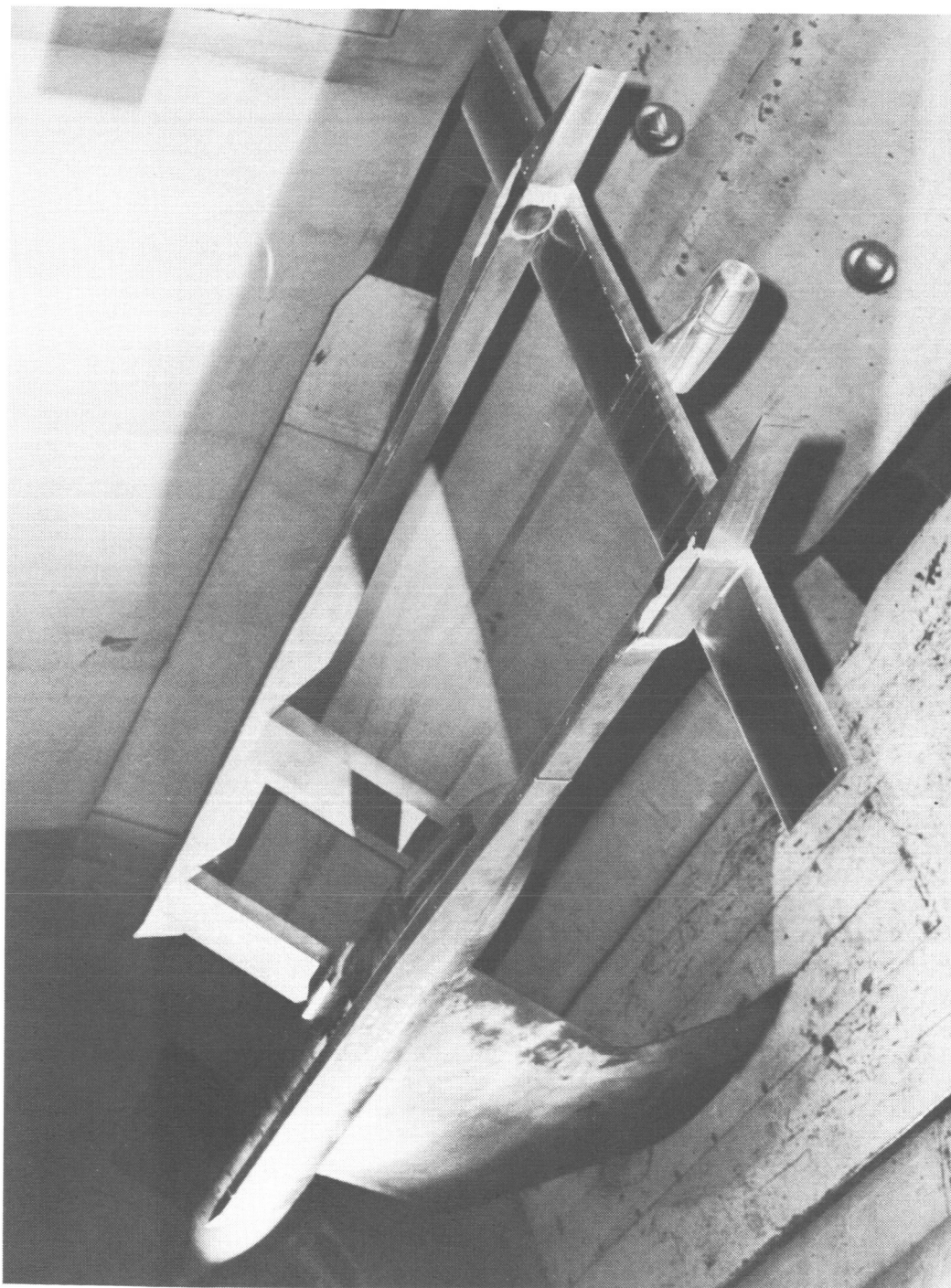


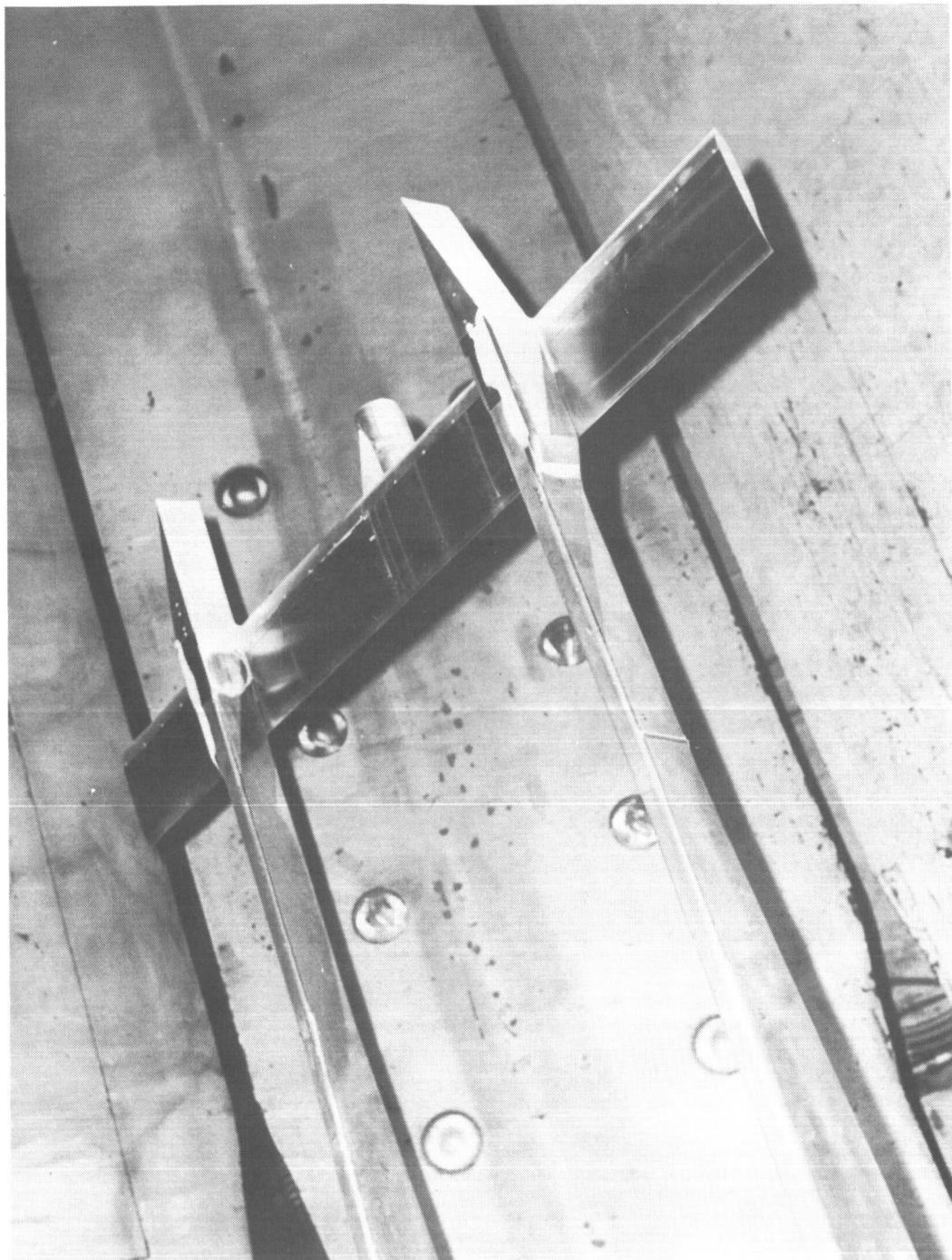
Figure 1.- Major elements of wing-pylon-nacelle flow field.



L-80-9105

(a) Front top view.

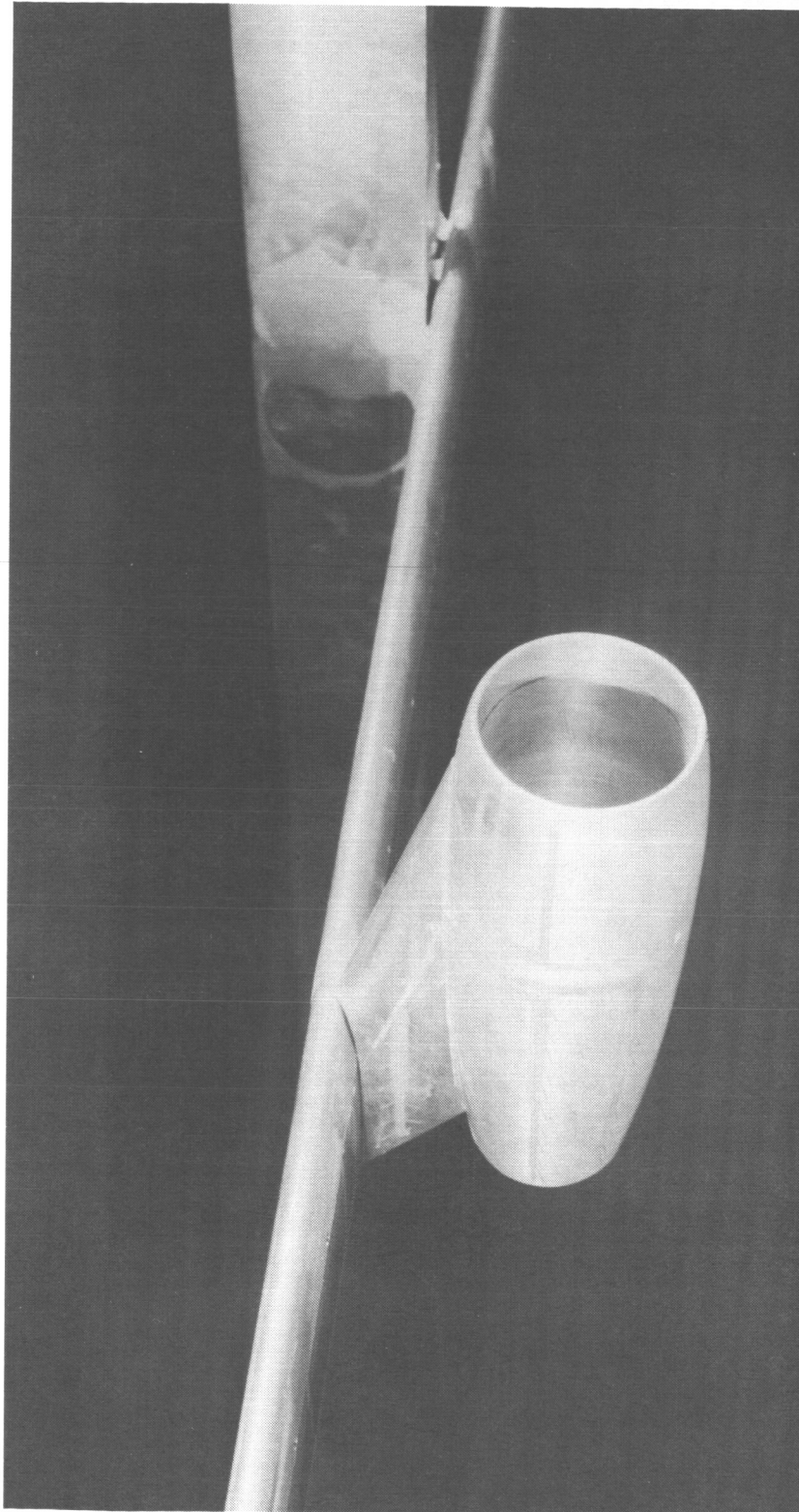
Figure 2.- Photographs of model installed in Langley 16-Foot Transonic Tunnel.



L-80-9106

(b) Rear top view.

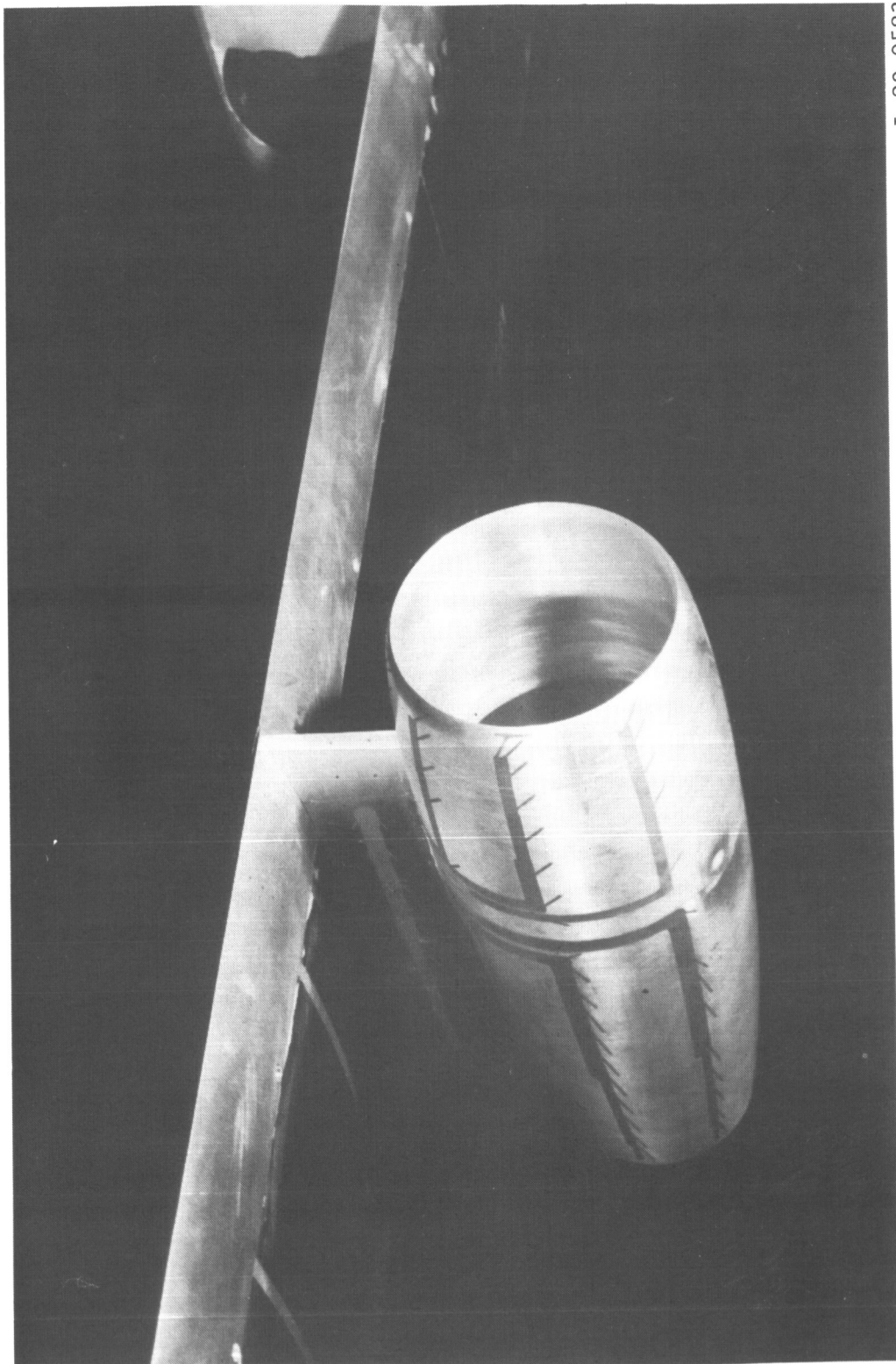
Figure 2.- Continued.



L-80-9754

(c) Front bottom view of swept-pylon configuration.

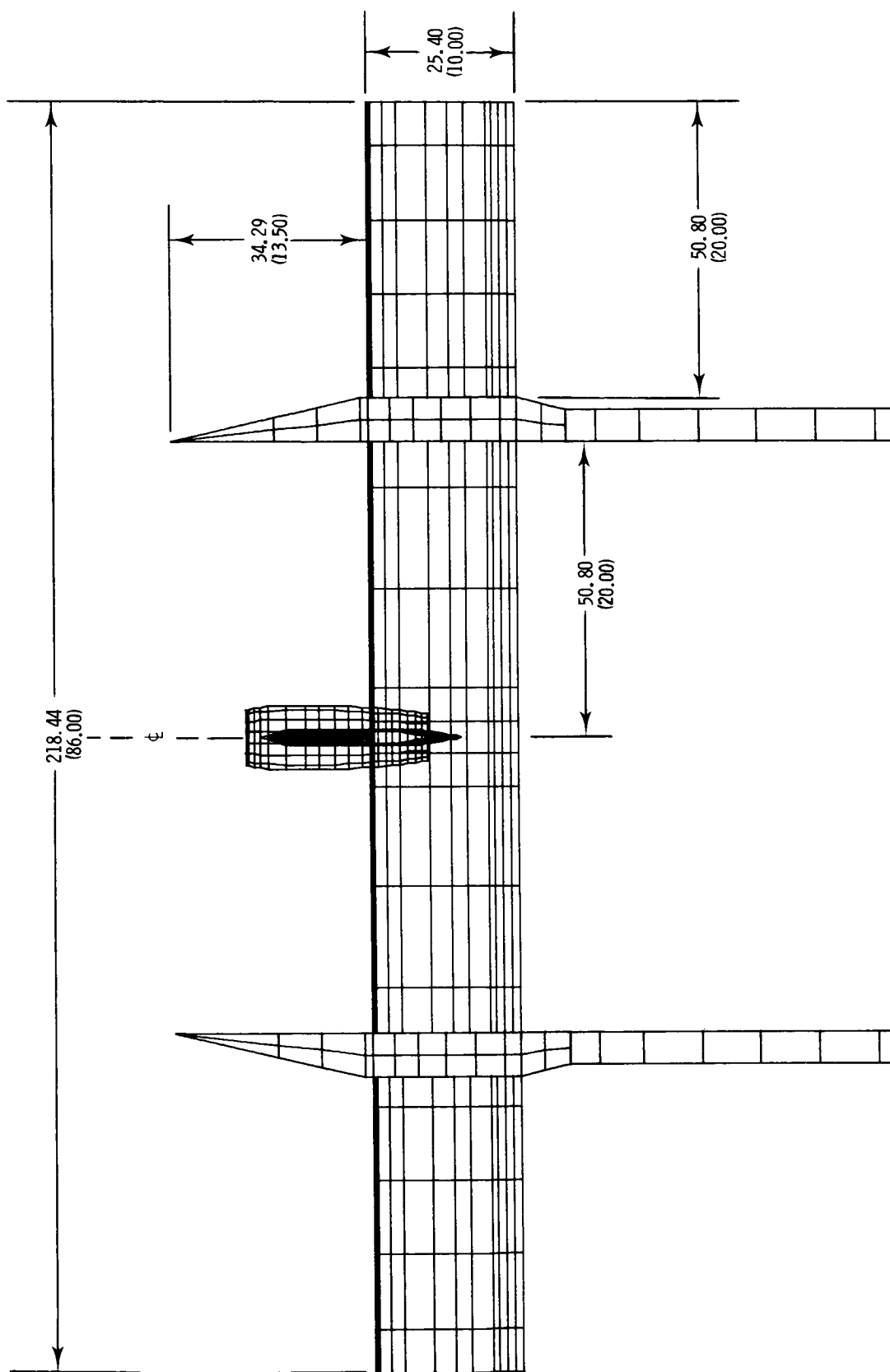
Figure 2.- Continued.



L-80-9583

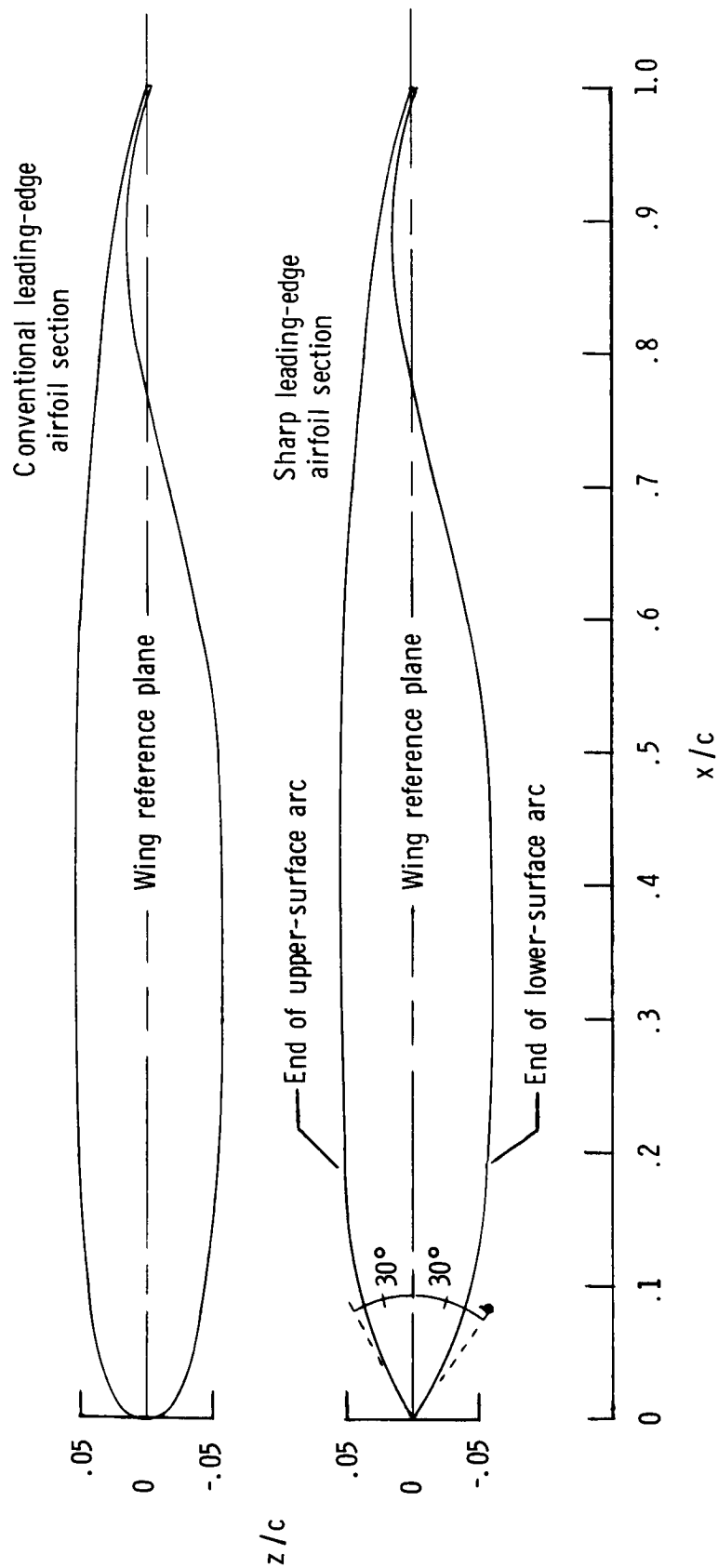
(d) Front bottom view of unswept-pylon configuration.

Figure 2.- Concluded.



(a) Planform view.

Figure 3.- Model sketch and wing coordinates. Linear dimensions are in cm (in.).



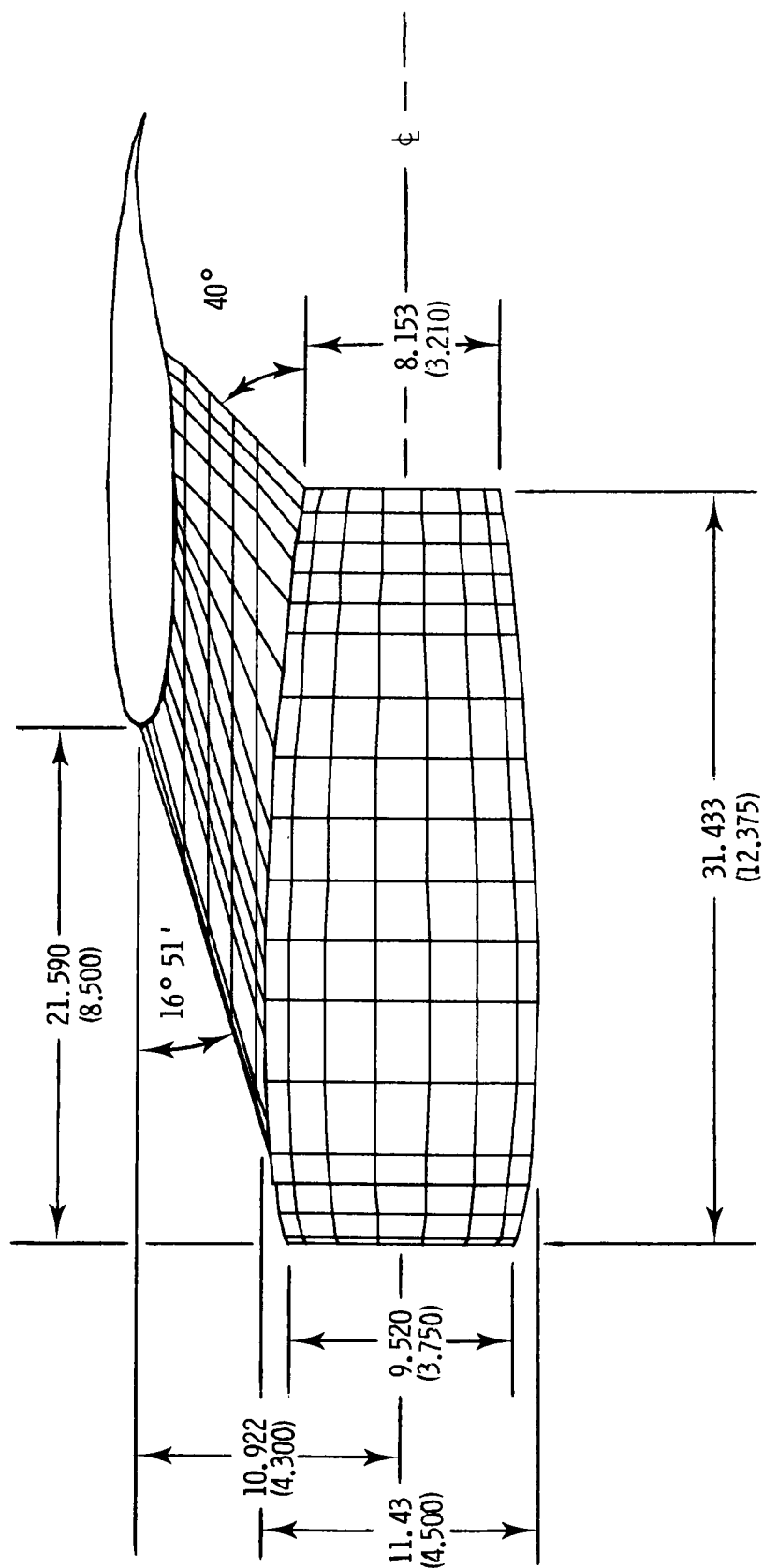
(b) Airfoil sections and coordinates.

Figure 3.-- Continued.

x/c	Conventional leading edge		Sharp leading edge	
	z/c		z/c	
	Upper surface	Lower surface	Upper surface	Lower surface
0.0075	.0176	-0.0176	0.0042	-0.0043
.0125	.0215	- .0216	.0069	- .0070
.0250	.0276	- .0281	.0132	- .0133
.0375	.0316	- .0324	.0189	- .0190
.0500	.0347	- .0358	.0240	- .0243
.0750	.0394	- .0408	.0327	- .0332
.1000	.0428	- .0444	.0395	- .0402
.1250	.0455	- .0472	.0444	- .0455
.1500	.0476	- .0493	.0476	- .0491
.1750	.0493	- .0510	.0493	- .0510
.2000	.0507	- .0522	.0507	- .0522
.2500	.0528	- .0540	.0528	- .0540
.3000	.0540	- .0548	.0540	- .0548
.3500	.0547	- .0549	.0547	- .0549
.4000	.0550	- .0541	.0550	- .0541
.4500	.0548	- .0524	.0548	- .0524
.5000	.0543	- .0497	.0543	- .0497
.5500	.0533	- .0455	.0533	- .0455
.5750	.0527	- .0426	.0527	- .0426
.6000	.0519	- .0389	.0519	- .0389
.6250	.0511	- .0342	.0511	- .0342
.6500	.0501	- .0282	.0501	- .0282
.6750	.0489	- .0217	.0489	- .0217
.7000	.0476	- .0154	.0476	- .0154
.7250	.0460	- .0098	.0460	- .0098
.7500	.0442	- .0047	.0442	- .0047
.7750	.0422	- .0001	.0422	- .0001
.8000	.0398	.0037	.0398	.0037
.8250	.0370	.0069	.0370	.0069
.8500	.0337	.0093	.0337	.0093
.8750	.0300	.0107	.0300	.0107
.9000	.0255	.0111	.0255	.0111
.9250	.0204	.0102	.0204	.0102
.9500	.0144	.0074	.0144	.0074
.9750	.0074	.0025	.0074	.0025
1.0000	-0.0008	- .0048	-0.0008	- .0048

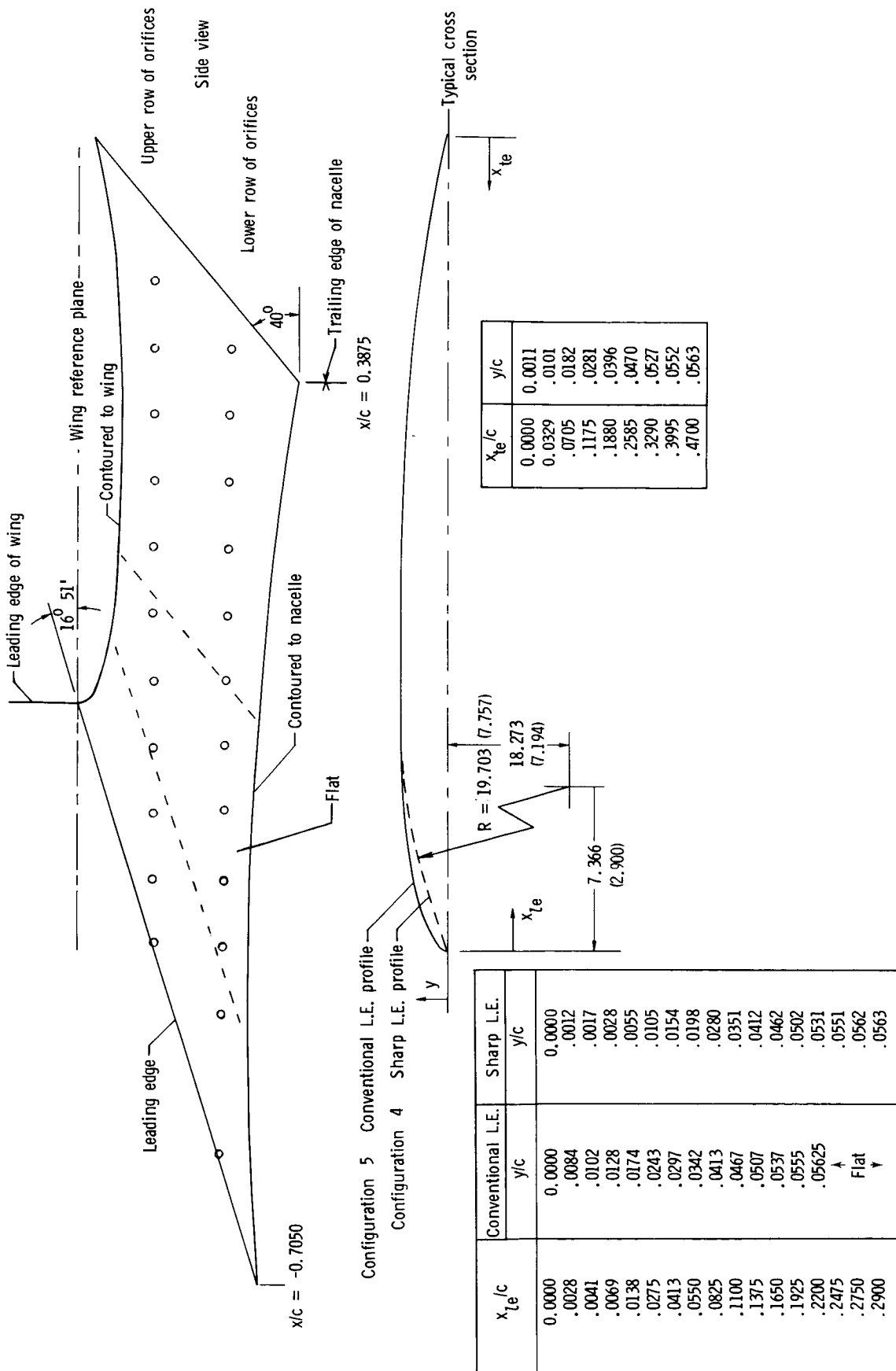
(b) Concluded.

Figure 3.- Concluded.



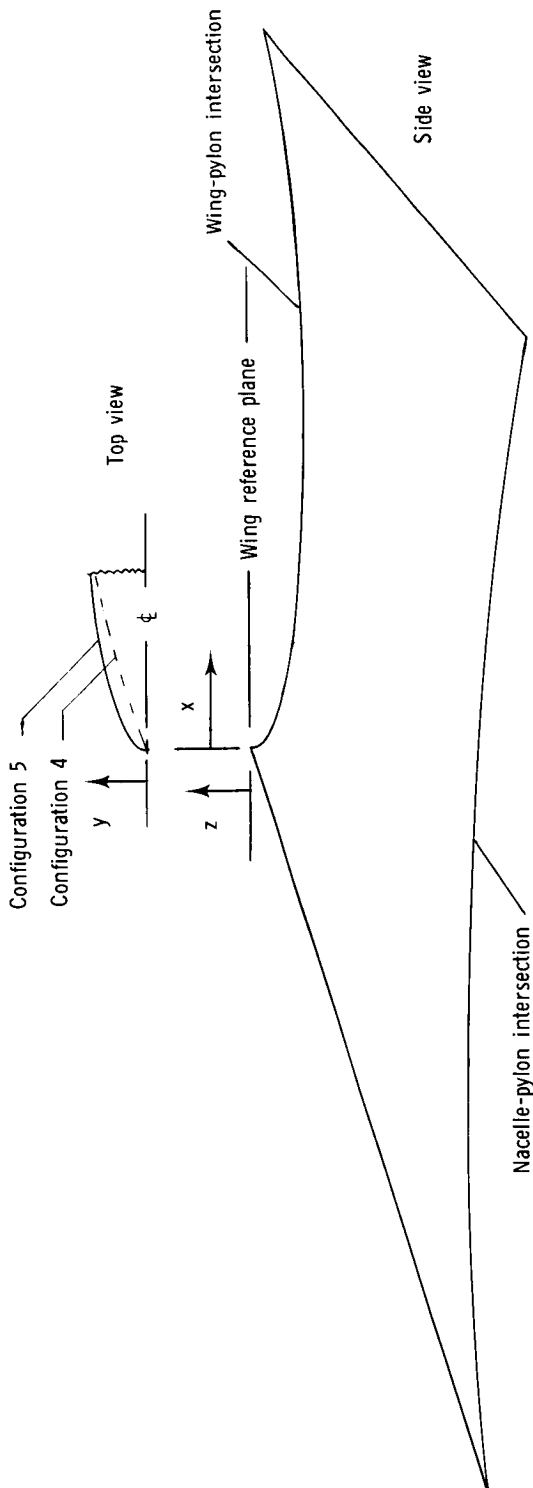
(a) Planform view.

Figure 4.- Sketches of swept pylon and nacelle. Linear dimensions are in cm (in.).



(b) Coordinates and intersections.

Figure 4.- Continued.



CONFIGURATION 4

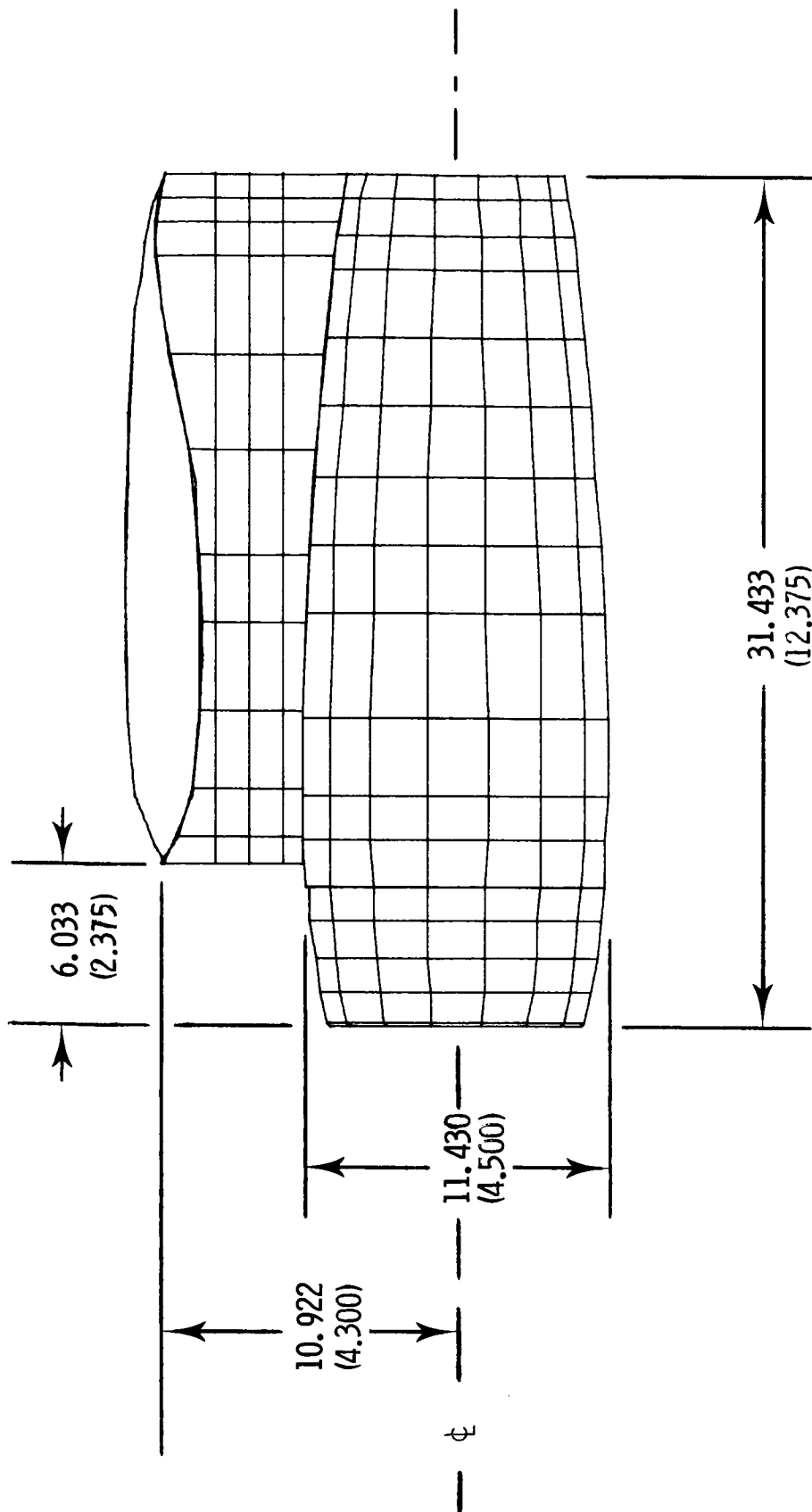
WING-PYLON			NACELLE-PYLON		
x/c	y/c	z/c	x/c	y/c	z/c
0.	0.	0.	-0.7050	0.	-0.2136
.002	.0009	.001	-.6820	.0123	-.213
.013	.0052	.007	-.6480	.0270	-.210
.047	.0173	.023	-.5100	.0513	-.205
.080	.0273	.035	-.4550	.0563	-.205
.130	.0396	.046	-.3925	.0563	-.205
.182	.0487	.051	-.3300	.0563	-.204
.272	.0535	.054	-.2430	.0563	-.208
.310	.0508	.055	-.1750	.0563	-.213
.340	.0481	.055	-.0350	.0544	-.212
.369	.0453	.055	.0879	.0506	-.223
.415	.0406	.054	.1995	.0396	-.243
.489	.0291	.050	.2700	.0281	-.251
.540	.0182	.047	.3170	.0182	-.257
.583	.0101	.042	.3546	.0101	-.263
.612	.0011	.035	.3875	.011	-.269

CONFIGURATION 5

WING-PYLON			NACELLE-PYLON		
x/c	y/c	z/c	x/c	y/c	z/c
0.	0.	0.	-0.7050	0.	-0.214
.002	.0174	.010	-.6820	.0174	-.211
.013	.0297	.020	-.6480	.0297	-.209
.047	.0537	.035	-.5100	.0537	-.205
.080	.0563	.042	-.4550	.0563	-.205
.130	.0563	.047	-.3925	.0563	-.205
.182	.0555	.051	-.3300	.0555	-.204
.272	.0535	.054	-.2430	.0535	-.208
.310	.0508	.055	-.1750	.0563	-.213
.340	.0481	.055	-.0350	.0544	-.222
.369	.0453	.055	.0879	.0506	-.233
.415	.0406	.054	.1995	.0396	-.243
.489	.0291	.050	.2700	.0281	-.251
.540	.0182	.047	.3170	.0182	-.257
.583	.0101	.042	.3546	.0101	-.263
.612	.0011	.035	.3875	.0011	-.269

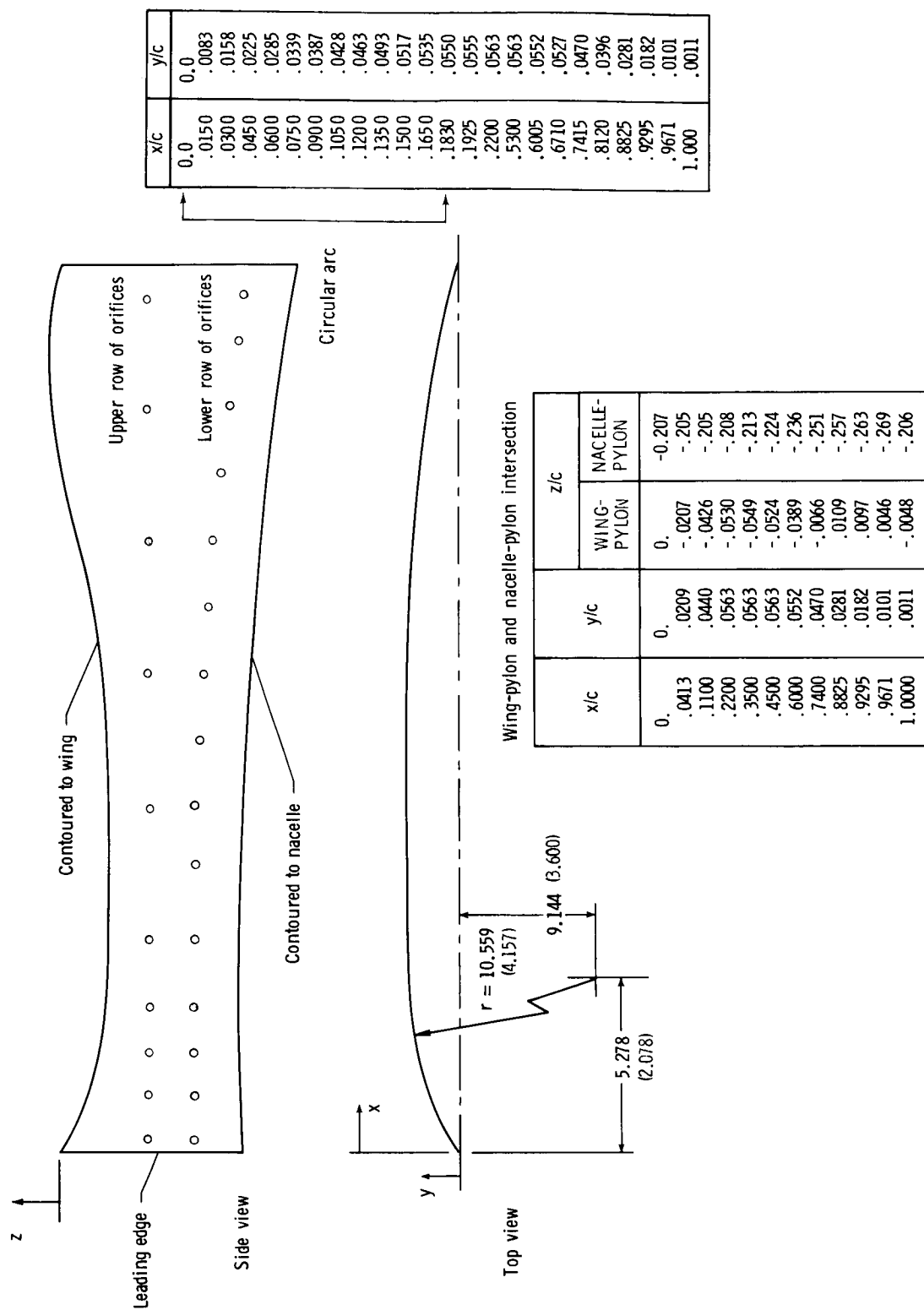
(b) Concluded.

Figure 4.- Concluded.



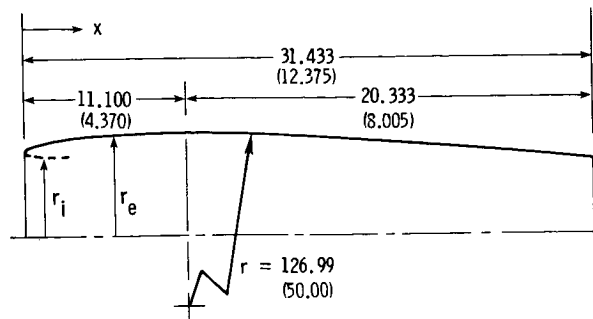
(a) Planform view.

Figure 5.- Sketches of unswept nacelle-pylon configuration. Dimensions are in cm (in.).



(b) Coordinates and intersections.

Figure 5.- Concluded.



Conventional inlet-nacelle			
External		Internal	
x/c	r_e/c	x/c	r_i/c
0.00	0.1874	0.00	0.1874
.0007	.1892	.0007	.1850
.0014	.1893	.0014	.1841
.0020	.1904	.0020	.1834
.0034	.1913	.0034	.1824
.0051	.1922	.0051	.1814
.0068	.1929	.0068	.1808
.0084	.1936	.0084	.1802
.0101	.1943	.0101	.1799
.0135	.1955	.0147	.1795
.0169	.1966	Straight line	
.0236	.1985	.0844	.1807
.0338	.2010	Circular arc	
.0506	.2044	$r/c = 1.5018$	
.0675	.2072	.1673	.1844
.0844	.2096	.2025	.1877
.1012	.2118	.2700	.1972
.1181	.2138	.3038	.2019
.1350	.2155	.3375	.2050
.1519	.2171	.7005	.2050
.1688	.2185	Circular arc	
.2025	.2209	$r/c = 1.107$	
.2362	.2227	.7475	.2030
.2700	.2240	Straight line	
.3038	.2248	1.2375	.1575
.3375	.2250	Circular arc	
.4370	.2250	$r/c = 4.9997$	
Circular arc		$r/c = 4.9997$	
1.2375	.1605		

Sharp inlet-nacelle			
External		Internal	
x/c	r_e/c	x/c	r_i/c
0.00	0.1874	0.00	0.1874
Circular arc		Circular arc	
$r/c = .1299$		$r/c = .0569$	
.0385	.2021	.0293	.1798
.0506	.2044	Straight line	
.0675	.2072	.0844	.1807
.0844	.2096	Circular arc	
.1012	.2118	$r/c = 1.5018$	
.1181	.2138	.1673	.1844
.1350	.2155	.2025	.1877
.1519	.2171	.2700	.1972
.1688	.2185	.3038	.2019
.2025	.2209	.3375	.2050
.2362	.2227	.7005	.2050
.2700	.2240	Circular arc	
.3038	.2248	$r/c = 1.107$	
.3375	.2250	.7475	.2030
.4370	.2250	1.2375	.1575
Circular arc		Circular arc	
$r/c = 4.9997$		$r/c = 4.9997$	
1.2375	.1605		

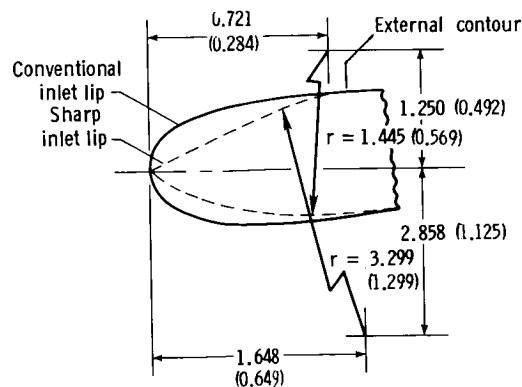


Figure 6.- Coordinates of inlet and nacelle.

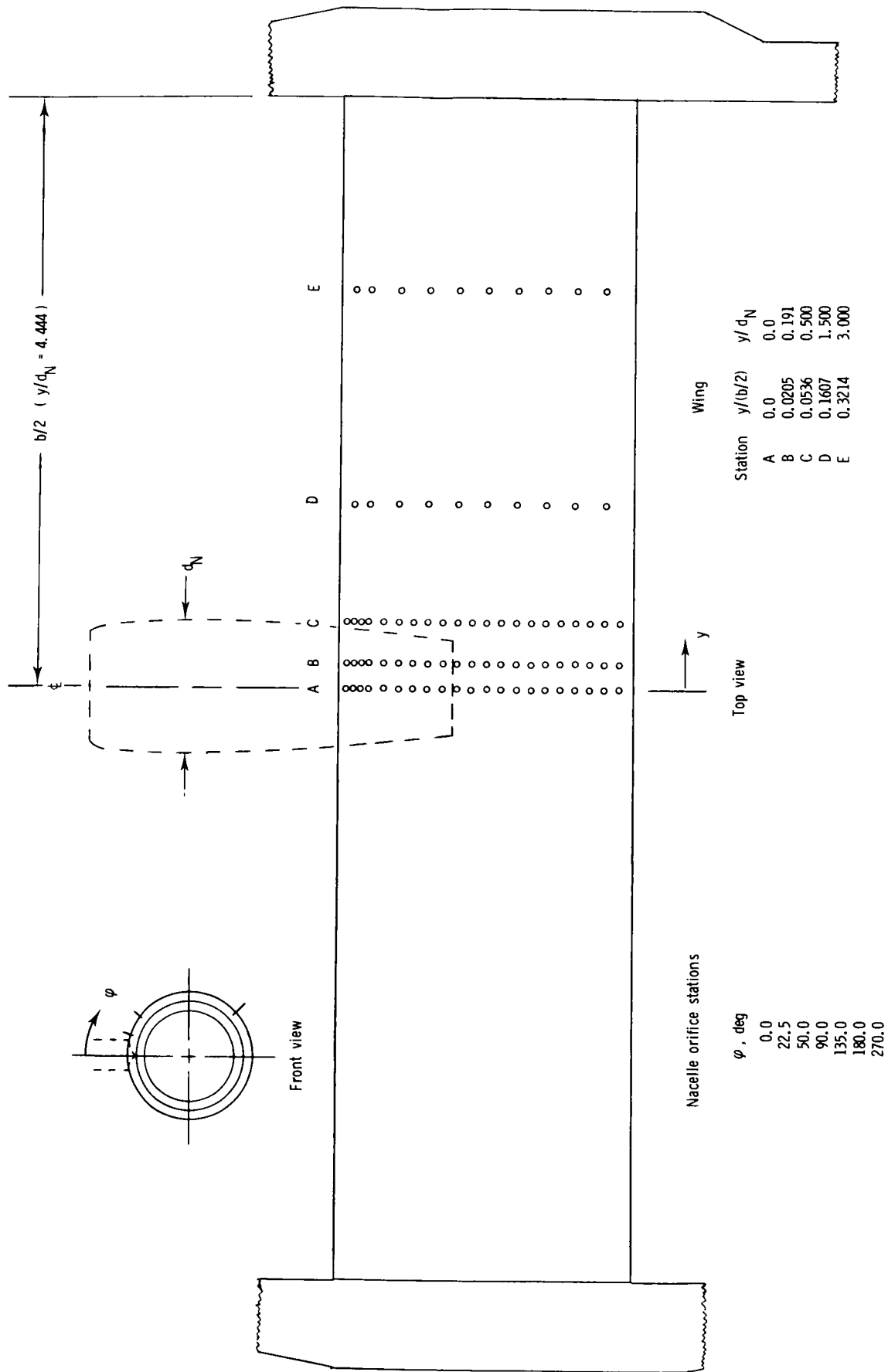
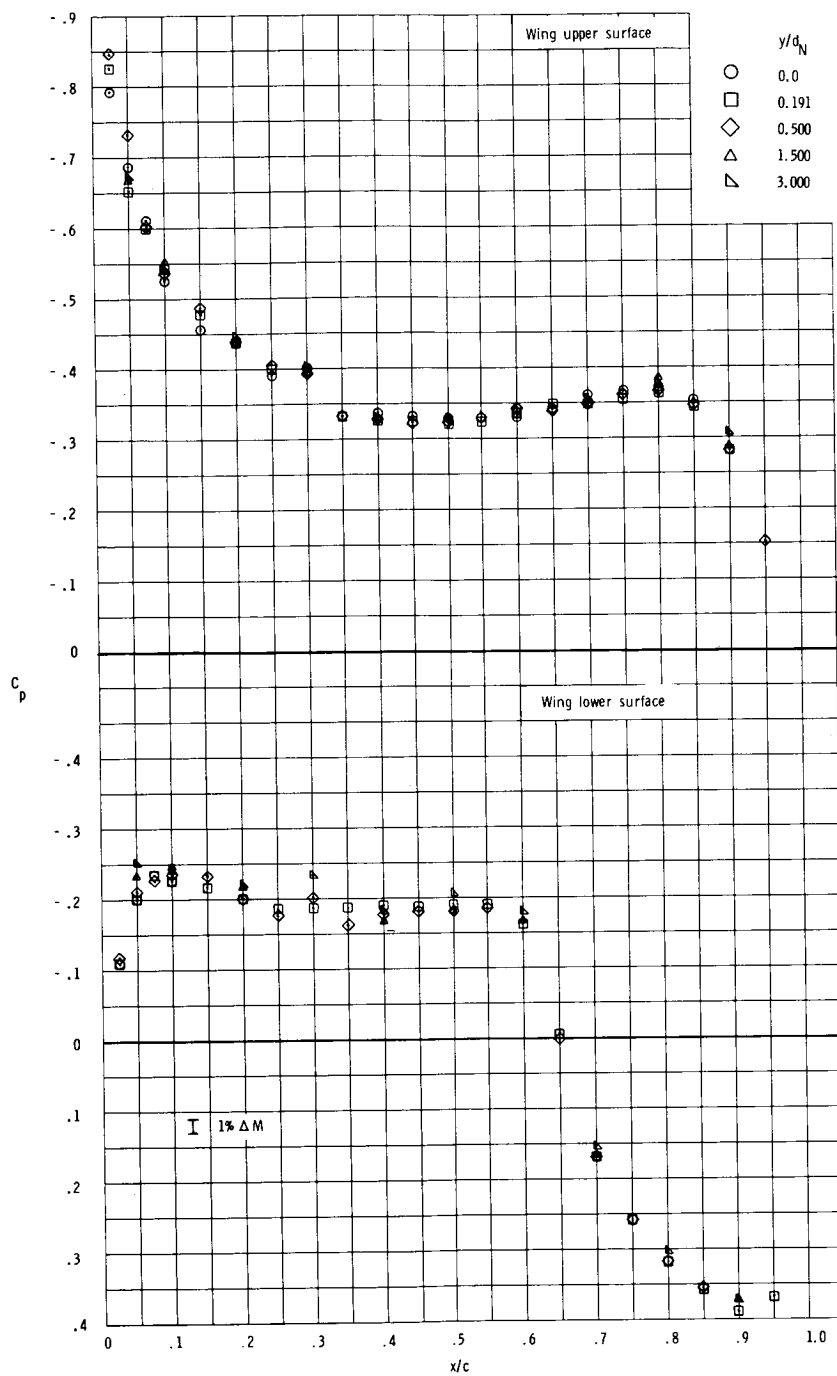
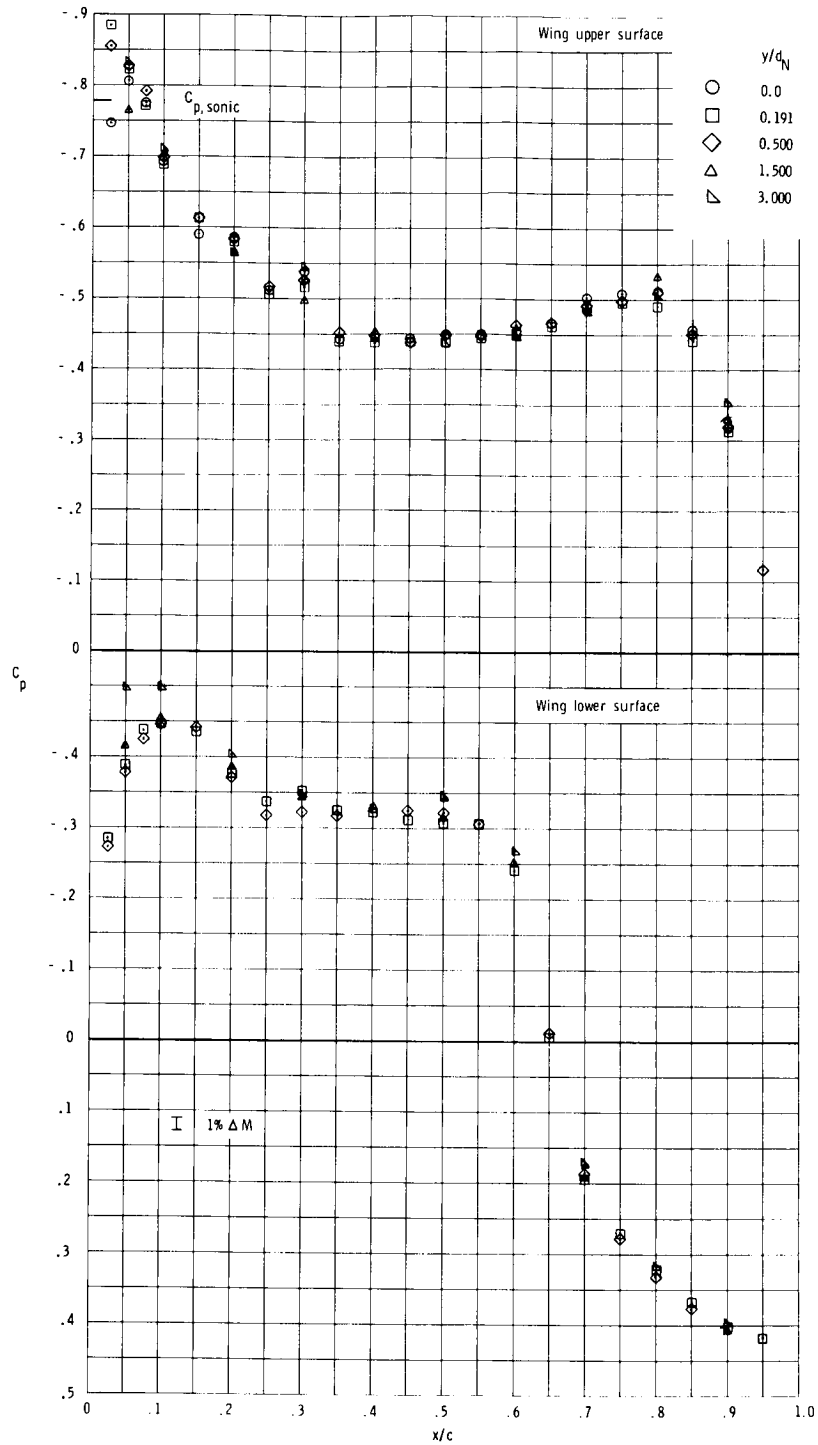


Figure 7.- Placement of model static-pressure orifice and key for nacelle and wing.



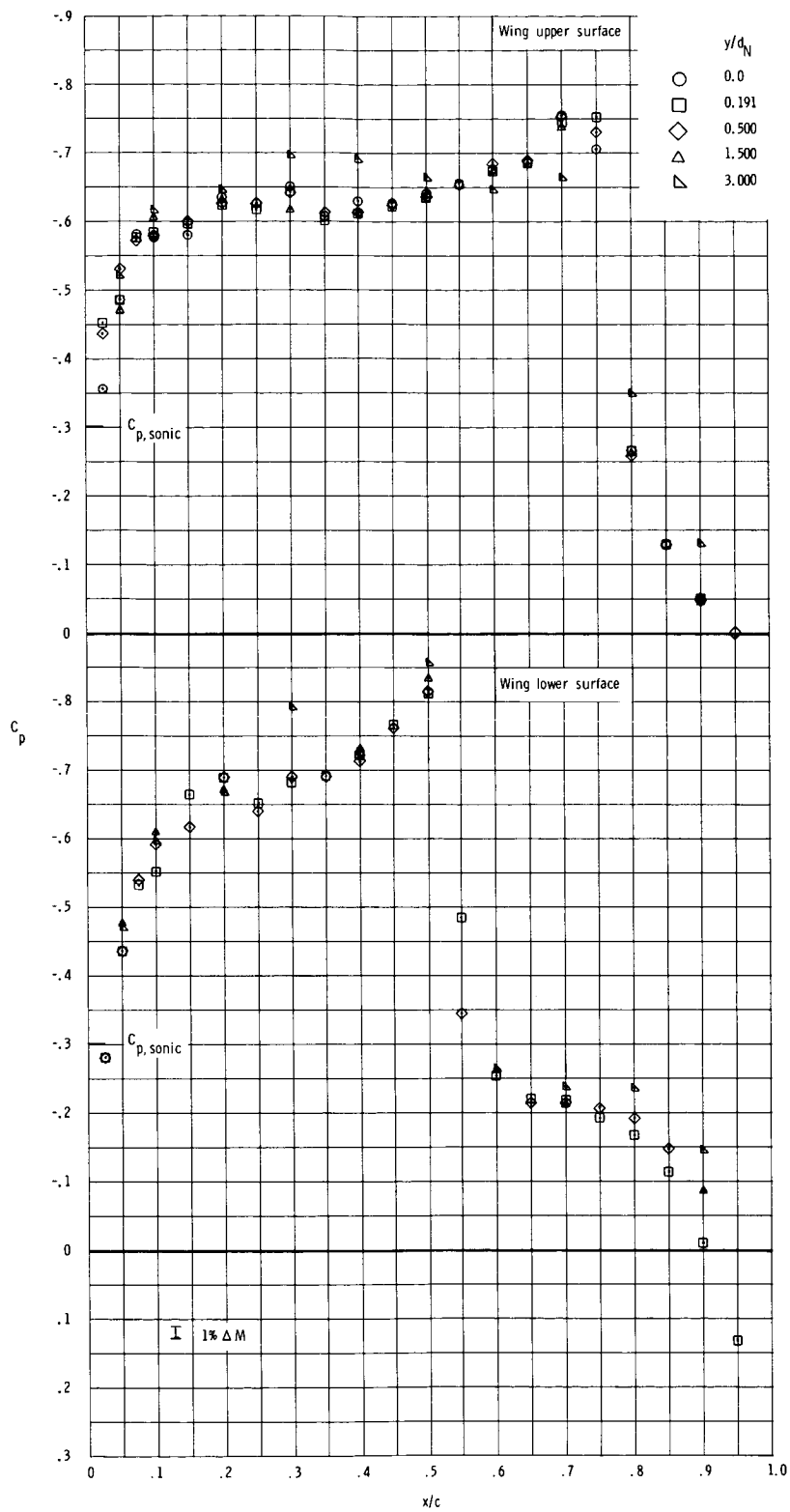
(a) $M = 0.20$; $\alpha = 0^\circ$.

Figure 8.- Experimental chordwise pressure distributions for configuration 1.



(b) $M = 0.70$; $\alpha = 0^\circ$.

Figure 8.- Continued.



(c) $M = 0.85$; $\alpha = 0^\circ$.

Figure 8.- Concluded.

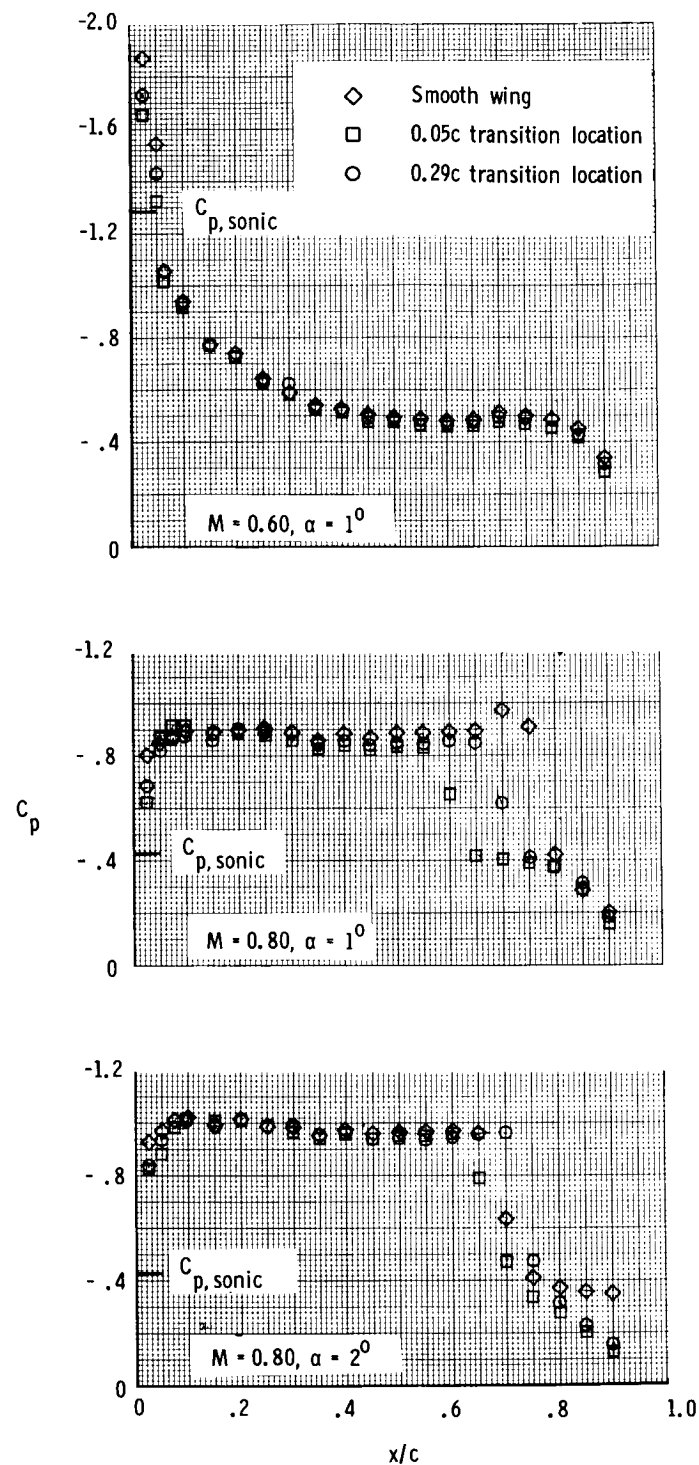


Figure 9.- Variance in experimental wing upper-surface static pressures with boundary-layer transition-strip location for clean wing with $y/(b/2) = 0.0$.

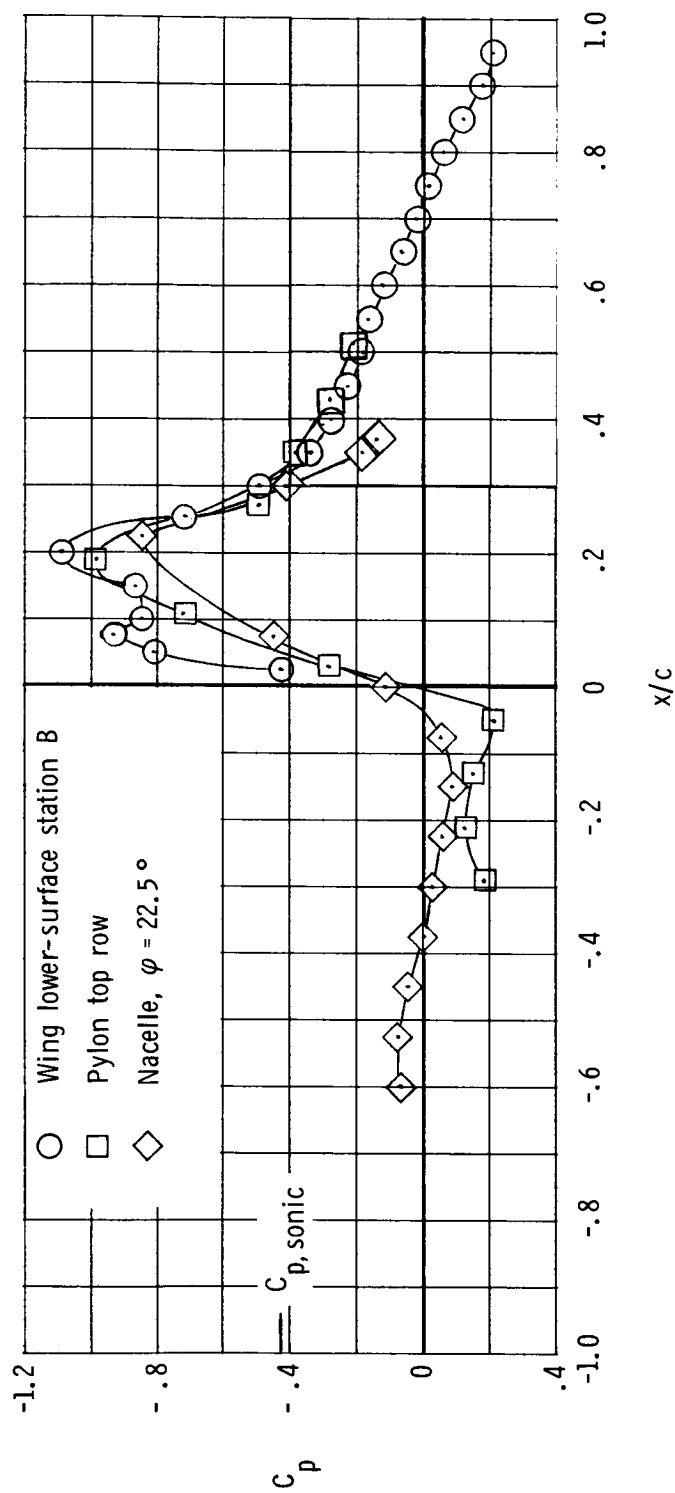
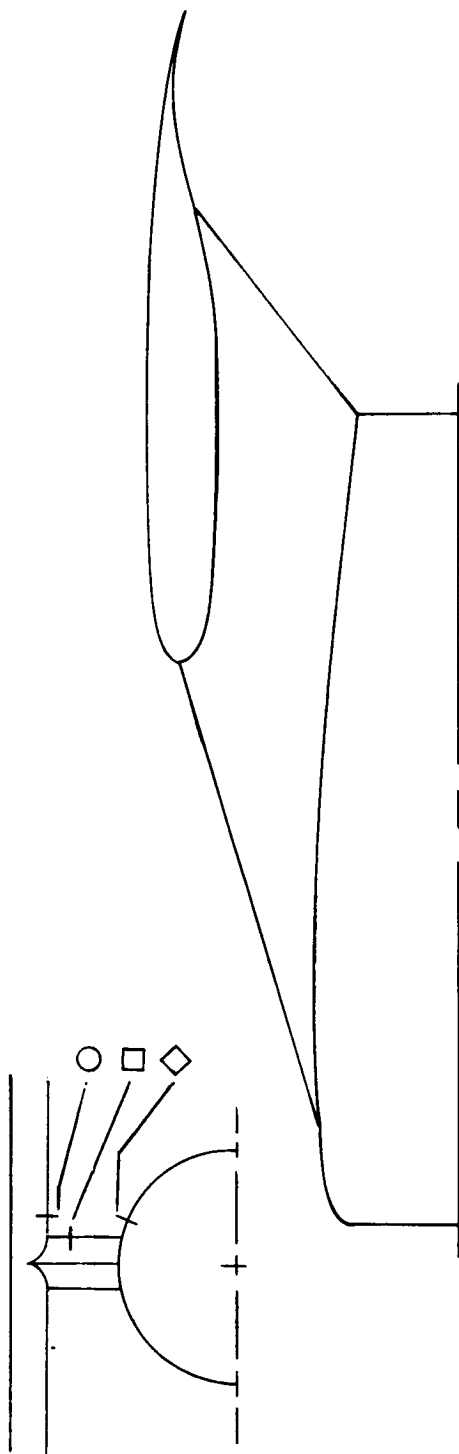


Figure 10.- Correlation of static-pressure coefficients of wing, pylon, and nacelle in junction region for configuration 5 at $M = 0.80$ and $\alpha = 0^\circ$.

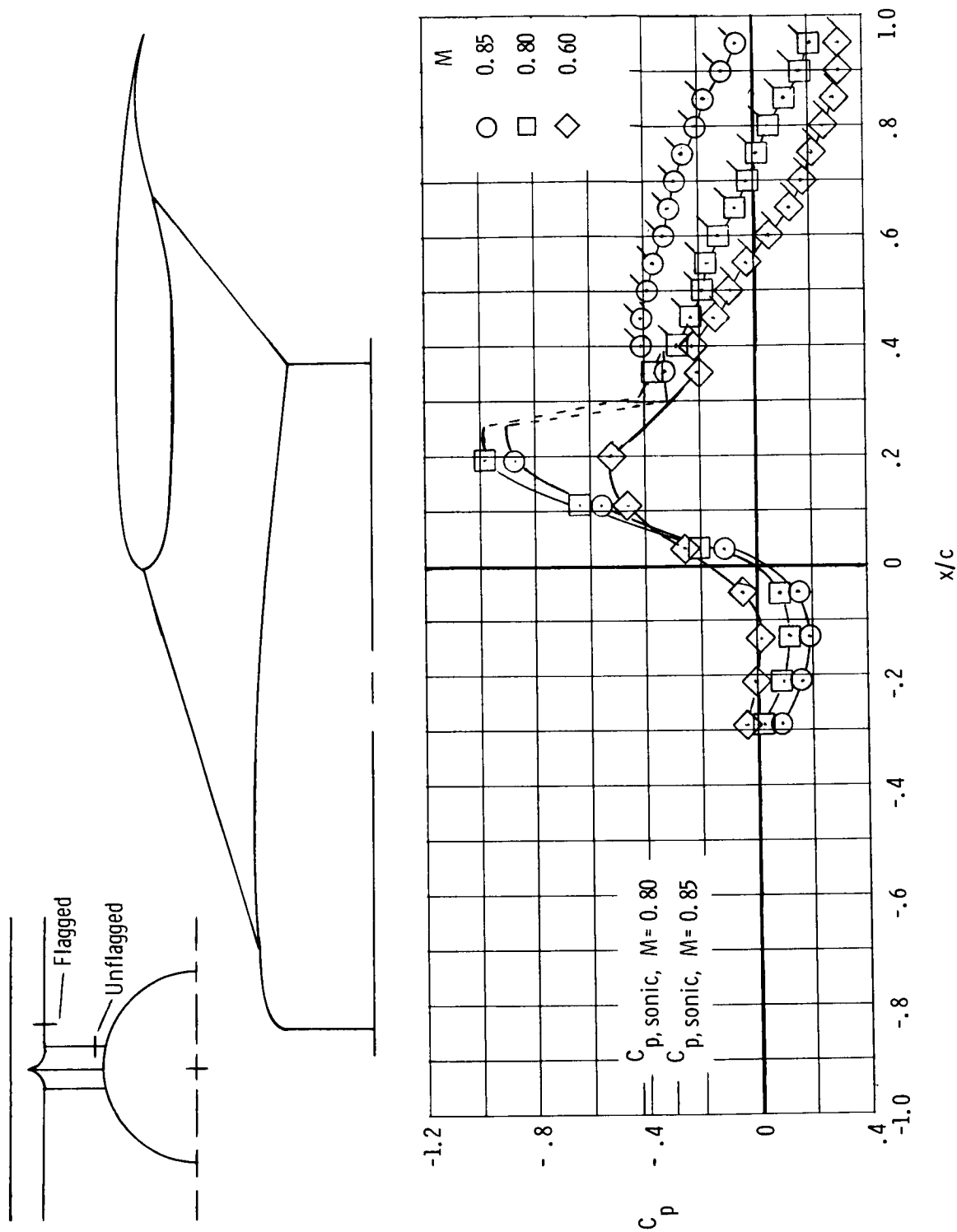


Figure 11.- Effects of Mach number on static-pressure coefficients of pylon bottom row (unflagged symbols) and wing lower-surface station B (flagged symbols). Configuration 5; $\alpha = 0^\circ$; $x_t = 0.29c$.

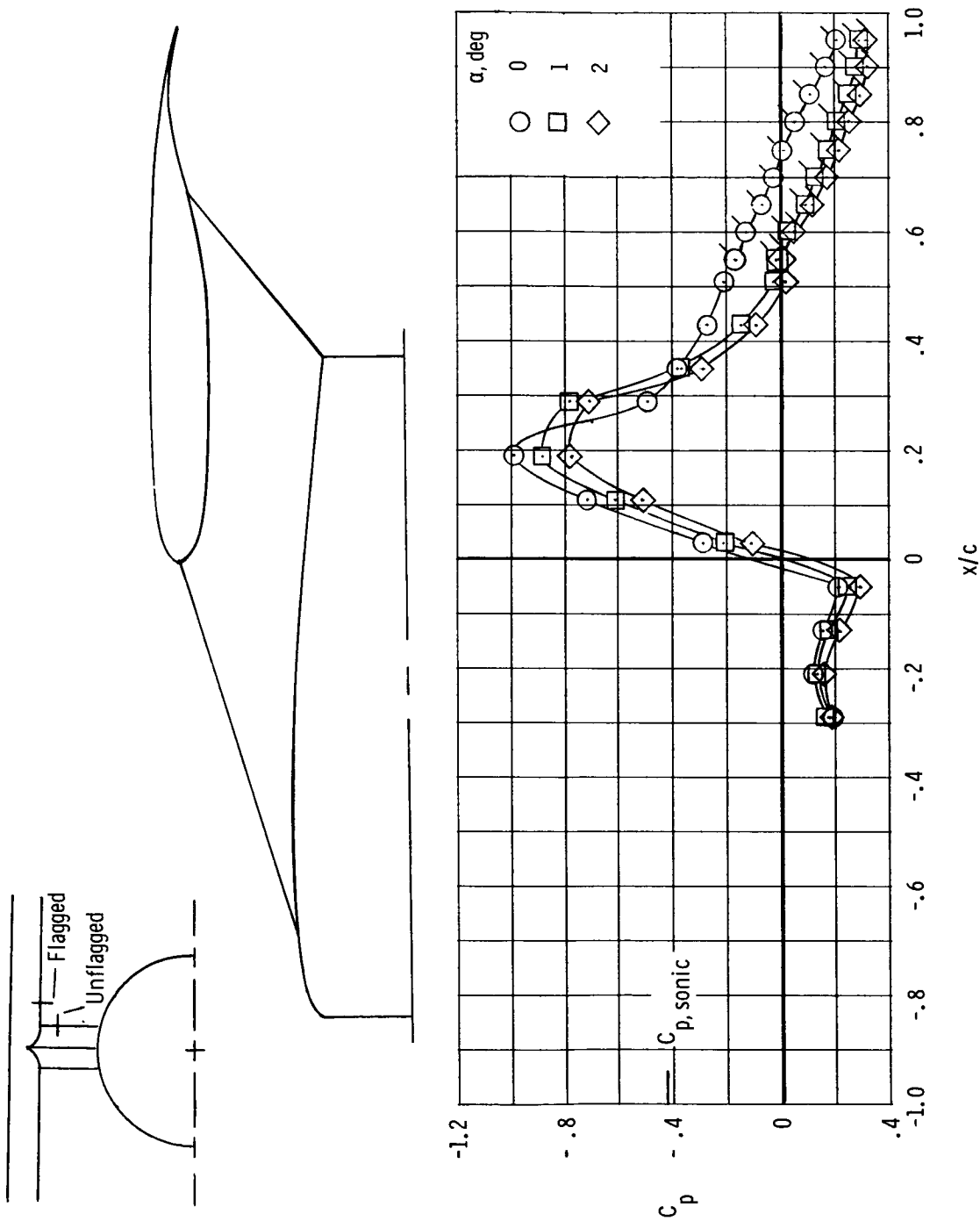


Figure 12.- Effects of angle of attack on static-pressure coefficients of the pylon top row (unflagged symbols) and wing lower-surface station B (flagged symbols). Configuration 5; $M = 0.80$; $x_t = 0.29c$.

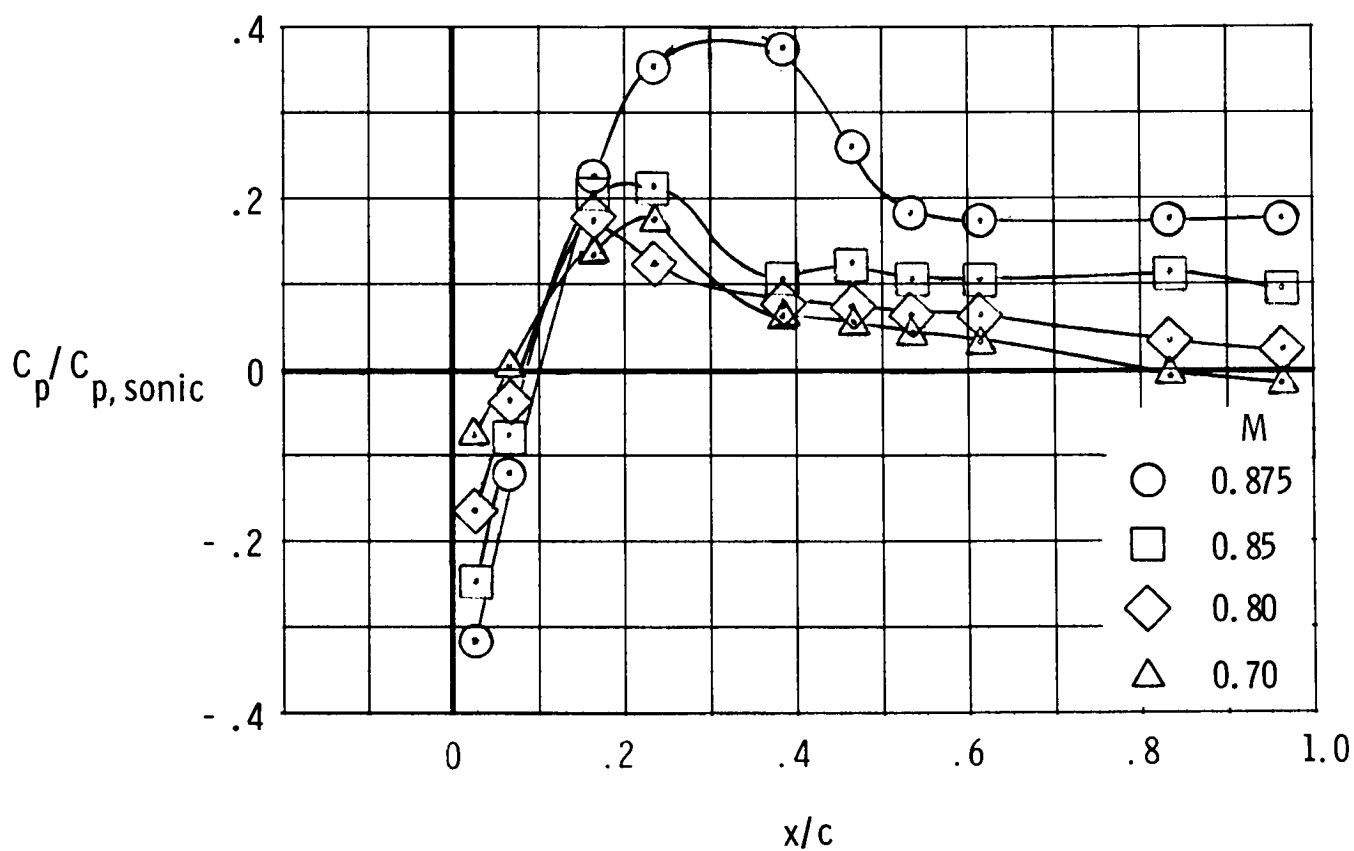
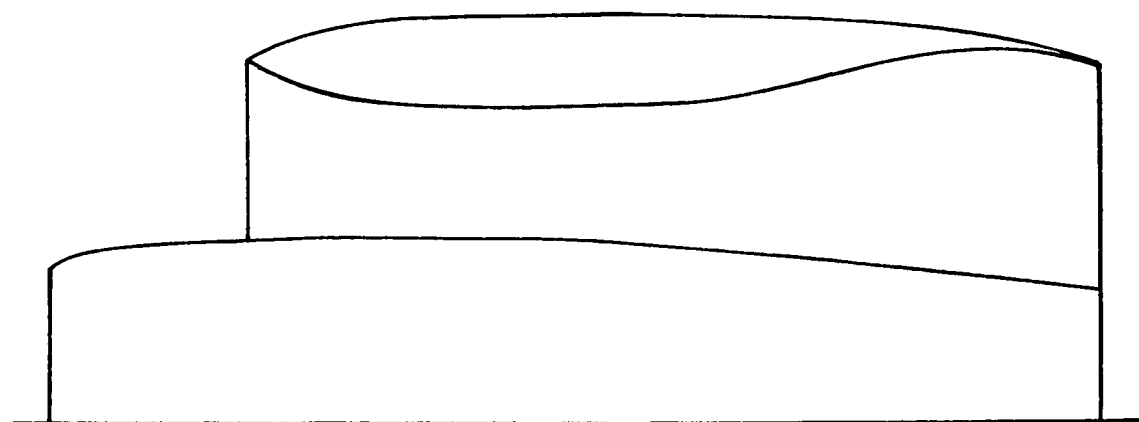


Figure 13.- Effects of Mach number on unswept-pylon static pressures. $\alpha = 0^\circ$; $x_t = 0.29c$; bottom row.

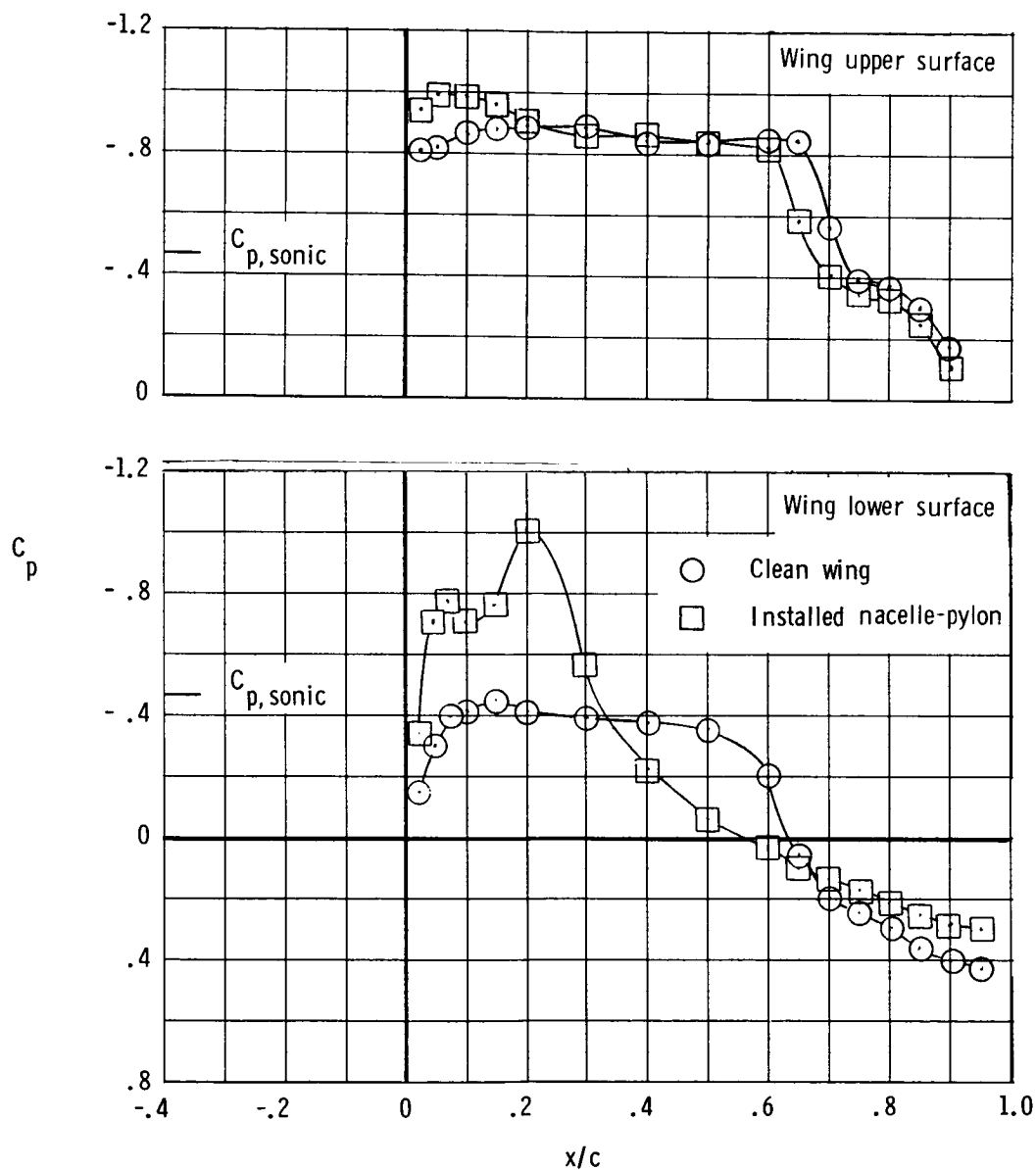
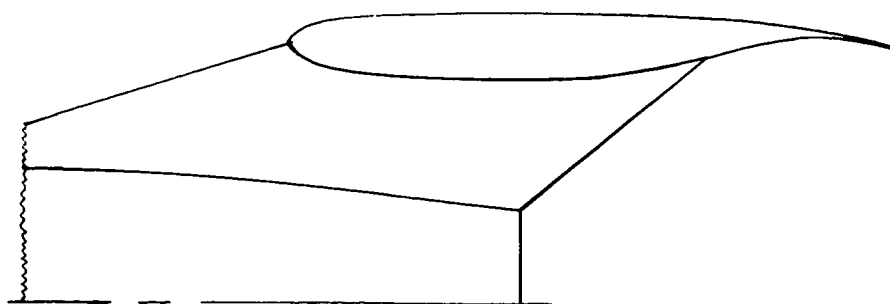
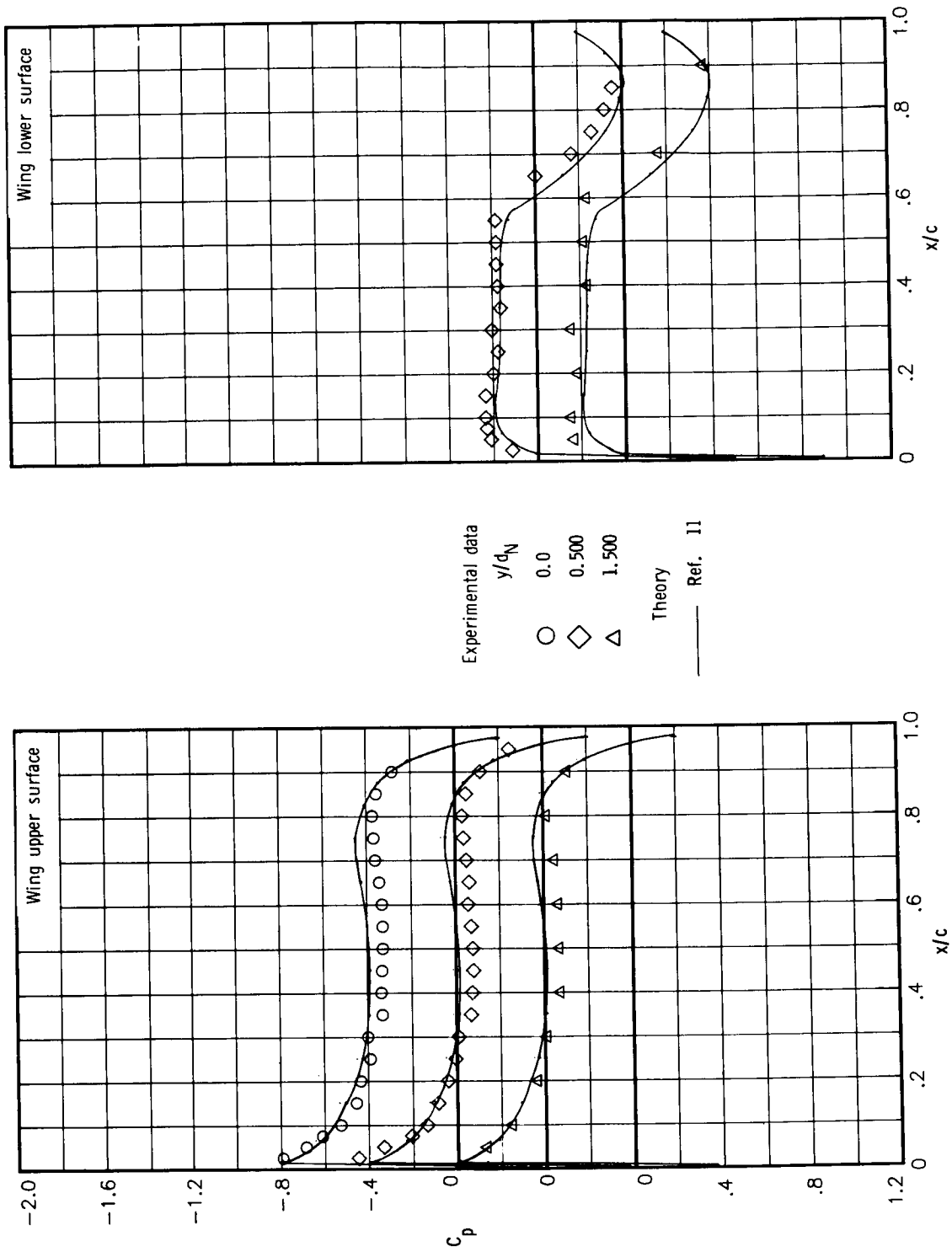
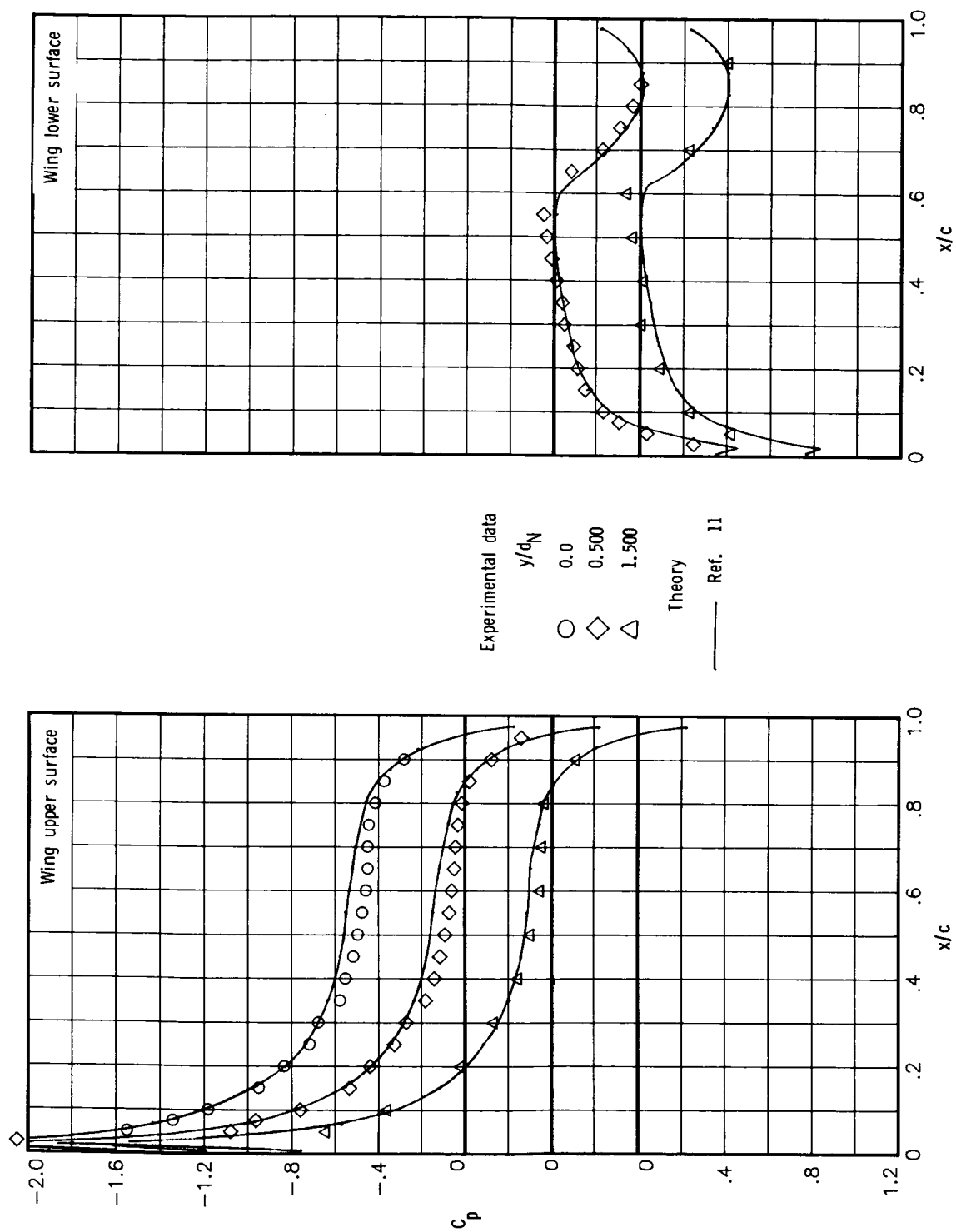


Figure 14.- Effects of swept nacelle-pylon installation on wing static pressures.
 $M = 0.80$; $\alpha = 1^\circ$; $x_t = 0.29c$; $y/d_N = 0.191$.



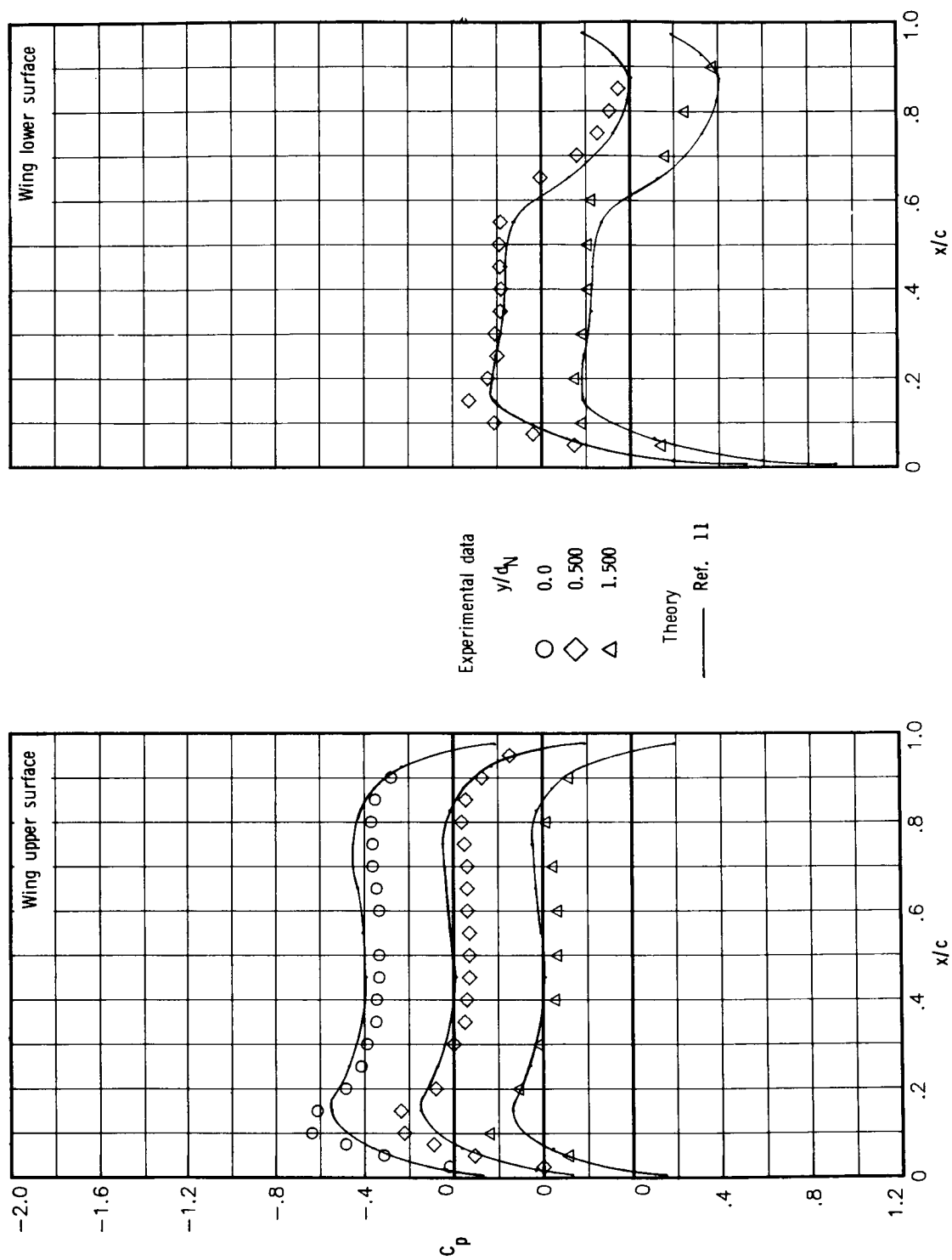
(a) $\alpha = 0^\circ$; configuration 1.

Figure 15.- Comparison of experimental and analytical pressure coefficients at $M = 0.20$ and $x_t = 0.29c$.



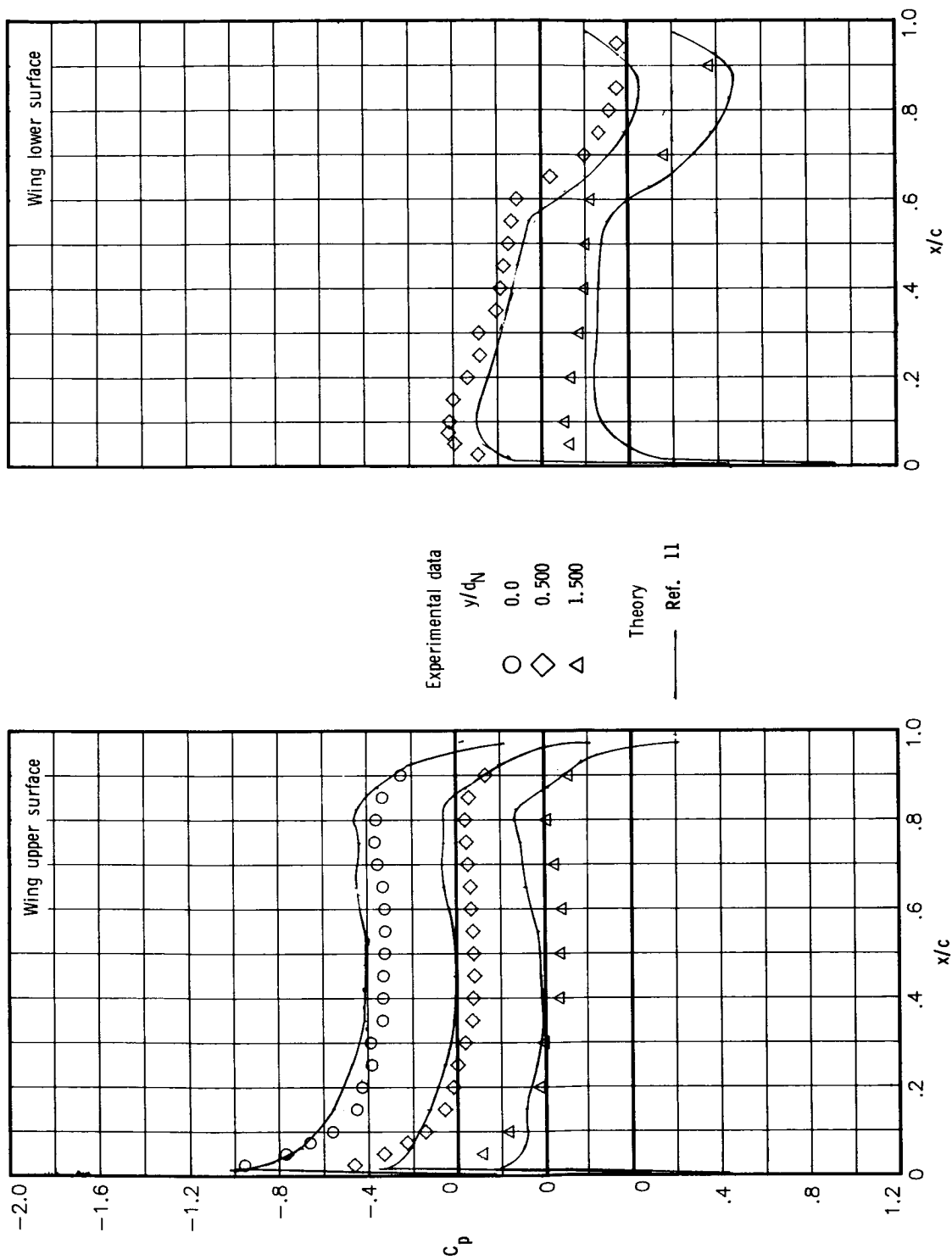
(b) $\alpha = 5^\circ$; configuration 1.

Figure 15.- Continued.



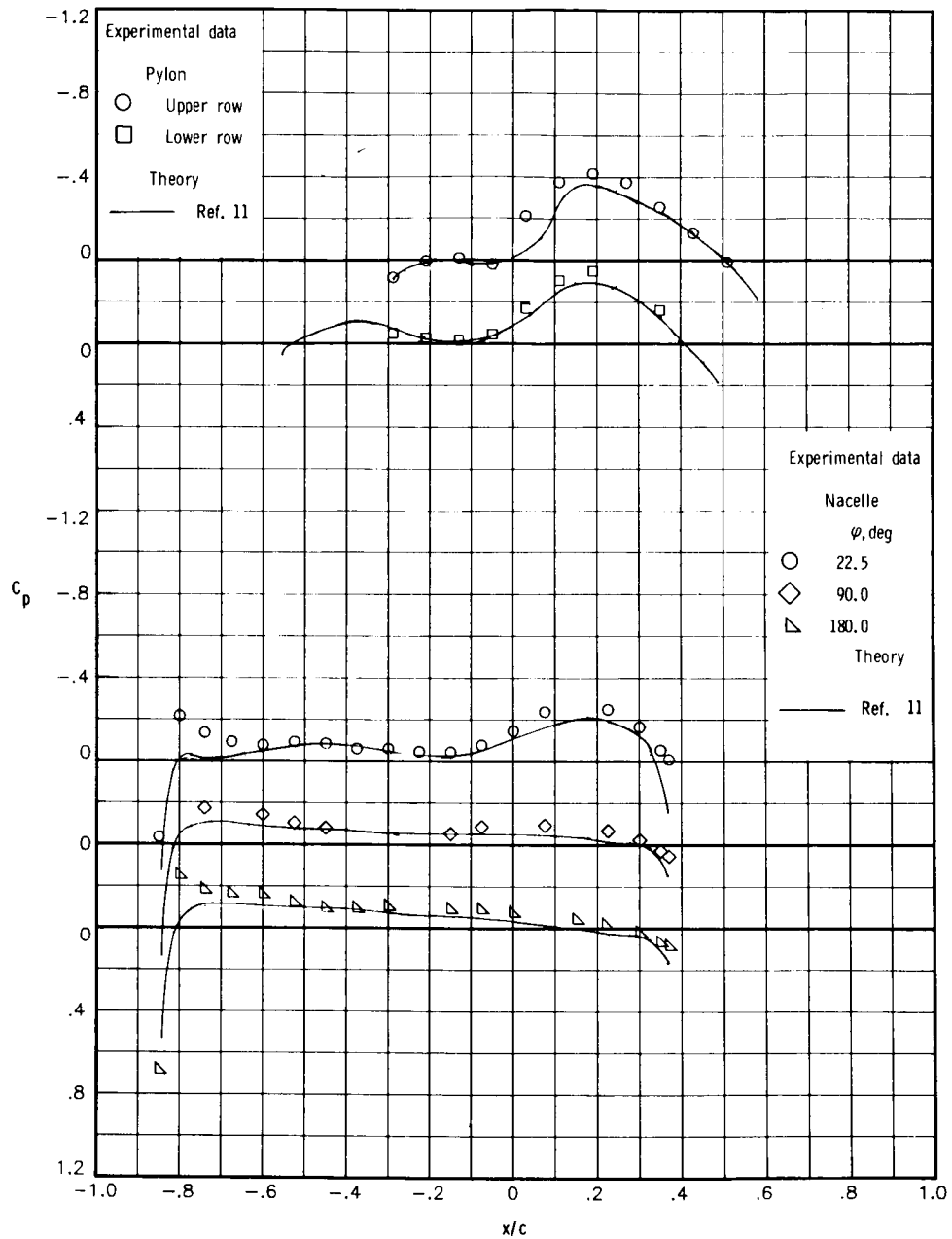
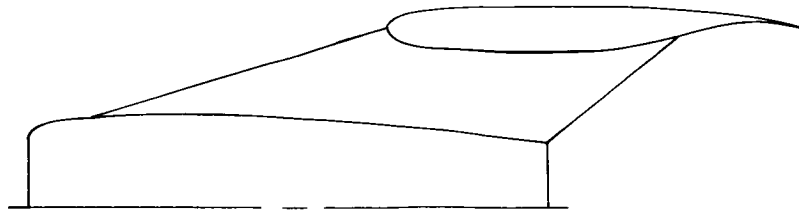
(c) $\alpha = 0^\circ$; configuration 2.

Figure 15.- Continued.



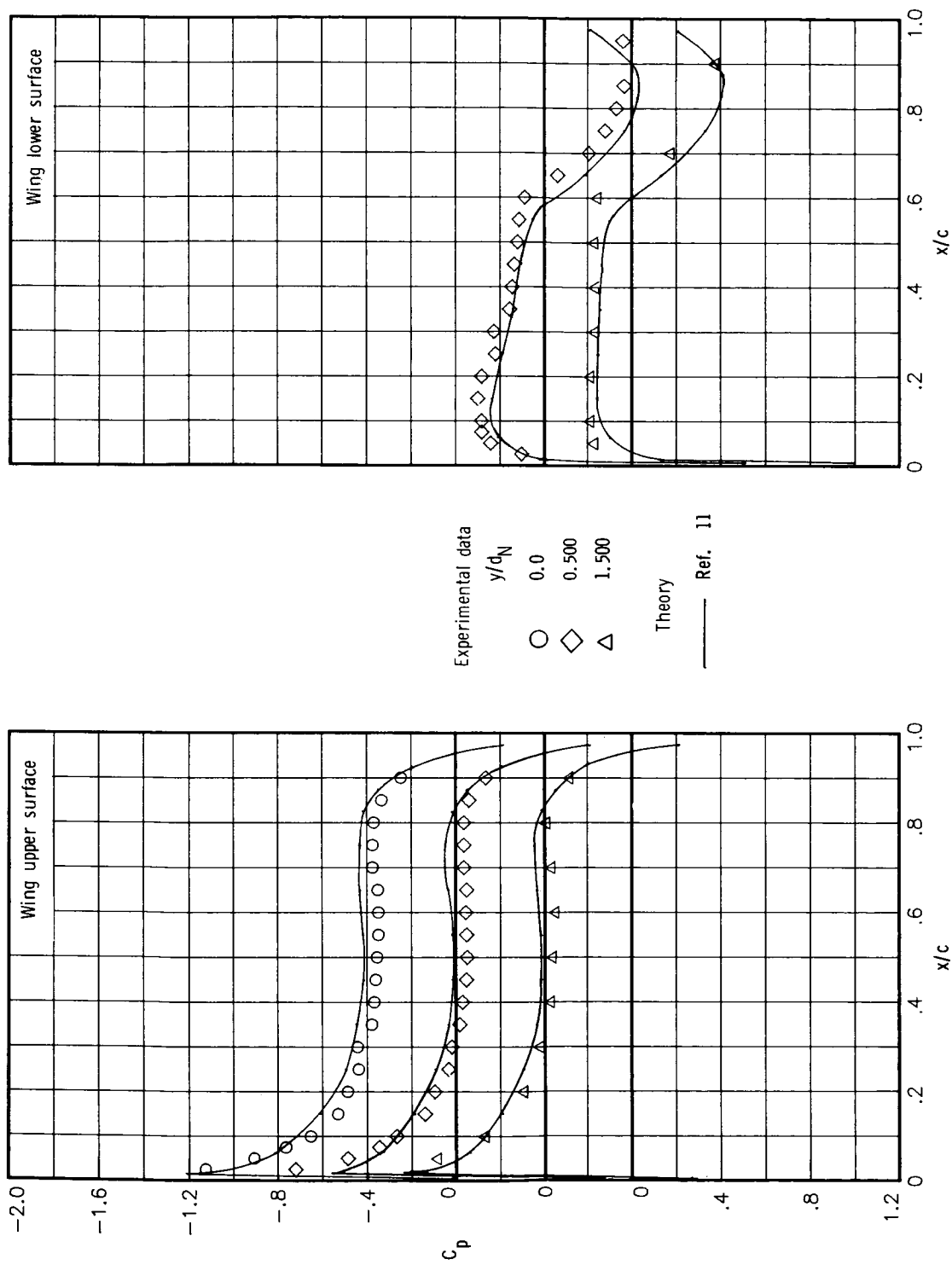
(d) $\alpha = 0^\circ$; configuration 5.

Figure 15.- Continued.



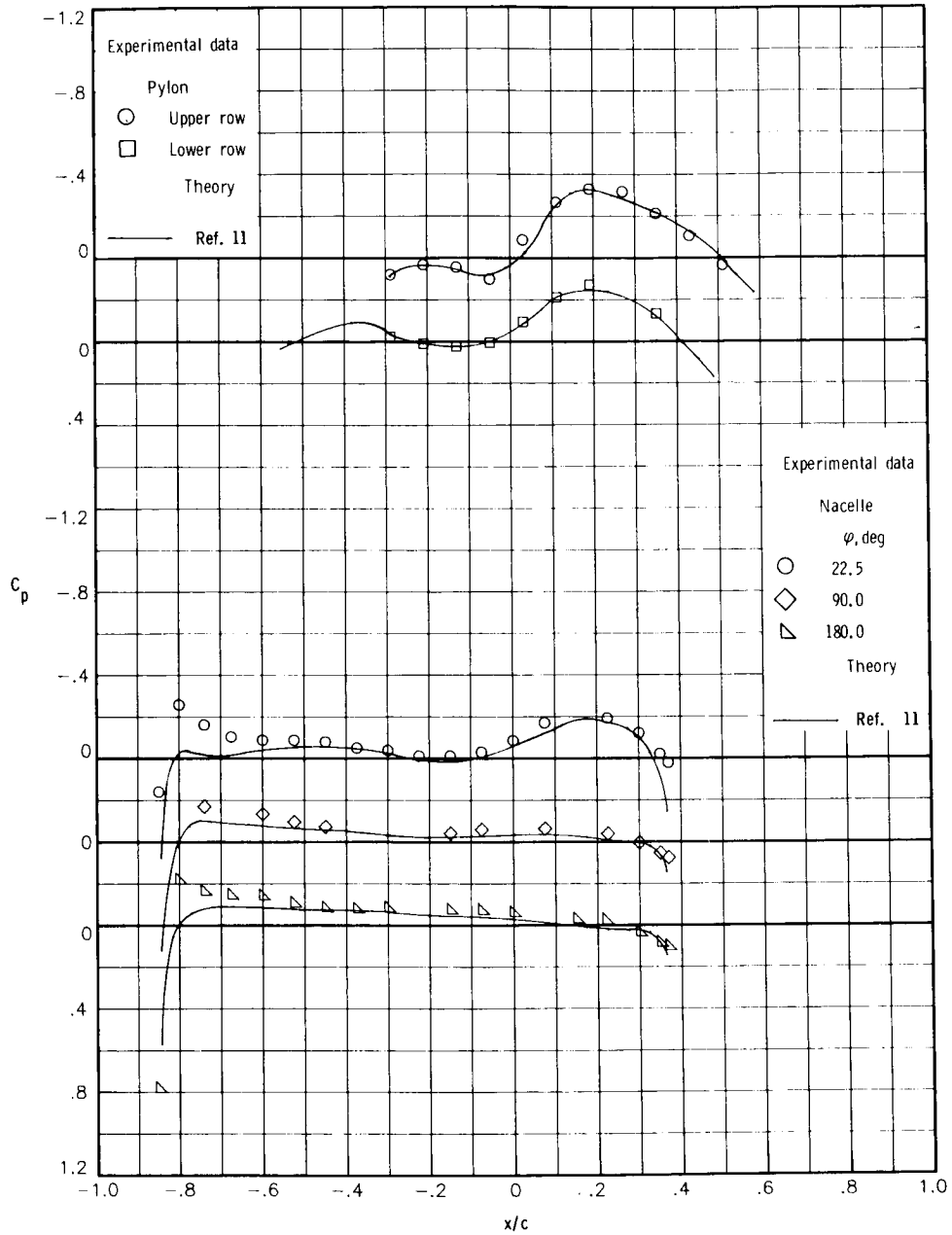
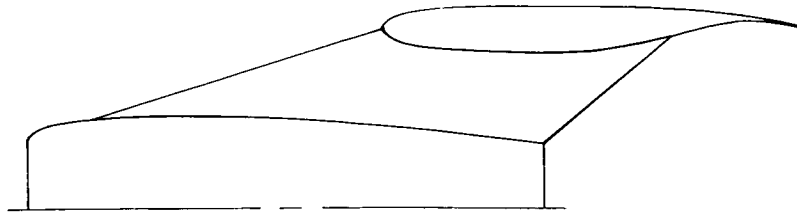
(d) Concluded.

Figure 15.- Continued.



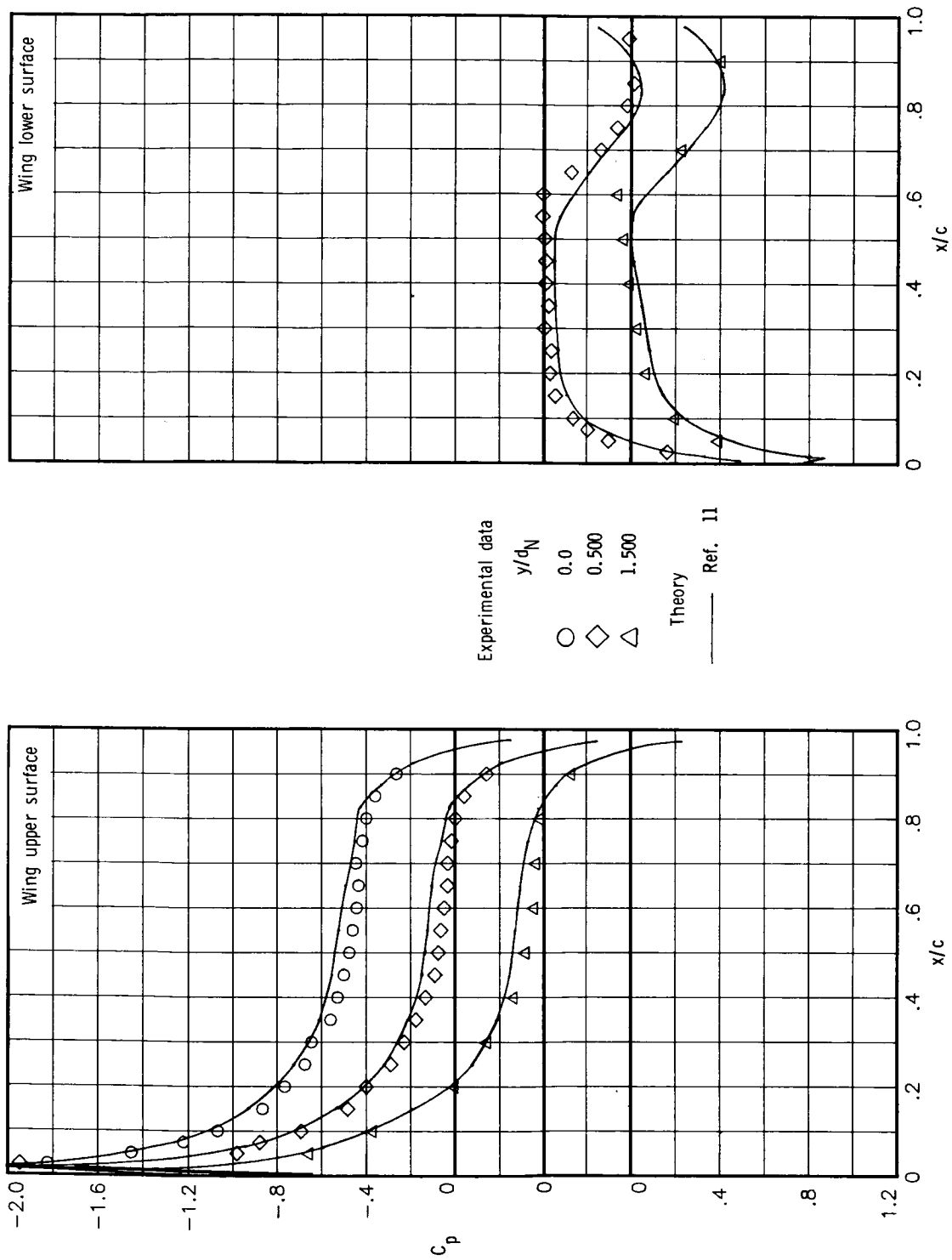
(e) $\alpha = 1^\circ$; configuration 5.

Figure 15.- Continued.



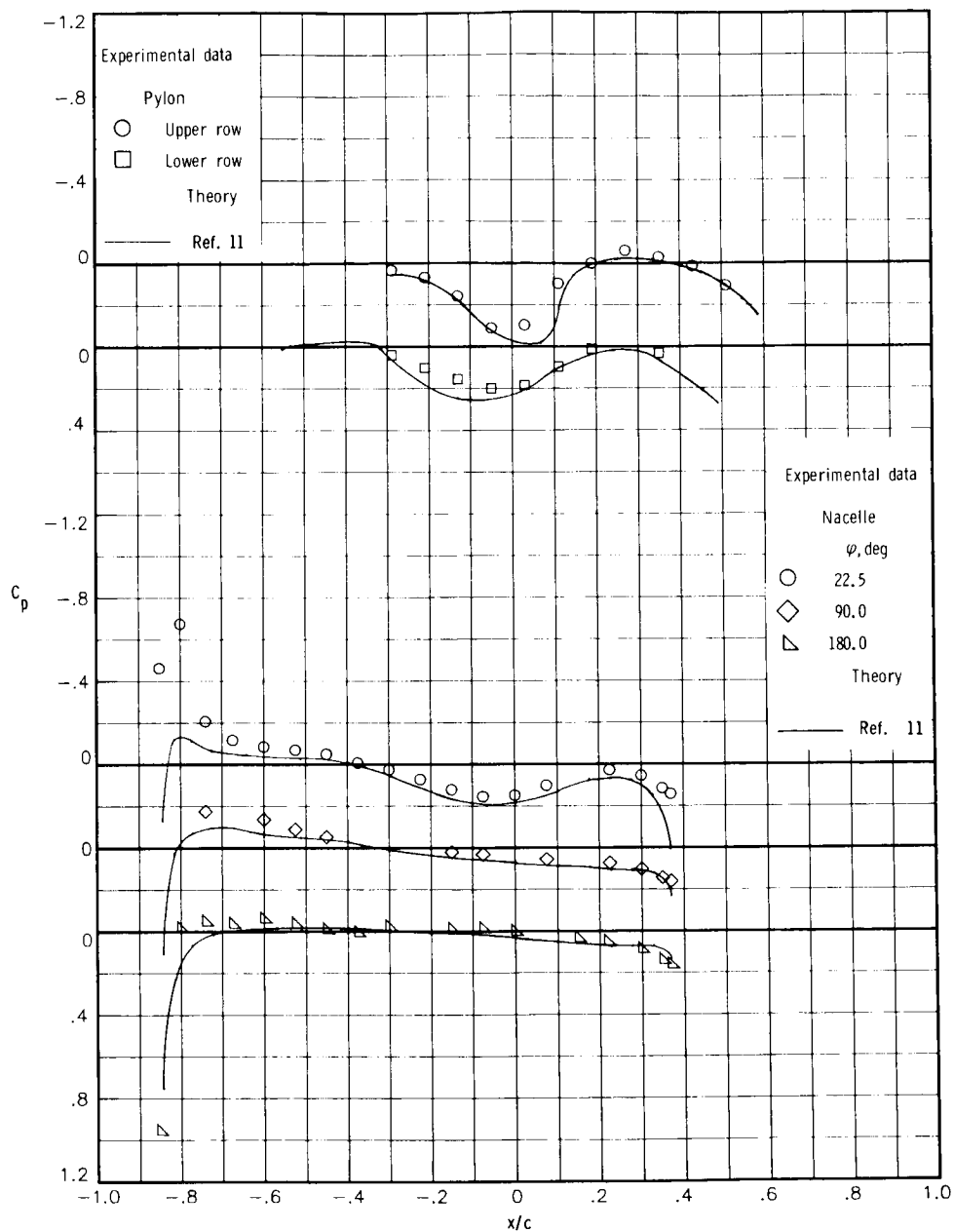
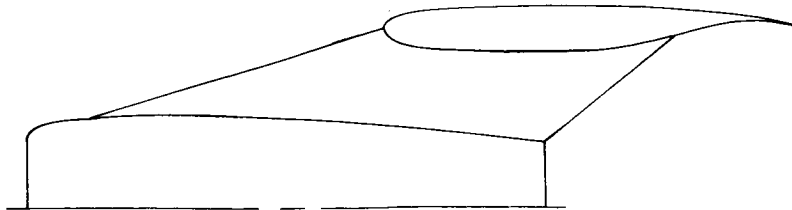
(e) Concluded.

Figure 15.- Continued.



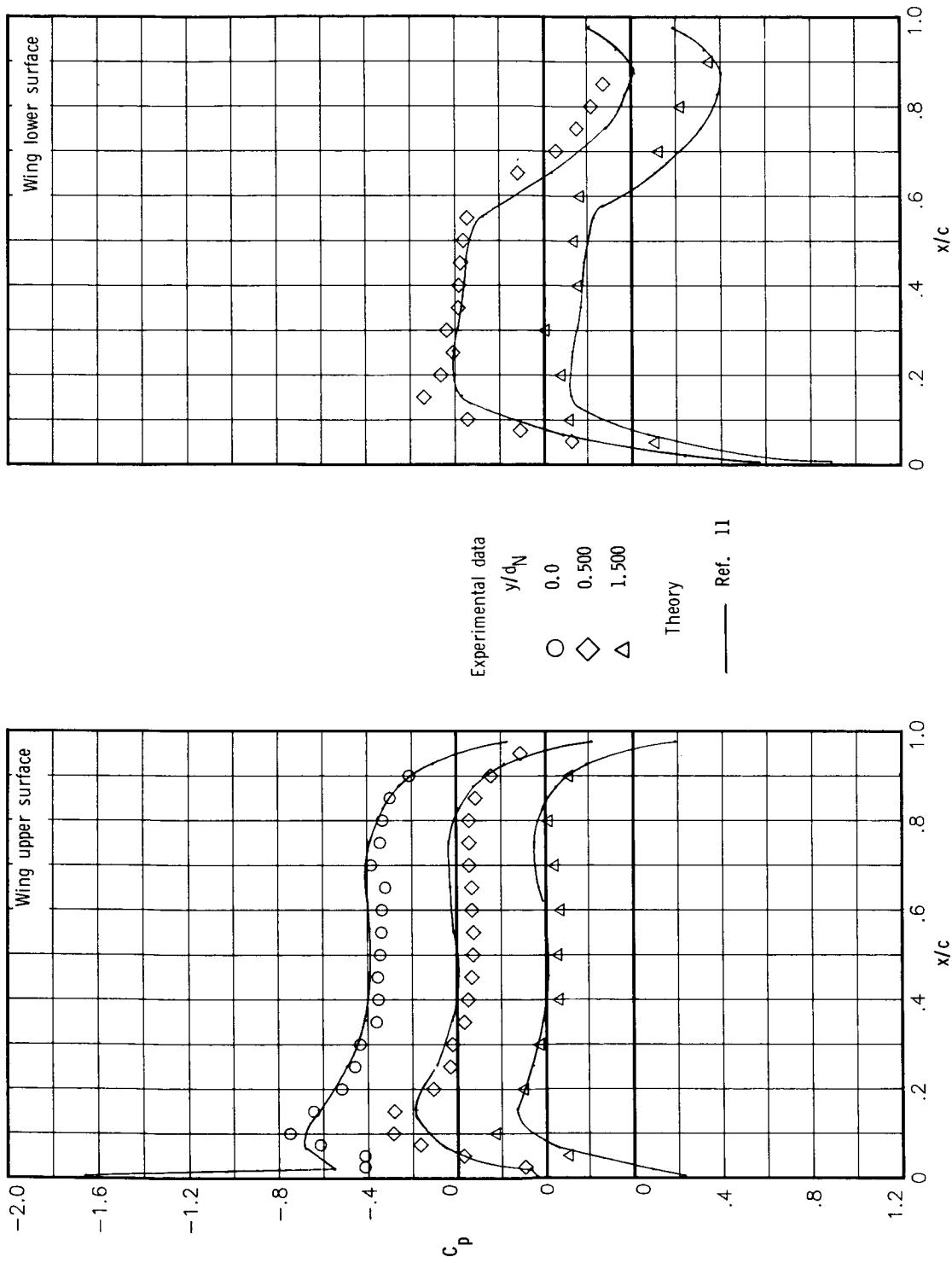
(f) $\alpha = 5^\circ$; configuration 5.

Figure 15.- Continued.



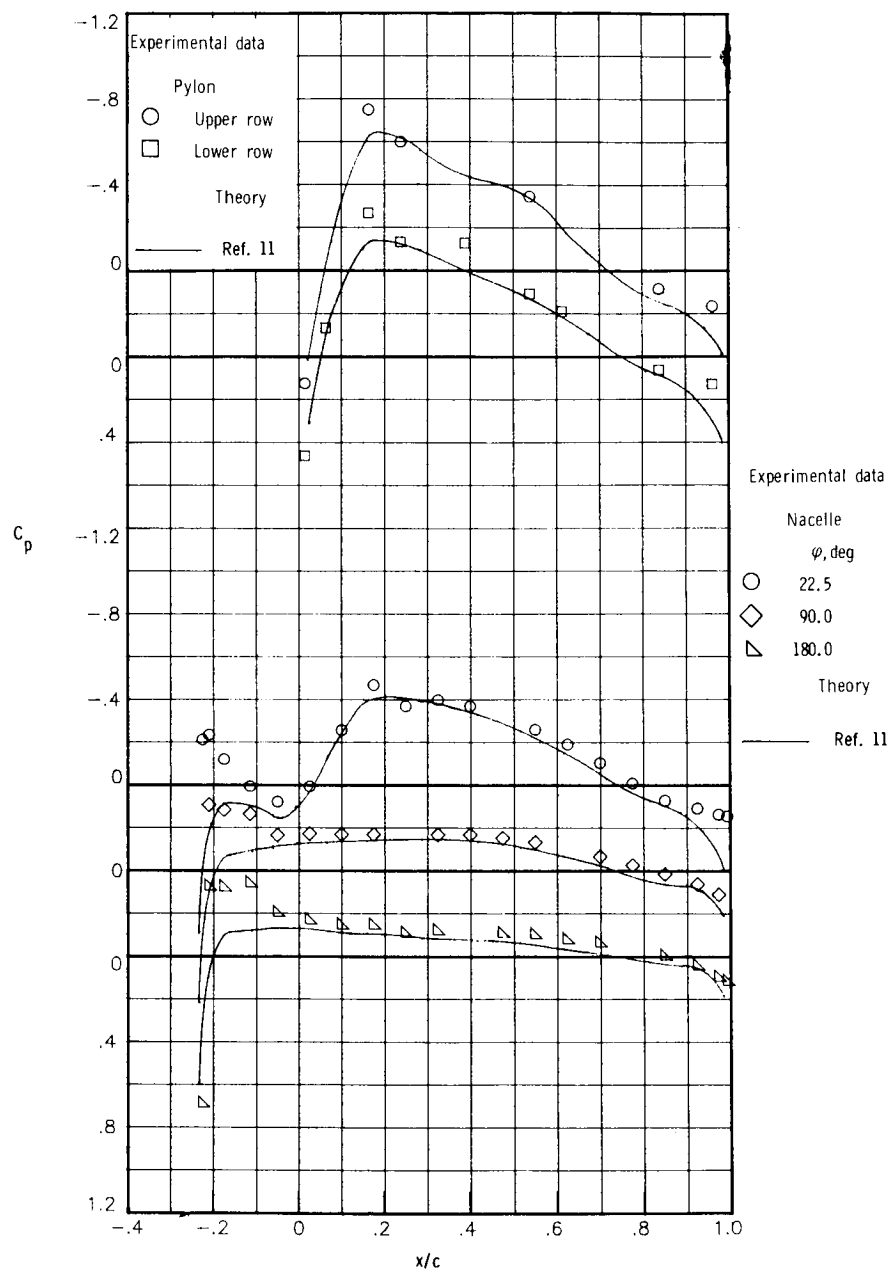
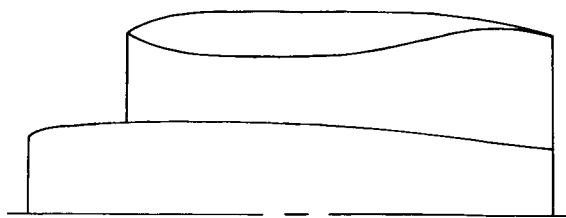
(f) Concluded.

Figure 15.- Continued.



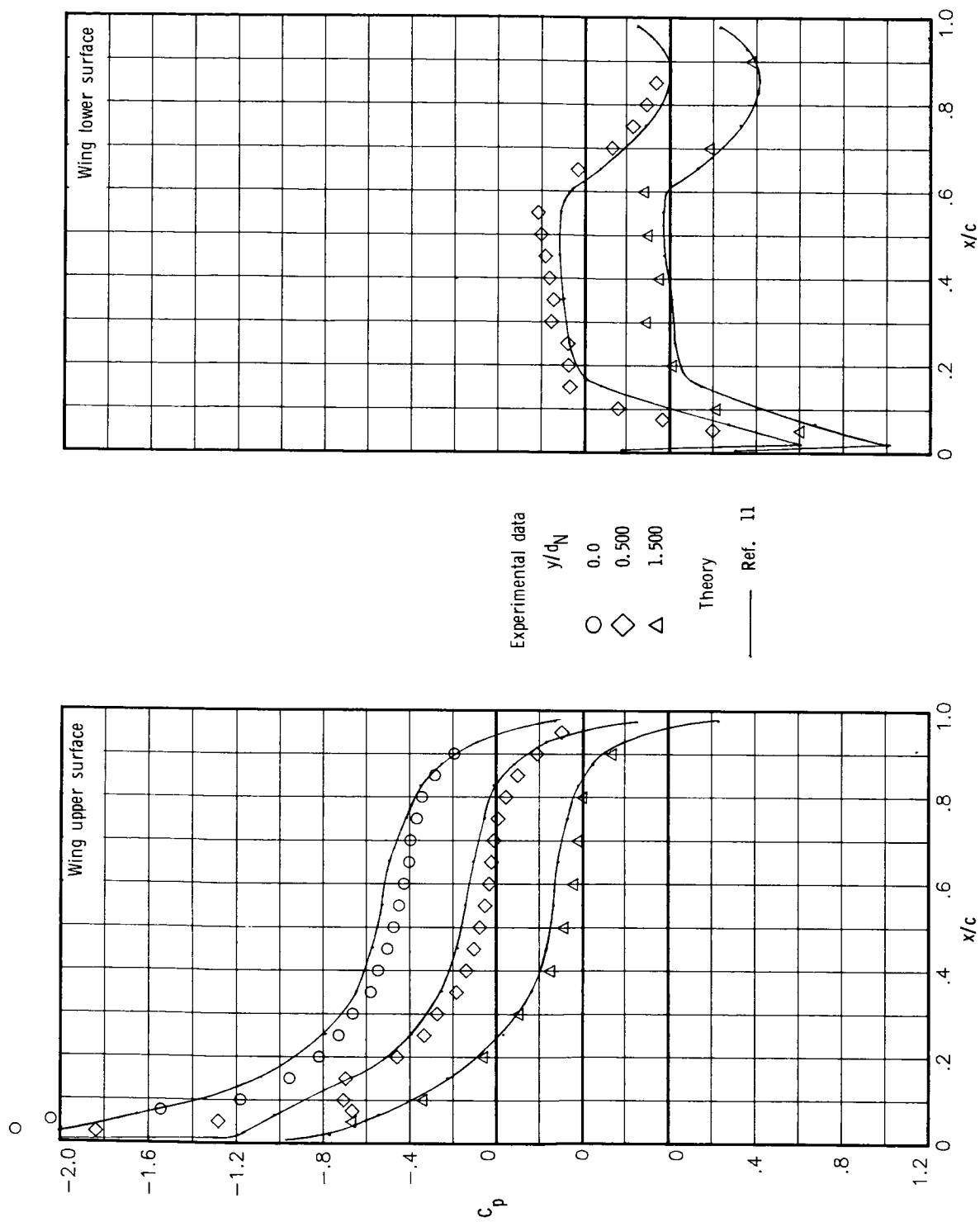
(g) $\alpha = 0^\circ$; configuration 3.

Figure 15.- Continued.



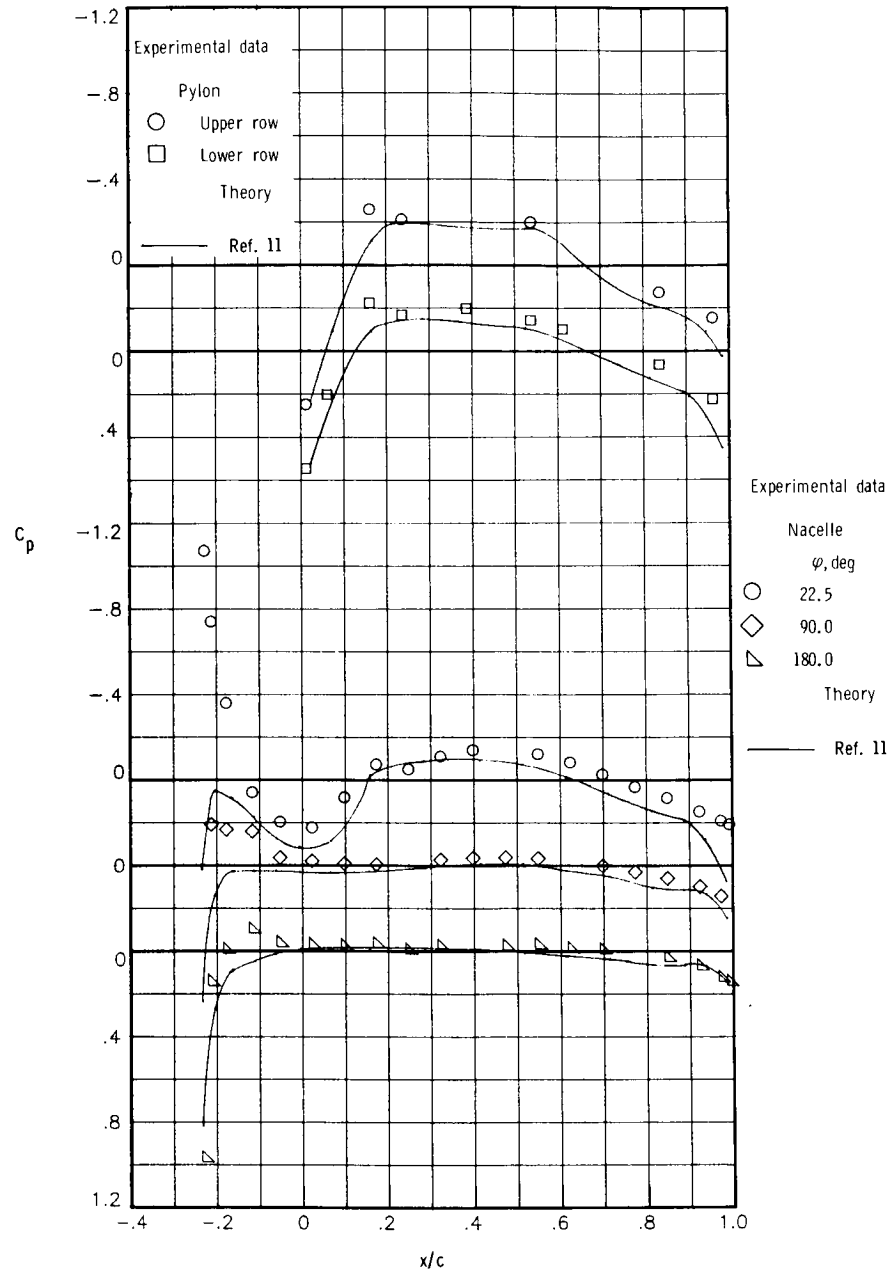
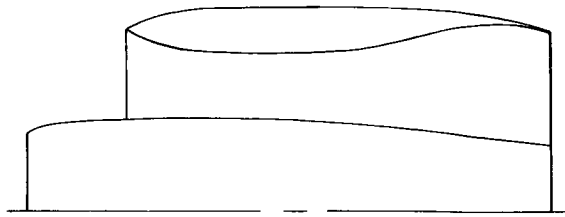
(g) Concluded.

Figure 15.- Continued.



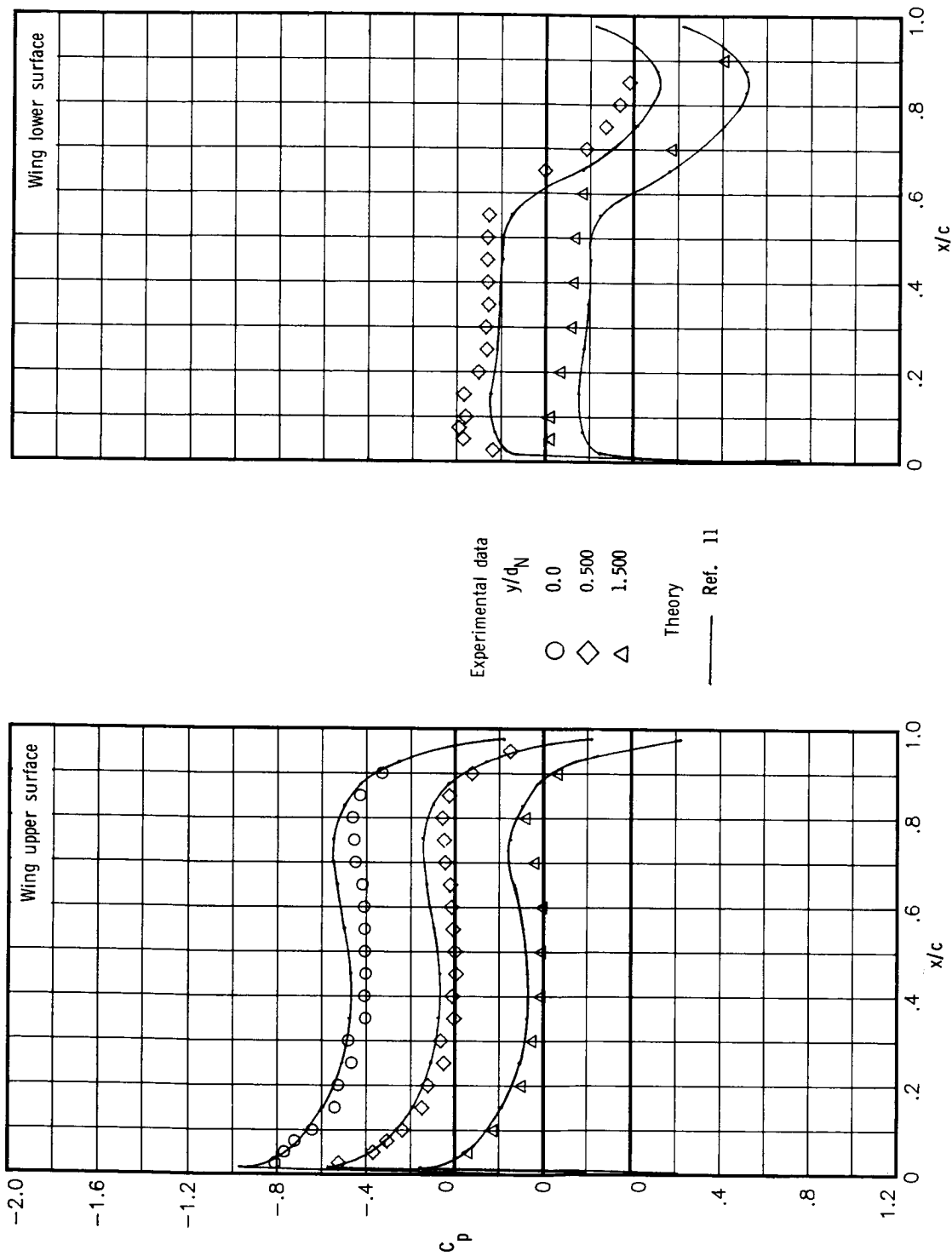
(h) $\alpha = 5^\circ$; configuration 3.

Figure 15.- Continued.



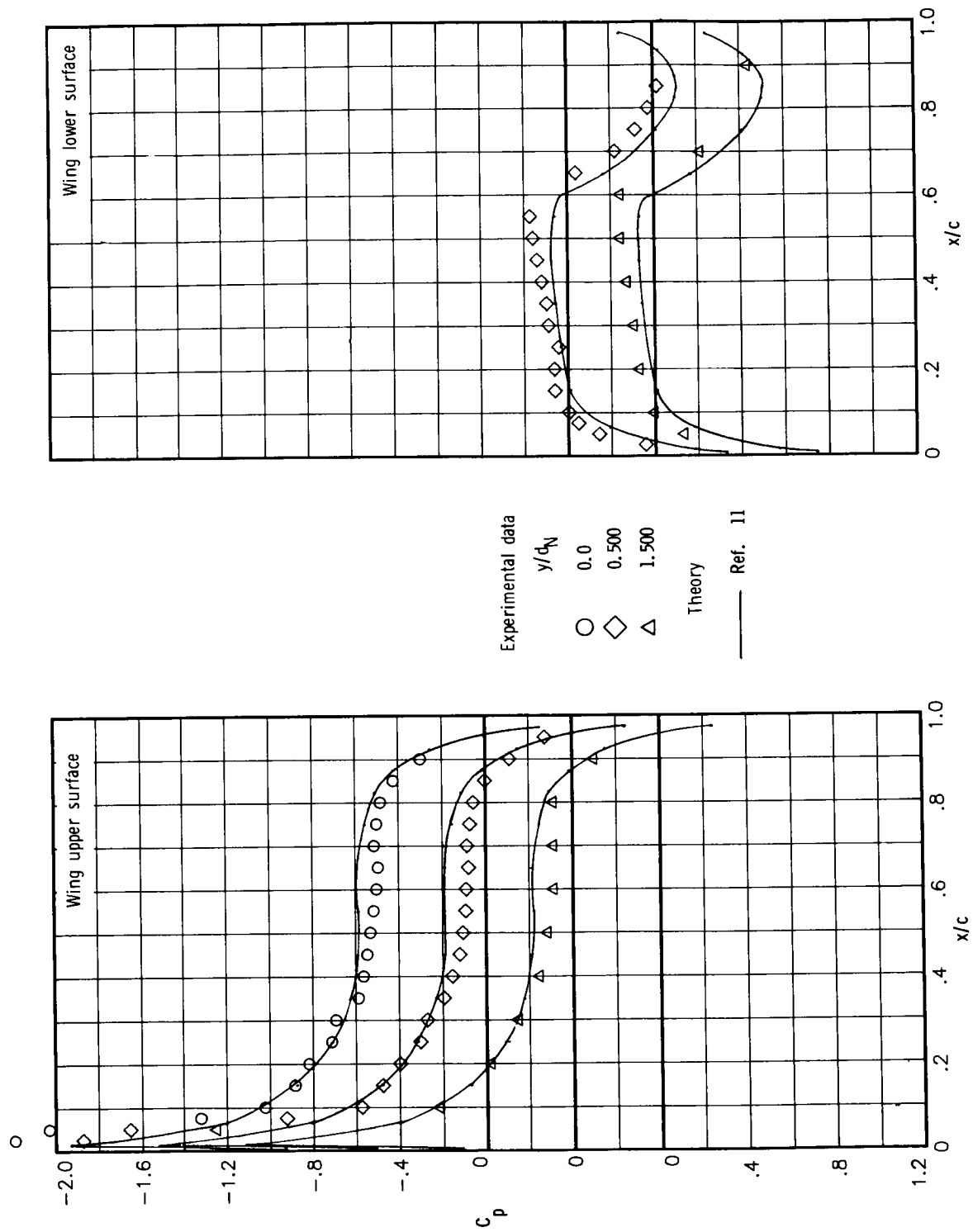
(h) Concluded.

Figure 15.- Concluded.



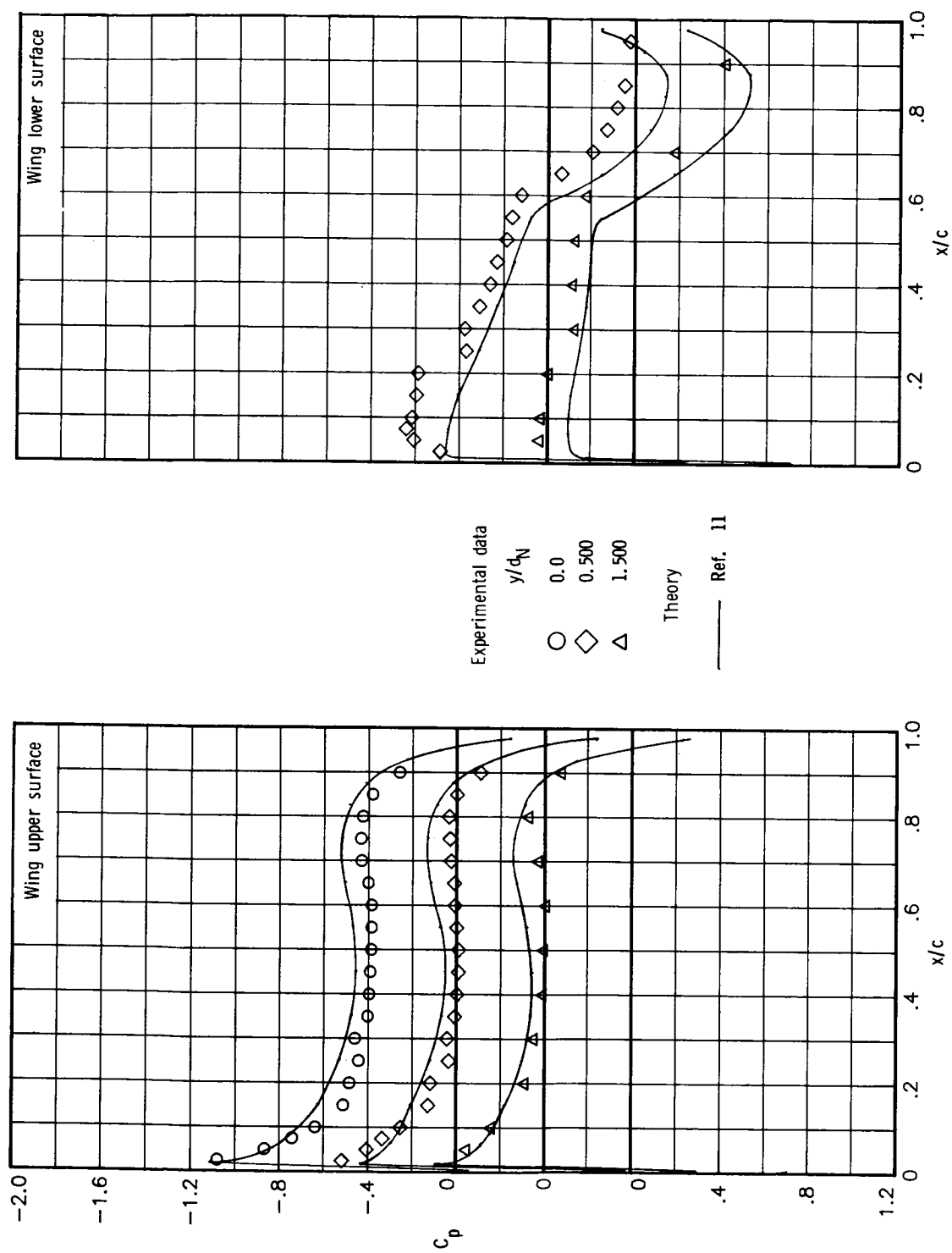
(a) $\alpha = 0^\circ$; configuration 1.

Figure 16.- Comparison of experimental and analytic pressure coefficients at $M = 0.60$ and $x_t = 0.29c$.



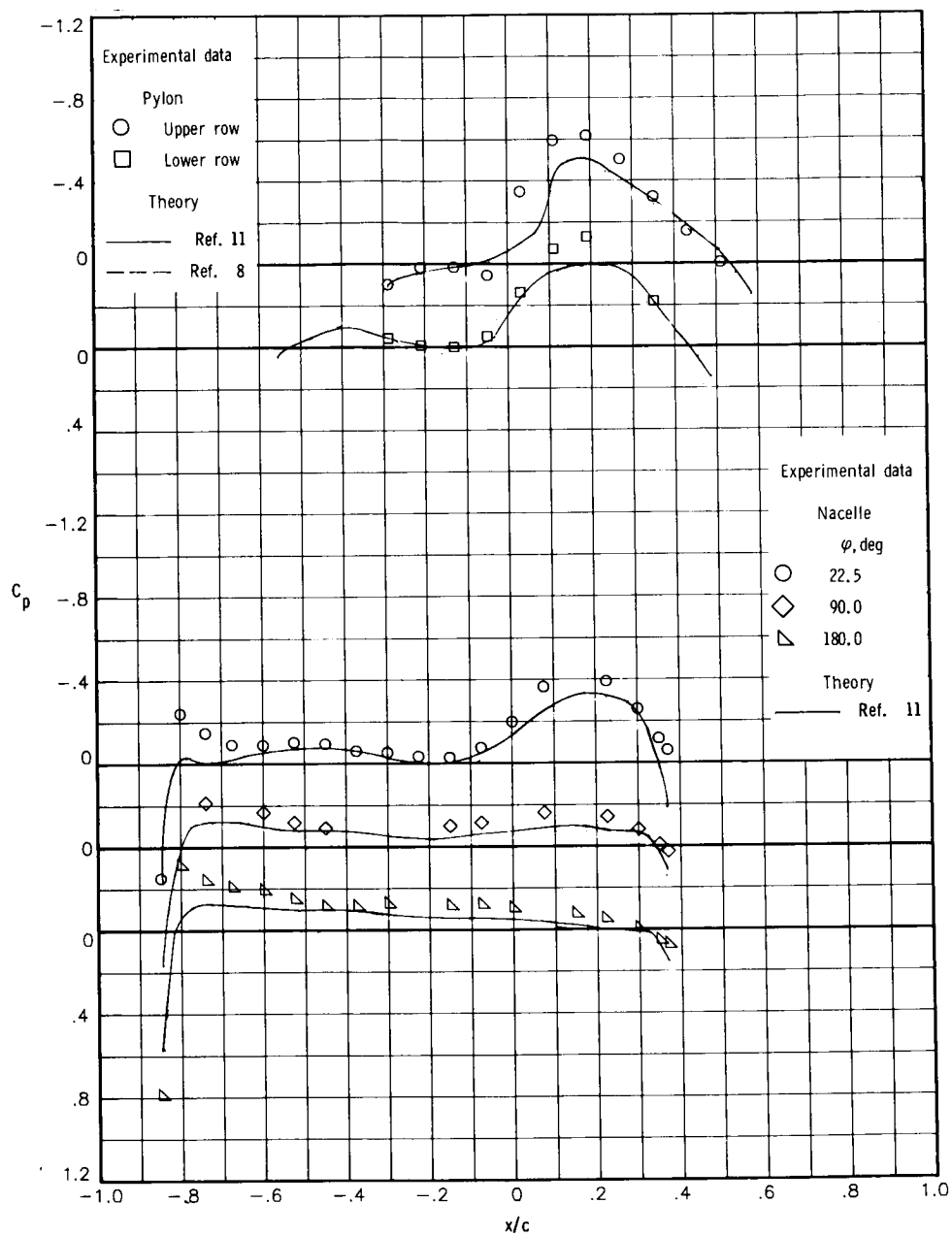
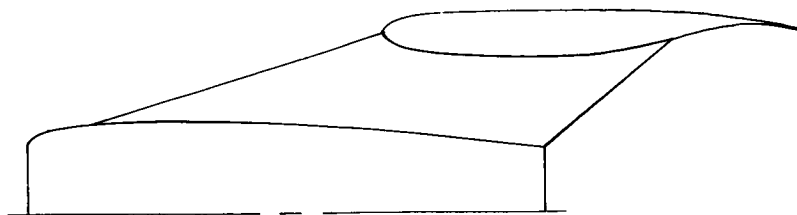
(b) $\alpha = 3^\circ$; configuration 1.

Figure 16.- Continued.



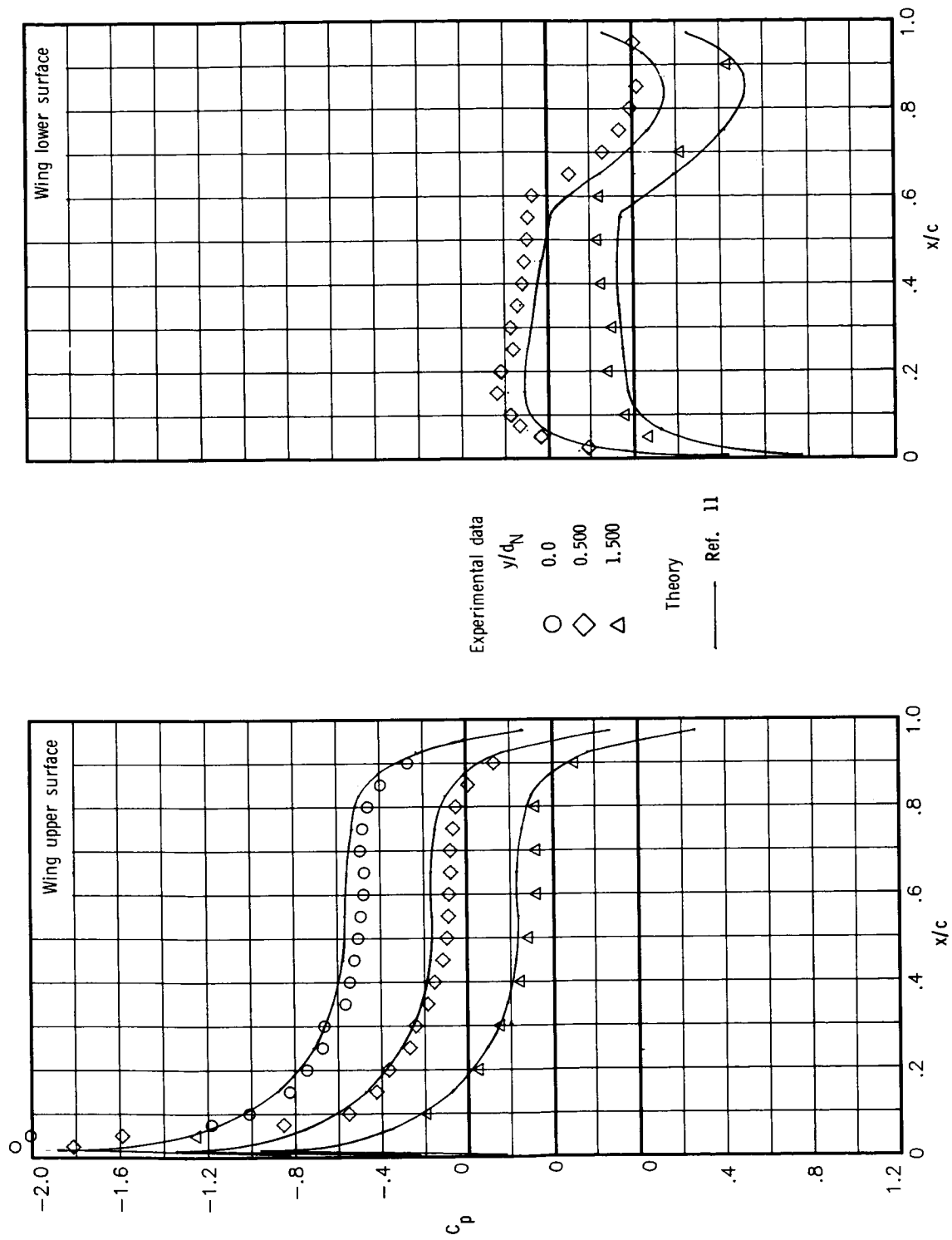
(c) $\alpha = 0^\circ$; configuration 5.

Figure 16.- Continued.



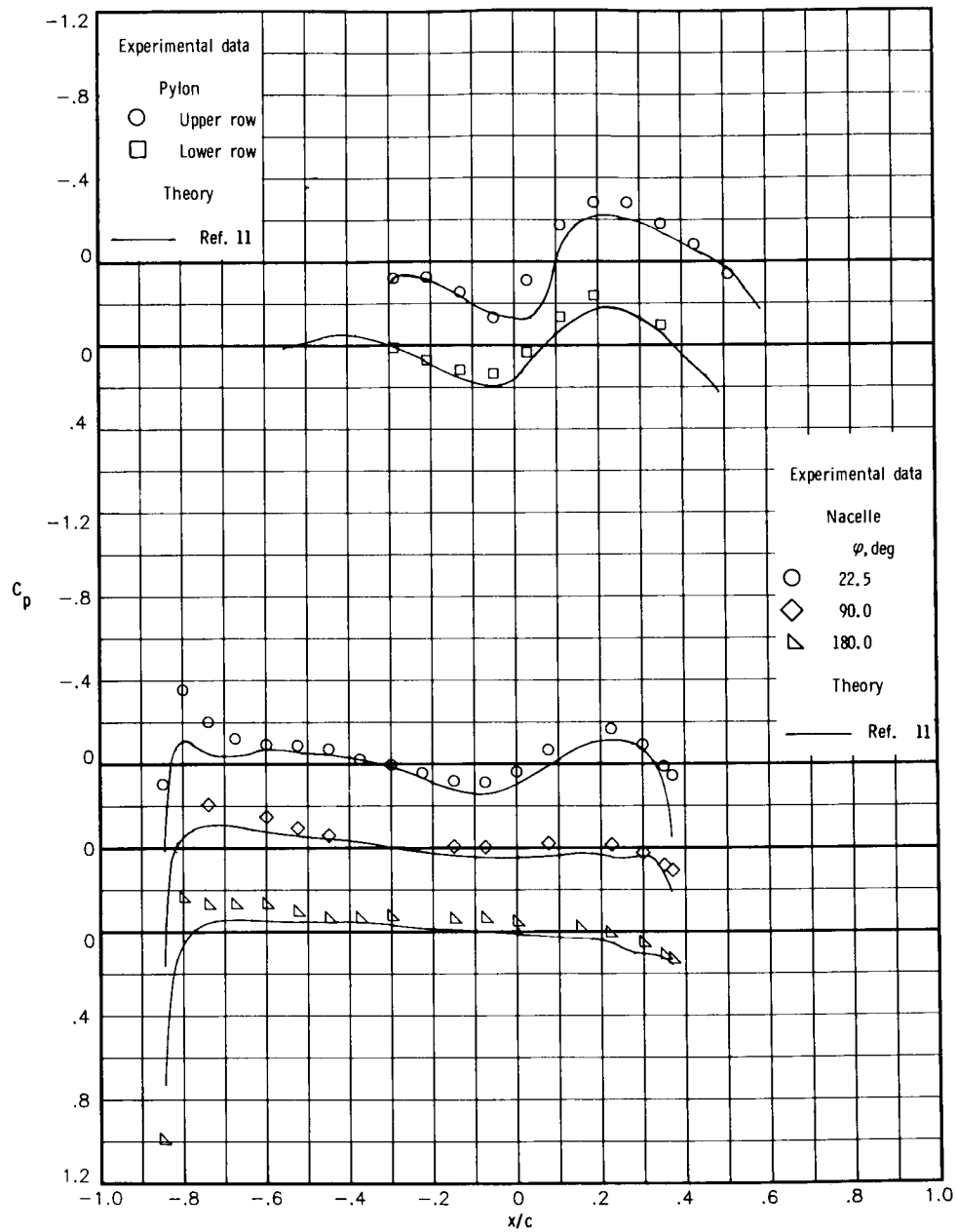
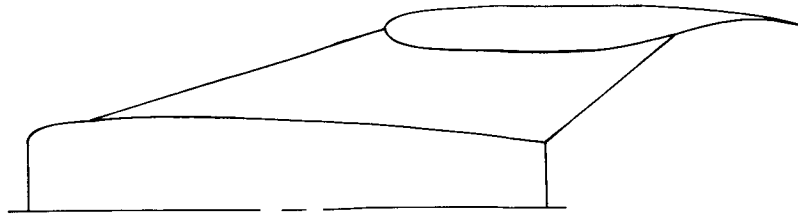
(c) Concluded.

Figure 16.- Continued.



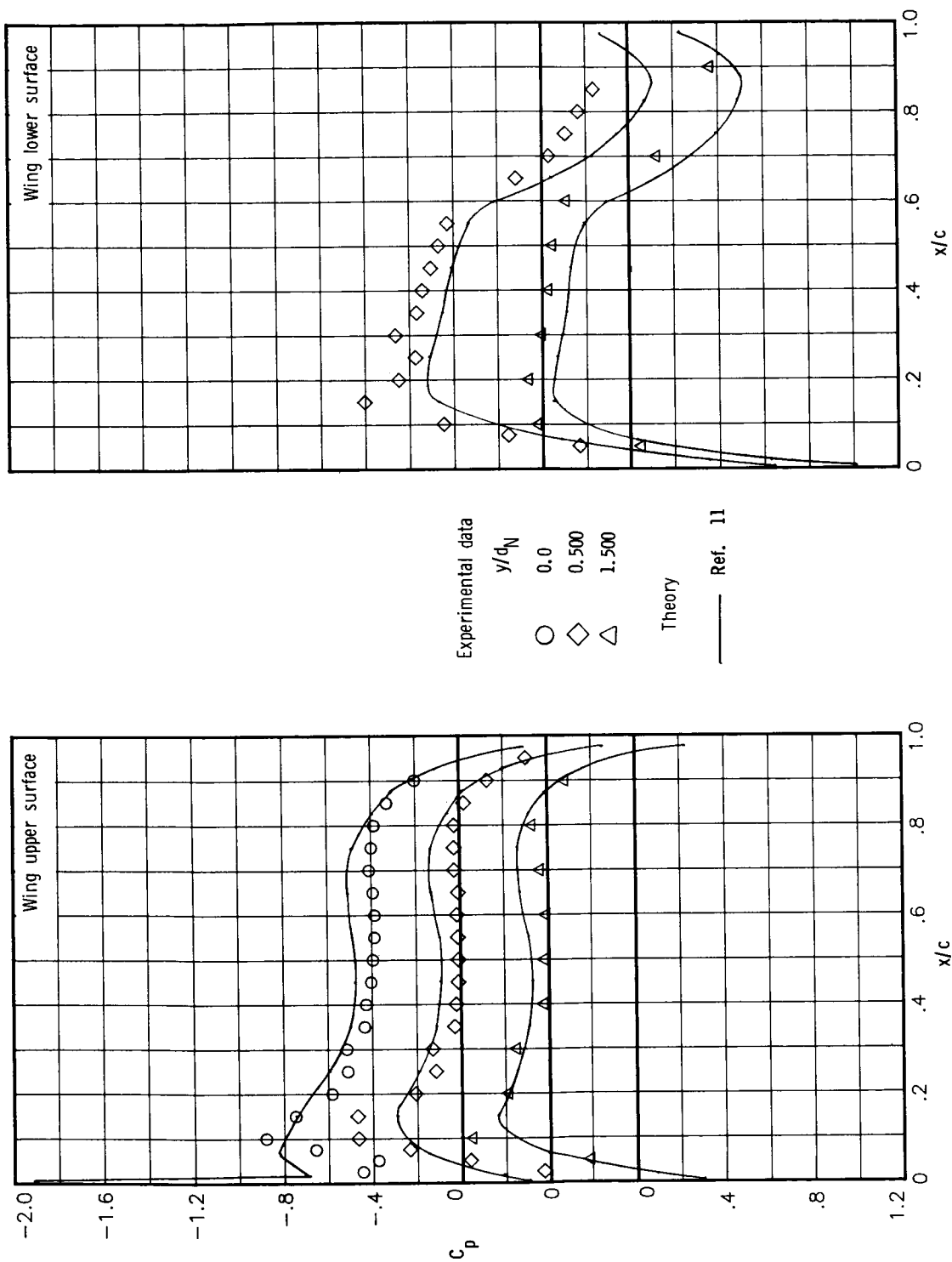
(d) $\alpha = 3^\circ$; configuration 5.

Figure 16.- Continued.



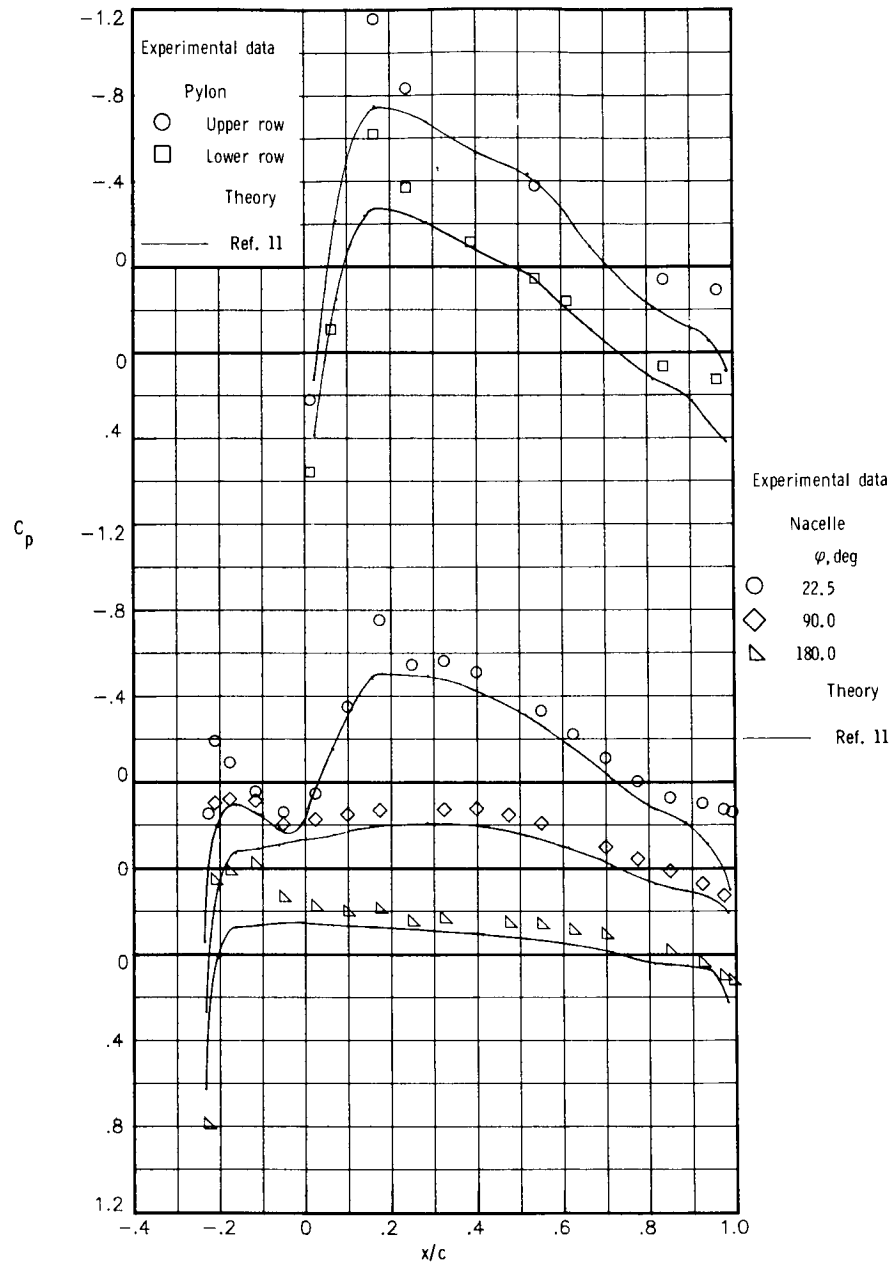
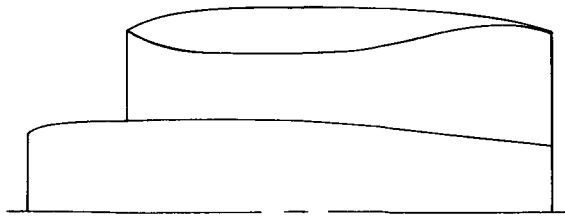
(d) Concluded.

Figure 16.- Continued.



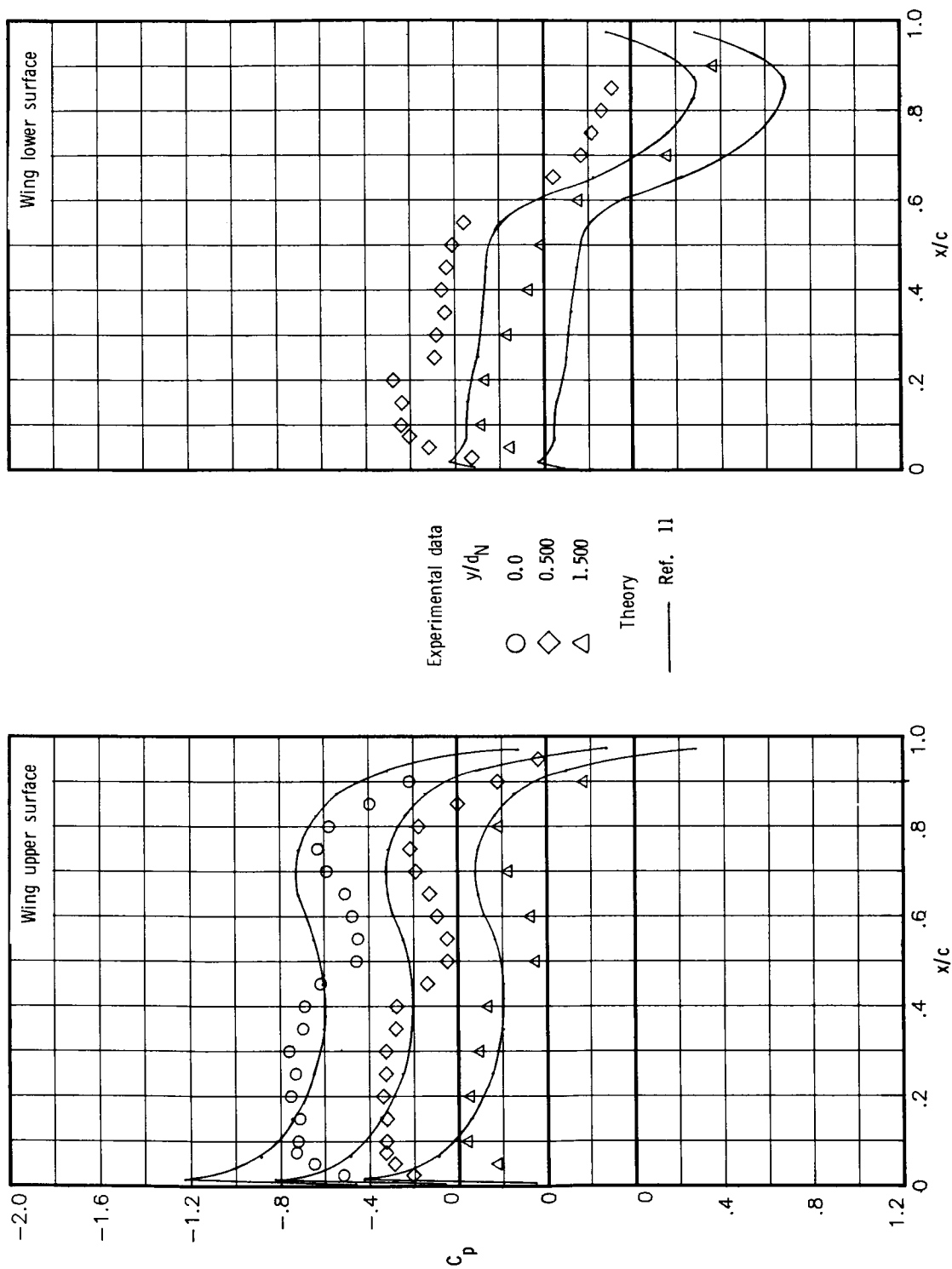
(e) $\alpha = 0^\circ$; configuration 3.

Figure 16.- Continued.



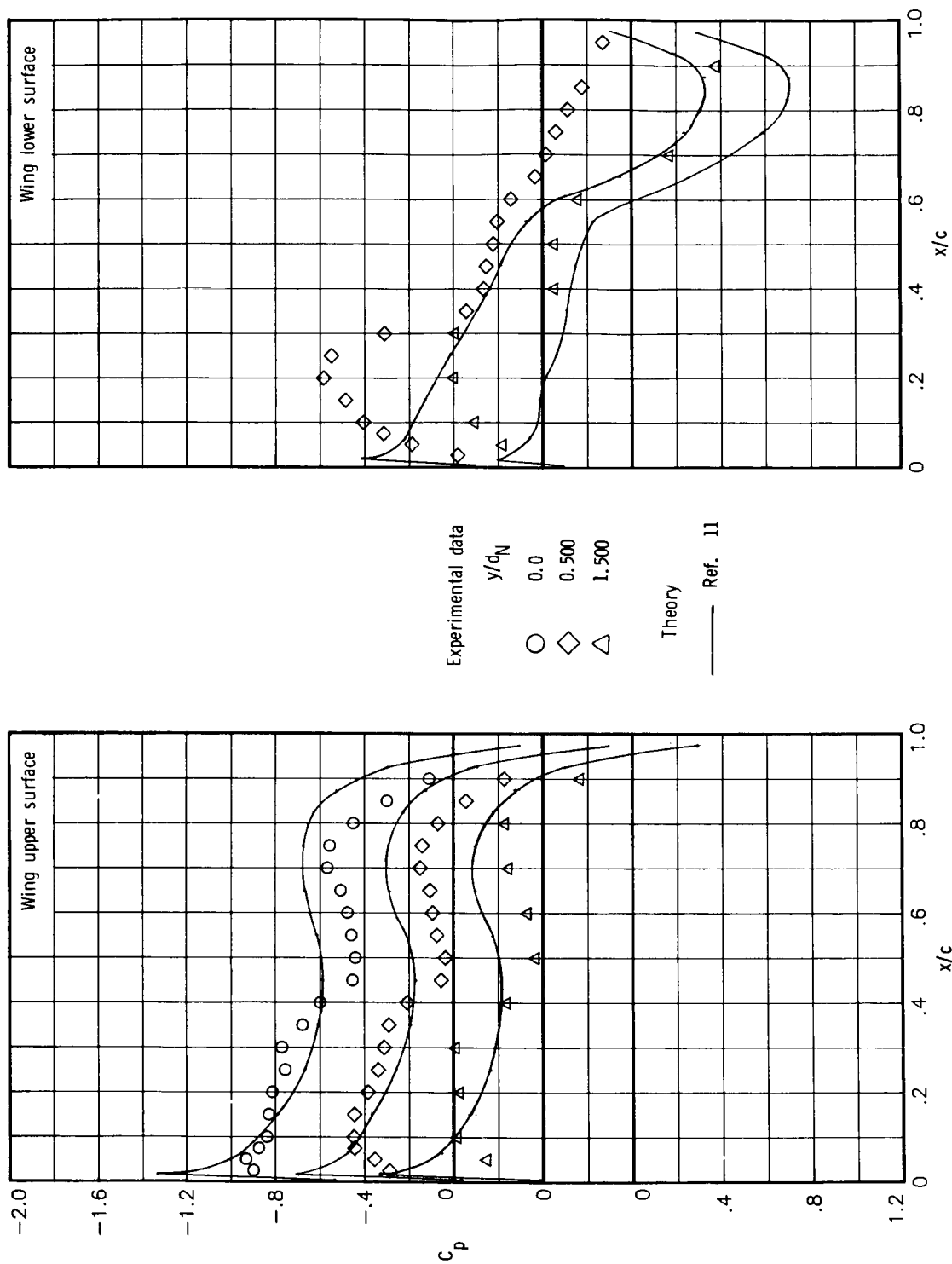
(e) Concluded.

Figure 16.- Concluded.



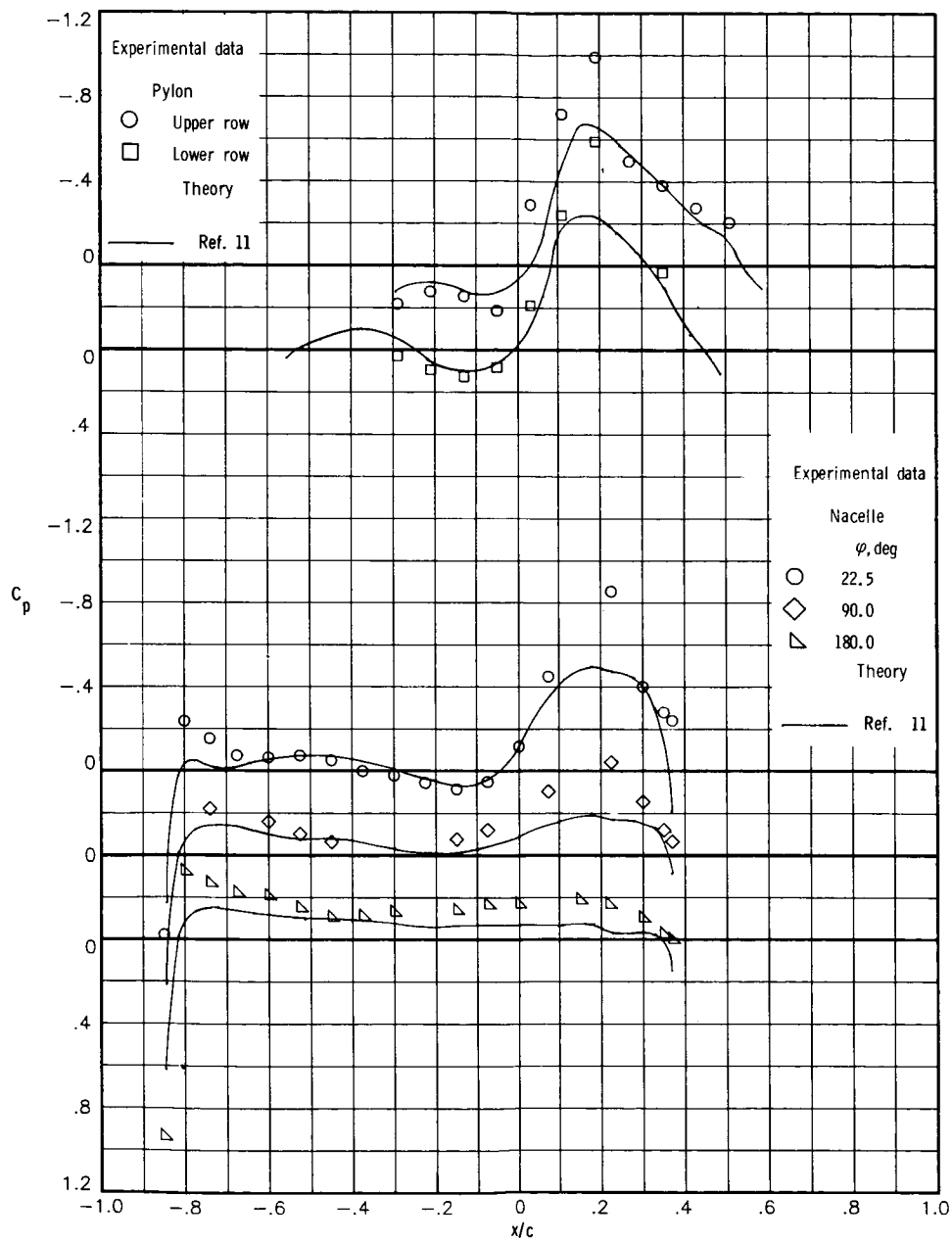
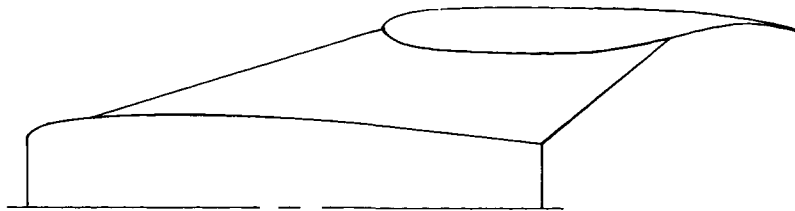
(a) $\alpha = 0^\circ$; configuration 1.

Figure 17.- Comparison of experimental and analytical pressure coefficients at $M = 0.80$ and $x_t = 0.29c$.



(b) $\alpha = 0^\circ$; configuration 5.

Figure 17.- Continued.



(b) Concluded.

Figure 17.- Continued.

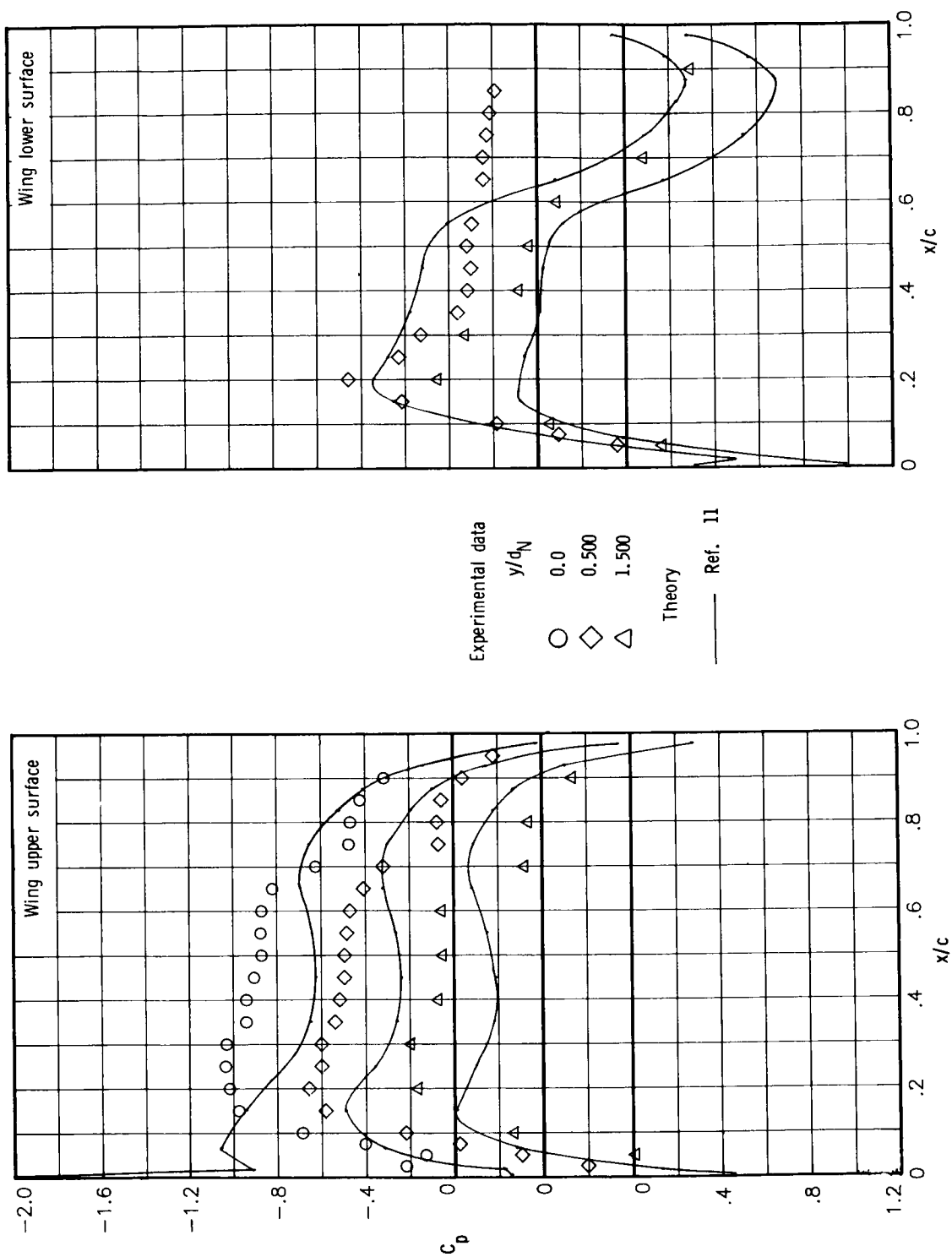
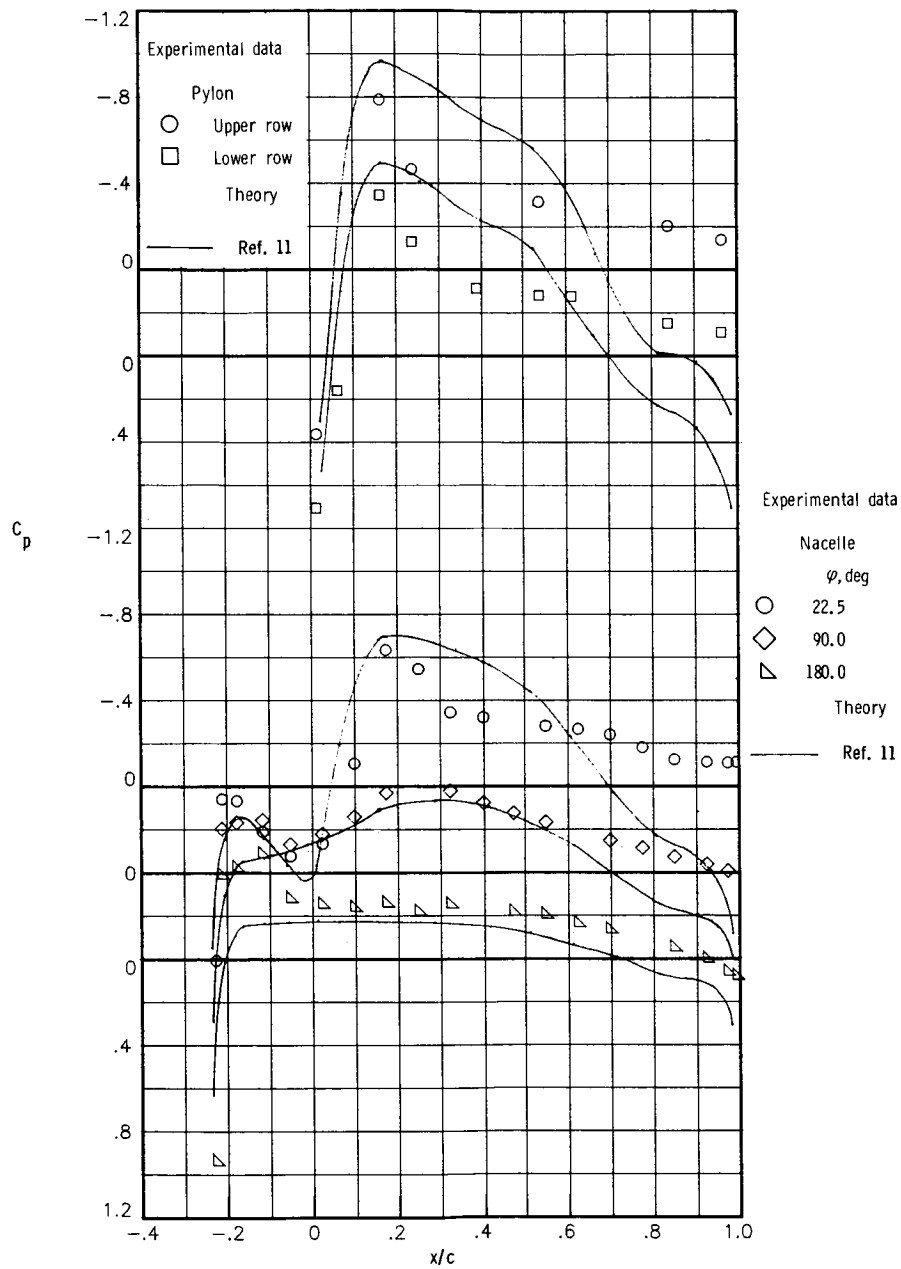
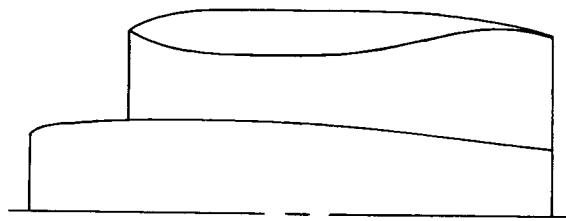
(c) $\alpha = 0^\circ$; configuration 3.

Figure 17.- Continued.



(c) Concluded.

Figure 17.- Concluded.

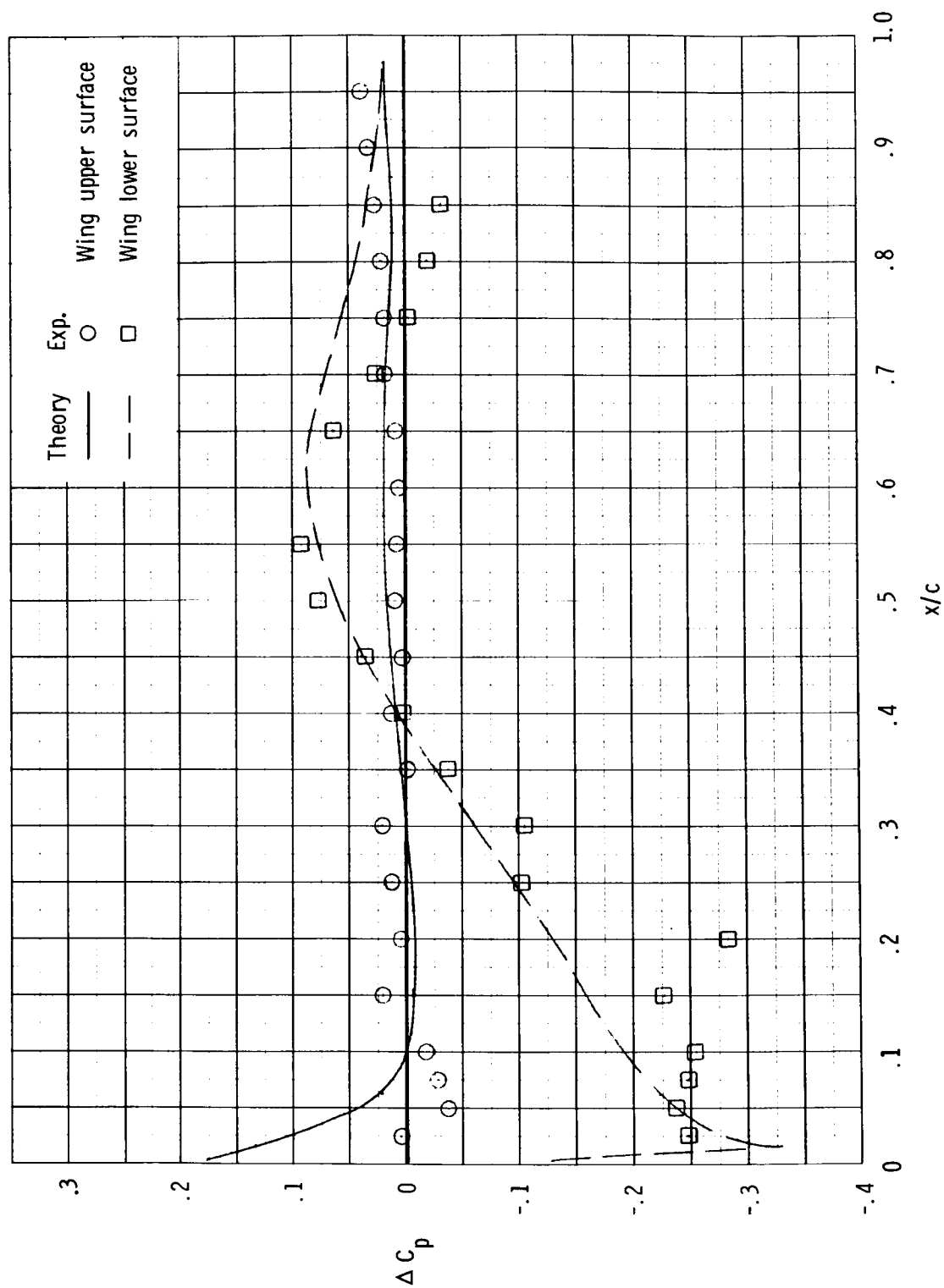


Figure 18.- Comparison of experimental and analytical increments in wing-surface static pressures caused by installation of swept-pylon configurations 1 and 5. $M = 0.60$; $\alpha = 0^\circ$; $x_t = 0.29c$; $y/d_N = 0.500$.

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